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ASSESSMENT OF NDE RELIABILITY DATA

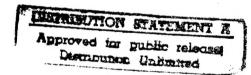
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Prepared for

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

NASA Lewis Research Center Contract NAS-3-18907

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16. Abstract

Twenty sets of relevant NDE reliability data have been identified, collected, compiled, and categorized. Three relevant on-going programs are being monitored for future usage. A criterion for the selection of data for statistical analysis considerations has been formulated. A model to grade the quality and validity of the data sets has been developed. Data input formats, which record the pertinent parameters of the defect/specimen and inspection procedures, have been formulated for each NDE method. A comprehensive computer program has been written to calculate the probability of flaw detection at several confidence levels by the binomial distribution. This program also selects the desired data sets for pooling and tests the statistical pooling criteria before calculating the composite detection reliability. Probability of detection curves at 95 and 50 percent confidence levels have been plotted for individual sets of relevant data as well as for several sets of merged data with common sets of NDE parameters.

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ASSESSMENT OF NDE RELIABILITY DATA

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Final Report

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NATIONAL AERONAUTICS & SPACE ADMINISTRATION Lewis Research Center Cleveland, Ohio

Contract NAS 3-18907

Foreword

This final report covers the work performed under Contract NAS-3-18907 from July 1974 to September 1975. The study was accomplished by General Dynamics Corporation, Fort Worth Division, Fort Worth, Texas and Vanderbilt University, Nashville, Tennessee. The program was managed by Dr. B. G. W. Yee of General Dynamics, with Dr. J. C. Couchman and Dr. F. H. Chang serving as principal investigators. Valuable contributions were made to the development of the statistical analysis by Dr. G. H. Lemon and computer programming by J. S. Kunselman and T. Walker. Dr. P. F. Packman of Vanderbilt University served as associate program manager. The program was under the technical direction of Mr. S. J. Klima, NASA Lewis Research Center, Cleveland, Ohio.

Participants in this program are indebted to many people in the NDE community for furnishing data to this program and for providing consultation.

Thank you!

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SUMMARY

The overall objective of this program is to assess available nondestructive testing data for the determination of the sensitivity and reliability of state-of-the-art production NDE methods for flaw detection on metallic materials. This program was separated into four different tasks. They were:

Task I Acquisition of Information

Task II Screening and Separation of Data by NDE Method and Ma-

terial

Task III Statistical Determination of

NDE Reliability

Task IV Reporting

Twenty sets of relevant NDE reliability data have been identified, collected, compiled, and categorized. relevant on-going programs have also been identified. A criterion for the selection of data for statistical analysis considerations has been formulated. A model to grade the quality and validity of the data sets has been developed. Data input formats, which record the pertinent parameters of the defect/specimen and inspection procedures, have been formulated for each NDE method. A comprehensive computer program has been written to calculate the probability of flaw detection at several confidence levels by the binomial This program also selects the desired data distribution. sets for pooling, and tests the statistical pooling criteria before calculating the composite detection reliability. Probability of detection curves at 95 and 50 percent confidence levels have been plotted by NDE technique and material type for individual sets of data as well as for merged data.

I. INTRODUCTION

In order to apply linear-elastic fracture mechanics to structural design, NDE has to show at a high level of confidence that no flaw larger than a specific size exists in the structure. To establish the minimum detectable flaw size, many companies and organizations have conducted NDE demonstration programs. Most of these demonstration programs have been conducted in the production and field-service environment, but some have been conducted in the laboratory environment.

The results obtained from demonstration programs are lacking in universal agreement. This lack of agreement is not surprising because each company or organization may use a different NDE procedure, different personnel, different procedures and parameters to generate the test flaws, different flaws and material types, and even different statistical analysis procedures. There appears to be a need to (1) collect much of the available NDE reliability data, (2) closely examine all the parameters that could affect the detection reliability, (3) compare the parameters used by each organization to obtain the data, and (4) attempt to identify the parameters that most likely cause observable differences in detection reliability. It appears worthwhile to obtain a composite detection reliability for each NDE method, material type, and flaw type by pooling data obtained from several sources. At the same time, the merits and shortcomings of several statistical analysis procedures should be carefully examined and the procedure most suitable for the analysis of NDE reliability data should be selected. Any needs for improved methods should be identified.

The program reported on in this document was intended to collect all available NDE data, screen and separate the data by NDE method and material, perform statistical analyses, and evaluate the state-of-the-art in NDE reliability.

II. ACQUISITION OF INFORMATION

The acquisition of NDE reliability related data, identification of on-going programs, and preparation of a bibliography of the acquired data is discussed in this section.

2.1 Acquisition of NDE Reliability Related Data

Twenty-three sets of potentially useful data have been identified during this program and twenty sets were acquired. Three sets of data involve on-going programs and the data were not made available for inclusion in this study. Of the twenty sets of data received, only seven sets were statistically analyzed. The thirteen sets of data that were not statistically analyzed were rejected because they did not satisfy the selection criterion discussed in Section III of this report or the owner refused to permit the data to be used.

2.2 Bibliography

The twenty sets of NDE reliability related data that have been acquired are listed below. Some of the references are private data in which case only company or committee report numbers are available. Several references to government funded programs, which have not been published can only be identified by sponsoring agencies and the name of an individual at the company where the work was conducted. Copies of these data can be obtained by either contacting an individual within the sponsoring agency or an individual associated with the company.

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- 20. Raatz, C. F., Senske, R. A. Woodmansee, W. E., et al. "Detection of Cracks Under Installed Fasteners," AFML-TR-74-80, April 1974.

2.3 On-Going Programs

There are several on-going programs that are currently known. Of these, only three are government funded and their results will be available to the public for analysis. The data from the privately funded programs may never be made available for public analysis.

The three government funded on-going programs are:

- 1. Crack detection reliability on welded plates and structures, Martin Marietta, Ward Rummel, sponsored by NASA/Johnson Space Center, NAS 9-13578.
- Crack detection reliability on actual aircraft structures at the depot level, Lockheed, GA., W. Lewis, sponsored by Kelly Air Force Base.
- 3. Crack detection reliability on bolt holes on F-111 fatigue tested structures, General Dynamics, Fort Worth Division, B. G. W. Yee, Sponsored by SMALC.

III. SCREENING AND SEPARATION OF DATA BY NDE METHOD AND MATERIAL

This section describes the development of data selection criteria, separation and categorization of data, development of a model to grade quality of the data, and the development of a data input format for each NDE method.

3.1 Criteria for Selection of Data for Statistical Analysis

All of the reliability related NDE data are not necessarily suitable for statistical analysis. Some are lacking in the documentation of certain key pertinent parameters, such as the defect dimension, defect type, NDE method, etc. Statistical analysis of data when the key pertinent parameters are not documented would be marginal in value. A data selection criteria is needed to screen the data and prejudge the suitability of the data for statistical analysis. Such a criteria is necessarily subjective because it involves human judgment of data value or usefulness. It is felt that such a subjective criteria will still be useful to screen out data having marginal statistical value and to eliminate lost time in processing the data.

To be eligible for statistical analysis, a set of data must satisfy the following conditions:

- a) An NDE procedure or specification must accompany the data which clearly describes the equipment and the parameters used so that the data may be reproduced in other facilities (assuming the same equipment or its equivalent is used).
- b) The defect dimensions and specimen geometry must be well documented so that data may be statistically analyzed and compared to defect detection in the proper defect size range. When artificial methods of defect fabrication are used, at least ten percent (10%) of all defects in a given set must be destructively tested to obtain the defect dimensions. For methods that are used to produce multiple defects in a specimen or methods that are questionable for producing controllable defect dimensions, at least fifty percent (50%) of all defects in a given set must be destructively tested to verify the defect dimensions.

The model that was developed to grade the quality of the data was very subjective. It consisted of assigning weighting factors and summing up the weighting factors to obtain an overall factor that is an index of data quality. The model is included in Appendix A.

Data pooling tests that were coded in the NASA data processing code (reproduced in Appendix B) provided a means for isolating input errors and identifying any data set that may be suspect. The tests are derived from binomial statistics equations which are developed in the next section.

The computer code developed for processing the NDE reliability data is described in Appendix C. Included is a complete description of data input formats and parameter keys.

3.2 Separation and Categorization of Data Sets

The twenty sets of data listed in the Bibliography in Subsection 2.2 can be separated into three categories. Table 3-1 describes the data separated into the three categories and the status of these data sets. The first category is the data that appear to satisfy the criteria discussed in Subsection 3.1, and they will be considered for statistical analysis. There are seven (7) sets of data in this category. category is data that probably could be used if permission to use the data is granted by the rightful owner of the data. There The third category is the are two sets of data in this category. data that are either lacking in the inspection procedure documentation or defect dimension documentation. There are eleven sets of data in this category. The data sets in each of the three categories are presented in Table 3-1.

A large majority of the data were obtained on thin flat plates which contain fatigue cracks or weld defects. There are few sets of data that were obtained with relatively complex shaped specimens such as a T, I, or H shape. In order to gain a better understanding of the availability of data on material type, defect type, and specimen complexity, a table (Table 3-2) is constructed to categorize the data sets according to test specimen complexity. Within each data set, a brief description of the material type, NDE methods, and defect type is presented.

The materials and shapes for which valid data were obtained and statistical analysis results are computed are tabulated later in Section V (RESULTS).

Table 3-1 STATUS AND CATEGORY OF DATA CONSIDERED FOR NDE RELIABILITY ASSESSMENT

Data that satisfy the criteria discussed in Subsection 3.1 and were statistically analyzed	Data that lack owner permission use	Data that were not statistically analyzed due to the lack of sufficient inspection procedure or defect dimension documentation or both
References: 2 4 8 10 11 16	References: 5 Pressure Vessel Research Committee approval needed before data can be used 6 Same as 5 above	References: 1 7 9 12 13 14 15 17
		70

Table 3-2 Data Grouping According to Specimen Thickness and Complexity

References: 1 2219-T87 Al up to 1 cm thick of the carboratory environment of atigue cracks in plates and vended plates of the plates of thick of the carboratory environment of atigue cracks in flat plates of thick of the carboratory environment of atigue cracks in flat plates of thick of atigue cracks in flat plates of a flates of	Actual Aircraft Structures	References:	11 7178-T651 A1	o Naturally occurring	tatigue cracks in bolt	o Eddy current	o Laboratory and depot level	13 6A1-4V-Ti and 7075-T6A1 o Fatigue cracks off C-5A	Pylon (actual and	simulated) o Ultrasonics and X-ray	14 7075-T6 A1	acks off	wing boxes (actual and simulated) o Ultrasonics and eddy		20 KC-135 lower wing skin panel and laboratory	specimens with ratigue cracks under installed	fastener o Ultrasonic	o Laboratory and field environment	
lat Plates and Simple eferences: 2219-T87 Al up to 1 thick o Cracks in plates and welded plates o Ultrasonics, penetra eddy current, and X- o Laboratory environme 2219-T87 Al up to 1 thick o Fatigue cracks in f1 plates o Ultrasonic, penetran eddy current, and X- o Laboratory environme (mostly) 2219-T87 Al up to 1 thick b Fatigue cracks in f1 plates o Ultrasonic, penetran eddy current, and X- thick b Fatigue cracks in f1 plates o Ultrasonic, penetran eddy current, and X- claboratory environme (mostly) Laboratory environme (mostly)	 lindrical, I, H, T and derately Complex Shapes	ferences				and X-ray			X-ray, and magnetic									cracks	
	Flat Plates and Simple Shape	References:	2219-T87 Al up to 1	thick Cracks in plates	_			2219-T87 Al up to 1 thick	Fatigue cracks in	•		(mostly)	2219-T87 A1 up thick	Fatigue cracks in			(mostly)		

Table 3-2 (Continued)

Flat Plates and Simple Shape	Cylindrical, I, H, T and other Moderately Complex Shapes	Actual Aircraft Structure
5 Steel forging up to 25 cm thick o Induced and naturally occurring defects o Ultrasonics o Laboratory environment		
6 Steel welded-plates up to 28 cm thick o Induced weld defects o Ultrasonics and X-ray o Laboratory environment	18 D6ac Steel o Induced forging defects o Ultrasonics, magnetic particle and rubber o Production environment	
7 2219-T87 welded plates of 0.62 and 1.25 cm thick o All types of weld defects o Ultrasonic and X-ray		
8 2219-T87 Al up to 1 cm thick o Fatigue cracks in flat plates o Ultrasonic, penetrant, eddy current, and X-ray o Production environment (mostly)		
9 2219-T87 and 2014-T6A1 up to 2.5 cm thick 6A1-4V-Ti and 5A1-2.5 Sn Ti up to 1½ cm thick o Fatigue cracks in flat plates weld defects in plates	ν,	

Table 3-2 (Continued

er Actual Aircraft Structure							
Cylindrical, I, H, T and other Moderately Complex Shapes							
Flat Plates and Simple Shape	o Ultrasonic, penetrant, X-ray, and eddy current o Laboratory environment	12 2219-T87 Welded plates of 1.25 and 2.5 cm thick o All types of weld defects o Ultrasonic and X-ray	16 6A1-4V-Ti up to 13 cm thick o Ultrasonic o Flat bottom holes and induced defects	6A1-4V-Ti Diffusion Bonded thin plates o Induced defects o Penetrant	Al Samples o Fatigue cracks in flat plates o Penetrant	6A1-4V-Ti o Fatigue cracks in flat plates o Penetrant	Steel o Induced weld defects o Ultrasonics

Table 3-2 (Continued)

Actual Aircraft Structure			19 o D6ac, 4340, and other Steel Alloys Actual Aircraft Struc-	c Particl		
Cylindrical, I, H, T and other Moderately Complex Shapes						
Flat Plates and Simple Shape	Al, Ti, and Steel Plates (multiple) o Gracks in fastener holes o Eddy current All in production environ- ment	17 Similar to those described in Reference 16	19 o Déac Steel Plate Delta Scan Ultrasonics			

This section describes the binomial statistical method, cumulative schemes, statistical pooling procedure, and digital computer code for computing NDE reliability.

4.1 Introduction

There are four possible outcomes from any nondestructive inspection of an item: (1) detection of a defect that is present, (2) non-detection of a defect that is present, (3) detection of a defect that is not present (false indication), and (4) non-detection of a defect that is not present. Because of these four possible outcomes, any single inspection may be called a quadrinomial event. Although it is recognized that false indications of defects and true indications of non-defective items (cases 3 and 4) are of practical significance to both the manufacturer and the customer, it is beyond the scope of this investigation to develop a straightforward statistical method for handling the quadrinomial event.

Preliminary indications show that most NDE reliability investigations have neglected to report information concerning either false indications of defects, or true indications that specimens contained no intentionally induced flaws. However, the data input format discussed in the previous section provides for storage of information concerning false indications for future use when more of these data become available.

Cases (1) and (2) involve either a detection or non-detection of a defect that is known to exist. This event can best be described statistically by applying the binomial distribution. The Normal, Chi-square, and Poisson distributions are sometimes used as approximations to the binomial. Their applicability to the problem of NDE reliability is to be considered in the later sections of this report.

4.2 Application of Binomial Distribution

An event that has only two possible outcomes is referred to as a binomial event. Suppose, for example, an experiment in NDE is performed where N specimens, all containing identical flaws, are routed through an ultrasonic inspection system. Suppose further, that the system capability does not change throughout the entire inspection process, i.e., each specimen is evaluated independently of the others. Let p equal the true, (but as yet unknown) probability of detecting each flaw and q=1-p be the probability of missing each flaw. Assuming p remains the same for all specimens, the random variable X can be defined as being the number of flaws that are detected in any given experiment. X, then, is referred to as a binomial random variable with parameters N and p. Its possible values are 0, 1, 2, ..., N. Equivalently, it can be said that X has a binomial distribution. The probability of obtaining any one of the N+1 possible values of X from such an experiment is described by the following equation:

$$P(X=n) = {N \choose n} p^{n} q^{N-n}, \qquad n=0,1,...,N$$
where
$${N \choose n} = \frac{N!}{n!(N-n)!} .$$
(1)

The sum of all the possible values for equation (1) is equal to unity and can be written as follows:

$$(p+q)^{N} = \sum_{n=0}^{N} {n \choose n} p^{n} q^{N-n} = 1$$
 (2)

The probability of detecting n or more flaws can be found by summing equation (1) over all the values of X for which $X \ge n$. Thus,

$$P(X \ge n) = \sum_{i=n}^{N} {n \choose i} p^{i} q^{N-i}.$$
 (3)

4.2.1 Confidence Interval Estimates of the True Probability of Detection

The objective is to estimate the true proportion of defects of a particular type and size that can be detected by a given NDE method. The best single estimate, \bar{p} , of the true

detectable proportion is the number of flaws detected divided by the total number of flaws present:

$$\bar{p} = \frac{n}{N} \tag{4}$$

If the binomial experiment (N, n_k , p) is repeated an infinite number of times and if P_k is computed each time, then the average of all the p_k s will be equal to p. So \bar{p} is an unbiased estimator of the true probability of crack detection.

Since the probability is small that \bar{p} will exactly equal p on any specific replication of the experiment, an interval estimator that will contain p most of the time is considered useful. A lower one-sided confidence interval can be used to estimate a lower bound on the true probability of crack detection. The lower bound, p_1 , for n detections of N cracks is computed as follows:

Solve
$$\alpha = \sum_{i=n}^{N} {n \choose i} p_1^i (1-p_1)^{N-i}$$
 (5)

for p₁, where

$$\alpha = 1 - G \tag{6}$$

The following interpretation can be given for a 100G percent lower binomial confidence interval. If the binomial experiment (N, n_k , p) is repeated many times (theoretically infinite), and if p_{1k} is computed each time, then 100G percent of these lower confidence intervals will be equal to or lower than p. Thus, there is 100G percent "confidence" that the lower bound, p_1 , computed for any specific binomial experiment will be less than or equal to the true probability of crack detection. The choice of G is arbitrary and depends on how sure one needs to be that the true probability of detection is in the interval from p_1 to 1, (the larger the value of G, the smaller the calculated lower bound p_1 will be).

^{*} n_k is the number of successes in the Kth replication of a binomial experiment consisting of N measurements.

4.2.2 Sample Size Determination

The objective is to determine the sample size required to estimate the lower confidence limit and confidence level. This can be accomplished by utilizing equation (6). By specifying the confidence level, G, and the lower confidence limit p_1 , a set of values (which must be integers) can be computed for N and n. Each combination of N and n in this set indicates the number of inspections and the number of detections required to achieve the specified probability of detection (POD) at the stated confidence level. For example, if G and p_1 are chosen to be 0.95 and 0.9 respectively, equation (5) becomes

$$0.95 = 1 - \sum_{i=n}^{N} {n \choose i} (0.9)^{i} (0.1)^{N-i}.$$
 (7)

One of the combinations of N and n is 29 and 29 respectively. This represents the smallest sample size that can be utilized to meet the minimum specified values for G and p_1 . The next smallest sample size is N=46. In this case n must equal at least 45 to achieve 90% probability of detection at 95% confidence level. The higher the reliability requirements, of course, the larger the sample size required.

Equation (5) can also be used to calculate the number of added NDE tests required to upgrade an existing batch of data in the hope of achieving higher reliability estimates. Equation (5) takes on the form

$$1-G = \sum_{i=n+\delta-\epsilon}^{N+\delta} \binom{N+\delta}{i} p_1^i (1-p_1)^{N+\delta-i}$$
 (8)

where $\boldsymbol{\delta}$ is the required number of additional tests and $\boldsymbol{\epsilon}$ is the maximum number of additional misses (nondetections). For example, if an experiment consisting of 29 inspections and 28 detections was performed, the reliability (equation 5) is 90% probability of detection at 80% confidence level. If a 95% confidence level is desired, the additional data requirements are indicated by equation (8). Thus, δ = 17 with ϵ = 0 represents the minimum added sample required to upgrade the existing data.

4.3 Comparison of Alternative Statistical Procedures

Table 4-1 lists the probability of detection at the 95% confidence level for four commonly used probability analysis methods, (Binomial, Normal, Poisson, and Chi-Square). The comparison is on the basis of 30 trials with 10, 15, 20, 25, 29 and 30 detections. The Binomial gives the exact results. The Normal approximation gives confidence values greater than the Binomial for large successes (30/30), and very nearly equal to the Binomial for intermediate successes (15 or 20/30). The Poisson approximation gives values less than the Binomial method. It is a good approximation only for cases with large number of trials with small number of successes. For 30 trials, the Chi-Square method gives values less than the Binomial for 10 to 30 successes. Like the Poisson, the Chi-Square values approach those of the Binomial for less than 10 successes in 30 trials.

However, a closer approximation to the Binomial for both the Poisson and Chi-Square can be achieved by using a conditional approach. By first calculating the upper one-sided confidence limit for the probability of missing a flaw (q_{μ}) , the lower one-sided confidence limit for $P_1 = 1 - q_{\mu}$ can then be approximated when 1 - n < n. By using this conditional approach, a new set of values is calculated and included in Table 4-1 under the conditional approach column. These new values give a much better approximation to the Binomial when the number of successes is high.

Since the Poisson and Chi-Square approximations yield almost the same values, only one set of values was calculated under the new conditional approach.

For larger or smaller number of trials, the above comparisons is not necessarily true but the comparison generally shows the importance of analyzing binomial measurements with binomial statistics.

Table 4-1 Comparison of Probability of Detection Values Obtained by Approximation to the Binomial Distribution

NUMBER OF SUCCESSFUL	POINT ESTIMATE OF	LOWER ONE	SIDED CONF	LOWER ONE SIDED CONFIDENCE LIMITS AT 95% CL	AT 95% CL	CONDITIONAL APPROACH
DETECTIONS	PROBABILITY OF PERFORMANCE	T ATMORATOR	7 13 50 0 75	1.00		POISSON AND
IN 30 IKIALS	OF DETECTION (P)	(Exact)	NOKMAL	FOLSSON*	CHI-SQUARE *	CHI-SQUARE
10	0.333	0.193	0.192	0.181	0.181	0.181
15	0.500	0.339	0.350	0.308	0.308	0.308
20	0.667	0.501	0.525	0.441	0.442	0.435
25	0.833	0.681	0.721	0.578	0.579	0.650
29	296.0	0.851	0.913	0.691	0.691	0.842
30	1.000	0.905	1,000	0.715	0.720	0.900

*No conditional information was used in obtaining these numbers, see "Reliability Management Methods and Mathematics," Lloyd and Litow, Prentice Hail, 1962, pp. 218-219

4.4 Data Cumulation Methods

The calculated value for the lower confidence limit (probability of detection, p₁, at some selected confidence level, CL) is influenced by the total number of measurements (sample size). In order achieve a high POD at a high CL. such as 90% at 95% confidence level, a minimum of 29 measurements have to be made without a miss for a given flaw size. Because of the high costs involved, it is generally not economical to make 29 or more measurements for each flaw size for the entire range of flaw sizes of interest. At the same time, in the inspection of actual structural components, it is unlikely to have 29 or more measurements at any specific flaw size. As a result, size interval grouping has been used in order to obtain a sufficient number of measurements to compute a high p₁ at a high CL and to smooth over the flaw sizes that have no measurements. A cumulation plan permits the accumulation of data over a range of flaw sizes for computing a p1 which is representative of that range.

Several cumulation procedures were considered. These procedures include (1) the range interval "RI", (2) the overlapping 60 points "OSP" and (3) the procedure developed under this contract which will be called the optimized probability method "OPM." Other procedures evaluated are reported in Reference 1. For the same set of data, the computed p1 can be considerably different depending on which of these cumulative procedures is used. Generally, NDE reliability is demonstrated by inspecting specimens containing flaws distributed uniformly over a wide flaw size range. The smallest flaws should be virtually nondetectable and the largest flaws should be 100% detectable. First, the raw data set is arranged in order of increasing flaw size with the appropriate outcome indicated for each measurement. The various cumulative procedures are applied as follows:

(a) Range interval method. The data are separated into groups of equal flaw size increments. The probability of detection at the one sided lower confidence limit is computed for each group separately and plotted as a histogram bar. A conservative pl curve can be obtained by connecting the upper right hand corner of the bars. This procedure is most appropriate for computing probabilities of detection and confidence limits when large numbers of data are available so that all histogram bars represent large data samples. It is important to note that sample size (N) may vary widely between intervals.

- (b) Overlapping sixty point method (N = 60). One begins by combining detection results for the largest sixty crack sizes. The POD is calculated for this interval and plotted at the largest flaw size within the size range spanned. The next data increment is obtained by starting at the median flaw size of the first interval and combining the data for the next smaller 60 cracks. The POD of the second set is plotted at its largest flaw size and the process is repeated until all data have been combined. Each interval overlaps its adjacent interval by 30 data points. Note that the sample size is the same for each interval, but the breadth of the interval can vary widely.
- (c) Optimized Probability Method. The ordered NDE data are grouped into J (a computer input number) intervals of successively increasing size range. The POD of the largest size range is computed at some desired confidence level. next smaller size range data are combined with the first and the POD of the second grouping is computed. This process is continued until J probabilities of detection are computed, and the largest value of POD obtained is plotted at the largest flaw size contained in the corresponding composite grouping. The largest flaw size interval is removed from consideration and this procedure is repeated starting from the next to largest flaw size grouping. The pattern is repeated until J probability of detections can be plotted. Note that if J is sufficiently large, the POD curve produced by this method will be the upper envelope of either the RI or the OSP method. The advantage to this method is that the sample size (N) is maximized, and results in the maximum possible value of p1 for the available data. A basic assumption is that the larger the flaw, the more detectable it is.

V. RESULTS

Of the twenty sets of data collected, only seven sets were statistically analyzed. These seven sets appeared to meet the criteria stated in Section III of this report. of these seven sets, in turn, was subdivided into subsets and analyzed according to the NDE method and material types. Appendix D contains a computer listing of NDE data and detection reliability results that were analyzed during this Table 5-1 summarizes the results of the statistical There is a total of one hundred and twelve subsets analysis. and the subset number is given in the first column of Table 5-1. The second column gives data source. The third and fourth columns identify the material and defect type respectively. The fifth and sixth columns identify the specimen geometry and NDE method respectively. The seventh column lists some of the pertinent parameters, and the eighth column gives the crack length at which 90% probability of detection at 95% confidence level POD90(CL95) was first achieved. The shortest crack length that reached POD90(CL95) by either the OPM or OSP scheme was used and is herein referred to as the threshold level. many instances, the POD(CL95) became smaller than 90% at crack lengths above the threshold level. This is particularly true for the OSP scheme (see Appendix D). For more details on the NDE parameters and inspection procedures that were used to acquire each set or subset of data, one has to refer to the original data reference.

The first set of data analyzed is from Martin Marietta (Contract NAS 9 12276). Four subsets, one by ultrasonic surface wave, one by penetrant, one by eddy current, and one by the X-ray method, were available before the specimens were chemically etched and another four subsets were available after chemically etching the test specimens.

The second and third sets of data analyzed are from Rockwell International-Space Division (Contract NAS 9-14000) and General Dynamics, Convair Aerospace Division (Contract NAS 9-12326). This set of data includes data from Contract NAS 9-12276 with the specimens in the "after-etched" condition.

SUMMARY OF NDE DATA STATISTICALLY ANALYZED
CAUTION: The crack lengths in the POD90(CL95) column
are not intended to be used for design purposes
unless the user demonstrates a similar capability TABLE 5-1

CRACK	POD90 (CL95)	.665	1	.211	.274	,	.665	.737	.221	669	.358	.282	.737	.737	.333	.737	.340	1.36	665		707.	.333	.320
PERTINENT PARAMETERS	Po Before Etch, and with 3 Ops; 10 MHz	Same as above, Uresco P151, K410, D499C SAME, as above;NDT-3;	Same as above	After Etch, and with 3 Ops.	Same as above	Same as above	Same as above	After Etch, and by Operator 0	Operator P	Operator R	Operator S	After Etch, and by Operator H	Operator I	Operator J	Operator K	Operator L	Operator M	•	After Etch, and by Operator T	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		Operator W	Operator X
NDE METHOD	Ultrasonic- Surface Wave	Liquid Penetrant Eddy Current	X-ray	Ultrasonic- Surface Wave	Liquid Penetrant	Eddy Current	X-ray	Ultrasonic- Shear Wave (Surface Wave)	Hz	ī	for GD Convair	Penetrant D495A	for RI - SD						Eddy Current	Model 2,154		ac .	
SPECIMEN GEOMETRY	Flat Plates							Flat Plates												:			
DEFECT TYPE	Fatigue Cracks			•				Fatigue															
MATERIAL TYPE	2219-T87 A1	•	•					2219-T87 }-							•			٠,	•	-	-		
DATA SOURCE	Martin Marietta NAS 9 12276							Rockwell Inter. Space Div. (NAS)	Marietta, and	GD CONVAIL					,				-	٠.	-		
DATA SUB-SET	1	0 E	7	5	9	7	œ	6	01	12	13	14	15	. 91	17	13.	19:	20	21	. 5	77	24	25. "

*Data Set Numbers are Included on the Corresponding Computer Output in Appendix D.

TABLE 5-1 SUMMARY OF NDE DATA STATISTICALLY ANALYZED (Continued)

																	<u> </u>									-
CRACK	POD90 (CL95			•	•		.737	•	1	.201	.237	٠	.282	.183	.170	:178		.127	.173	.203		.173	.211	.218	- ·	147
DEPTINENT PARAMETERS	1	After Etch and by	•	Operator B	Operator C	Operator D	Operator E	Operator F	Operator G	After Etch and Merged Results of 5 Operators	••	Results of 7 Operators	After Etch and Merged Results of 5 Operators	Used P5F-1 Penetrant	Used P5F-2	lised P5F-2 and D100	oper	Used P5F-2.5	Used P5F-2.5	Used P5F-2.5, NQ-1	P5F-2 and		Simulate Welded Flaws	Simulate Flaws in	CAT TOTAL TAIL TO SERVICE STATE OF THE SERVICE STATE STATE OF THE SERVIC	Liquid Penetrant Merged Kesuits Ifom the Four Data Source on Al
doman aux	NUE METHOD	X-ray								Ultrasonics	Liquid Penetrant After Etch	•	Eddy Current	Liquid Pene.								Ultrasonic, Shear Wave		,		Liquid Penetran
SPECIMEN	GEOMETRY								:	Flat Plate				Flat Plate					Flat Plates		Flat Plates	Welded Flat			Flat	Flat Plates
DEFECT	TYPE									Fatigue	ST S			Fatigue	Cracks				Fatigue	Cracks	Fatigue Cracks	Fatigue/ Cracks	2222	Faofone	Cracks	Fatigue
	MATERIAL TYPE									2219-T87 A1				T1-6A1-4V					7075-T6511 A1		PH13-8 Mo. St.	T1-6A1-4V	4330V St.	PH17-4 St.	PH17-4 St.	2219-T87 and 7075:T6511 A1
	DATA SOURCE	·								RI-SD, Martin	Marietta, &GDC			RI-B-1 Division	(TFD-72-925)	(TFD-72-1005)	(TFD-72-1515)	(mm 79 703)	(TED-72-757)	(TFD-73-532)	(TFD-73-532) (TFD-73-496)	(TFD-73-371)	/men_73_379)	(TFD-73-140)	(TFD-72-768)	RI-SD, MM, GDC, and RI-B1
DATA	SUB-SET	56		27	000	0 0	67	3 5	32	33		34	35,	36		37	38	ç	5 6	41,	42	777	,	45	47	87

TABLE 5-1 SUMMARY OF NDE DATA STATISTICALLY ANALYZED (Continued)

	2								_																	
CRACK		,	1	•	•				1	•			•	•		,				,		,		•		.178
PERTINENT PARAMETERS		Production Condition	Laboratory Condition	=	•		= =	: :		=	Hand Hald R C Drobe of	Insp. at	Depot Level	Results of Team 2	(Atypical of 3 more teams)	Results of Team 4 (A-	typical)	Merged results of 5 teams	Production Insp.	=		=		2		Laboratory Insp.
NDE METHOD	Liquid Penetrant	Magniflux ZL-2, ZE-3, and ZP-4		Mag, Particle	Ultrasonic	Shear Wave 5 MHz	Ultrasonic	Snear wave	X-ray	X-ray	Eddy Current								Liquid Penetrant	ZL-2A and 30A;		ZE-4B;		ZP-9B	Plus Others	Liquid Penetrant
SPECIMEN GEOMETRY	Cylindrical	She11	7.62 cm in dia.	0.64 cm thick							Fatione KC-135 Wines								Tandem T	Solid Cyl.,	Threaded	Hollow Cyl.,	Filleted	Hollow Cyl.,	Solid Cyl.	Tandem T
DEFECT TYPE	Fatigue Cracks										Fatiene	Cracks							Forge. Closed			=		=		E
MATERIAL TYPE	4330V St.		7075-T6511 A1	4330V St.	4330V St.		7075-T6511 A1		/0/2-T6511 AT	4330V St.	7178-T651 AT								2024-T6 A1	4340 M St		*		=	=	2074-T6 A1
DATA SOURCE	Lockheed GA	(TR-68-32)									Boeing, W.Kan.	3							Boeing Comm.	Airplane Co.		(TR-74-241)				
DATA SUB-SET	. 07	50	51	52	53		24	u u	0	26	57	-			- 1	58		. 65	09	19		. 62		63	99	65

TABLE 5-1 SUMMARY OF NDE DATA STATISTICALLY ANALYZED (Continued)

DATA SUB-SET	DATA SOURCE	MATERIAL TYPE	DEFECT	SPECIMEN GEOMETRY	NDE METHOD	PERTINENT	PERTINENT PARAMETERS	CRACK LENGTR
							PC	POD90 (CL95)
99		4340 M St	Forged Closed	Solid Cyl., Threaded	ZL-2A and 30A;	Laboratory Insp.	Insp.	. 584
67		=	DM,Slot	Hollow Cyl.,	ZE-4B;	=	=	787
				Filleted			•	
89		=	=	Hollow Cyl.,	ZP-9B;			ı
69		=	=	1.,F11.	Plus Others	•		.229
70	•	=	=					905.
71.		2024-T6 A1	=	Tandem T	Ultrasonics	Production Insp.	Insp.	,533
72 .		4340 M St.	:	Solid Cyl. Threaded	ed 5 and 10 MHz			
73		=	=	Hollow Cyl.	Shear and Surface	O)		
				Filleted				.330
. 42		=	=	Hollow Cyl.	Wave			,
75.			=	Solid Cyl.,				
				llet				8/1.
92		=	=	= '				.356
77		**	=	Solid Cyl.	Ultrasonics	Laboratory	Insp.	,356
78.		=	=	" "Fillete	Filleted 5 and 10 MHz			.229
79		=	=	Hollow Cyl.	Shear and Surface	ø		.254
80		=	=	" Filleted Wave	d Wave			.330
81		2024-T6 A1	=	Tandem T				.432
82		4340 M St.	=	ld Cyl.	ray	Production Insp.	Insp.	1
~ &			=	", Filleted	pa			
84			= :	low "				•
82			=	=				•
98	•		=	Solid ", Threaded	pa			1
87		2024-T6 A1	=	Tandem T				ı
. 88		4340 M St	:	Solid Cyl.	X-ray	Laboratory	Insp.	•
68		4340 M St		Hollow ", Filleted				1
06				Solid ", Filleted	pa			
91()		-		Hollow ",	-		,	

TABLE 5-1 SUMMARY OF NDE DATA STATISTICALLY ANALYZED (Continued)

PERT INENT PARAMETERS LENGTH	РОПЭ	1 .279		. 330			:	 Insp305	.329		.178	.254	Insp400	. 762		•	Ç.	. 330		Insp.	.508	.356	.584						
PERT INENT		· -	Production Insp.					Laboratory Insp.					Production Insp.	1 52b						Laboratory									
NDE METHOD			Mag, Particle	ered		eted	aded	Mag. Particle	ered	pap	ted		Eddy Current	Filleted ED400 and	at f=100 KHz	eced				Eddy Current	ED4	.at f=100 KHz	ted				_		
SPECIMEN GEOMETRY		Tandem T	id cyl.		Hollow "	" , Filleted	Solid " , Threaded	Solid Cyl.	Hollow " , Fillered	Solid ", Threaded	" ", Filleted	Hollow "	Solid Cyl.	" " F111	Hollow Cyl.	" ", Filleted		Tandem T		Tandem T	Solid Cyl.	Hollow Cyl., Fil.	Solid ", Filleted	S					·
DEFECT			Forge	Closed	ייייי איייי	=	=	=	=	=	=	=	=	:	:	=	=	:		=	=	=	=	=		 			
MATERIAL TYPE		2024-T6 A1	4340 M St.	=	=		=	 =	=				 . ,		=	=		2024-T6 AI	;	=	4340 M St.	=	=					**********	
DATA SOURCE								:					• ,		, ,-	4							;	;	•				
DATA SUB-SET		92	93	96	95	96	65	86	99	100	101	102	103 / 1	104	105	106	į	107		108	109	110.	111	112					

There are five subsets of data obtained with the ultrasonic method, each subset for a different inspector, three inspectors for Contract NAS 9-12276, one for NAS 9-14000, and one for NAS 9-12326. There are seven subsets of data obtained with the liquid penetrant method, each subset by a different inspector, three inspectors for NAS 9-12276, three for NAS 9-14000, and one for NAS 9-12326. There are five subsets of data obtained with the eddy current method, each subset by a different inspector, three inspectors for NAS 9-12276, one for NAS 9-14000, and one for NAS 9-12326. There are seven subsets of data obtained with the X-ray method, each subset by a different inspector: three inspectors for NAS 9-12276, three for NAS 9-14000, and one for NAS 9-12326. Subsets 33, 34 and 35 were obtained by merging the data of five ultrasonic inspectors, seven penetrant inspectors, and five eddy current inspectors. The X-ray data were not merged because the results are not worthy of further consideration.

The fourth set of data analyzed is from Rockwell International B-1 Division, B-1 NDI Demonstration Program. There were twelve subsets of data in this set. The fifth set of data analyzed is from Lockheed, Georgia, AFML Report No. TR-68-32. There are eight subsets in this data set. The sixth set of data analyzed is from Boeing Company of Wichita, Kansas. This set of data was obtained from eddy current inspection of bolt holes, after pulling off the bolts, from actual aircraft parts.

The seventh set of data analyzed is from the Boeing Commercial Airplane Company, AFML Report No. TR-74-241. There are fifty-three subsets in this set of data.

Data subset number 48 was obtained by merging the data from Martin Marietta (NAS 9-12276), Rockwell Int.-Space Division (NAS 9-14000), General Dynamics' Convair Division (NAS 9 12326), and Rockwell Int.-B-1 Division for aluminum for the liquid penetrant method.

All the data described in this report were taken from a total of nine alloys of aluminum, titanium, and steel. These alloys are described in Table 5-2. The predominant amount of data were taken with the four aluminum alloys.

Table 5-2 Alloys with Valid Data

Aluminum	Stee1	Titanium
2219-T87	PH13-8MO	6A1-4V
7075-T6511	4330V	
2024-Т6	PH17-4	•
7178-T651	4340M	

VI. DISCUSSION OF RESULTS

The tabulated and graphical computer output format (Appendix D), is convenient for assessing the state-of-the-art in NDE reliability. It contains:

- (1) the binomial experiment results (N,n) in each of 32 crack size intervals (tabulated in units of Mils and plotted in both Mils and cm.);
- (2) the 50 percent lower confidence limit for each interval POD(CL50
- (3) the 95 percent lower confidence limit POD(CL95); and
- (4) the number of new measurements required in each interval to demonstrate a 90 percent probability of detection at a 95 percent confidence level "POD90(CL95)".

These data are useful for comparing the effects of material, inspector, and inspection parameters on NDE reliability and seeing where data deficiencies lie.

As can be seen in Figure D-la the POD fluctuates widely throughout 32 ranges for this set of data because of the variation in the number of measurements, N, reported in each range. Twenty of the ranges contain less than 29 measurements while six ranges contained no measurements at all. The number of measurements per range can be increased by broadening the flaw size interval per range, but there is still no assurance that all of the broader ranges will contain measurements. Fluctuation in the probability of detection due to the variation in the number of measurements in a range is a shortcoming of the range interval method.

Figure D-1b shows the tabulated and graphic results obtained by using the optimized probability method (OPM) for the set of data in Figure D-1a. As can be seen, the number of measurements available for each interval is much larger than for the range scheme. The thirty-first interval lists 183 detections out of 183 measurements resulting in a POD of 98% plotted at 1.18 cm. The POD computed with the OPM increases monotonically with increasing crack length and does not fluctuate as in the range scheme. The POD at 95% CL reaches 89% for the first time at .32 cm for the OPM versus .358 cm for the range scheme.

Figure D-1c shows graphic and tabulated results using the overlapping 60 points scheme. In this scheme, the number of measurements for each POD calculation is constant (60). Fluctuation in the POD can be attributed primarily to human operator and inspection process variations. The POD at 95% CL reaches 89% for the first time at .328 cm, which is intermediate between the OPM and the RI results.

6.1 Effects of Material, Source, and Inspection Parameters

It is apparent from data subsets 1 through 8 in Table 5-1 that the sensitivity is increased for ultrasonic, liquid penetrant, and X-ray techniques after the test specimens were chemically etched. This increase is reasonable since the crack openings are enlarged. The sensitivity is decreased, however, for eddy current method after chemically etching the specimens. This effect is difficult to explain.

Data subsets 9 through 32 show that the difference in the POD90(CL95) obtained by different inspection operators within a company vary as much as those obtained by companies which use different inspection parameters and procedures. Several observations can be made with subsets 9 - 14: (1) Operators P and S in subsets 10 and 13 can be considered to be model operators. Not only did they achieve a smaller POD90(CL95) threshold level than other operators, they were able to maintain the POD(CL95) at a relatively constant (and high) level for crack lengths above the threshold level (Figures D-10 and D-13). (2) Subsets 10 and 13 show that the sensitivity of the ultrasonic method increases or remains constant for increasing crack length, particularly for crack lengths above the threshold level. (3) Fluctuations in the POD(CL95) for crack lengths above the threshold level are primarily caused by fluctuations in operator efficiency and less likely by the sensitivity of the NDE method.

Subsets 14-20 (Figures D-14 to D-20) show more fluctuation in the POD(CL95) due to variation in operator efficiency than to types of penetrant or procedures used by the three different companies that made the measurements. There is a large variation in the threshold level of the seven operators. Three operators demonstrated threshold levels around .33 cm, another three demonstrated the level at .737 cm, and the seventh one demonstrated it at 1.35 cm. One should keep in mind that most of these POD (CL95) curves are relatively flat and the POD(CL95) is greater than 80% from .254 cm on to the longer crack length.

Like the ultrasonic and penetrant results, the eddy current results showed considerable differences in the threshold level for the five different operators. In fact Operator V did not demonstrate POD90(CL95). However, he did demonstrate POD88(CL95) at a crack length of .737 cm. Operator T demonstrated POD90(CL95) at a crack length of .665 cm. However, he reached POD88(CL95) at a crack length of .399 cm.

X-ray is not a good inspection technique for reliably detecting fatigue cracks. Data sets, 26 to 32 show that only one operator can achieve POD90(CL95). This forecasts that X-ray techniques will not be viable procedures for detecting fatigue cracks. However, this does not necessarily mean that the technique should not be used to detect other defects such as porosity.

Merging data provides a means for observing overall detection reliability trends. When the data for five ultrasonic operators were merged (subset No. 33) and calculated with the OSP scheme the POD(CL95) reached 95% at a crack length of 0.66 cm (see Figure D-33c). However, it fell below 95% at higher crack length, and it reached a minimum of 79% at a crack length of 0.810 cm. When the results of seven penetrant operators were merged (subset No. 34) and calculated with the OSP scheme, the POD(CL95) reached 92% at a crack length of 0.26 cm (see Figure D-34c), but it fell below 90% between 0.26 cm and 0.63 cm. It peaked to 95% at 0.65 cm, then fell below 90% and reached a minimum of 81% at a crack length of 1.12 cm. When the results of five eddy current operators were merged (subset No. 35) and calculated with the OSP scheme, the POD(CL95) reached 92% at a crack length of 0.32 cm (see Figure D-35c). But like the ultrasonic and penetrant results, the POD(CL95) for the eddy current fell below 90% at higher crack length and it reached a minimum of 81% at 0.88 cm.

Data subsets 36 through 47 (Figures D-36 to D-47) are from the B-1 NDI demonstration program. These data show that the POD(CL95) either increases monotonically or remains constant with increasing crack length once the threshold level is reached. In several data sets no further measurements were made once the desired threshold level was demonstrated. It is interesting that the threshold level as well as a large portion of the POD(CL95) curve does not differ greatly for several types

of penetrant applied to materials such as Ti-6A1-4V, 7075-T6511 A1, PH 13-8 Mo steel, 4330V steel, and PH 17-4 steel.

Data subset 48 represents the merged results of four data sources (RI-Space Division, Martin Marietta, General Dynamics Convair, and RI-B-1) on 2219-T87 Al by the liquid penetrant method on flat plates.

Data subsets 49 through 56 were obtained with fatigue cracks in 4330V steel and 7075-T6511 A1 having a cylindrical shape (7.62 cm in diameter and .62 cm thick). Data were taken in both the laboratory and production environments. None of the curves for these 8 sets of data attained POD90(CL95), however, the penetrant results for 7075-T6511 A1 did demonstrate POD87(CL95) at a crack length of 1.0 cm, POD87(CL95) at a crack length of 1.17 cm, and POD89(CL95) at 1.47 cm. The POD for ultrasonic shear wave reached a value of POD86(CL95) at a crack length of .80 cm and POD87(CL95) at a 1.27 cm.

Data subsets 57, 58, and 59 were obtained with eddy current inspection of bolt holes during tear-down inspection of KC-135 wings at the depot level. The same holes were inspected by five separate inspection teams. The inspection results from four of the five teams were about the same. Figure D-57 shows the results of team number two which is representative of the four teams. Figure D-58 shows that the detection reliability of team number four is considerably different than the other four teams. Figure D-59 shows the merged results of the five teams. None of the five teams achieved POD90(CL95).

Data subsets 60 through 112 (Figures D-60 to D-112) were obtained from the Practical Sensitivity Limits program which was conducted by Boeing Commercial Airplane Company and supported by the NDE Branch of Air Force Materials Laboratory. inspection results were obtained in both the laboratory and production lines of several Boeing Divisions. Divisions used different NDI procedure or specification, as well as different NDI techniques. For example, ultrasonic shear and surface waves of 5 and 10 MHz were used, different liquid penetrant systems in either the Group V or VI category were used, etc. Because the results were obtained under different NDE procedures, it is difficult to make comparison between results from different data sources. Furthermore, the majority of the defects used in this program were forging cracks instead of fatigue cracks, nevertheless the length of the crack at POD90(CL95) found by each of the NDI techniques is comparable, in most cases, to those found at other companies using fatigue cracks.

6.2 Statistical Analysis Procedures

All NDE measurements are considered to be binomial and describable by binomial distribution functions and binomial statistics. Because of this consideration, lower one sided confidence limits and data deficiencies can be rigorously computed.

The only factor which reduces mathematical rigor is the crack length parameter. It is necessary to group NDE data into a size interval that will contain a population of 29 successes without a miss in order to show 95 percent confidence that the probability of detection exceeds 90 percent. If a miss is encountered, it is necessary to have a higher population (see Table 6-1) in order to demonstrate POD90(CL95).

Table 6-1

Interval Sample Size and
Successes Required to Achieve POD90(CL95)

No. of	Interval Population	No. of
Misses	Required, N	Successes, n
. 0	29	29
1	46	45
2	61	59
3	76	73
4	89	85
5	103	98
6	116	110
7	129	122
8	142	134
9	154	145
10	167	157

Once an acceptable size interval has been chosen that will produce a reasonable population, there remains a problem of selecting the crack length within the range at which to plot the computed POD(CL95). For conservatism, the POD curves in this report are plotted through the maximum crack length in the interval. The crack interval is also indicated by a horizontal line connecting maximum and minimum crack sizes.

Three procedures for determining the probability of detection for the lower one-sided confidence level are compared in this report. Described earlier in Section 4.3, they were the range interval (RI) method, the optimized probability method (OPM) and the overlapping sixty point method (OSP). The optimized probability method is preferred over the RI or OSP but requires more computational time. The main claim to success for the OSP method is that it forces groupings to be large enough to provide the capability for achieving POD95 (CL95). The OPM should provide the upper envelope to both the RI and the OSP methods.

6.3 Data Deficiencies

The data described in this report represents more than 30,000 measurements, yet there are many gaps in the probability of detection for different defect types, material types, and specimen geometry by the five NDE methods (ultrasonics, liquid penetrant, eddy current, X-ray, and magnetic particle). most data are available for fatigue cracks in flat plates of 2219-T87 aluminum which reasonably characterize the reliability of all five NDE methods. There are also several sets of data for fatigue cracks in aluminum cylinders, forging cracks in extruded aluminum (tandem T configuration), and fatigue cracks in bolt holes of actual aluminum aircraft parts. There is a notable data deficiency for steel. Most of the steel data are for fatigue and forged cracks in cylinders. There are. however, four sets of data for fatigue cracks in flat plates of steel. Only five sets of data are available for titanium and all of them were taken on flat plates.

With the exception of the B-1 NDE demonstration data most of the data exhibit a gap over some range of the crack lengths. A perfect example of this is the data from Martin Marietta, Rockwell International-Space Division, and General Dynamics-Convair Division. All these three companies made measurements using the same set of specimens which contains two gaps. As can be seen from the crack size distribution curve in Figure 6-1, there are no cracks with length between 0.47 to 0.59 cm and between 1.03 to 1.17 cm. Thus, no cracks can be detected in these two gaps and the probability of detection, regardless of which cumulative scheme is being used, is affected. A set of specimens without gaps in the crack size distribution curve is needed to study the actual probability of detection in future programs.

6.4 Application to Fracture Mechanics

The threshold crack length (the shortest crack length that reached to POD90(CL95)) that was detected by the different NDE techniques is given in Table 5-1. However, as can be seen from the curves in Appendix D, the POD (CL95) in many instances falls below 90% for some crack length longer than the threshold level. Thus the potential user is cautioned against using these threshold crack lengths in the design In the first place, the POD can vary from company to company even when the same inspection procedures were used because of training procedures and using different inspectors. In the second place, some of these POD curves do not have a zero or positive slope for crack length above the threshold level. The decrease in the POD above the threshold level could be a real phenomenon and might not necessarily be attributed to human factors. There are scatter reports, in the case of liquid penetrant, that indicate penetrant washes out of wide cracks before the developer is applied.

In the fracture mechanics design process, one must be assured with some probability at some confidence level that no cracks larger than the assumed size can be present in the structure. Thus, for fracture mechanics to be a viable design process, the probability of detection as a function of crack length must be monotonically increasing or at least levels off above some selected threshold crack length.

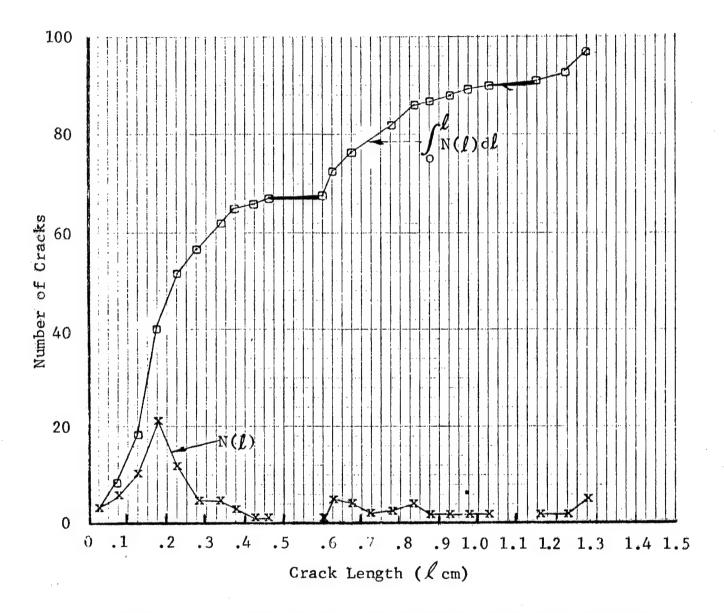


Figure 6-1 Differential and Integral Crack Distribution Function for the Martin Marietta Demonstration Program. 2219-T87.

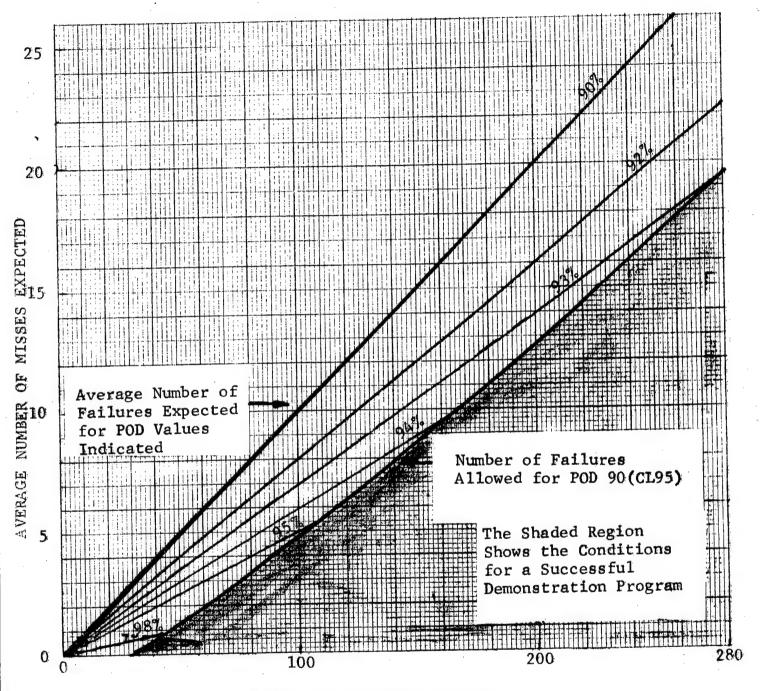
6.5 Optimum Demonstration Program

In order to determine the reliability of an NDE technique without operator influence, the optimum demonstration program will have to employ computer automation in scanning specimens and interpreting NDE signals. It should incorporate ultrasonic, eddy current and penetrant inspection into each demonstration program.

The data in this report can be used to estimate the sample size required for the optimum demonstration program. For example, if one can afford to test 100 specimens and requires a 50-50 chance of success in demonstrating POD90/CL95 then one would have to select a data set representative of this expected capability and select a crack length for his demonstration program for which the probability of detection is at least 95%. If he can afford only 45 specimens then he must choose a crack length for which the probability of detection is at least 98%. This can be seen in Figure 6-5 where number of misses corresponding to POD90(CL95) is plotted against the number of specimens tested. The shaded region is the region of success for which POD 90(CL95) has been met or exceeded. The broken lines show the number of failures expected for various values of POD (98, 95, 94, 93, 92, and 90%).

Assume that data set # 1 (Figure D-1) is for a geometry and NDE method that is most representative of those required in a demonstration program. These data demonstrate a POD(CL95) at .345 cm. Figure D-1a would lead one to believe that POD 90(CL95) might be demonstrated at about .27 cm for which 38 out of 39 cracks were detected. Figure D-1a shows that one can be 50 percent confident that the probability of detection exceeds 95% for this case if his NDE techniques are as good as those used to produce data set # 1. Figure 6-2 shows that if 110 test cracks were fabricated and tested and that less than 5 cracks were missed, then the demonstration program would be successful.

The previous example is indicative of the usefulness of these data as a means for minimizing the number of specimens required for achieving a successful demonstration program. The common practice of manufacturing a large number of crack specimens for which there is little chance to achieve POD 90(CL95) will be eliminated by this approach to designing an optimum demonstration program.



NUMBER OF SPECIMENS TESTED

Figure 6-2 Conditions for Success in Selecting a Sample Size that will Potentially Demonstrate POD90(CL95)

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR

VII. CONCLUSIONS

This program was sponsored by NASA as a service to the new fracture mechanics approach to structural design whereby system reliability may be related to the expectation that flaws larger than a specific size may exist in the structure. The program objectives were to (1) collect all available NDE reliability data, (2) examine the validity of the data, (3) select and computerize a statistical procedure for analyzing the data, and (4) generally assess the current state-of-the-art in NDE. We feel that the surface has been broken by this program but vigorous follow-on work would be very valuable to the field of structural design.

Some of the specific conclusions that can be drawn from the contents of this report and the experience gained in performing this study are:

- (1) The human factors influence stands out as dominating in influencing NDE reliability. Although a company can demonstrate a 90 percent probability of detecting a crack at a lower one sided confidence limit at the 95 percent confidence level, there is no guarantee that this is indicative of a continuing capability unless some form of automation has replaced human judgment. This then indicates a need for some type of periodic audit of capability.
- (2) Although more than 30,000 measurements have been filed in a computer library and analyzed to determine the NDE reliability of X-ray, ultrasonics, eddy current, and penetrants, there are still many gaps in the data: There are no eddy current data in the data bank for titanium nor are there much data for crack lengths near .5 or near 1.2 cm. These data deficiencies cause large uncertainties in the POD plots for these crack lengths.
- (3) Binomial statistics are recommended for analyzing NDE reliability data. This report contains a comprehensive discussion of the preferred analysis model.
- (4) The optimized probability method (OPM) of data cumulation is preferred over the range interval or the overlapping sixty point scheme for computing probability of detection. The optimum probability model which was developed during this program is discussed in this report.

- (5) Data show that the X-ray technique is not viable for detecting tight fatigue cracks.
- (6) In order to obtain optimum results from an NDE demonstration program, special care must be taken in the design of flaw size distribution. The cumulative distribution must always maintain a positive slope or there will be gaps without data which can have deleterious effects on the POD curves as a function of flaw sizes.
- (7) The data presented in this report are offered as a guide for manufacturers who intend to perform an NDE reliability demonstration program and can identify data sets that represent their expected NDE capabilities. However, due to the human factor influence on POD, the data cannot be used as design information. Each company must develop reliability numbers for their own use.
- (8) A continuing program to maintain an NDE reliability data file and to optimize the data processing procedures formulated during this program would be useful to the field of fracture mechanics. As the data base becomes larger it will begin to support parametric studies of the influence of NDE variables.

APPENDIX A

A MODEL TO GRADE THE QUALITY OF THE DATA SETS

The thoroughness of the characterization of the defectspecimen and the documentation of the inspection procedures affects the quality and the usefulness of the data sets. A designer will not have confidence in using the data unless the inspection procedures and defect dimensions are sufficiently documented so that it can be reproduced in future inspections.

The model described in this section will only relate to the quality of each set of data. It will not address the question of applicability. That is, a set of data obtained on flat

plates will not be graded on the basis of its applicability to the design of complex structures.

The model is empirical and the weighting factors assigned to the various known pertinent parameters are rather arbitrary. However, it does represent a first attempt to quantitatively evaluate the quality of a data set. Each of the pertinent parameters in this model is listed in the Input Data Format which will be discussed in the next subsection. The grade for a given set of data can be tallied in the computer by checking the entries to the columns containing these pertinent parameters. A score of one hundred (100) corresponds to a perfect set of data. A perfect set of data is one where all the pertinent parameters are documented.

The preliminary model to grade the data quality for the ultrasonic, eddy current, liquid penetrant, magnetic particle, and X-ray methods is given in Table A-1. The model is divided into two major groups. Group A is the characterization of specimens and defects and it is common to all techniques. Group B is the documentation of the inspection and is different for different NDE techniques. The first eight parameters in this last group are common to all techniques. Each technique has about eleven parameters that are considered pertinent to adequately document the inspection. Parameter Al is the description of the specimen geometry and defect location. This is considered a key parameter and a value of 7 points is arbitrarily assigned to it. If no description about the specimen geometry and defect location is given, that data point or set of data will receive no points. Those parameters that are marked by an asterisk are considered key parameters. If any of them is not recorded, the data point or set will be considered for possible exclusion from statistical analysis.

Table A-1

THE GRADING FACTORS USED TO GRADE NDE DATA QUALITY

Α.	Spe	cimen - Defect Characterization Informati	on	35
	1.	Specimen geometry complexity and defect location	ric .	7
	2.	Crack dimension verification	*	10
	3.	Defect type	*	3
	4.	Specimen surface condition	*	3
	5,	Material characterization (thermo-mechanical history)		3
	6.	Material inside defect	*	3
В.	Ins	pection Documentation		65
	1.	False indication recording		4
	2.	Inspector qualification level		3
	3.	Knowledge of defect orientation, location, and presence by inspector	ን ዮ	9
	4.	Use of proof load	* %	3
	5.	Use of control specimens without crack		3
	6.	Method of data recording		3
	7.	Method of scanning specimens		3
	8.	Insp. environment	*	3
	9.	Number of insp. prior to this insp.		3
	10.	Parameters recorded by nondestructive testing standards		
		a. Radiography		38
		1) Radiographic source	*	3
		2) Ref. standard		3

10. (Continued)

	3)	Detector type *	4
	4)	Voltage *	4
	5)	Current *	4
	6)	Exposure time *	4
	7)	Source/film distance *	4
	8)	Angle of entry *	3
	9)	Film development parameters	3
	10)	Radiograph density *	3
	11)	Radiographic equipment type	3
ъ.	U1t	rasonics	38
	1)	Ultrasonic method *	4
	2)	Frequency *	4
	3)	Transducer type and size *	4
	4)	Reference standard type and size *	4
	5)	Angle of incidence (inside the * material)	3
	6)	Equipment type	3
	7)	Gate alarm level (% of ref. signal) *	4
	8)	Gain setting (% of screen saturation) *	4
	9)	Type of coupling	3
	10)	Index interval *	3
	11)	Contact or Immersion	3

10. (Continued)

c.	Edd	y Current		38
	1)	Method of Scan	ř	3
	2)	Coil size	%	4
	3)	Coil arrangement and shape	2,4	3
	4)	Frequency	*	4
	5)	Reference type and size	*	4
	6)	Equipment type	•	3
	7)	Index interval	*	3
	8)	% of meter response (Real Part)	*	4
	9)	% of meter response (Imaginary Part)	*	4
	10)	Lift-off compensation		3
	11)	Signal processing		3
d.	Pen	etrant		38
	1)	Penetrant type	*	4
	2)	Developer type	*	4
	3)	Classification of penetrant (group	2
	4)	Emulsifier type	*	4
	5)	Pre-insp. surface cleaning and penetrant removal	d *	3
	6)	Method of penetrant application	on	3
	7)	Dwell time	*	4
	8)	Development time	*	4

10. (Continued)

	9)	Wash time		4
	10)	Light type and intensity at specimen surface		3
	11)	•		3
e.	Mag	netic Particle		38
	1)	Type of current used	*	4
	2)	Current level (Amperes)	*	4
	3)	Method of magnetization		4
	4)	Direction of magnetization	ボ	4
	5)	Magnetic flux density	*	4
	6)	Magnetic particle type and size	*	3
	7)	Magnetic particle density		3
	8)	Type of liquid vehicle		3
	9)	Method of particle application		3
	10)	Equipment type		3
	11)	Dwell time (seconds)	*	4

APPENDIX B

DATA POOLING

B.1 Data Pooling by NDE Method and Parameters

Data that meet the preliminary criterion as described in Subsection 3.1 are input to the computer for statistical analysis. Then data from several sets are pooled and analyzed if they have a common set of parameters. Data from different NDE methods are not to be pooled. For a given NDE method such as ultrasonic shear wave at 5 MHz, reliability curves will be plotted for a material type, a defect type, an environment, a specimen geometry and defect location, and either before or after some enhancement such as a proof test. Composite reliability curves can be plotted by pooling data with different parameters, such as pooling 5 and 10 MHz shear wave data, laboratory and production data, flat plate and cylindrical shell data, etc.

B.2 Statistical Pooling Criteria

The NDE data that is compiled for this contract was collected from different sources using different calibration factors, different equipment, different personnel, different environments, etc. Each set of data therefore contains unique source characteristics that preclude indiscriminate pooling for reliability calculations.

A statistical pooling criteria has been developed to safeguard against mistakes or inconsistencies in the data which would produce abnormal statistical results. The data pooling technique is based solely on the binomial distribution and is described below as a procedure which can be implemented on a computer. The procedure consists of the following four steps.

(1) The best single estimate, \bar{p}_c , for probability of detection

$$\bar{p}_{c} = \frac{\sum_{k=1}^{M} n_{k}}{\sum_{k=1}^{M} N_{k}}, \qquad (9)$$

where M is the number of data sets to be pooled.

(2) The best single estimate of the true probability

$$\bar{p}_{k} = \frac{n_{k}}{N_{k}} \quad , \tag{10}$$

of each of the data subsets is computed.

(3) Consider the binomial distribution function for each data set (N_k, n_k) having a true probability of detection given by p_c . The two-sided probability, α_2 , that (N_k, n_k) and all less likely outcomes are possible is computed from

$$\alpha_2 = 2 \sum_{i=0}^{n_k} {N_k \choose i} \bar{p}_c^i (1-\bar{p}_c)^{N_k-i}$$
 (11)

if
$$\frac{n_k}{N_k} < \bar{p}_c$$

or by

$$\alpha_{2} = \sum_{i=1}^{N_{k}} \binom{N_{k}}{i} \quad \bar{p}_{c}^{i} \quad (1-\bar{p}_{c})^{N_{k}-i}$$

$$if \quad \frac{n_{k}}{N_{k}} > \bar{p}_{c} \quad . \tag{12}$$

(4) All data sets having a value of α_2 less than a reference value α (computer input value) are removed as candidates for pooling.

The choice of α' is somewhat arbitrary and depends upon the acceptable risk. The data sets that will be rejected from pooling for a given α' value will be reviewed. If no abnormalties are found within each set of data (i.e., no mistakes in data recording, or other possible means of causing the probability of detection to be normally high or low),

a new value of α' will be tried and one that will permit the data sets to be pooled with the data base will be selected. Thus, the value for α' may be governed by operator judgment of the validity of the data.

Table B-1 is an example which contains six hypothetical binomial experiments (assuming all sets have the same measurement parameters). (For this example α' was chosen to be .05). The α_2 values for sets C, E, and F are very low. The α_2 values for sets C and E were calculated using Equation (11) and for set F was calculated using Equation (12). For set C one is only 2.22% confident that one out of eight measurements is successful. For set E one is only 0.86% confident that one out of ten measurements is successful. For set F one is only 0.11% confident that seven out of seven measurements are successful. The confidence is too low for measurements to be pooled with those of sets A, B, and D.

Upon rejecting sets C, E, and F, new α_2 values are calculated for sets A, B, and D using the \bar{p}_c (16/40). These three sets have comparable confidence limits and they will be pooled.

Table B-1
SIX SETS OF DATA FROM DIFFERENT SOURCES
(A-F) TESTED FOR POOLING

Source	N Number of Measurements	n Number of Successes	a ₂ (Alpha)
Α	24	9	.4085 (accepted)
В	8	3	.3612 (accepted)
C	8	1	.0222 (rejected)
. D	8	4	.3572 (accepted)
E	10	1	.0086 (rejected)
F	7	7	.0011 (rejected)
TOTAL	65	25	
A	24	9	.3641 (accepted)
В	8	3	.3361 (accepted)
D	8	4	.3834 (accepted)
TOTAL	40	16	
1	·		

APPENDIX C

NASA. FTN COMPUTER CODE

NASA. FTN is written in Fortran and organized with a main routine performing all executive functions. Main does no data manipulation, it simply calls other routines which manipulate the data.

The flow diagram Figure C-1 illustrates the run sequence of the program. Two inputs required for the program are the <u>Master Data File</u> and a <u>Problem Input Deck</u>. At the end of a production run, the program lists the computational results.

C-1 Master Data File

The master data file consists of a sequence of 'data sets'. Each data set grouping has a common combination of conditions under which a series of specimens were inspected. These conditions are described in two 256 word records (header records). The results of the inspection are called data points and are stored after the two header records. Each data point is 8 words long and contains: crack ID, detect code, crack length, crack depth, measured crack length, measured crack depth, surface finish and thickness. The 'data records' consists of 256 words which contain 32 data points in each. Each data set is labeled by a number which is unique to the data set. The data set number and the number of data records per data set are stored in the header records of each data set. The master data file is stored on magnetic tape in binary format, 256 words per record.

C-2 Program Flow (Ref. to Table C-1)

The first step in running the program is to read all the program options. The subroutine then reads the master data file to extract, data points for analysis. These data are stored on a local disk file for up to 10,000 data points. Each data point on this data file is described by four words, crack size, detect/no detect, crack ID, and data set number. The input options have already (either implicitly or explicitly) described a set of common conditions. Subroutine Order then rewrites the local data file in increasing crack size.

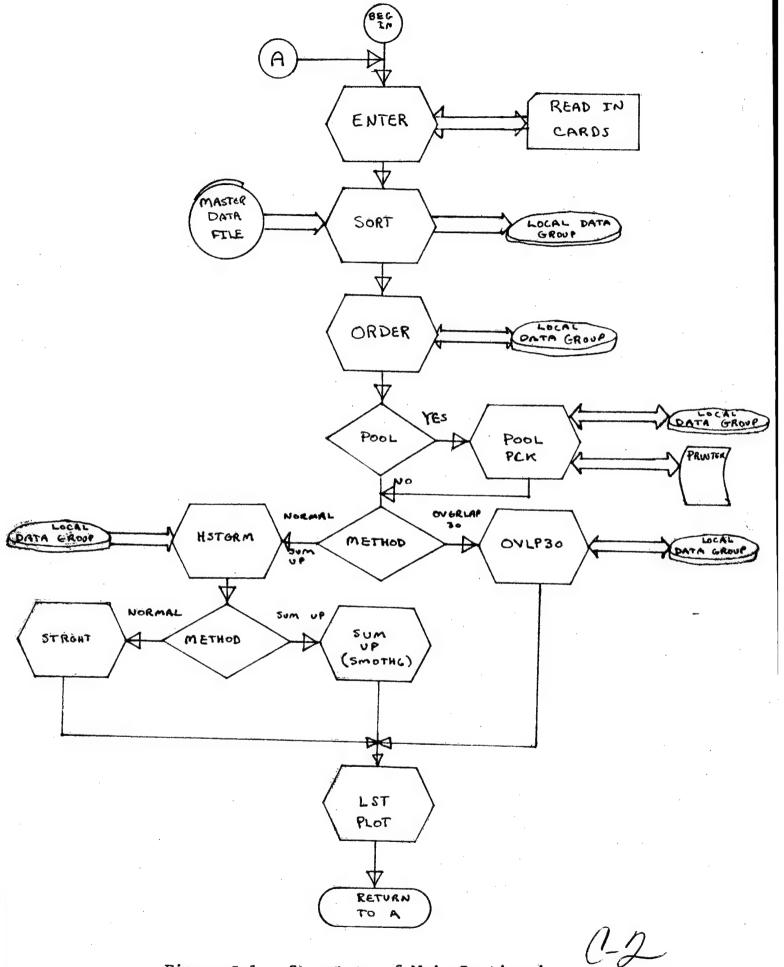


Figure C-1 Structure of Main Routine in NASA. FTN Code

Table C-1

Glossary of Programs and Subroutines Used in NASA, FTN

NASA		Main program to control the program steps flow in Figure
ENTER	-	Routine to read in cards which contain the program options. The options are stored in commons for use by other routines.
SORT	-	Routine that reads the master data file, sorts out data for analysis, then writes this data on local disk file.
ORDER	-	Routine to sort the data on the local data file so that the order of the data will be by increasing crack size.
POOL	-	Routing to determine whether a particular data set belongs to the rest of the sets being considered.
POOLR	-	Routine to calculate probability of data sample belonging to a group having a known probability of detection.
RJCT	- .	Routine to zero out data points of a data set that has been rejected on statistical grounds.
PCK	-	Routine to remove 'holes' in the local data file that are created by RJCT and to form a packed data file.
HSTGRM	-	Routine to sort data into 32 size interval groups.
OVLP30		Routine to perform an analysis of the data by using the procedure known as the "OSP" scheme.
SMOTHG		Routine to perform an analysis of the data by using the procedure known as the OPM.

Table C-1 (Continued)

STRGHT - Routine to perform analysis of the data by using the procedure known as the "RI" scheme.

BIN - Routine to calculate POD at a specific confidence limit for a binomial experiment consisting of N measurements and n successes.

DEFJC - Routine to calculate the number of successful experiments which must be performed in order to know the lower limit probability of detection within a given confidence limit.

LST - Routine to list the results of the analysis.

CKNRT - Routine which performs paging of the local data file.

If pooling is desired, routine POOL can be used to calculate the probability that each data set statistically belongs to the rest of the data sets that are collected together. Any data set that gets rejected is removed from the local data file by zeroing the data points. Routine PCK takes the local data file and re-packs it so that there are no zeroed points. All data sets removed are listed on the printer. The method of analysis determines the route the program takes next. If 'overlapping 60' is chosen routing OVLP30 "OSP" takes the local data file and processes the data directly. Routing HSTGRM is used to divide the data points into 32 groups which have equal crack size intervals ranging between a minimum and maximum input crack size. All crack sizes falling below the minimum are included in the first interval and all data falling above the maximum are included in the thirty second interval. Either the routine STRGHT "RI" or the SMOTHG "OPM" analysis procedure can be selected to analyze the data. The results are then listed. The program then returns to the beginning to read more cards. If there are no more problems to be solved, the program terminates.

C-3 Sorting Option

The program has two sort options. The first option uses the data set number directly. (See Figure 4-2). The user specifies the number of data sets to be considered, then specifies the data set numbers. The second option (Figure 4-3), makes use of the header card information which includes 26 words or numerical codes for the alphanumeric data that is contained in the header records. By supplying values for these words, each header record can be compared to this 'master' input. No zero values are compared. Each time a match is made, that data set is stored on the local data file. A maximum of 5 master header records can be combined per run.

C-4 Program Limitations

The maximum crack size that can be stored and analyzed is 2.54 cm. The maximum number of data points that can be combined into a set is 1000. The maximum number of measurements that can be analyzed by the subroutine "DEFIC" which computes data deficiencies is 120.

IDSN = NUMBER OF DATA SETS TO BE SELECTED

JOSN = ARRAY CONTAINING THE DATA SET NUMBERS

DSN = DATA SET NUMBER (READ FROM MASTER DATA FILE)

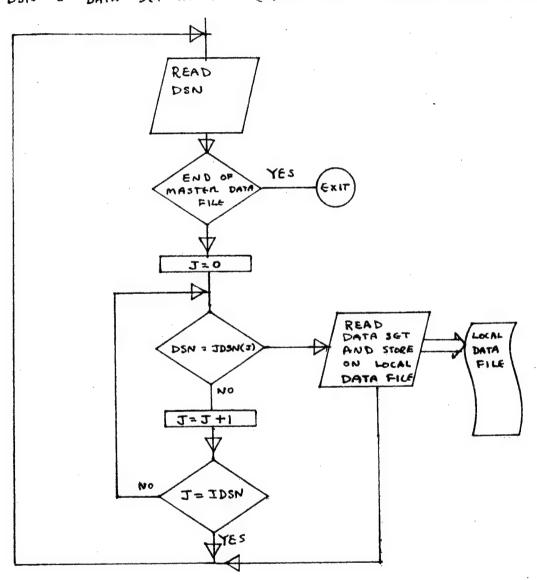


Figure C-2 Sort Subroutine Flow When Using Data Set Number Only for Sorting

C-5 Local Data File

The local data file is a direct access file which occupies a continuous block or disk space. A paging method has been incorporated to avoid executive disk I/O. The subroutine <u>CKNXT</u> is used for paging. At any given time, 1024 data points may be held in the core, therefore a disk access does not have to be made each time a new data point is addressed. The local data file has ten pages available for program use.

C-6 Ordering Method

A subroutine names "ORDER" reorders the local data file entries in increasing crack length then rewrites the ordered data file in place of the original unordered local data file. This ordered data is then used as input for data pooling and statistical analysis.

C-7 Lower One-Sided Confidence Limits Calculations

The binomial Equation (5) is solved by setting G equal to the confidence level desired and determining the value of p_1 which satisfies the equation. A Fortran IV program which has been made operable in a PDO 11/45 computer to perform this function is given in Figure 4-5. p_1 is the probability of detection at the lower confidence limit.

The procedure used to determine how many more measurements are required in each interval for the 95% confidence curve to reach the 90% probability of detection level is to solve Equation 8 for and . Figure 4-6 gives the listing of a Fortran IV program to perform this calculation.

C-6-a

ICCN = NUMBER OF CONTROL CARDS (MAXIMUM OF S)

JCCN (26, ICCN) = ARRAY OF CONTROL CARD IN FORMATION THAT

IS INPUT

KCC (26) = ARRAY OF CONTROL CARD INFORMATION FOR EACH DATA SGT

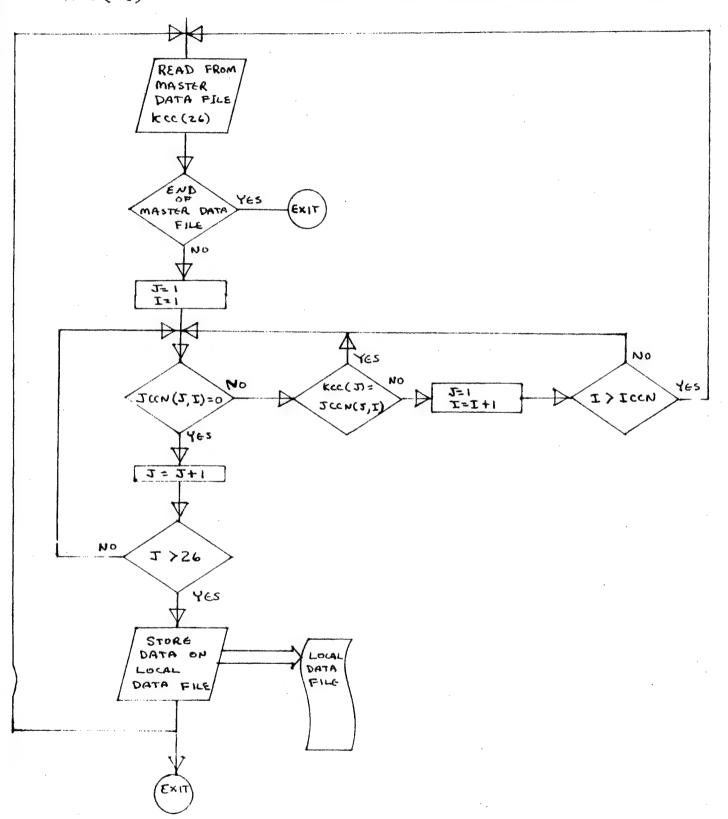


Figure C-3 Sort Subroutine Flow When Using Master Control Records to Sort on Master Data File

```
SUBROUTINE BINCAR1, AR2, AR3, AR10)
 1 IF(AR2)2,2,4
 2 AR10=0.0
 3 RETURN
 4 IF(AR2-AR1)7,5,5
 5 AR10=(1.0-AR3)**(1.0/AR1)
 6 RETURN
 7 ATT=2.0*AR2
 8 IF(ATT-AR1)9,9,12
 9 AR4=AR2-1.0
10 AR5=-1.0
11 GO TO 15
12 AR4=AR1-AR2
13 AR3=1.0-AR3
14 AR5=-1.0
15 AR10=0.5
16 AR6=1.0
17 ARS=0.0
18 AR9=1.0
19 AR11=AR1
20 AR7=(AR10**AR8)*((1.0-AR10)**(AR1-AR8))
21 IF(AR8-AR4)22,27,22
22 AR8=AR8+1.0
23 AR9=AR9*AR11/AR8
24 AR11=AR11-1.0
25 AR7=AR7+AR9*(AR10**AR8)*((1.0-AR10)**(AR1-AR8))
26 GO TO 21
27 IF(AR3-AR7)28,28,30
28 AR20=AR10-AR5/(2.0**(AR6+1.0))
29 GO TO 31
30 AR20=AR10+AR5/(2.0**(AR6+1.0))
31 CCC=ABS(AR7-AR3)
32 IF(CCC-0.0001)36,36,33
33 AR6=AR6+1.0
34 AR10=AR20
35 GO TO 17
36 IF(ATT-AR1)6.6.37
37 AR10=1.0-AR10
38 RETURN
39 END
```

Figure C-4 Computer Code Used to Calculate Lower One-Sided Confidence Limit

```
SUBROUTINE DEFICCARLAR2, FFF, AR4)

1 A=AR1-AR2+FFF
2 AR4=0.8
3 IF(A)4,4,9
4 AR4=29.8-AR1
5 IF(A)4,4,9
6 AR4=8.8
7 RETURN
9 X=1.8
11 C=1.8
12 CCC=AR1+AR4
13 AR5=8.9 X*CCC
14 C=(C/X)*(AR1+AR4+1.8-X)
15 DDD=AR1+AR4-X
15 DDD=AR1+AR4-X
15 DDD=AR1+AR4-X
15 DDD=AR1+AR4-X
16 AR5=AR5+(8.1*X)*XC*(8.9*XDDD)
17 IF(A-X)18,18,21
18 IF(B.85-AR5)19,19,7
19 AR4=AR4+1.8
11 C AR4 EQ. 180. > RETURN
28 GO TO 9
22 GO TO 14
23 END
```

Computer Code Used to Calculate Additional Measurements Required to Achieve 90% Probability of Detection at 95% Confidence Limit Figure C-5

C-8 DATA INPUT FORMAT

The data are filed in mass memory by using a 'data set' concept. All the common control parameters for a group of data points are loaded together. There are 26 such control parameters. The first 14 are the same for all NDE methods, the next 12 are different for each of the five NDE methods. The results of the inspection follow the control parameters. Eight numbers are used to describe each point. (Crack ID, detect code, crack length, crack depth, measured crack length, measured crack depth, surface finish, thickness).

A typical control parameter listing is shown in Table C-1. Nine hundred eighty four flaw measurements have these parameters in common. The data in Table C-1 can be compacted for computer storage into the format shown in Table C-2. A special data set listing like that shown in Table C-1 can be prepared from the line of integers shown in Table C-2 by using the parameter key included at the end of this appendix. Four of the 984 data points for data set # 1 were entered onto each of 246 lines in the format depicted in Table C-3. Each complete data set is terminated by a 0000 entry in Columns 1 through 4 of the digital computer data sheet.

DATA SET NUMBER : 1

MERTIN MARIETTA DETECTION OF FATIGUE CRACKS ALUNINAM 2219-T87 RAUMINAM 2219-T87 FATIGUE CRK. AND CYCLE-REMOUE EDM NA CLIMITALIFIED ACCORDING TO MIL-STD-453 PRODUCTION STD. INSP. / MULTIPLE FLOW SPEC. STD. INSP. / MULTIPLE FLOW SPEC. REDAMNICAL SCAN AND INDEX FLAT BOTTOM HOLE ATR ATR
NDE METHOD COMPANY NAME PROGRAM TO MATERIAL DEFECT TYPE OPERATOR TO QUALIFICATION TINSP ENUIRONMENT TINSP PROCEDURE DATA RECORD TYPE MODE OF SCAN REFERENCE STANDARD DEFECT MATERIAL PART GEOMETRY
-06.40.00.00.00.00.00.4 0400604900

15) 1 ULTRASONIC METHO
16) 4 FREQUENCY
17) 1 XMITTER TYPE/SIZ
18) 1 RECEIVER TYPE/SIZ
19) 1 EQUIPMENT TYPE
20) 0 GAIN SET 2 OF SIZ
22) 1 TYPE OF COUPLING
23) 1 ANGLE OF INCIDEN
24) 1 INDEX INTERIGE

LSE ECHO

THOTOENCE : ZO. CO IN

Table C-2 Data Set Entry for Compact Storage of Data Set Parameters

Computer Input:

DIGITAL COMPUTER DATA

2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 21 22 23 24 25 26 27 28 29 30 31 32 33 34 35 36 37 36 39 40 44 45 46 47 48 49 50 51 52			1	
2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 21 22 23 24 25 26 88 87 2		Hr		
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2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 21 22 23 24 25 26 88 87 2		94		
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2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 21 22 23 24 25 26 88 87 2		3.	+	
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Computer Output:

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3 1 1

2 4 3

1 1 2

2 1 1

5 4

Table C-3 Flaw Size Data Entry Format Showing Last Four Entries with End of File Designated by "0000"

Computer Input:

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	1	Blank
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9 79	0	Thickness
78	N	
75 77	7	Surface Finish
75 7	10	Measured Crack
74 7	- 37	Depth
73	2	Crack Depth
1 72	1.0	
0 71	0	Measured
9 70		Crack Length
69 6	0	•
67 68	5	
9 99	0	Crack Length
65	٩	Detect Code
64	0	
63	0	Crack I.D.
62		*
9		
9 09		
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9	0	
29 60	200	
58 59 60	2200	
57 58 59 60	22200	
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Computer Listing

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AND CYCLE/RENDUE

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SPECIAL KEY FOR EDDY CURRENT DATA

- 1 NOE METHOD
- 2 COMPANY NAME
- 3 PROGRAM ID
- 4 MATERIAL
- 5 DEFECT TYPE
- 6 OPERATOR ID
- 7 QUALIFICATION
- 8 INSP. ENUIRONMENT
- 9 INSP. PROCEDURE
- 10 DATA RECORD TYPE 11 MODE OF SCAN
- 12 REFERENCE STANDARD
- 13 DEFECT MATERIAL
- 14 PART GEOMETRY
- 15 EQUIPMENT TYPE
- 16 DIAMETER OF COIL
- 17 CONF. SHAPE OF COIL
- 18 FREQUENCY
- 19 TYPE OF EC RESPONSE
- 26 LIFT-OFF COMP.
- SIGNAL PROCESSING
- 22 INDEX INTERVAL

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ve=5224557858282

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SPECIAL KEY FOR LIQUID PENETRANT DATA

- 1 NOE METHOD
- 2 COMPANY NAME
- 3 PROGRAM ID
- 4 MATERIAL
- 5 DEFECT TYPE
- 6 OPERATOR ID
- 7 QUALIFICATION
- 8 INSP. ENVIRONMENT
- 9 INSP. PROCEDURE
- 18 DATA RECORD TYPE
- 11 MODE OF SCAN
- 12 REFERENCE STANDARD
- 13 DEFECT MATERIAL
- 14 PART GEOMETRY
- 15 PENETRANT TYPE
- 15 DEVELOPER TYPE
- 17 CLASS OF PENETRANT
- IS REMOVER EMULSIFER
- 19 APPLICATION METHOD
- 20 DHELL TIME
- 21 DEVELOPING TIME
- 22 HASH TIME
- 23 LIGHT INTENSITY
- 24 REMOUAL PRE-CLEANING

DEVELOPER TYPE 1 URESCO D4990

MAGNAFLUX ZP-4A ,-5,4-13, 4 URESCO URESCO D499C

PENETRANT TYPE 1 URESCO P-151

CLASS OF PENETRANT
1 GROUP 1
2 GROUP 3
4 GROUP 4
5 GROUP 5
6 GROUP 5

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR

APPLICATION METHOD I HAND BRUSH

รายการเกาะ ใช้ - B HASH TIME

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SPECIAL KEY FOR MAGNETIC PARTICLE DATA

- 1 NOE METHOD
 2 COMPANY NAME
 3 PROGRAM ID
 4 MATERIAL
 5 DEFECT TYPE
 6 OPERATOR ID
 7 QUALIFICATION
 8 INSP. ENVIRONMENT
 9 INSP. PROCEDURE
 10 DATA RECORD TYPE
 11 MODE OF SCAN
 12 REFERENCE STANDARD
 13 DEFECT MATERIAL
- 15 EQUIPMENT 16 PROCEDURE 17 MAG. PARTICAL TYPE 18 SUSPENSION TYPE

PART GEOMETRY

SPECIAL KEY FOR RADIOGRAPHY DATA

- 1 NOE METHOD
- 2 COMPANY NAME
- 3 PROGRAM ID
- 4 MATERIAL
- 5 DEFECT TYPE
- € OPERATOR ID
- 7 QUALIFICATION
- 8 INSP. ENUIRONMENT
- 9 INSP. PROCEDURE
- 10 DATA RECORD TYPE
- 11 MODE OF SCAN
- 12 REFERENCE STANDARD
- 13 DEFECT MATERIAL
- 14 PART GEOMETRY
- **15** RADIOGRAPHY
- 16 EQUITIPMENT TYPE
- 17 SOURCE ENERGY
- 18 SOUPCE STRENGTH
- 19 HINDOW MATERIAL
- DO TYPE OF FILM
- **21 EXPOSURE TIME**
- 22 SOURCE TO FILM DISTA
- 23 ANGLE OF INCIDENCE
- 24 DENSITOMETER PEADING

EQUI IPMENT

EQUITPMENT

555

TO FILM DISTA

SPECIAL KEY FOR ULTRASONICS DATA

- 1 NOE METHOD
- 2 COMPANY NAME
- 3 PROGRAM ID
- 4 MATERIAL
- 5 DEFECT TYPE
- 6 OPERATOR ID
- 7 QUALIFICATION
- 8 INSP. ENVIRONMENT
- 9 INSP PROCEDURE
- 10 DATA RECORD TYPE
- 11 MODE OF SCAN
- 12 REFERENCE STANDARD
- 13 DEFECT MATERIAL
- 14 PART GEOMETRY
- 15 ULTRASONIC METHOD
- 16 FREQUENCY
- 17 XMITTER TYPE/SIZE
- 18 RECEIVER TYPE/SIZE
- 19 EQUIPMENT TYPE
- 20 GAIN SET % OF SS
- 21 ALARM SET % OF SS
- 22 TYPE OF COUPLING
- 23 ANGLE OF INCIDENCE
- 24 INDEX INTERUAL

7	T	T	在山
RECEIVER	FLAT	FLAT	FEAT
REC		(v)	M

EQUIPMENT TYPE 1 UM 715 , 18N PULSER/RECIEVER

GAIN SET % OF SS 1 80 % SCREEN SATURATION

TYPE OF COUPLING

YPE OF CONTRACTOR STANDER STAN

817241978282828288

AURATION AURATION

N

APPENDIX D

COMPUTER OUTPUT - NDE RELIABILITY DATA

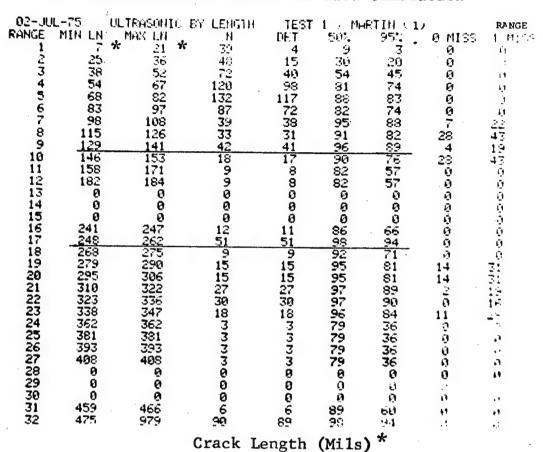
The data for Figures D-la, D-lb, and D-lc were obtained with an ultrasonic surface wave technique and at a frequency The abscissa is the surface length of a fatigue of 10 MHz. The NDE technique (ultrasonics) is given on the upper crack. left hand corner of the figures immediately after the date when the curves were plotted. Test 1, Test 2, or Test 3 is written on the same top line of the page. Test 1 refers to the range scheme where the lower one-sided confidence limit, was calculated for equal flaw size intervals. Figure D-la the flaw size interval is 0.038 cm. Test 2 (Figure D-1b) refers to the use of the optimized probability method to Test 3 (Figure D-1c) refers to the use of calculate the P_1 . the overlapping sixty point scheme to calculate the p₁. The top line of the computer output also contains the name of the company or the agency that published the report from which the data were taken. In Figure D-1 the company is Martin Marietta of Denver, Colorado. The test specimen martial type is identified in the figure caption on the bottom of the page or in Table 5-1 of Section V where the data set number can be matched with the number in parenthesis in the upper right hand side of each figure.

The first column of the tabulated results in these figures lists the range numbers. The second column gives the minimum crack length of each range and the third column gives the maximum crack length of each range.* The fourth column gives the total number of measurements or observations (N) for each size range and the fifth column gives the number of detections (n) for each size range. The probability of detection at 50% and 95% confidence level** is given in the sixth and seventh columns respectively. The last two columns list the number of new measurements that must be made with no misses and with one miss, respectively, to achieve 90% probability of detection at a 95% confidence level. The zeroes in these last two columns indicate that either no measurements need to be made or the number is too large for practical consideration.

^{*} Tabulated data give crack lengths in mils while the graphs give both mils and cm scales. This dual labeling is offered as a convenience for the design engineer.

^{**} Hereafter POD90(CL95) will be used to refer to a 95 percent confidence that the true probability of detection equals or exceeds 90 percent.

All the reliability curves discussed in this appendix are presented in the above format. Additional information such as flaw, type, material alloy type, contract or report identification number, and other pertinent parameters associated with the data used to generate these reliability curves and tables are included wherever possible.



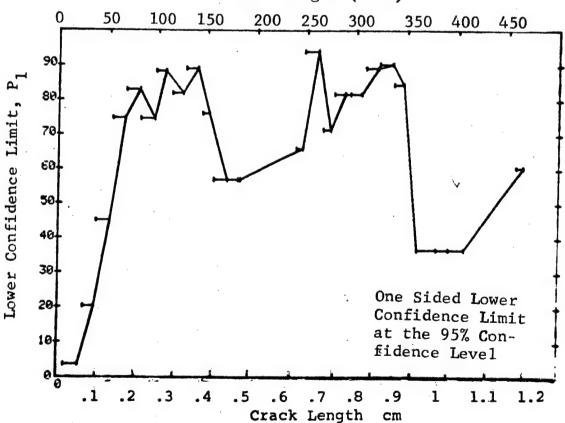


Figure D-1 Probability of Detection for 2219-T87 Al Using
Ultrasonic Surface Waves. Fatigue Cracks in
Flat Plates. Lab. Env.
D-B REPRODUCIBILITY OF THE

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(b) Optimum Probability Method of Data Cumulation

1 2 3 4 5 6 7 8 9 9 11 12 3 14 5 15 17 18 19 20 12 22 22 22 22 22 22 22 22 22 22 22 22	MIN 238488888888888888888888888888888888888	MAX LN 316 52 67 822 67 822 97 108 126 141 153 171 184 0 0 247 262 279 306 326 347 362 381 398 400 0	148 72 120 132 219 39 72 114 132 141 150 0 162 51 60 75 90 117 147 147 168 171 177 168	TEST DET 4 15 40 98 117 189 38 69 110 127 135 143 0 0 154 51 68 177 165 168 177 167 167 167 167 167 167 167 167 167	ୁ ଅଧିକ ପର୍ବ ପର୍ବ ବର୍ଷ ବର୍ଷ ବର୍ଷ ବର୍ଷ ବର୍ଷ ବର୍ଷ ବର୍ଷ ବ	RTIN 953 245 473 1889 222 1110 0 0 145 6 6 7 7 8 8 8 8 9 9 9 1 9 9 5 6 6 7 7 8 8 8 8 9 9 9 9 9 9 9 9 9 9 9 9 9 9	S	1 MISS 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
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Crack Length (Mils)*

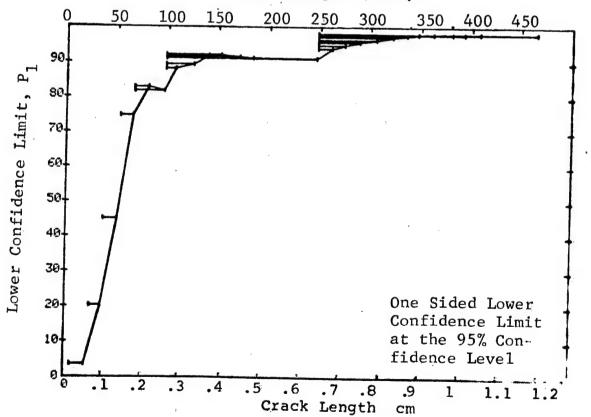


Figure D-1 (Continued)

02-JUI RANGE 12345678901123456789012 1123456789012 12322222223 123222222222222222222222	L-75 N	TRASONIC MAX LN * 9 20 30 40 45 51 561 667 70 773 77 803 87 95 102 115 129 136 158 2258 279 326 340 466 500	× ତ ଓ ତ ଓ ଉଦ୍ଧନ୍ଧ କରିଥିଲି । ଏହି ଓ ତ ଓ ତ ହେଉଛିଥିଲି । ଏହି ଓ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ	TEST DET 0004410160332433553555555555555555555555555555555	500 0 0 1 1 1 2 2 2 4 9 0 7 2 2 7 5 2 5 2 5 5 5 5 6 7 8 9 8 8 8 8 8 8 8 8 8 8 8 9 8 9 9 9 9	HT19500004973338095555555555555555555555555555555555	1) 000000000000000000000000000000000000	MI 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	с.
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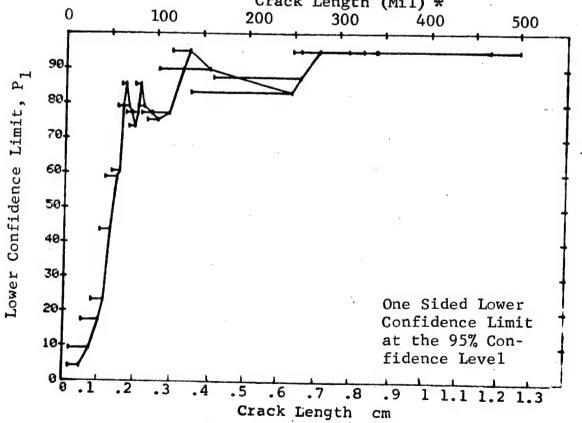
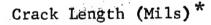


Figure D-1 (Concluded)

J.

02-10				TEST	1 , PEI		4FT.11 (2)	RANGE
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26	393	393	3	3	79	36	8	0
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28	. 0	Ø	. 0	Ø	0	Ü	₹.	
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. 32	474	979	90	63	2.1	00		



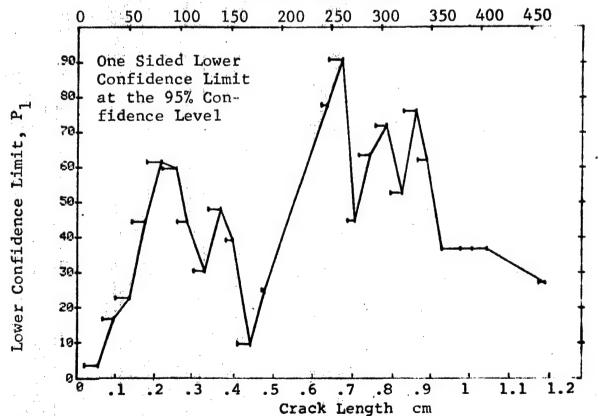


Figure D-2 Probability of Detection for 2219-T87 Al Using Liquid Penetrant. Fatigue Cracks in Flat Plates.
Lab. Env.

(b) Optimum Probability Method of Data Cumulation

02-JUU RANGE 1 2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17	NT-55548888488888888888888888888888888888	PENETRANT MAX LN % 21 % 36 52 67 82 97 108 126 141 153 171 185 0 0 247 262	N 39 48 120 120 132 219 251 333 351 369 0 12 63	TEST DET 4 13 36 63 91 151 174 252 215 226 229 234 0 0 12 62	2 500 500 500 500 500 500 500 500 500 500	NET / 95% 3 16 23 44 61 63 62 7 69 6 9 7 7 92	MARTIN (2) 0 MISS 0 0 0 0 0 0 0 0 0 0 0 0 0	MCD
18 19 20 21 22 23 24 25 26 27 28 29 30 31	241 241 241 241 241 241 241 241 241 241	275 290 306 322 336 347 362 381 393 408 0 0 466 979	72 87 102 129 159 177 180 183 186 189 0 195 285	69 82 96 115 142 157 160 163 166 169 0 0 173 256	ତି ବିଷ୍ଟି ବିଷ୍ଟ ବିଷ୍ଟି ବିଷ୍ଟ ବିଷ୍ଟି ବିଷ୍ଟି ବିଷ୍ଟ ବି	999934444444 664446999999999999999999999	4 16 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	17 22 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0

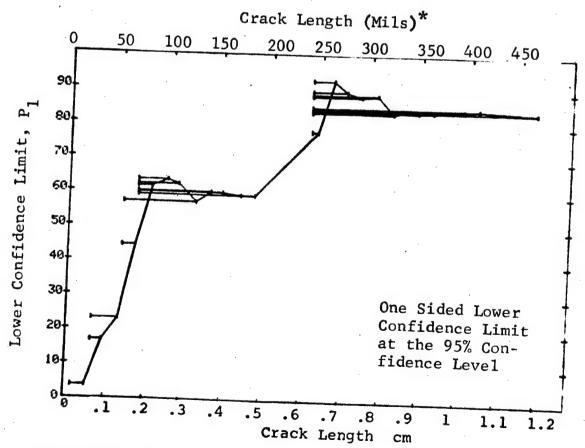
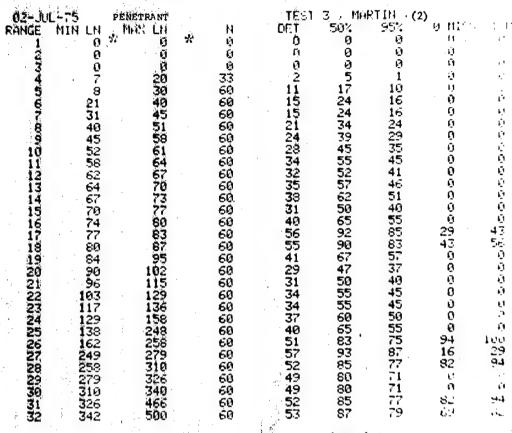


Figure D-2 (Continued)



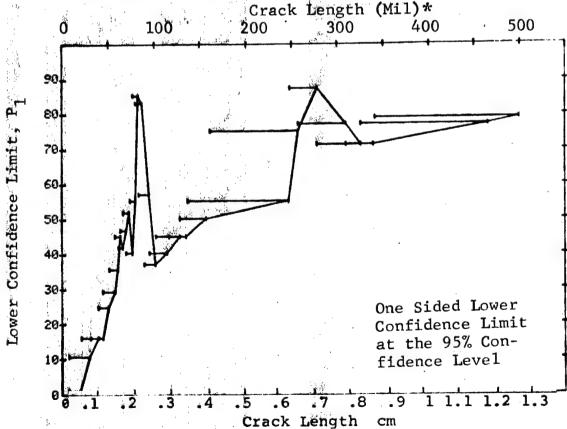
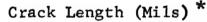


Figure D-2 (Concluded)

1 2 3 4 5 6 7 8 9 9 9 11 12 13 14 15 16 16 16 16 16 16 16 16 16 16 16 16 16	MIN LN * 25 38 54 68 83 98 115 129 146 158 182 0 0 241	DDY CURRENT MAX LU- 36 52 67 82 97 108 126 141 153 171 185 0 0 247 262	H 39 48 72 120 133 87 33 42 18 99 00 12	TEST DET 10 40 85 109 78 39 39 18 97 0	50% 11 20 54 70 81 88 98 98 99 96 92 71 0 0 86	07 CURPI 957: 55 11 45 63 75 82 92 78 82 84 71 45 9	0 MI (a) 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	(3) RANGE (4) SH (5) SH	•
17 18 19 20 21 22 23 24 25 26 27 28 29 31 32	248 268 279 295 310 323 338 362 381 393 408 0 459 475	262 275 290 306 322 336 347 362 381 393 408 0 9	51 94 157 238 133 133 133 133 133 133 133 133 133 1	51 9 13 14 26 30 16 3 3 3 3 3 9 0 6 85	98 92 89 93 93 95 79 79 79 0 0 89 93	24 71 70 72 83 99 68 36 36 36 36 0 0 69 88	6 6 6 6 6 6 6 6 6 6 6 6 7 8 8 8 8 8 8 8	00004600000000000000000000000000000000	



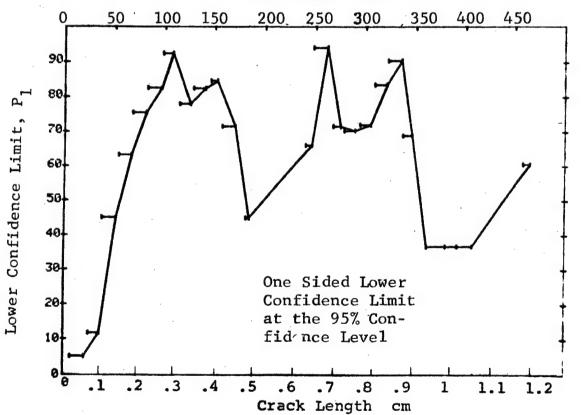


Figure D-3 Probability of Detection for 2219-T87 Al Using Eddy Current. Fatigue Cracks in Flat Plates.

Lab. Env. p-9

(b) Optimum Probability Method of Data Cumulation

07-JU RANGE 12345678901112314516789222222222222222222222222222222222222	MIN 25548398889999999999999999999999999999999	DY CURRENT MAX LH * 21 * 26 52 67 82 97 108 126 141 153 171 185 0 0 247 2275 290 322 336 3367 362 381 393 408 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	N 39 48 720 1337 72 1131 149 0 0 161 167 173 176 0 0 173 176 177 177 177 177 177 177 177 177 177	TEST DET 10 40 84 109 78 39 69 107 125 134 141 0 0 152 163 163 165 168 171 0 0 0	50 ପ୍ରତ୍ତର ପର୍ବତର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପର୍ବତର ପର ପର୍ବତର ପର ପର୍ବତର ପର ପର ପର୍ବତର ପର ପର ପର ପର୍ବତର ପର	95/15/25/23/99/11/90/00/00/45/33/34/33/34/40/00/99/99/99/99/99/99/99/99/99/99/99/99	ଞ୍ଚଳ ବ୍ୟବର ଅବର ଅବର ଅବର ଅବର ଅବର ଅବର ଅବର ଅବର ଅବର ଅ	MCD 1 MISS 5 0 0 0 0 7 7 0 0 0 0 0 0 0 0 0 0 0 0 0	
29	0	0	0	0	0	0	e)	0 0 0	

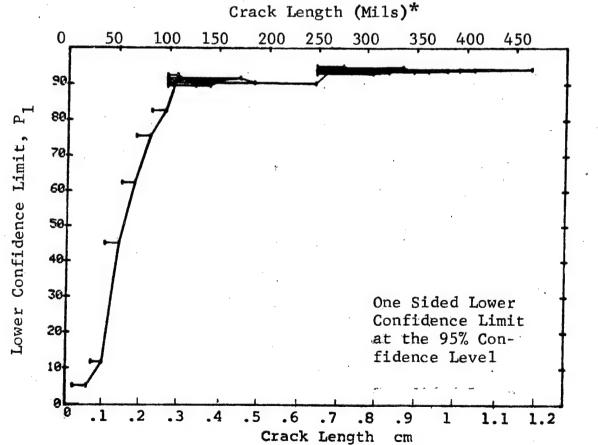


Figure D-3 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

02-JUL-75 NAX LN N 1 0 0 0 2 0 0 0 3 0 0 0 4 7 20 33 5 8 30 60 6 21 40 60 7 31 45 60 8 40 51 60 9 45 59 60 10 52 61 60 11 58 64 60 12 62 67 60 13 64 70 60 14 67 73 60 15 78 77 60 16 74 80 60 17 77 83 60 18 80 87 60 19 83 95 60 20 87 102 60 21 95 115 60 22 102 12	TEST 3 FEDDY CURRENT THAT (6) DET 580: 95% 9 MICO 1 179 0 0 0 0 9 0 0 0 9 4 11 4 9 9 11 17 10 0 0 9 12 14 9 9 22 35 26 9 9 33 54 43 9 0 0 36 59 48 9 0 0 37 67 9 69 9 9 44 77 77 67 9 9 9 45 47 77 67 9 9 9 56 92 85 29 47 57 93 87 16 29 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16 57 93 87 16 58 95 89 1 16
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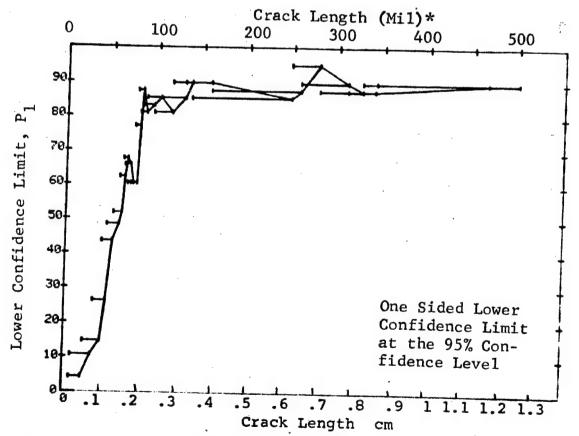


Figure D-3 (Concluded)

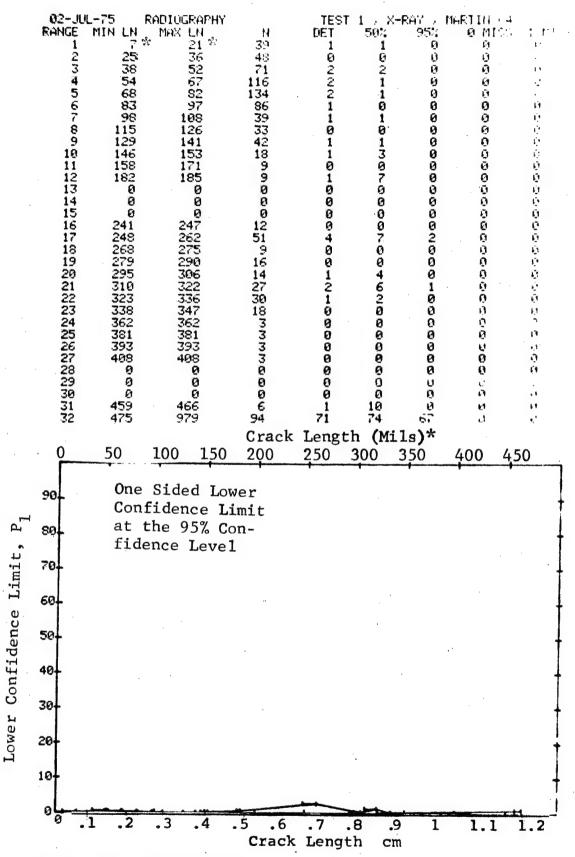


Figure D-4 Probability of Detection for 2219-T87 Al Using X-ray. Fatigue Cracks in Flat Plates. Lab. Env.

(b) Optimum Probability Method of Data Cumulation

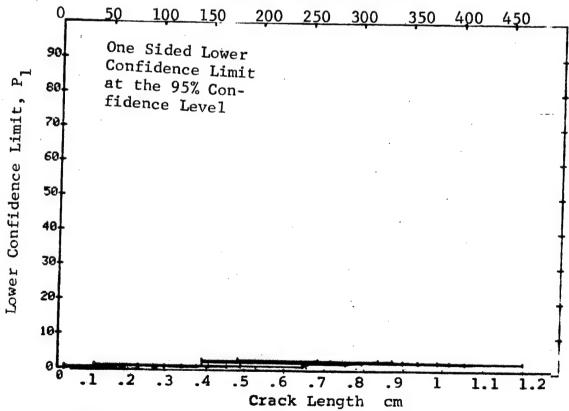


Figure D-4 (Continued)

.(c) Overlapping Sixty Point Method of Data Cumulation

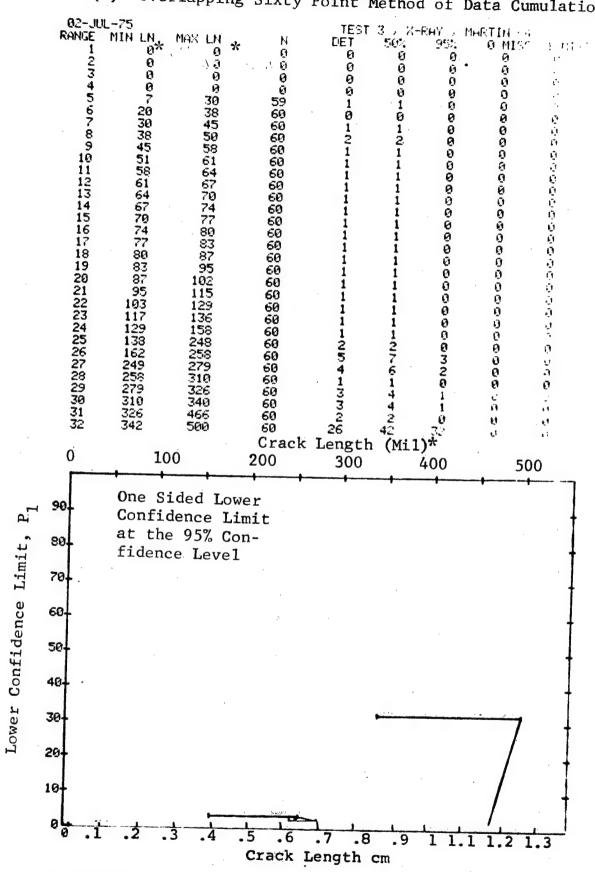


Figure D-4 (Concluded)

03-JUL RANGE 1 2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17	-75 NIN LN 7 25 38 468 83 91 129 146 158 162 0 0 0 1 248	ULTRASONIC MAX LN	BY LENGTH N 39 72 120 132 87 333 42 18 9 0 0 12 51	TEST DET 12 35 58 101 119 76 33 27 37 16 7 9 0	1, MAR 50% 271 79 89 86 87 98 85 71 90 86 98	TIH -5 -, 95% 180 71 777 84 79 717 676 45 71 00 66 94	9 00 00 00 00 00 00 00 00 00 00 00 00 00	RANGE 1 MISS 6 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
18 19 20 21 22 23 24 25 26 27 28 29 30 31	268 279 295 310 323 338 362 381 393 408 0 0 459 475	275 290 306 322 336 347 362 381 393 408 0 0 466 979	9 15 15 27 29 18 33 33 90 90	9 14 15 25 26 16 33 33 33 9 9 9 83	92 89 95 90 94 85 79 79 79 0 0 89 91	71 72 81 78 84 68 36 36 36 36 0 0 60 85	00 144 347 170 00 00 00 00 00 00 00 00 00 00 00 00 0	00043000000000000000000000000000000000

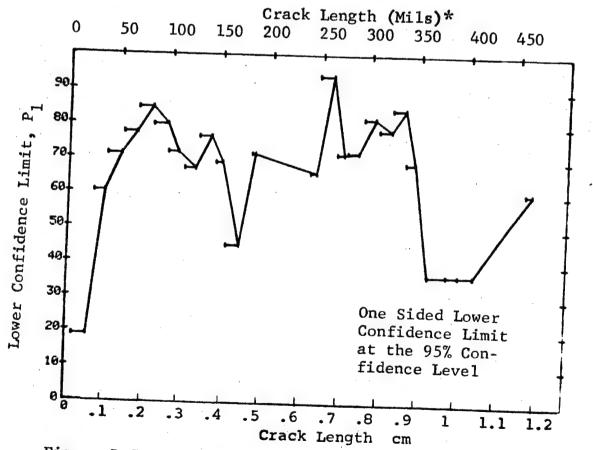


Figure D-5 Probability of Detection for 2219-T87 Al Using Ultrasonic Surface Waves. Etched Fatigue Cracks in Flat Plates. Lab. Env.

(b) Optimum Probability Method of Data Cumulation

93-JUI RANGE 123456789101123134151679	MIN LN ** 25:38 68 68 68 68 68 68 68 68 68 68 68 68 68	TRASONIC MAX LN & 21 & 36 & 36 & 36 & 36 & 36 & 36 & 36 & 3	BY LENGTH N 39 72 192 132 219 258 291 333 351 360 369 0 0 381 51	TEST DET 12 35 58 159 119 195 228 255 292 308 315 324 0 0 335 51	2, Mik 59:000000000000000000000000000000000000	TIN (5). 95% 18 60 71 77 84 84 83 84 84 86 80 84 84	0 MISS 00000000000000000000000000000000000	MCD 1 MI (S) 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
18 19 20 21 22 23 24 25 26 27 28 29 30 31	248 248 248 248 248 182 182 182 182 182 182 182 182	275 290 306 322 336 347 362 381 393 408 0 0 466 979	60 75 90 117 146 185 188 191 194 197 0 0 203 293	60 74 89 114 142 178 181 184 187 190 0 0 196 279	ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ ତ	95 93 94 93 93 93 93 93 93 93 93 93 93	ପ୍ରତ୍ର ପ୍ରତ୍	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0

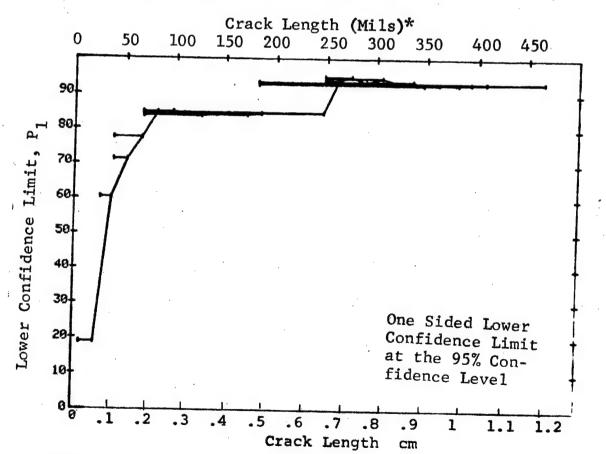
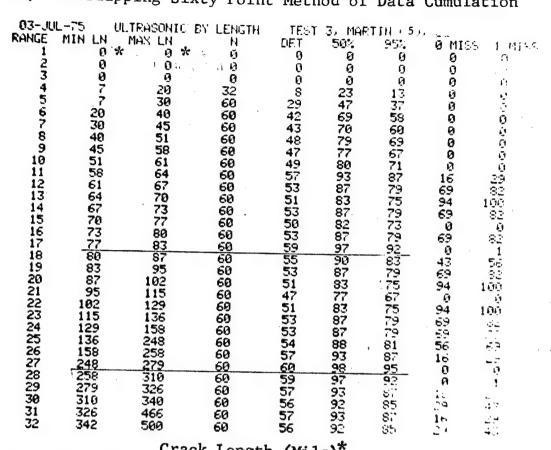


Figure D-5 (Continued)



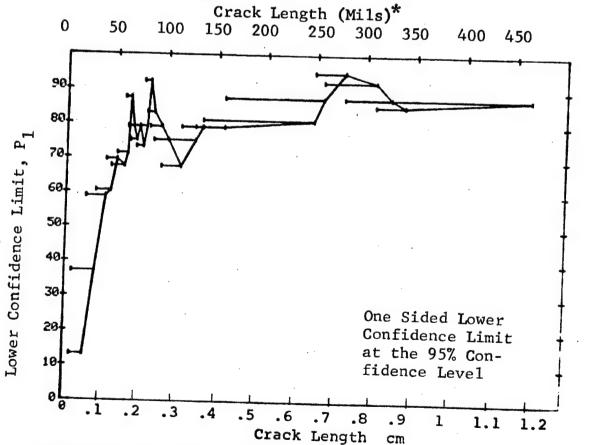


Figure D-5 (Concluded)

02-JU		PENETRANT		TEST			MHF71H 6	RANGE
RANGE	MIN LH	MAX EN	N	DET	50	95%	ម មាន	* * * * * * * * * * * * * * * * * * *
1		k 21 %	39	10	24	14	អ	' '
2	25/	- 36)48	31	63	51	·fi	
3	38	52	72	57	78	63	Ø	1
4	54	67	119	98	81	75	Ģ	-1
- 5	68	82	132	113	85	79	_6	- 11
5 6 7	83	97	119 132 87	79	90	84	5 5	P. 1
7	98 115	108	39 33	39 27	98	92	Ü	
\$ 9 10	115	126	33	27	.79	67	ų.	3
. 9	129	141	42 18 9 9	41	96	89	4	1.4
10	146	153	18	16	85	<u>68</u>	õ	4.1
11	158 182	171	9	9 9 0	92	71	9	· 9
12	182	185	9	9	92	71	ō.	ŭ
13	0	0	Ø	Ø	0	9	0 0	. U
14	0	Ø	Ø	ø	9	0	Ñ	in .
15	0	.0	Ø	Ø	0	0 66	<u>o</u>	ñ
16	241	247	0 0 12 51	11	86	55	<u> </u>	U .
17	248 268	262	51	51	98	94	_ @	8
18	268	275	15 15	9	92	71	0	19
19	279	290	15	15 15	95	81	14	21
20	295	306	15	15	95	81	14	41 45
21 22 23	310	322 336 347	27	25	90	78	34	4.4
22	323 338	336	30	30	97 85	. 90	0	
23	3 38	347	18	16	85	6 8	્રું ફ	Ľ.
24 25	362	362	3	33230	79	36	Q	9
25	381	381	3	3	79	36	Ø.	P.
26	393	393	3	2	50	13	ų į	12
27	408	408	3	3	79	36	ø	o .
23	0	Ð	0	0	9	0	P	n
29	Ø	Ø	0	Ø	0	Ü	_ <u>ç</u>	•
30	Ø	Ø	18 3 3 3 9 9	0	9	0	B	• 1
31	459	466	б	6	89	60	U	41
32	475	979	.91	91	99	96	1:	ઇ

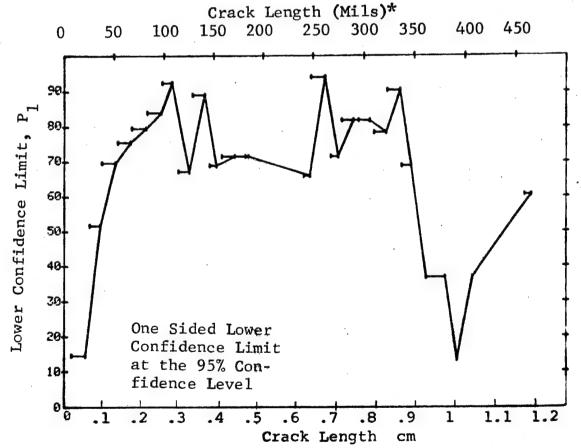


Figure D-6 Probability of Detection for 2219-T87 Al Using Liquid Penetrant. Etched Fatigue Cracks in Flat Plates. Lab. Env.

(b) Optimum Probability Method of Date Cumulation

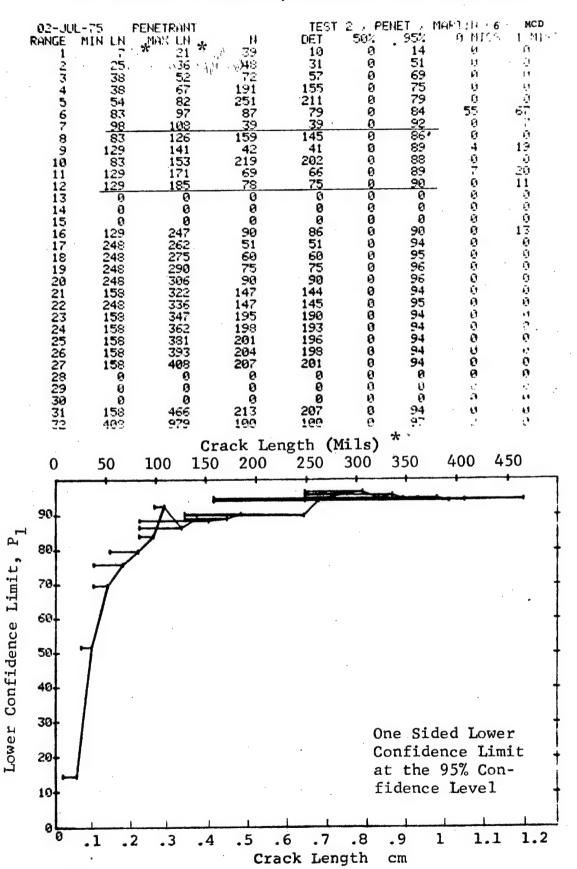


Figure D-6 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

1 23 4 5 6 7 8 9 9 9 9 9 11 12 13 14 15 16 7 18 18 18 18 18 18 18 18 18 18 18 18 18	-75 LN * 0 0 7 7 20 340 45 58 64 77 77 77 80	99 29 39 49 45 59 66 77 77 83 87	N 9 0 9 2 9 6 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9	TEST 0 0 0 7 24 8 2 7 4 8 8 8 3 8 4 5 5 8 5 7 5 7 5 7	3 PE 50% 0 0 0 20 20 20 20 20 20 20 20 20 20 20	NET - 1959 0 0 0 0 0 0 1 2 5 5 6 6 6 9 9 3 9 9 6 6 7 7 6 6 6 9 9 3 9 9 9 9 9 9 9 9 9 9 9 9 9 9	MATTH (6) 9 MI	
19 20 21 22 23 24 25 26 27 28 29 30 31 32	84 90 96 103 117 129 138 162 249 258 279 310 326 342	95 102 115 129 136 158 248 259 310 326 346 466 500	69 69 69 69 69 69 69 69 69 69	556 555 557 56 57 57 59 60 56 56 57 57 59	79978823378855337 9998899999999999	55391557722555555 66878889955555599	189173916539161618181919191919181818181818181818181	17-16-21-0-15-0-15-0-15-0-15-0-15-0-15-0-15-0

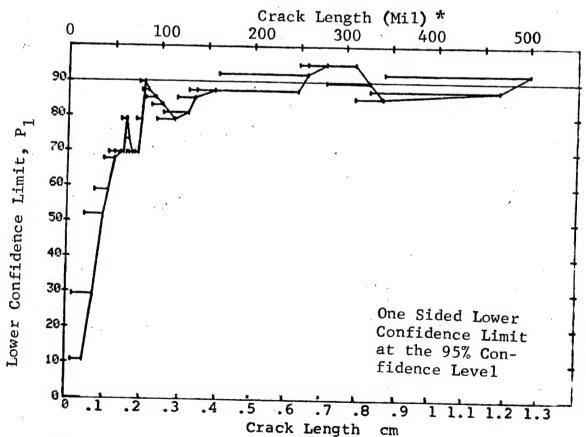


Figure D-6 (Concluded)

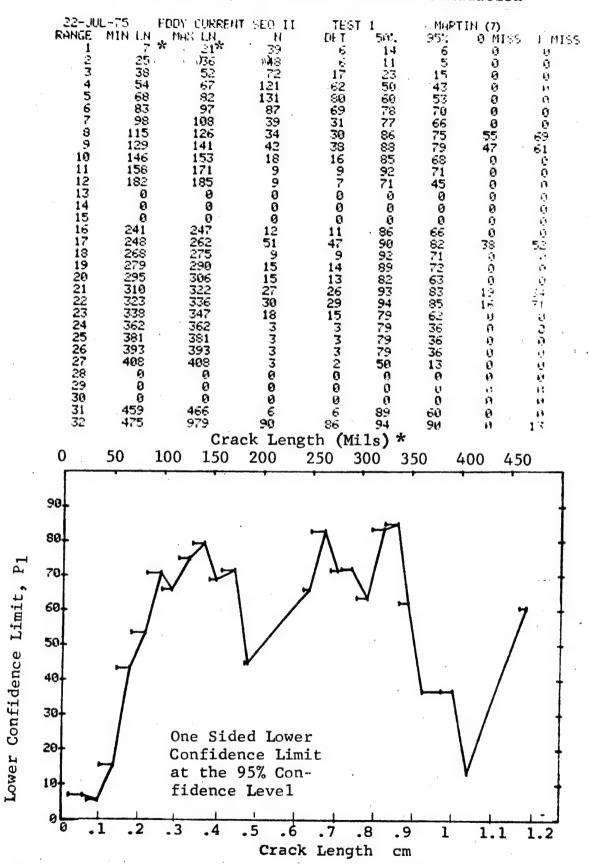
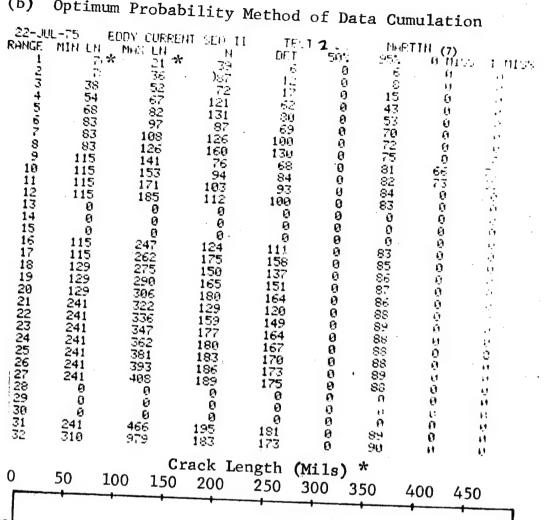


Figure D-7 Probability of Detection for 2219-T87 Al Using Eddy Current. Etched Fatigue Cracks in Flat Plates. Lab. Env.

Optimum Probability Method of Data Cumulation (b)



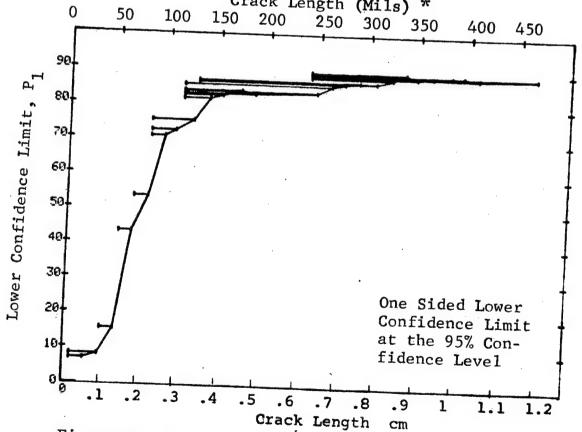


Figure D-7 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

22-JUL-75 EDDY CURRENT SEO II RANGE MIN LN	TEST 3 50% 50% 50% 50% 50% 50% 50% 50% 50% 50%	MARTIN (7) 95% 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9	1. 9.9.9.9.9.9.9.9.9.9.9.9.9.9.9.9.9.9.9
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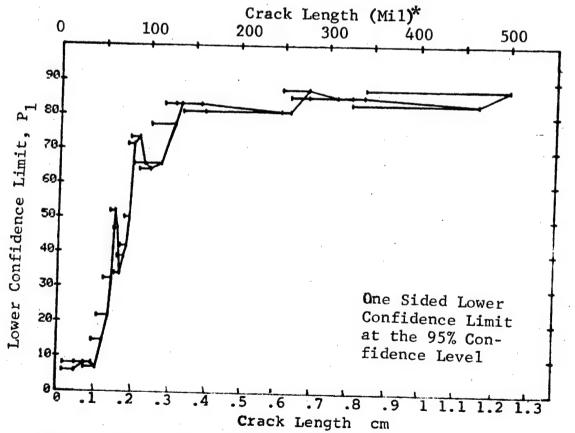


Figure D-7 (Concluded)

1234567890 1123456789 111231456789 1222222222222222222222222222222222222	MIN 23848385596838515968898595968898595968899595933388213889689959583338838968	ADIOGRAPHY MAX 236 522 978 126 127 127 185 185 185 185 185 185 185 185 185 185	N99920117933428990002195570833330000	TEST DFT 1 8 14 16 14 55 15 7 11 11 6 3 0 0 0 0 2 0 9 3 2 3 3 5 1 2 3 3 0 0 0	5415435170580800046226644900999009	-RAY 9 867 12782555 1594 9 8 8 8 7 9 1 2 3 5 2 1 1 3 3 4 9 8 8 8 7 7 9 7 6 5 3 6 6 2 1 3 3 6 6 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8	MARTIN : 8 0 MISS 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	
30 31 32	9 459 475	9 466 979	0 0 6 90	0 0 6 88	0			ii ii

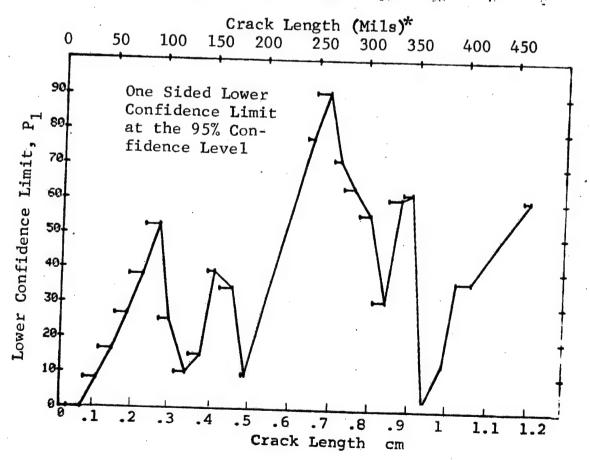


Figure D-8 Probability of Detection for 2219-T87 Al Using X-ray. Etched Fatigue Cracks in Flat Plates.
Lab. Env. D-24

(b) Optimum Probability Method of Data Cumulation

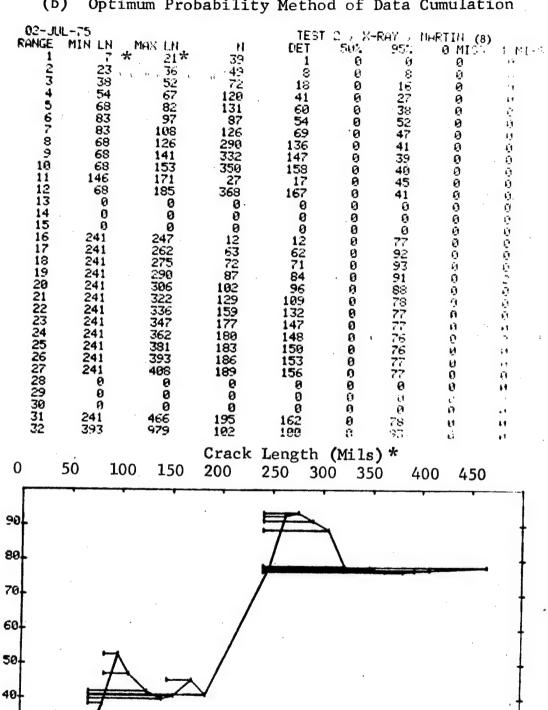


Figure D-8 (Continued)

. 3

.5

Lower Confidence Limit,

30

20-

10-

Crack Length

.8

.9

CM

One Sided Lower

fidence Level

1

1.1

1.2

Confidence Limit at the 95% Con-

(c) Overlapping Sixty Point Method of Data Cumulation

02-JUL-75 RANGE MIN LN MAX LH % 1 0/k 0 2 0 0 3 0 0 4 7 20 5 7 30 6 20 40 7 30 45 8 40 50 9 45 50 9 45 61 11 58 64 12 61 67 13 64 70 14 67 73 15 70 77 16 73 80 17 77 83 17 77 83 17 77 83 19 83 95 20 87 102 21 95 115 22 102 129 23 115 136 24 129 158 25 136 248 27 248 259 28 258 310 31 326 366 32 340 500	N ପ ସ ପ 2000 ପ ଅଟେ କେ	TEST 3 , % DET 50% 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	M 1442175007403407485235893 111232232356533407485235893 468	146T1MC) 146T1MC) 146T1MC) 146T1MC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 146TC) 1	
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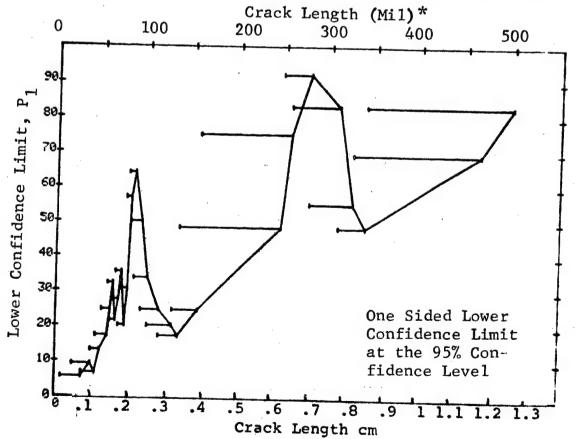


Figure D-8 (Concluded)

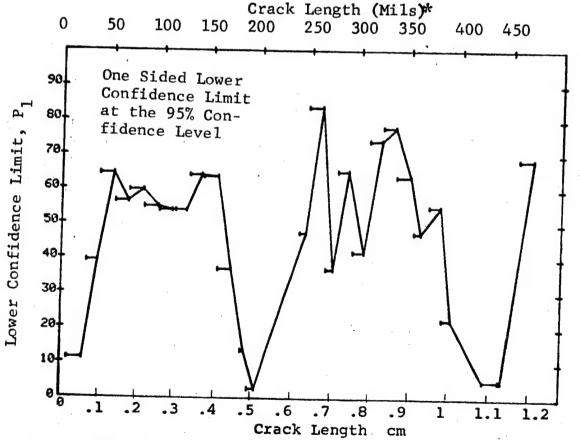


Figure D-9 Probability of Detection for 2219-T87 Al Using Ultrasonic Shear Wave. Etched-Fatigue Cracks in Flat Plates Measured by Operator O. Lab. Env. D-27

(b) Optimum Probability Method of Data Cumulation

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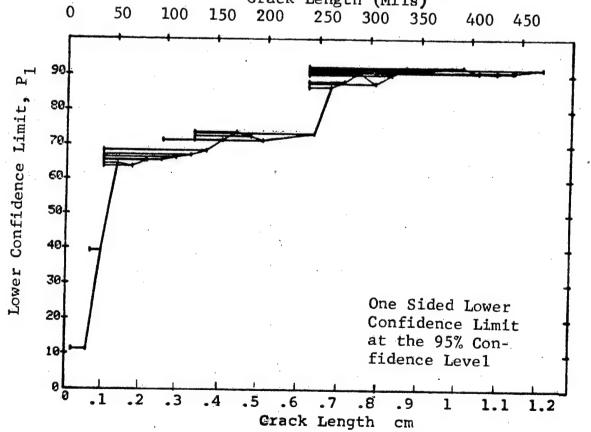


Figure D-9 (Continued)

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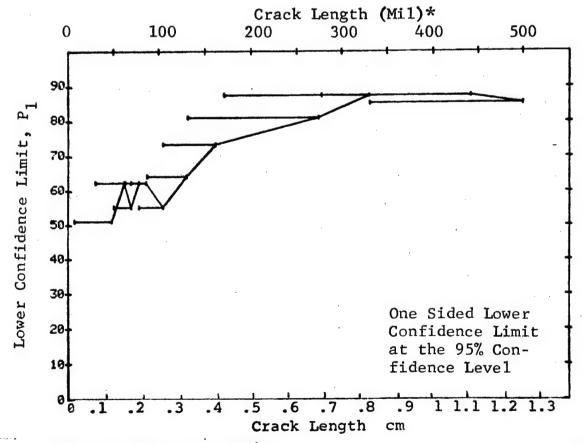


Figure D-9 (Concluded)

03-JUL-75 ULTRASONIC RANGE MIN LN MAX LH N 1 7 22 13 2 25 1666 13 3 38 52 23 4 54 67 46 5 68 82 53 6 83 97 39 7 98 111 17 8 115 126 17 9 129 141 19 10 143 157 15 11 158 171 3 12 182 185 3 13 190 197 2 14 207 207 1 15 0 0 0 0 16 241 247 4 17 248 262 17 18 268 275 3 19 279 290 6 20 295 306 6 21 310 322 10 22 323 338 352 11 24 356 362 4 25 370 381 5 26 384 393 2 27 408 408 1 28 426 426 1 29 442 442 3 30 444 450 2 31 458 472 7 32 474 979 59	TEST 1
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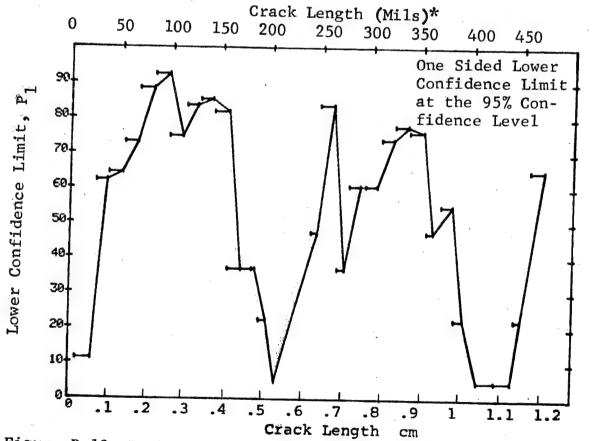


Figure D-10 Probability of Detection for 2219-T87 Al Using Ultrasonic Shear Wave. Etched-Fatigue Cracks in Flat Plates Measured by Operator P. Lab. Env.

(b) Optimum Probability Method of Data Cumulation

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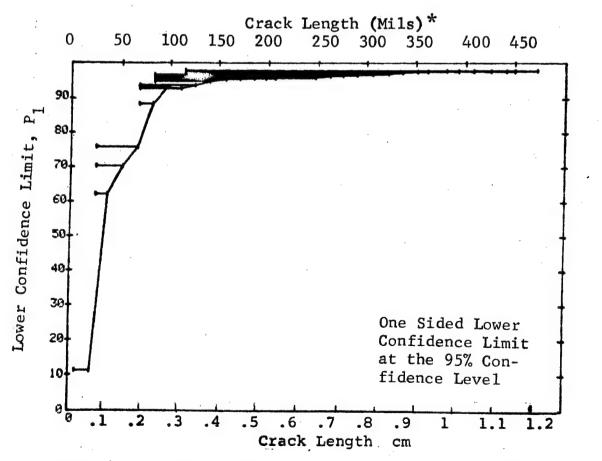


Figure D-10 (Continued):

for

(c) Overlapping Sixty Point Method of Data Cumulation

03-JUL-75 RANGE MIN LN 1	ULTRASONIC MAX LN 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	DET 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	95.000000000000000000000000000000000000	(10) MISS 1 MISS 1 P P P P P P P P P P P P P P P P P P
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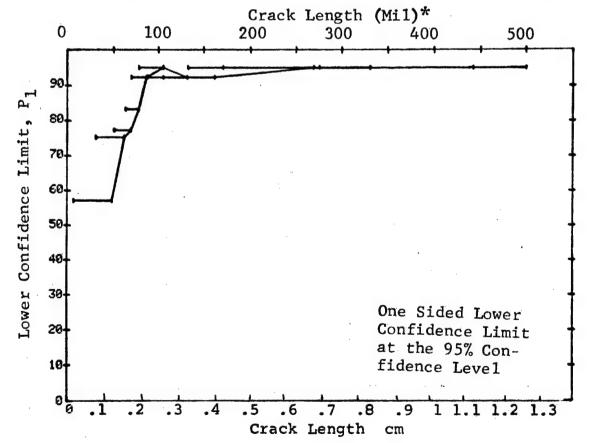


Figure D-10 (Concluded)

(b) Optimum Probability Method of Data Cumulation

03-JUL-75	DF 3 987 613 164 10 23 20 00 47 37 60 10 4 4 2 1 1 1 1 7 55	20 472 79 85 73 90 63 79 90 84 89 99 99 99 99 99 99 99 99 99 99 99 99	1015 MISS 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	(11) MISS 1 U O O U 9 O O 4 O O O O O O O O O O O O O O O O
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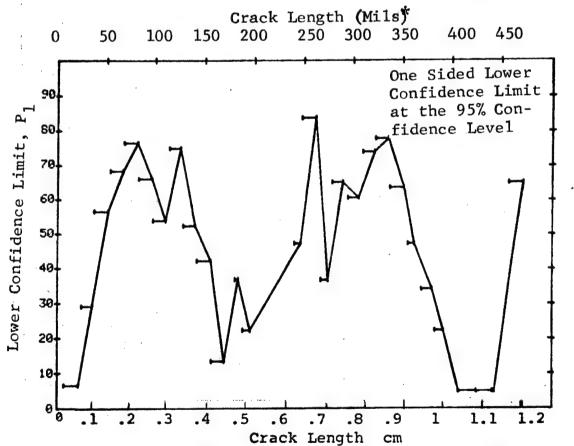


Figure D-11 Probability of Detection for 2219-T87 Al Using Ultrasonic Shear Waves. Etched Fatigue Cracks in Flat Plates Measured by Operator Q. Lab. Env.

(b) Optimum Probability Method of Data Cumulation

03-JUL-75 RANGE MIN LN 1 25 33 38 4 56 6 54 7 68 9 54 10 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54 11 54	* 22 * 56 52 67	N 138 244 753 138 1556 1296 2124 9 9 8 8 8 8 9 9 9 7 6 8 8 7 8 8 9 9 9 7 156	TEST UET 39:18 18:18:18:18:18:18:18:18:18:18:18:18:18:1	ି ୍ଟ୍ରପ୍ରତ୍ତର ପ୍ରତ୍ତର ପର ପ୍ରତ୍ତର ପରତ ପରତ୍ର ପର୍ତ ପର ପ୍ରତ୍ତର ପର ପ୍ରତ୍ତର ପ୍ରତ୍ତର ପରତ୍ତର ପର ପରତ୍ତର ପରତ୍ତର ପର ପରତ୍ତର ପର ପରତ ପରତ୍ତର ପରତ ପରତ୍ତର ପରତ ପରତ ପରତ ପରତ ପରତ ପରତ ପରତ ପରତ ପରତ ପ	CVNELL: 69686667777777765660969934534222333332	90°99 M6000000000000000000000000000000000000	(11) 1 MISS 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
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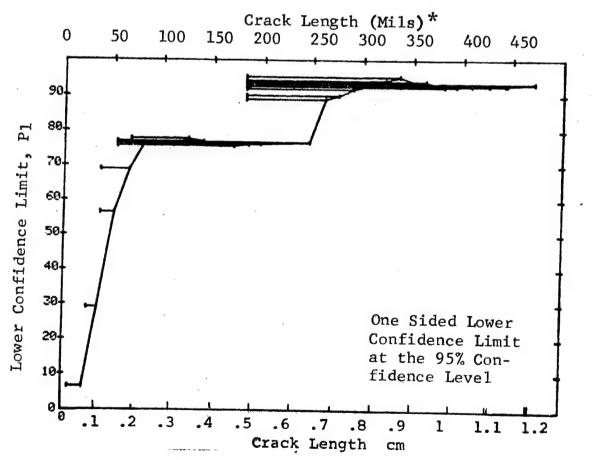


Figure D-11 (Continued)

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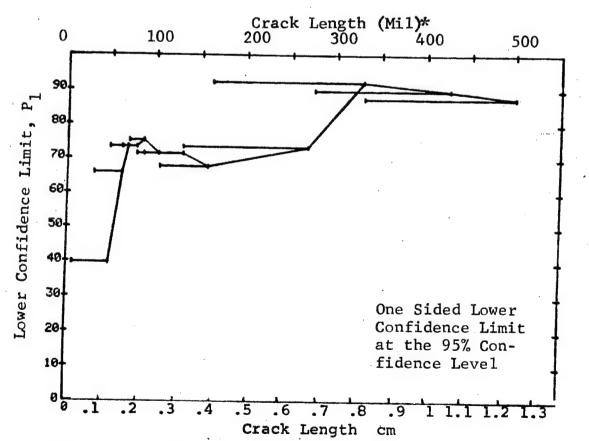


Figure D-11 (Concluded)

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24-JUL-75 ULTRASONIC RANGE MIN LN MAX LN 7 22 x 25 36 38 52 4 54 67 55 68 82 66 83 97 77 98 111 8 115 126 9 129 141 110 143 157 111 158 171 112 182 185 171 112 182 185 171 112 182 185 171 112 182 185 171 112 182 185 171 112 182 185 171 112 182 185 171 12 182 185 171 112 182 185 171 112 182 185 171 112 182 185 171 12 182 185 171 12 182 185 171 12 182 185 171 12 182 185 171 12 182 185 171 17 248 262 187 263 275 19 279 290 295 306 21 310 322 22 323 336 352 24 356 362 370 381 322 24 356 362 370 381 322 24 356 362 370 381 393 27 408 408 408 408 408 408 408 408 408 408	N382739777953320047376021145211111895	TEST DE 63 82 47 75 68 28 4 4 2 1 1 1 1 8 5 5 5 5 5 6 8 2 8 4 4 2 1 1 1 1 8 5 5 5 6 8 2 8 4 4 2 1 1 1 1 8 5 5 6 8 2 8 4 4 2 1 1 1 1 8 5 5 6 8 2 8 4 4 2 1 1 1 1 1 8 5 5 6 8 2 8 4 4 2 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	1, 545987349160900469909447488000012 5955555555555555555555555555555555555	POI 92563884757636200736509773742555585 926778677763620073650977374235585	CKWELL 55 CKWE MI 55 56229978888828978888828978888888888888888	1 M

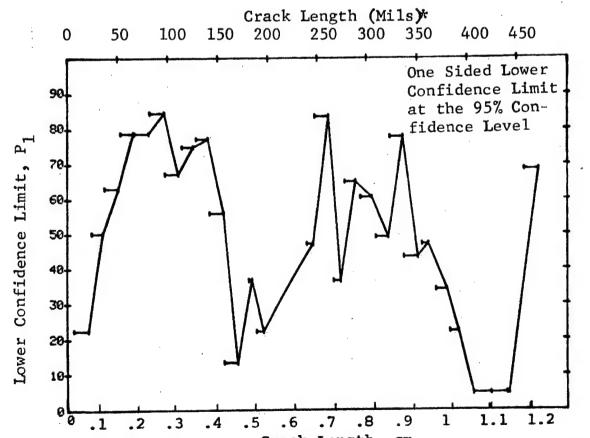
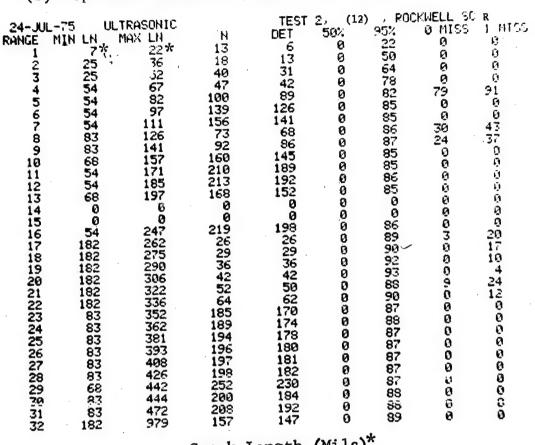


Figure D-12 Probability of Detection for 2219-T87 Al Using
Ultrasonic Shear Waves. Etched Fatigue Cracks in
Flat Plates Measured by Operator R. Lab. Env.
D-36

(b) Optimum Probability Method of Data Cumulation



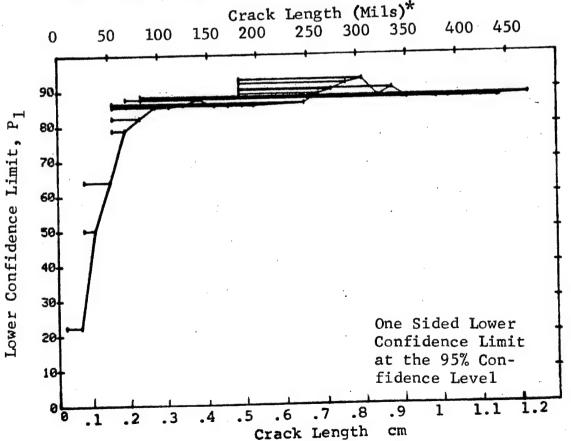
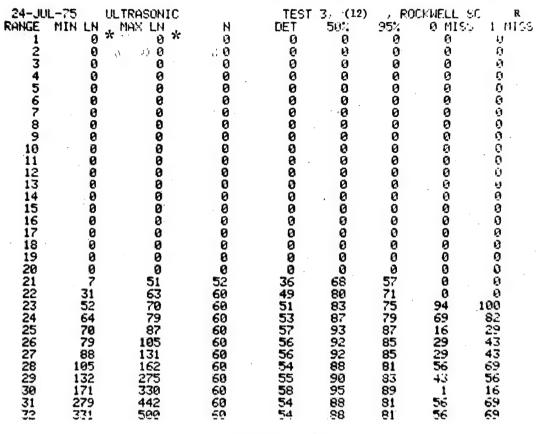


Figure D-12 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation



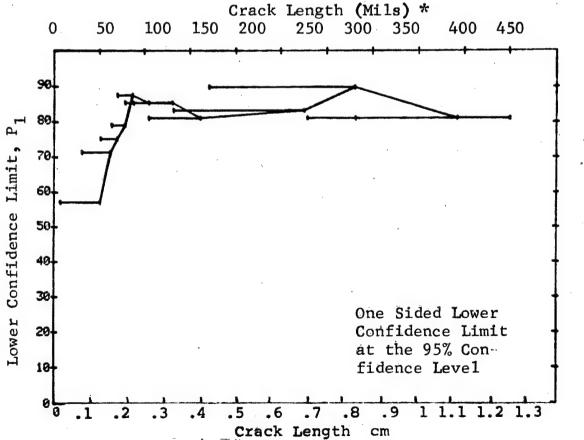


Figure D-12 (Concluded)

24-JUL-75 ULTRASONIC RANGE MIN LN MAX LN 22 25 36 36 38 52 4 54 67 5 68 82 66 83 97 77 98 111 8 115 126 9 129 141 110 1143 1157 111 158 171 112 182 185 13 190 197 14 0 0 0 15 0 0 0 16 241 247 17 248 262 18 268 275 19 279 290 20 295 306 21 310 322 22 323 336 23 338 352 24 356 362 25 370 381 26 384 393 27 408 408 28 426 426 29 442 30 444 444 31 458 472 372 474	N38365977953320047376021452111188	TEST DET 43178055671681433220031736601001442111118959	1, 507 507 67 67 83 99 99 99 99 99 99 99 99 99 99 99 99 99	95% 11 50 40 61 33 57 72 66 62 0 0 44 33 67 0 44 32 5 5 5 5 5 5 68 67 56 37 44 22 5 5 5 5 5 68	SUBSTANCE	NISS 100000000000000000000000000000000000
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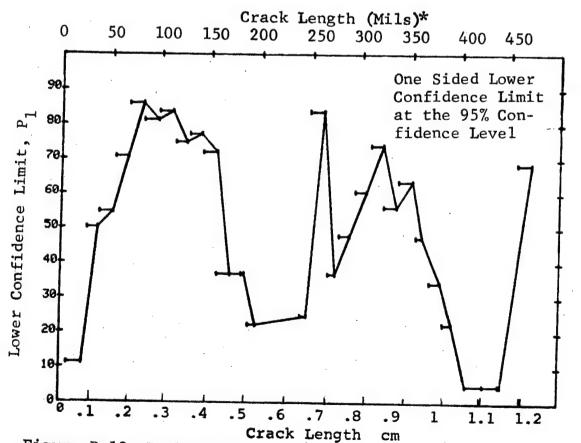
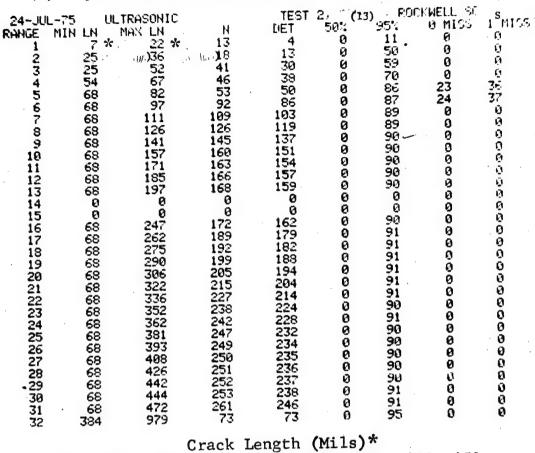


Figure D-13 Probability of Detection for 2219-T87 Al Using Ultrasonic Shear Waves. Etched Fatigue Cracks in Flat Plates Measured by Operator S. Lab. Env. D-39



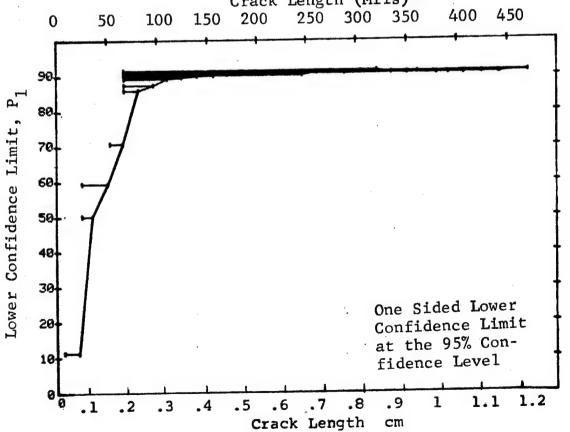
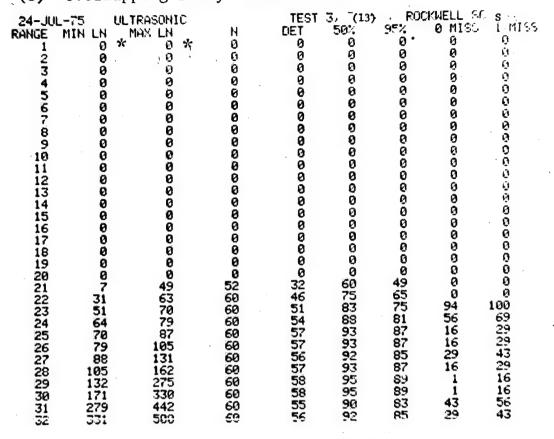


Figure D-13 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation



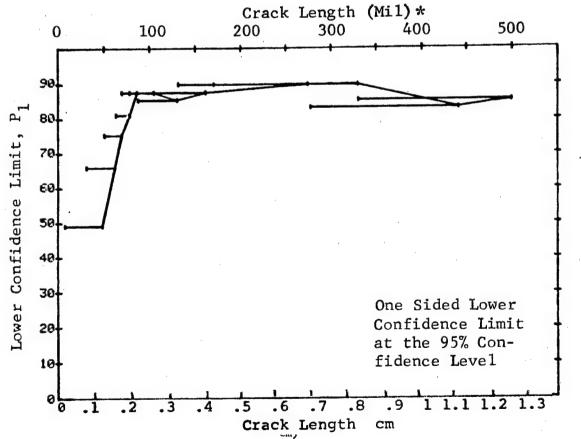


Figure D-13 (Concluded)

RANGE MIN LN MAX LN N DE 1 7 * 32 * 13 2 25	TEST 1, (14) ROCKWELL SC H ET 50% 95% 0 MT 1 MISS 4 27 11 0 0 9 47 29 0 0 0 21 88 75 38 53 39 83 73 0 0 0 47 87 78 63 75 37 95 88 8 23 17 96 83 12 29 15 84 67 0 0 0 18 91 77 27 42 14 89 72 0 0 0 14 89 72 0 0 0 14 89 72 0 0 0 17 96 83 12 29 16 83 79 36 0 0 0 17 96 83 12 29 17 96 83 12 29 18 91 77 27 42 19 0 0 0 0 0 0 0 10 0 0 0 0 0 0 0 11 89 66 66 0 0 0 11 86 66 0 0 0 11 88 76 18 35 11 50 5 0 0 0 11 50 5 0 0 0 11 50 5 0 0 0 15 98 98 95	
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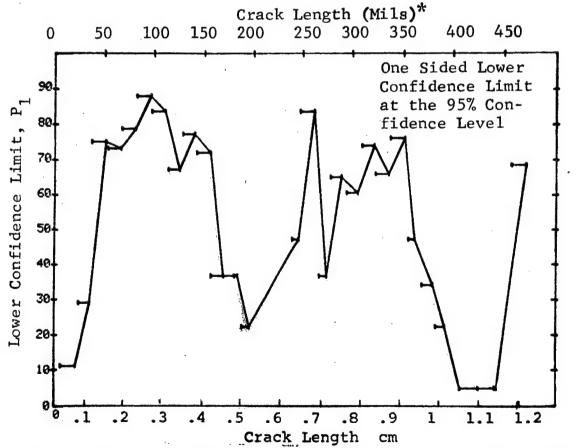


Figure D-14 Probability of Detection for 2219-T87 Al Using Liquid Penetrant. Etched Fatigue Cracks in Flat Plates Measured by Operator H. Lab. Env.

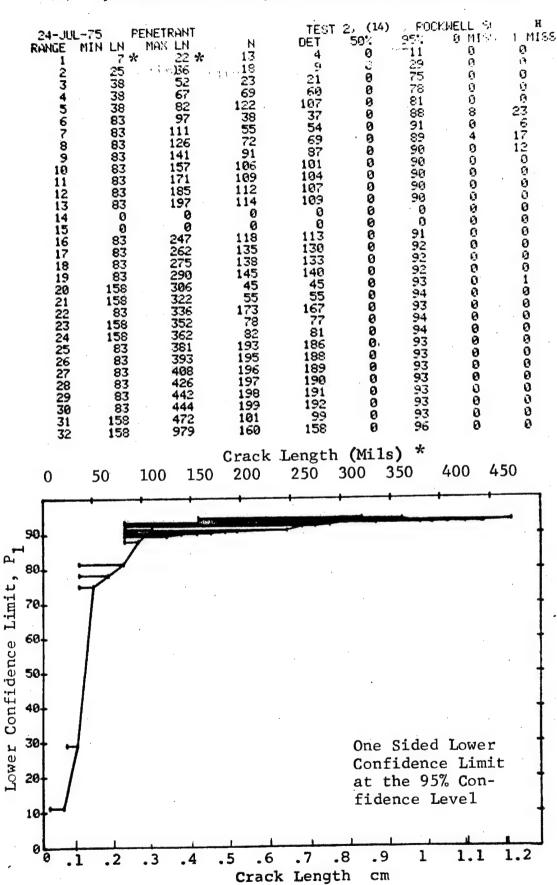


Figure D-14 (Continued)

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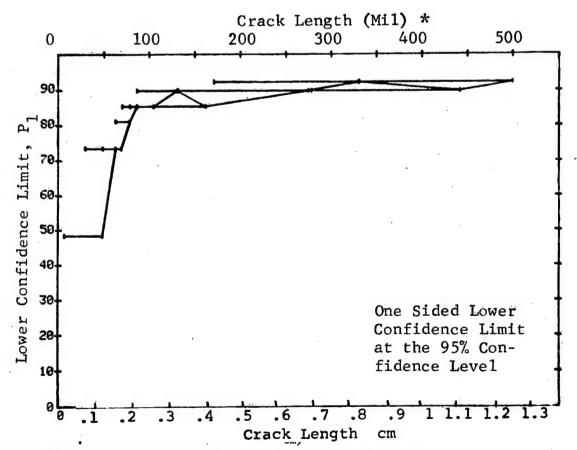
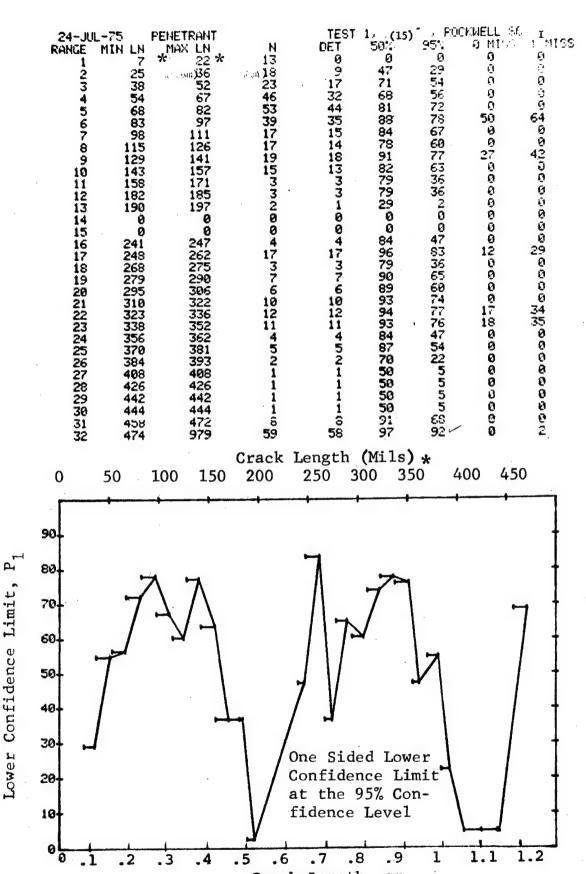
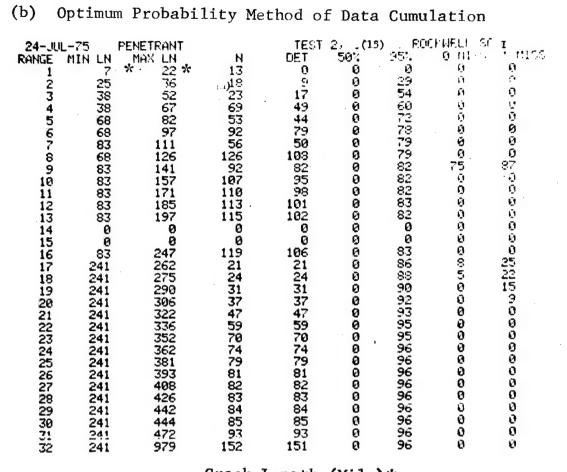


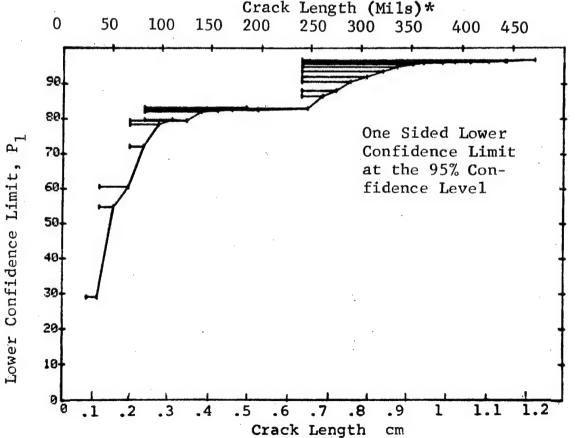
Figure D-14 (Concluded)

REPRODUCIBILITY OF THE



Crack Length cm
Figure D-15 Probability of Detection for 2219-T87 Al Using
Liquid Penetrant. Etched Fatigue Cracks in Flat
Plates Measured by Operator I. Lab. Env.





Figur^{PC} D-15 (Continued)

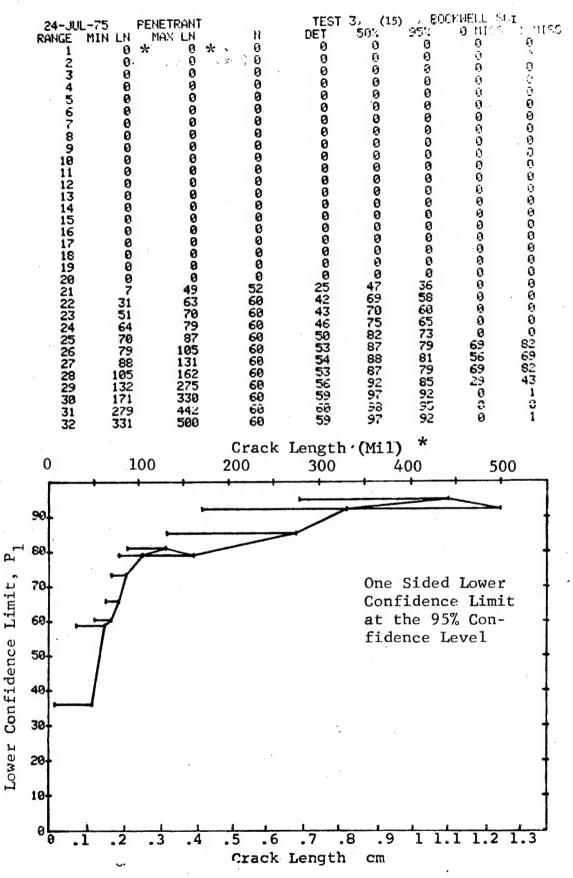


Figure D-15 (Concluded)

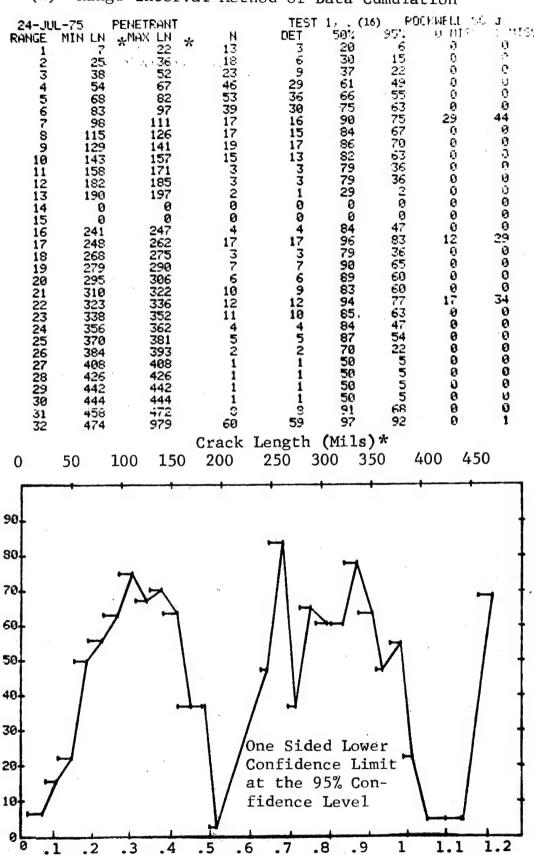


Figure D-16 Probability of Detection for 2219-T87 Al Using Liquid
Penetrant. Etched Fatigue Cracks in Flat Plates Measured
by Operator J D-48 Lab. Env.

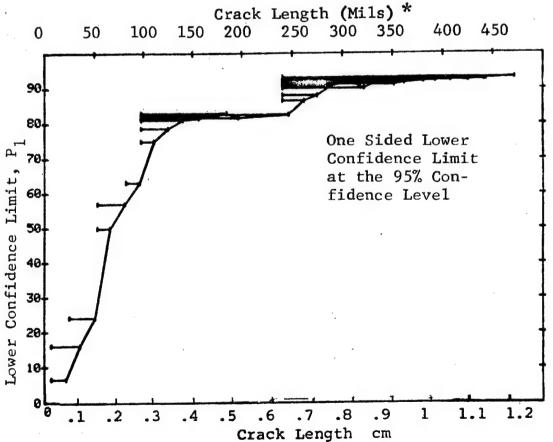
Crack Length

cm

Lower Confidence Limit,

(b) Optimum Probability Method of Data Cumulation

24-JUL-75 PENETRANT RANGE MIN LN MAX LN N DET 50% 95% 0 MILT 1 7 22% 13 3 0 6 0 0 2 7 16 81 9 0 16 0 0 3 25 52 41 15 0 24 0 0 4 54 67 46 29 0 49 0 0 5 54 82 99 65 0 57 0 0 6 83 97 39 30 0 63 0 0 6 83 97 39 30 0 63 0 0 6 83 97 39 30 0 63 0 0 7 98 111 17 16 0 75 0 0 8 98 126 34 31 0 78 0 0 9 98 141 53 48 0 81 50 63 9 98 141 53 48 0 81 50 63 10 98 157 68 61 0 81 61 71



Figurend-16 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

24-JUL-1 RANGE 1 23-45-67-89-101123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-145-167-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-189-01123-18	MIN LO 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	HETRANT * OOOOOOOOOOOOOOOOOOOOOOOOOOOOOOOOOOO	200808080808080808080808080808080808080	TET 00000000000000000000000000000000000	16) 16) 16) 17) 18) 18) 18) 18) 18) 18) 18) 18) 18) 18	95 95 95 95 95 95 95 96 96 96 96 96 96 96 97 97 97 97 97 97 97 97 97 97 97 97 97	CKMELT:	
30	171	330	68	58	95	89	1	16
31	279	442	68	58	95	89	1	16
32	331	500	68	58	95	89	1	16

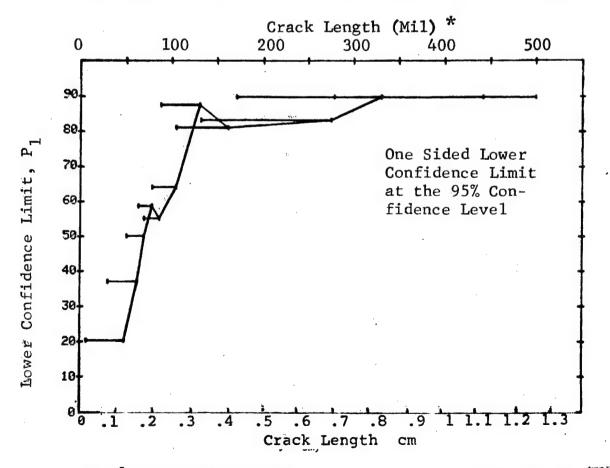


Figure D-16 (Concluded)

24-JUL-75 PENETRANT RANGE MIN LN NAX LN N 1 77	TEST 1, (17) POCKHELL SC K DET 50% 95% 0 MIC 1 MIC 1 3 20 6 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
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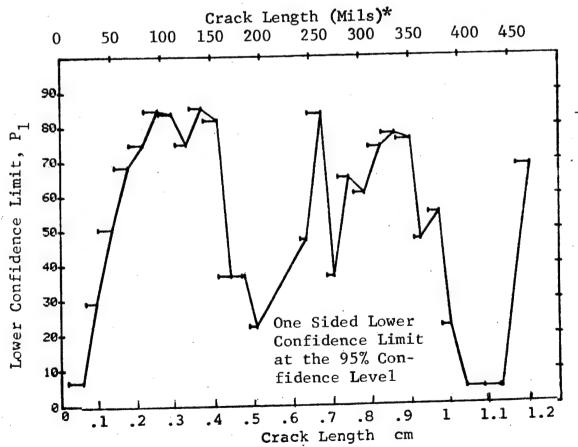
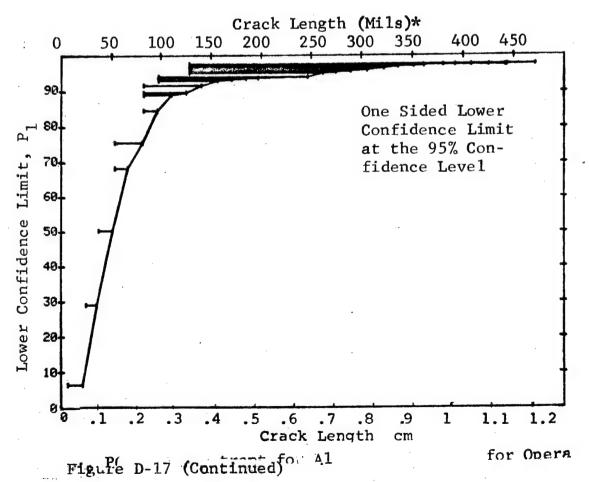


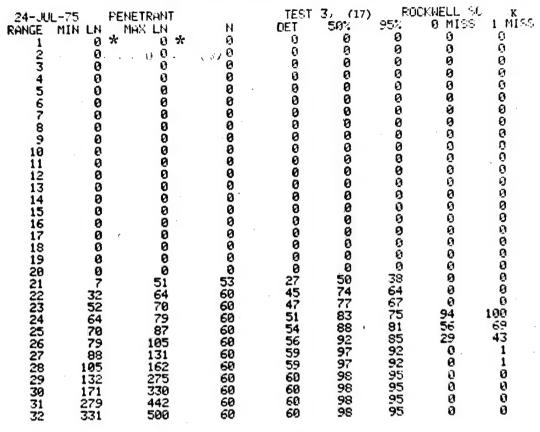
Figure D-17
Penetrant. Etched Fatigue Cracks in Flat Plates Measured by Operator K

Lab. Env.

D-51

24-JUL-75 PENETRANT RANGE MIN LN MAX LN H 1 7	TEST 2, (17) 95% 69 95% 69 16 37 83 754 00 99 33 37 54 00 99 99 99 99 99 99 99 99 99 99 99 99	CKWELL 90 k 1 mins 6 mins 7 mins 6 mi
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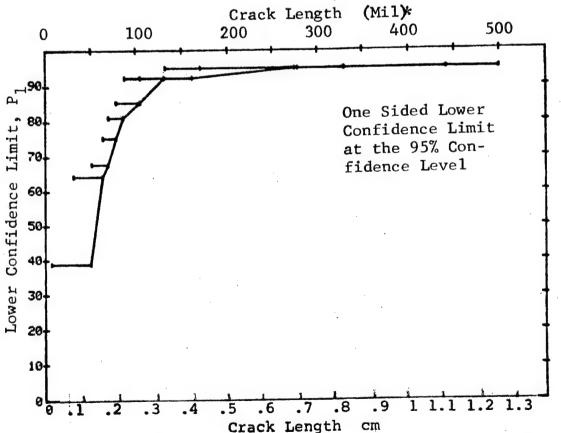


Figure D-17 (Concluded)

1 7	TEST 1, (18)
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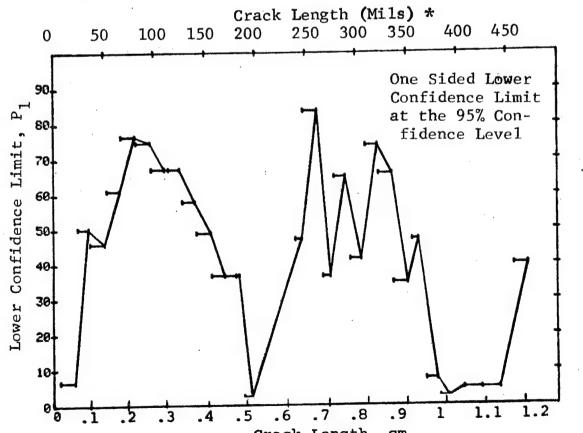
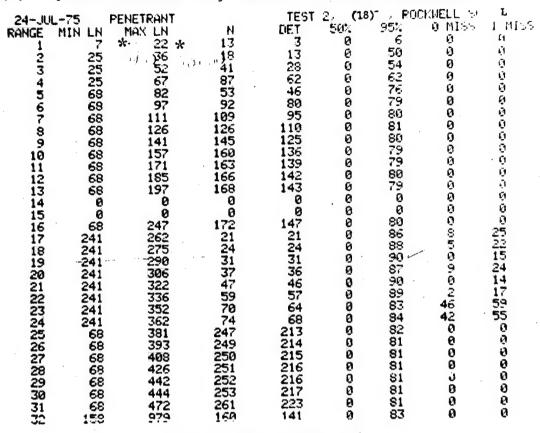


Figure D-18 Probability of Detection for 2219-T87 Al Using Liquid Penetrant. Etched Fatigue Cracks in Flat Plates Measured by Operator L. Lab. Env.



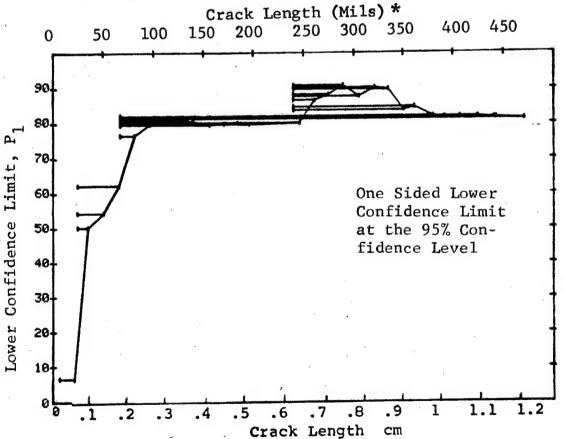
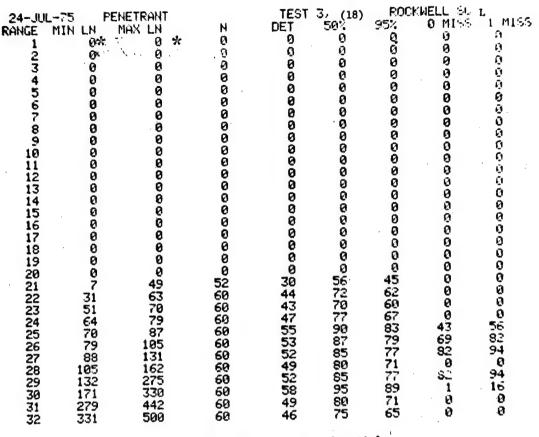


Figure D-18 (Continued) 11



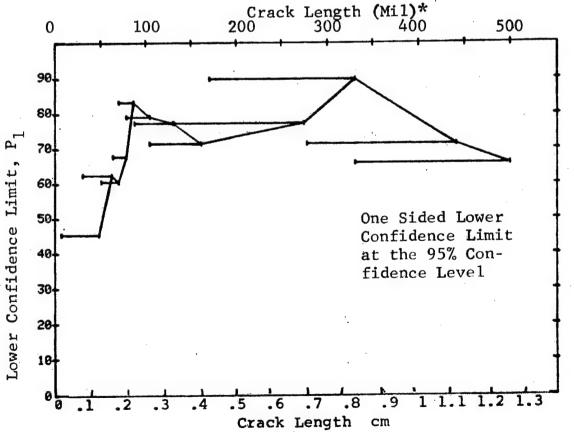
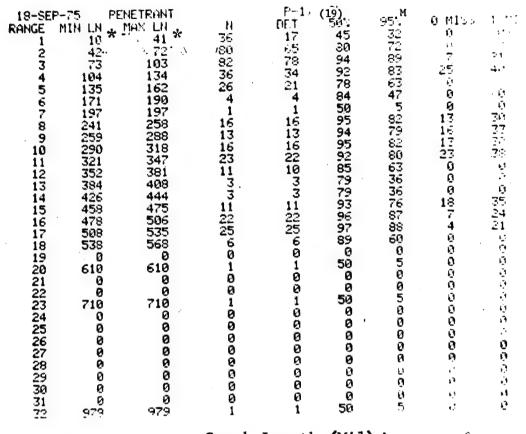


Figure D-18 (Concluded)



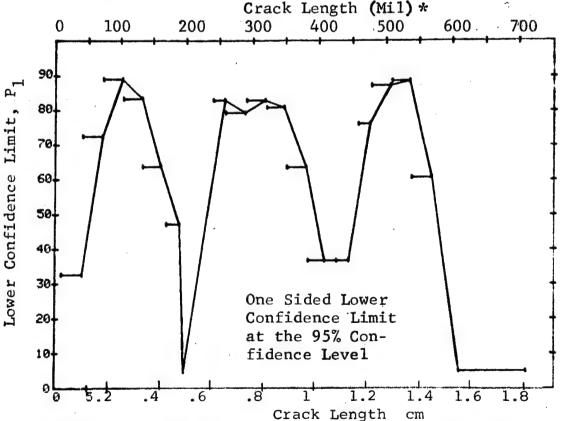


Figure D-19 Probability of Detection for 2219-T87 Al Using Liquid Penetrant. Etched Fatigue Cracks in Flat Plates Measured by Operator M. Lab. Env.

18-SEP-75 RANGE MIN LN 1 10 2 42 3 73 4 73 5 73 6 73 73 8 73 9 171 111 171 112 171 113 171 114 171 115 171 118 171 119 0 20 171 121 171 121 171 121 171 121 171 131 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 18 171 171 171 18 171 171 18 171 171 18 171 171 18 171 171 171 171 171 171 171 171 171 171	103 134 1162 1190 1197 1258 11 288 347 381 408 444 475 11 506 11 535 11 568 11 610 11 00 710 11	78 78 112 112 113 137 148 137 138 138 139 139 139 139 139 139 139 139) ୬ ୧୯୭ ୧୯୭ ୧୯୭ ୧୯୭ ୧୯୭ ୧୯୭ ୧୯୭ ୧୯୭ ୧୯୭ ୧୯୭	M 9 3 7 8 9 0 7 7 8 9 1 4 3 2 2 3 3 4 5 5 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0		
30 0	9 0	0 9	0 0 0	0	13	43

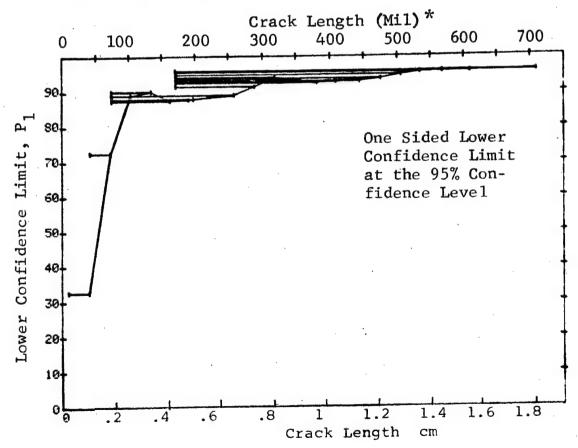


Figure D-19 (Continued) for ^1

REPRODUCIBILITY OF THE URICINAL PAGE IS POOR

18-SEP-RANGE M 18-SEP-RANGE M 12345567891011231441516718922122324425627289331	IN ** LN ** CO C	NETRANT NAME OF STREET OF	# 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5	T	9)\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\		NA 14 0 1 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	
28 29 30 31 32	134	283	60	55	99 98 95 95 98	8 3 349 89 95	43 1 1	56 1- 16

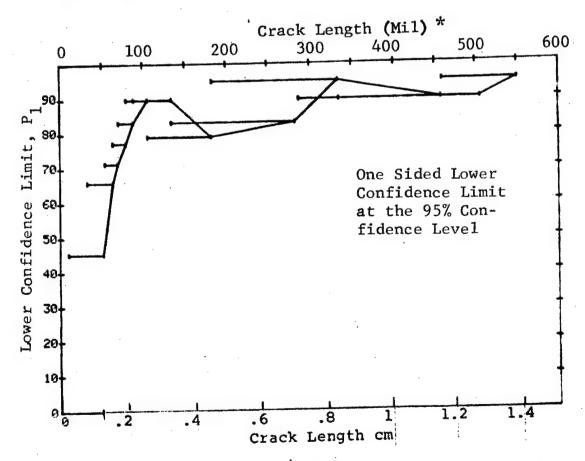


Figure D-19 (Concluded)

18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18-SEF 18	MIN 10 % 42. 73 104 135 177 241 9 290 322 3584 478 538 610 0 0 0 0 0 0 0 0	NETRANT ** 103 134 109 109 109 109 109 109 109 10	N7026651631631331225601001000000	P-1, DET 10022005 1533500 100 100 000 000 000 000 000 000 000	N 00:147744709949459932790000000000000000000000000000000000	31550694573939366686050050050000 31550694573939366886050050050000	0 15 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3	1 6 7 6 6 6 6 6 7 7 8 6 6 6 6 6 6 6 6 6 6
26 27 28 29 30 31 32	Ø	Ø Ø	0	9 9 9 9 9	0	00000005	0	0 0 0 0 0 0

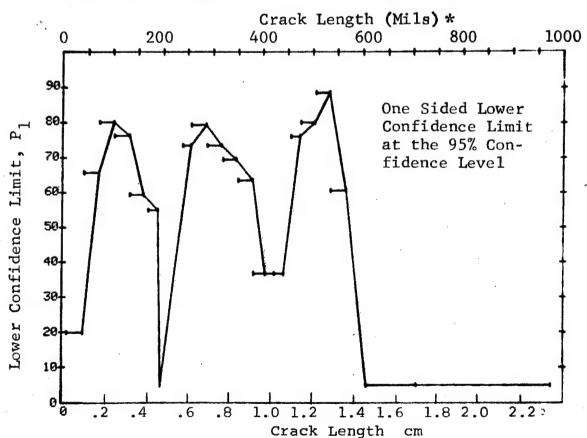


Figure D-20 Probability of Detection for 2219-T87 Al Using Liquid Penetrant. Etched Fatigue Cracks in Flat Plates Measured by Operator N. Lab. Env.

(b) Optimum Probability Method of Data Cumulation

18-SEE 18-SEE 18-SEE 12-3-4-5-6-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-01-12-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-14-15-16-7-8-9-03-13-13-13-13-13-13-13-13-13-13-13-13-13	7-75 PE MIN 10 *, 42 x40 73 73 73 73 73 171 171 171 171 171 171	NETRANT * 1034 1154 1159 1157 1158 1159 1159 1159 1159 1159 1159 1159	17.002 11449 1565 11449 1565 11244 1490 1665 11244 1490 1000 11244 1490 1000 1000 1000 1000 1000 1000 10	PT 2010 244 499 925 6739000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 7000 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 700 000 000	N (25) (25) (25)	**95626+127-866678925659999666699999999669999999999999999	8 0 0 0 0 0 0 1 1 0 0 1 1 0 0 0 0 0 0 0	
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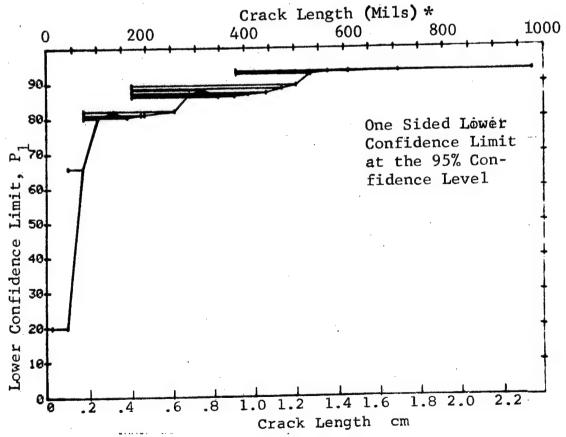


Figure D-20 (Continued)

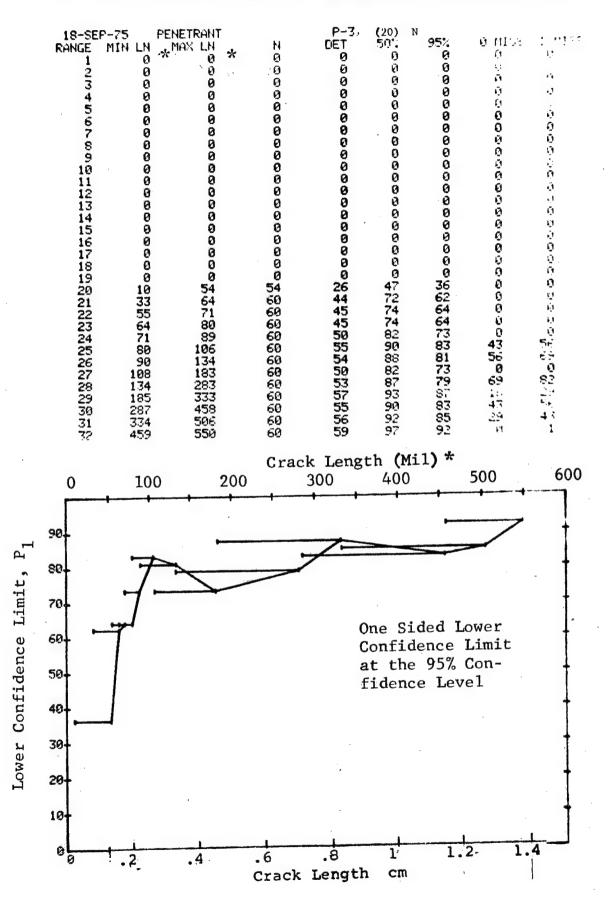


Figure D-20 (Concluded)

1234567890112345678901234567890	IN 238488385588 1129381294889557848895757848896644444444444444444444444444444444	EDDY CURRENT MAX LN * 22 * 36	NESS 4538879533201373760214521111	TEST 123 366 1667 153 32 0 1 3 7 6 0 2 1 1 4 5 2 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	1, 50.5 50.5 11646693506659790009699993437000096599999999999999999999999999999999	T) 95:023325488578166820568365047767452555568	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	000000304010000000000000000000000000000	
30 31 32	444 458 474	444 472 979	i 8 59	i 8 59	50 91 98	5 68 95	9 9	0 0 0	

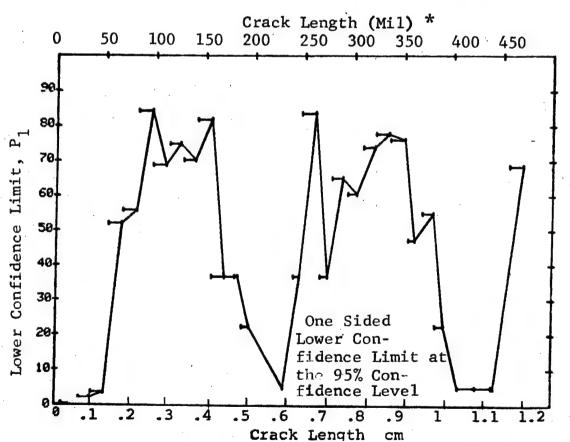


Figure D-21 Probability of Detection for 2219-T87 Al Using Eddy Current. Etched Fatigue Cracks in Flat Plates Measured by Operator T. Lab. Env.

24-JU		EDDY CURRENT		TEST			KWELL S		
RANGE	MIN LN	MAX LN	. <u>N</u>	DET	50%	95%	O MISS		55
1	- 7	ж <u>22</u> ж	13	1	0	9 2	@ 0	e O	
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3 4	54	31 27	46	30	ĕ	52	õ	й	
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ě	83	97	38	36	ĕ	84	23	38	
7	83	111	56	52	Ø-	84	33	47	
8	83 83	126	73	€8	Ø	86 86 88 88 88	30	43	
. 9	83	141	92	- 85	Ø	86	37	50	
10	83	157	107	100	0	88	0 0	9	
11 12	83 83	171	117	103	0 0	88	õ	0	
13	83	141 157 171 185 197	110 113 115	68 85 100 103 106 108	ĕ	88	0	9	
14	õ	é	Ö	Ø	ĕ	0	0	9	
15	83 83	9 232	116	109 27	Ø	83 89	0 2	0	
16	143	247 262 275	27	27	Ø	89	2	19 2 0	
17	143	262	44	44 47	0	93	0	2	
18	143	275	47	47	9	93 94	8	Ö	
19	143 143	290 304	54 60	54 60	9	9 5	ø	ĕ	
20 21	143	322	70	70	õ	9 5	ĕ	ĕ	
22	143	336	82	82	ĕ	96	8	ø	
23	143	352	93	93 97	0	96 96	0	.0	
24	143	352 362	97	97	0	96	0	0	
25 26	143	381 393	102	102 104 105	Ø	97	Ø	0	
26	143	393	104	104	9	97 97	9	0	
27 28	143	408 426	105 106	105	9 9	97	Ø.	ő	
29	143	442	107	107	ø	97	ŏ	ĕ	
30	143	444	108	108	ø	97	Ø	0	
31	143	472	116	116	Ø	97	0	0	
32	143	979	175	175	Ō.	98	9	Ю	

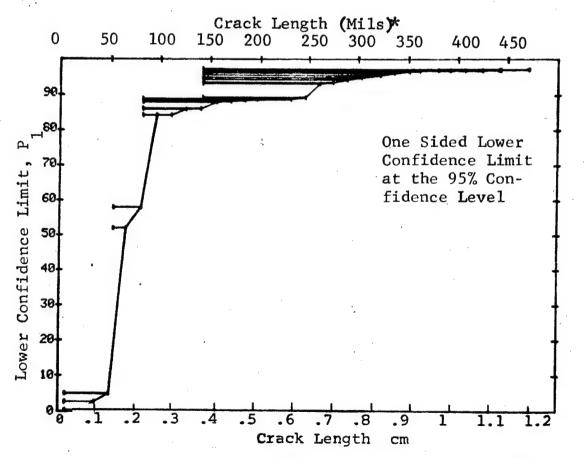


Figure D-21 (Continued)

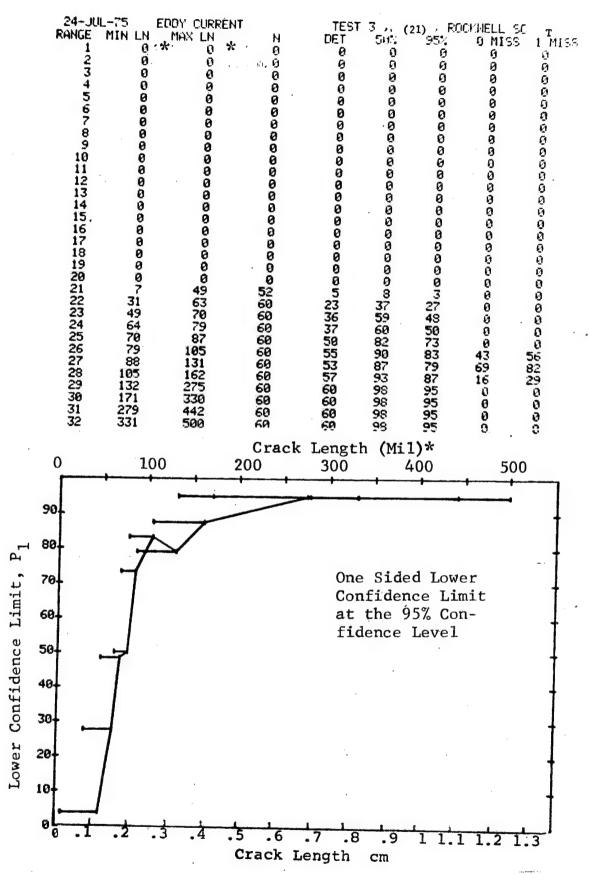


Figure D-21 (Concluded)

03-JUL-75 EDDY CURRENT RANGE MIN LN 1	10 12 11 4 5 2 1 1	1 - POCKMELL 50: 95: 20 30 15 34 76 36 98 98 98 96 98 81 36 98 81 36 98 81 36 98 82 96 99 84 82 86 90 97 77 77 77 77 77 77 77 77 77 77 77 77	00 00 00 00 00 00 00 00 00 00 00 00 00	(22) 1 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
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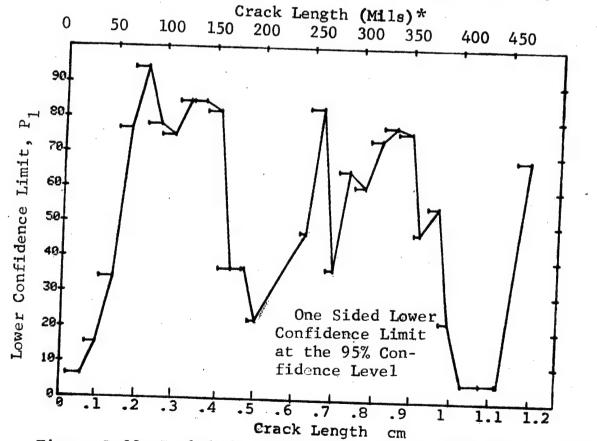


Figure D-22 Probability of Detection for 2219-T87 Al Using Eddy Current. Etched Fatigue Cracks in Flat Plates Measured by Operator U. Lab. Env.

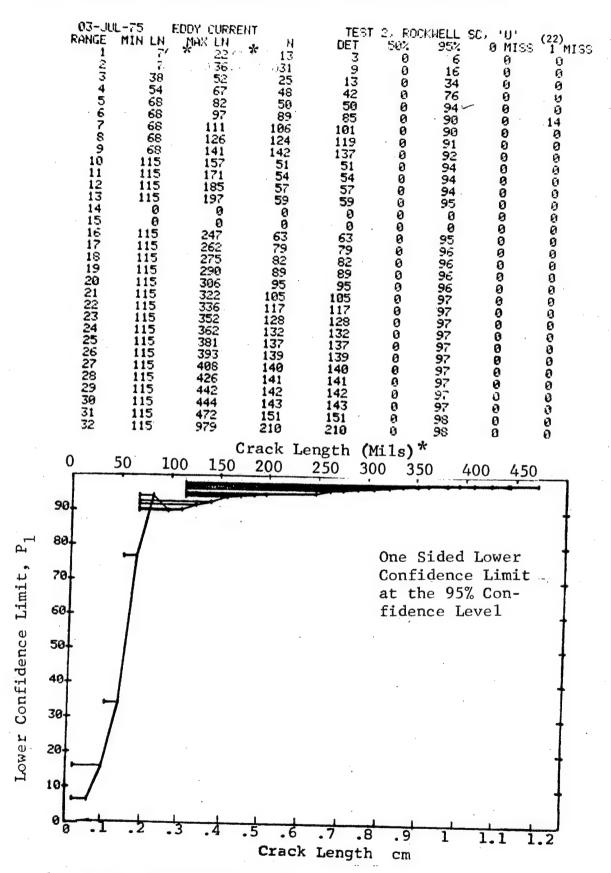
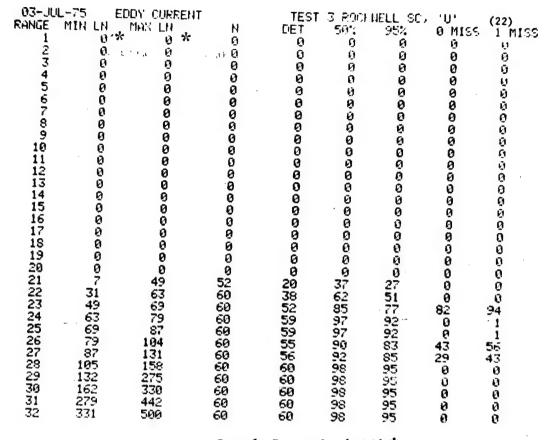


Figure D-22 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation



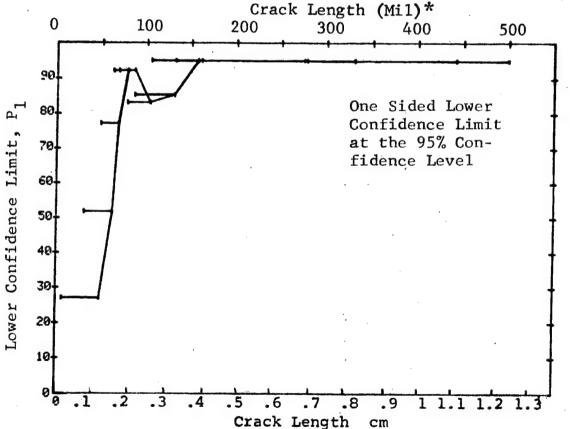


Figure D-22 (Continued)

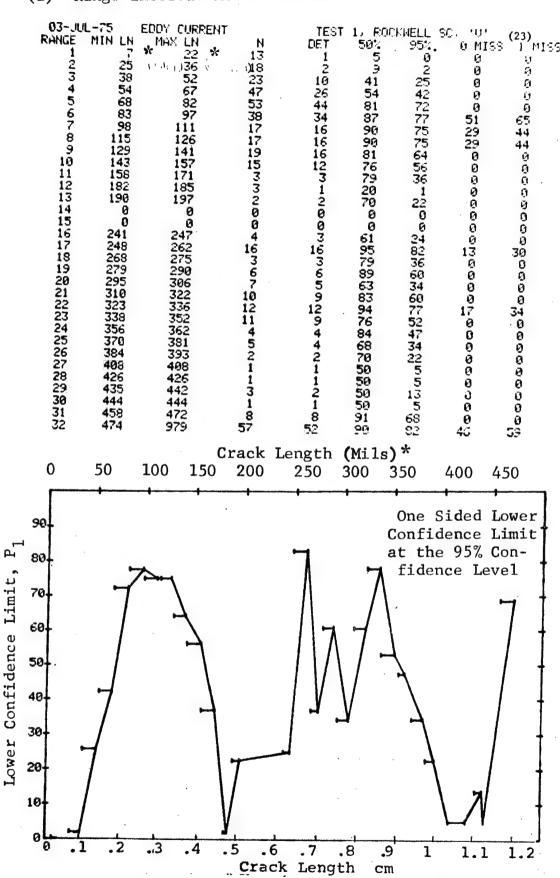


Figure D-23 Probability of Detection for 2219-T87 Al Using Eddy Current. Etched Fatigue Cracks in Flat Plates Measured by Operator V. Lab. Env.

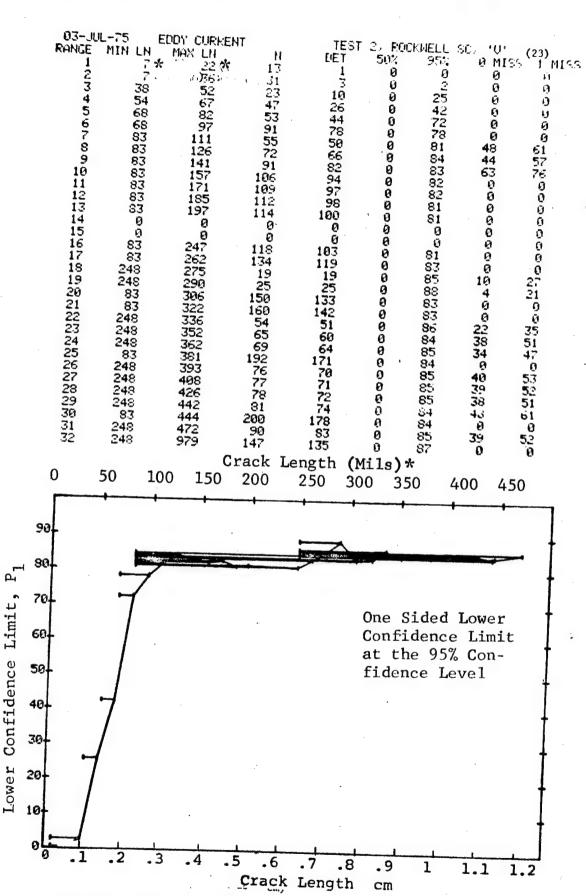
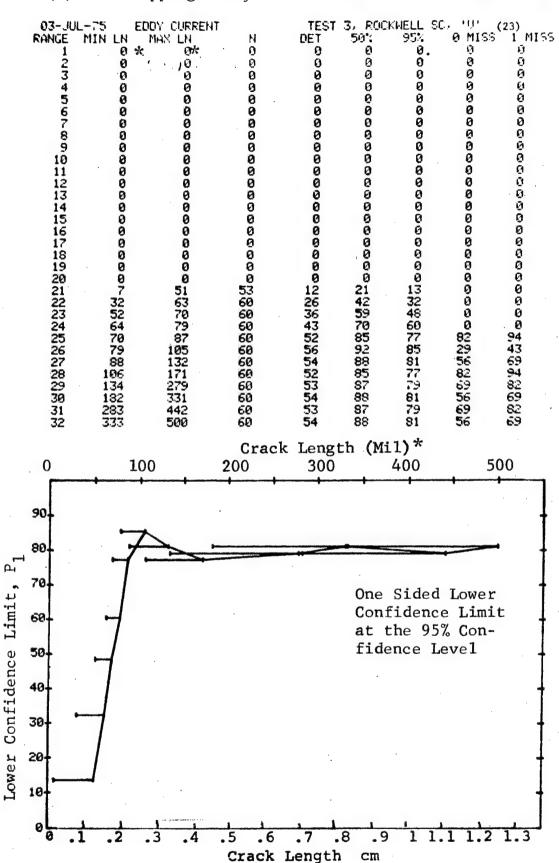


Figure D-23 (Contint ()



¹Figure D-23 (Concluded)

03-JUL-75 EDDY CURRENT RANGE MIN LN NAX LN N 1 7	DET 50 2 1 6 3 8 10 47 8 17 99 16 17 16 77 17 12 77 2	2 15 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9	(24) MISS 1 MISS 1 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
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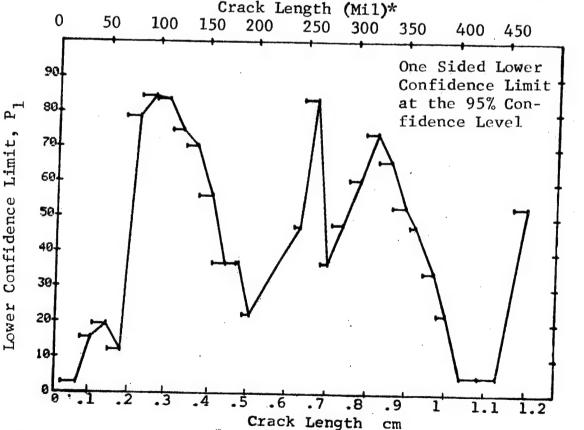
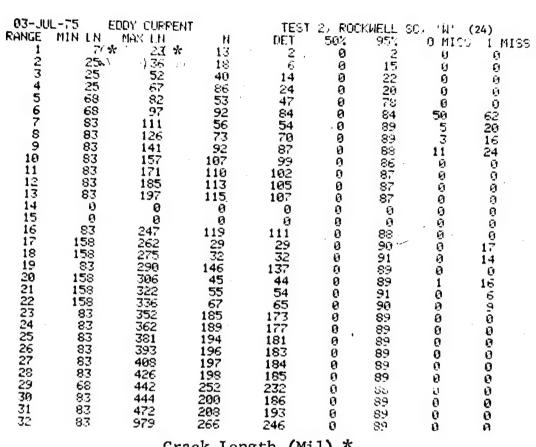


Figure D-24 Probability of Detection for 2219-T87 Al Using Eddy Current. Etched Fatigue Cracks in Flat Plates Measured by Operator W. Lab. Env.



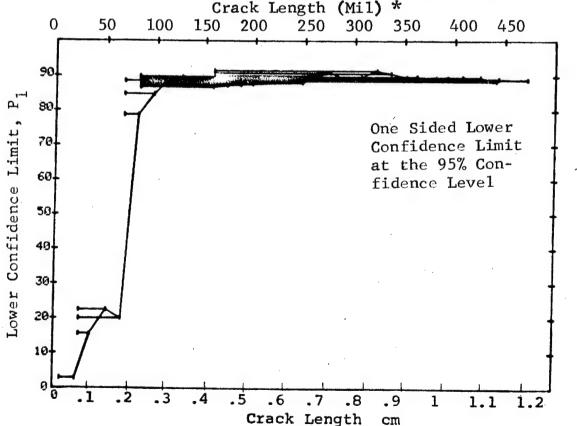


Figure D-24 (Continued)

tor

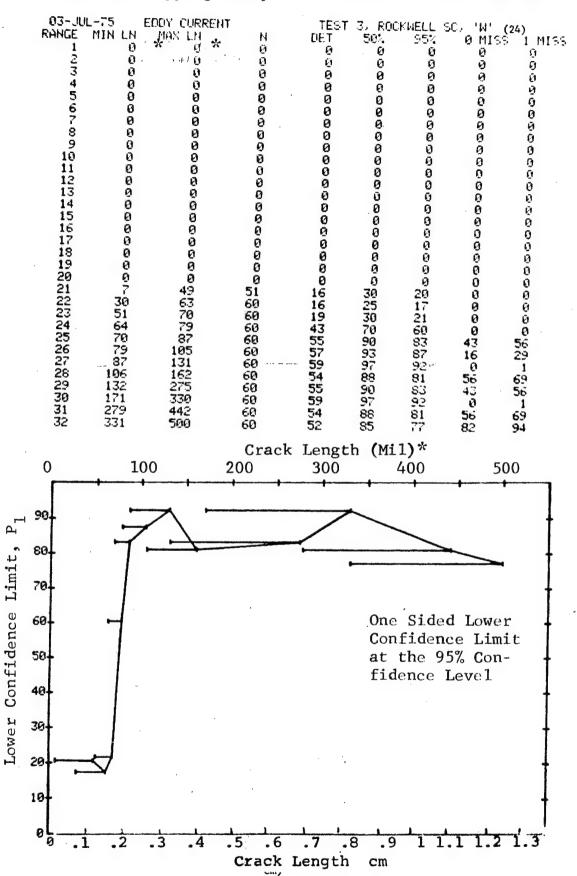


Figure D-24 (Concluded)

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR

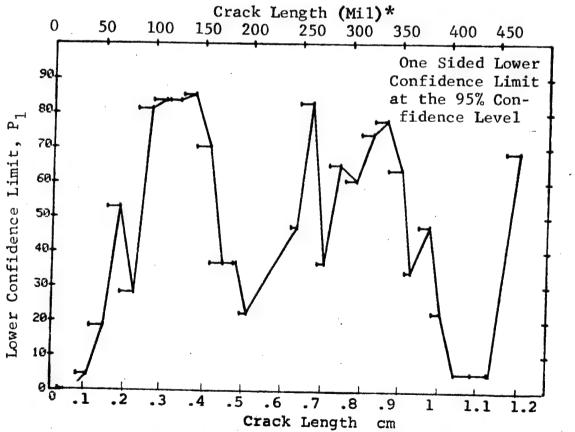


Figure D-25 Probability of Detection for 2219-T87 Al Using Eddy Current. Etched Fatigue Cracks in Flat Plates Measured by Operator X. Lab. Env.

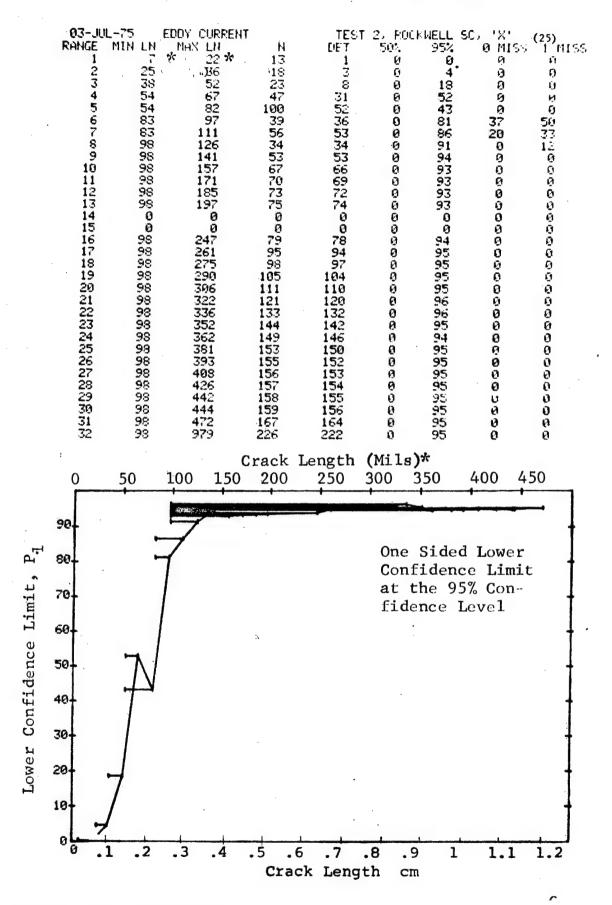
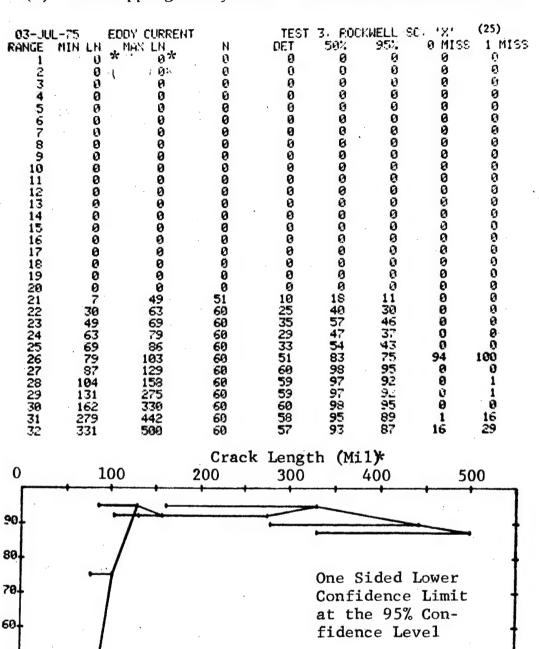


Figure D-25 (Continued)



Confidence Limit at the 95% Confidence Level

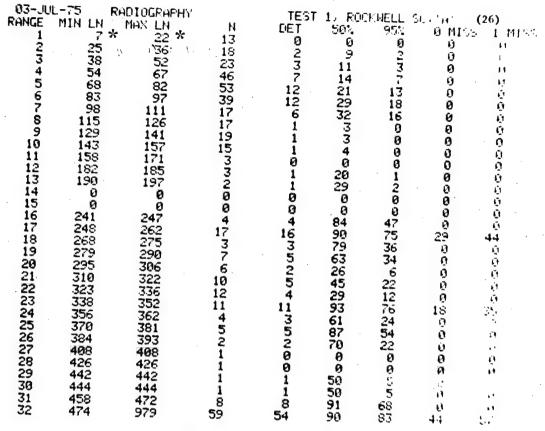
30

10

0 .1 .2 .3 .4 .5 .6 .7 .8 .9 1 1.1 1.2 1.3

Crack Length cm

Figure D-25 (Concluded)



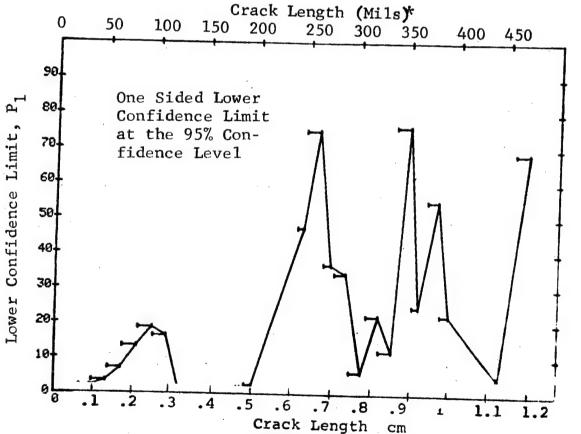
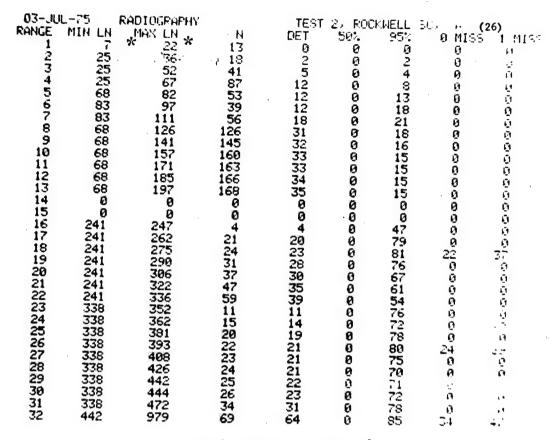


Figure D-26 Probability of Detection for 2219-T87 Al Using X-ray. Etched Fatigue Cracks in Flat Plates Measured by Operator A. Lab. Env.



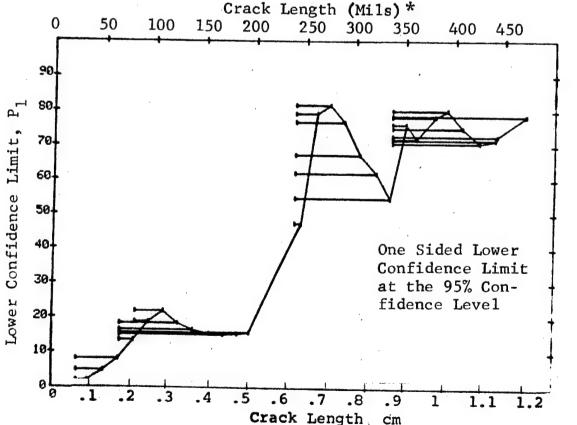
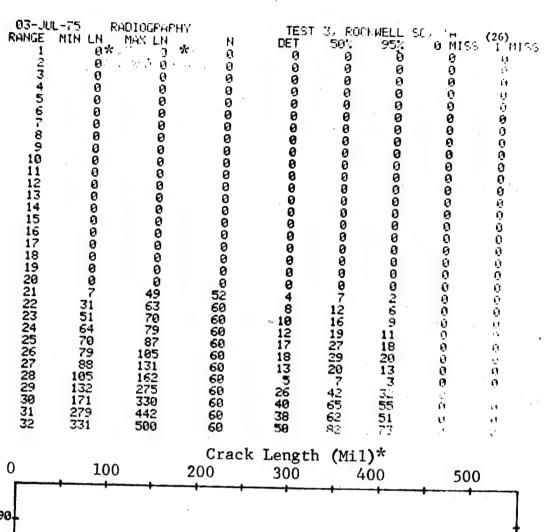


Figure D-26 (Continued)



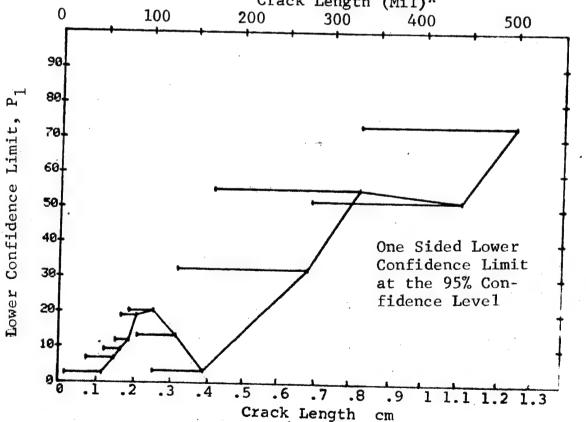


Figure D-26 (Concluded)

2 25 36 18 3 38 52 23 4 54 67 46 5 68 82 53 6 33 97 37 7 98 111 17 8 115 126 17 9 129 141 19 10 143 157 15 11 158 171 3 12 182 185 3 13 190 197 2 14 0 0 0 15 0 0 0 16 241 247 4 17 248 262 17 18 268 275 3 19 279 290 7 20 295 306 6 21 310 322 10 22 323 336 12 23 338 352 11 24 356 362 4 25 370 381 5 26 384 393 2 27 408 408 1 28 426 426 1 29 442 442 1 30 444 444 1 31 458 472 8 32 474 979 59	1 2 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
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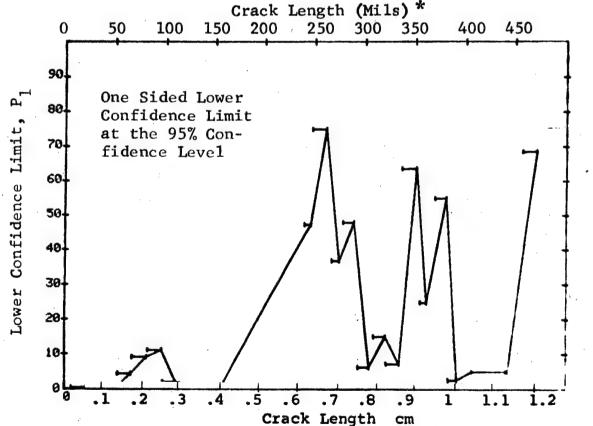


Figure D-27 Probability of Detection for 2219-T87 Al Using X-ray. Etched Fatigue Cracks in Flat Plates Measured by Operator B. Lab. Env.

D-81

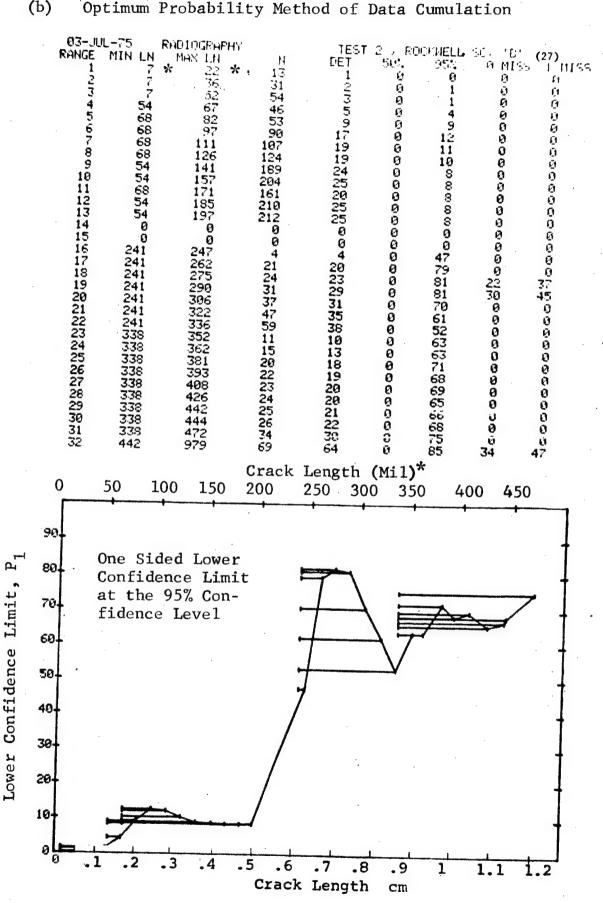


Figure D-27 (Continued)

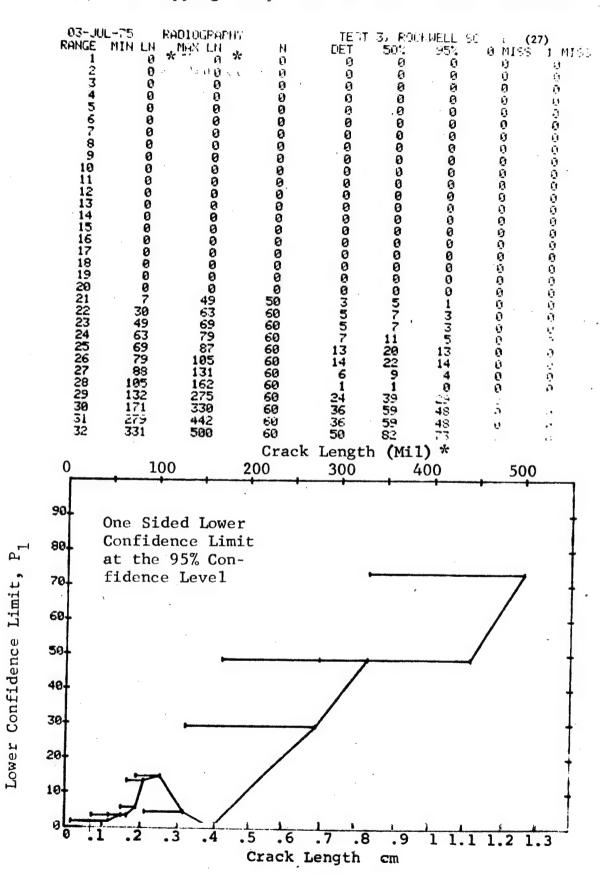
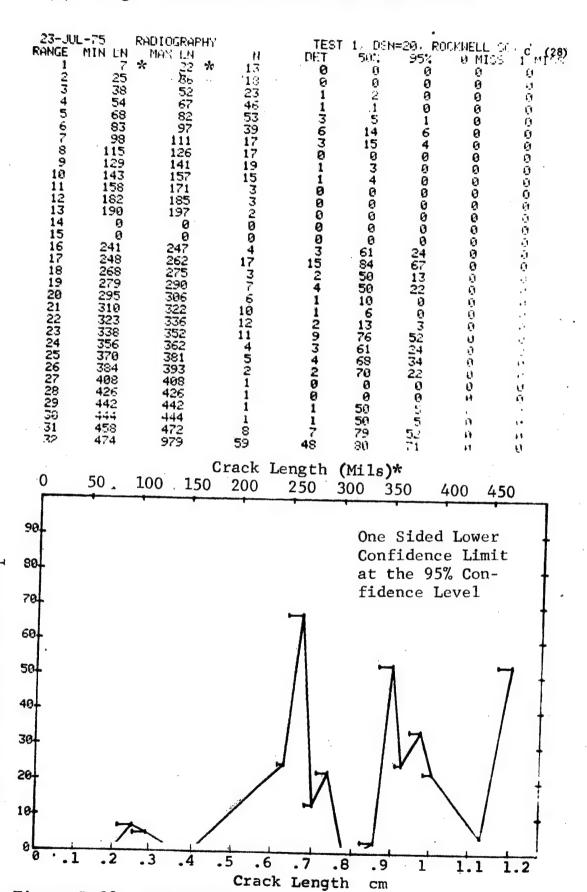


Figure D-27 (Concluded)

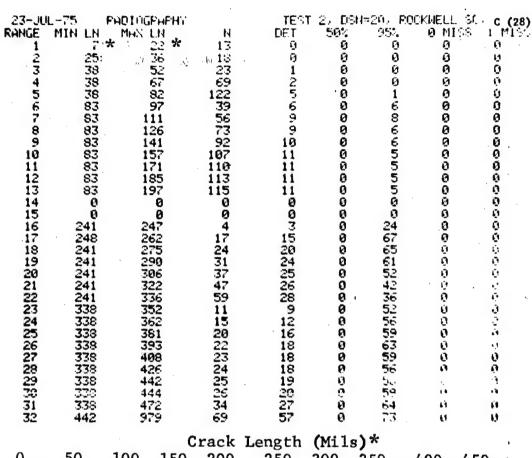


Limit,

Confidence

Lower

Figure D-28 Probability of Detection for 2219-T87 Al Using X-ray. Etched Fatigue Cracks in Flat Plates Measured by Operator C. Lab. Env.



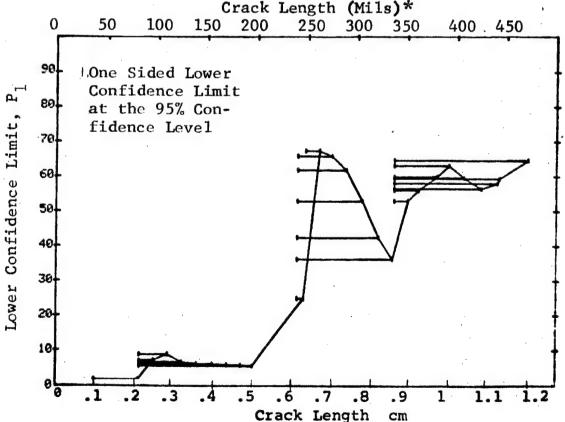


Figure D-28 (Continued)

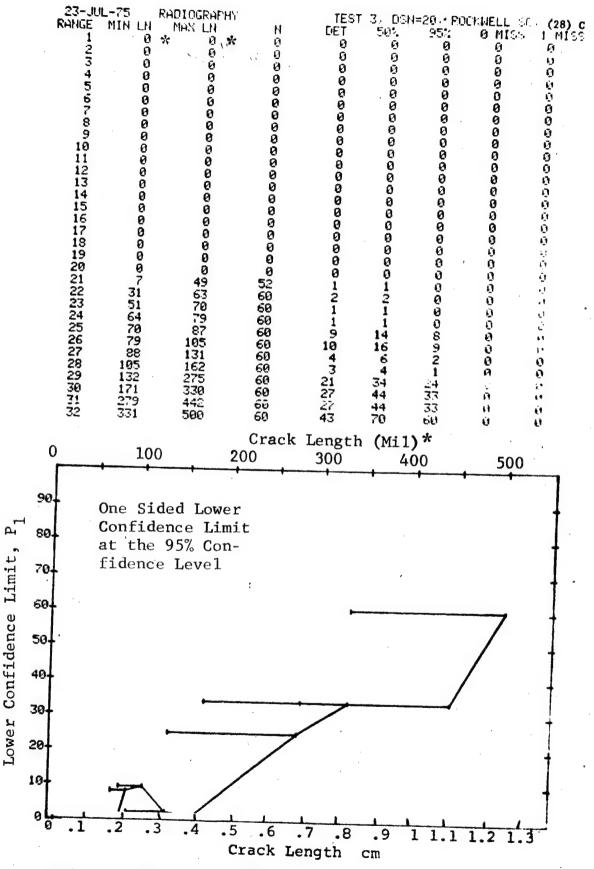
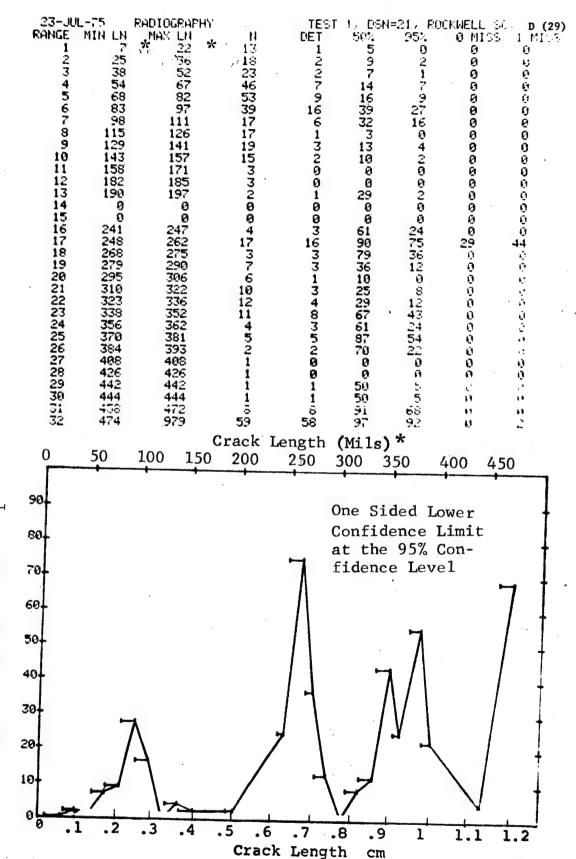


Figure D-28 (Concluded)



Lower Confidence Limit,

Figure D-29 Probability of Detection for 2219-T87 Al Using X-ray. Etched Fatigue Cracks in Flat Plates Measured by Operator D. Lab. Env.

D-87

23-JU 23-JU 23-45-67-89-0111234-15-67-89-01112314-15-67-89-01112314-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-14-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-01123-15-67-89-0100-0100-0100-0100-0100-0100-0100-0	H-75 MIN 5443 833338 998 1998 1948 2441 2441 2441 2441 2441 2441 2441 24	* **	3527 827 1121 157 1211 157 165 172 173 173 173 173 173 173 173 173 173 173	* 131546939573271113566732772455669 k	1 DF1 1 1 7 16 16 22 23	50° 0 0 0 0	95 0 2 3 7 0 7 8 2 0 9 9 8 8 8 9 9 7 5 8 5 5 8 4 8 4 9 5 5 5 5 5 5 7 9 8 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5	((((156)	
0	50	100	150	200	250		50	400	450	

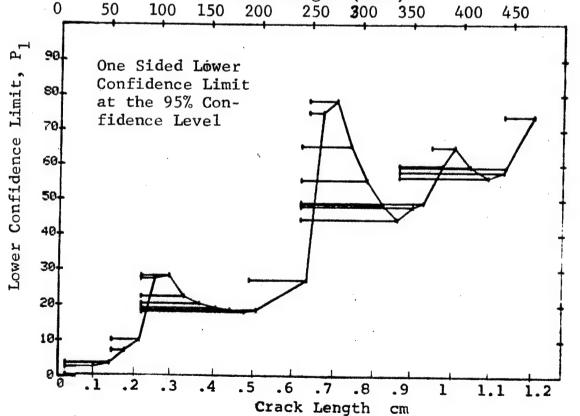
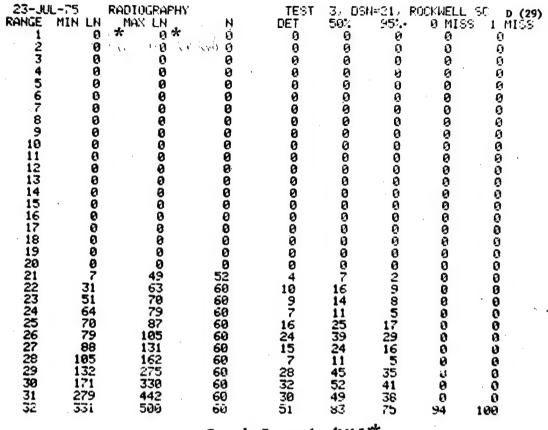


Figure D-29 (Continued)



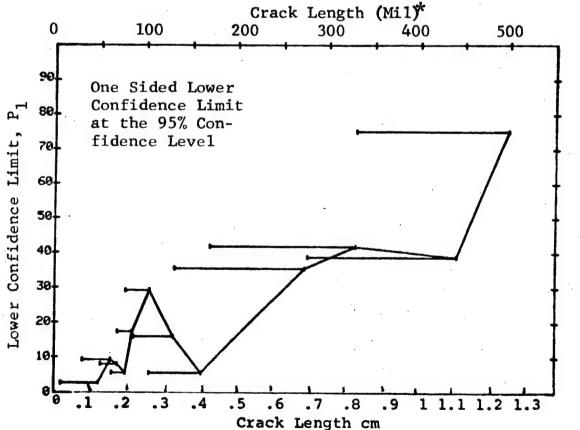


Figure D-29 (Concluded)

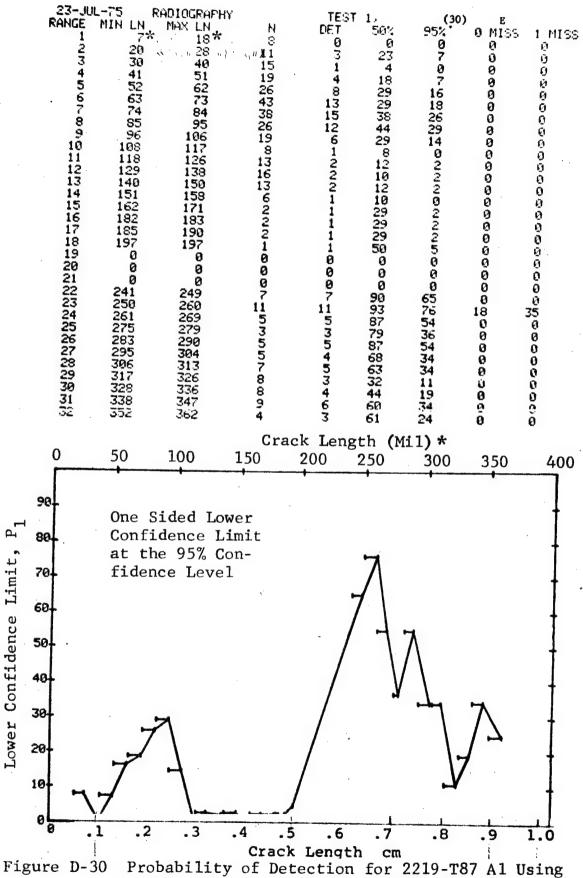


Figure D-30 Probability of Detection for 2219-T87 Al Using X-ray. Etched Fatigue Cracks in Flat Plates Measured by Operator E. Lab. Env.
D-90

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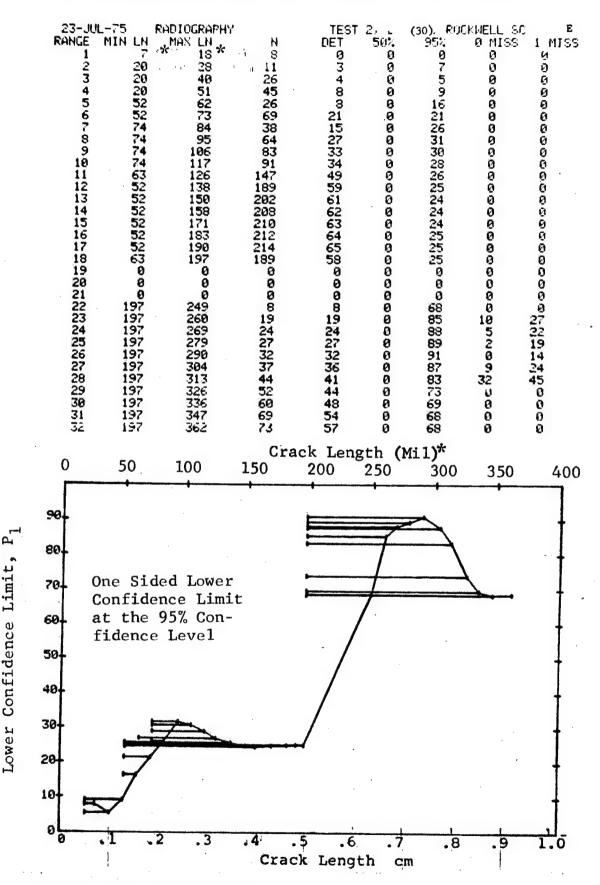
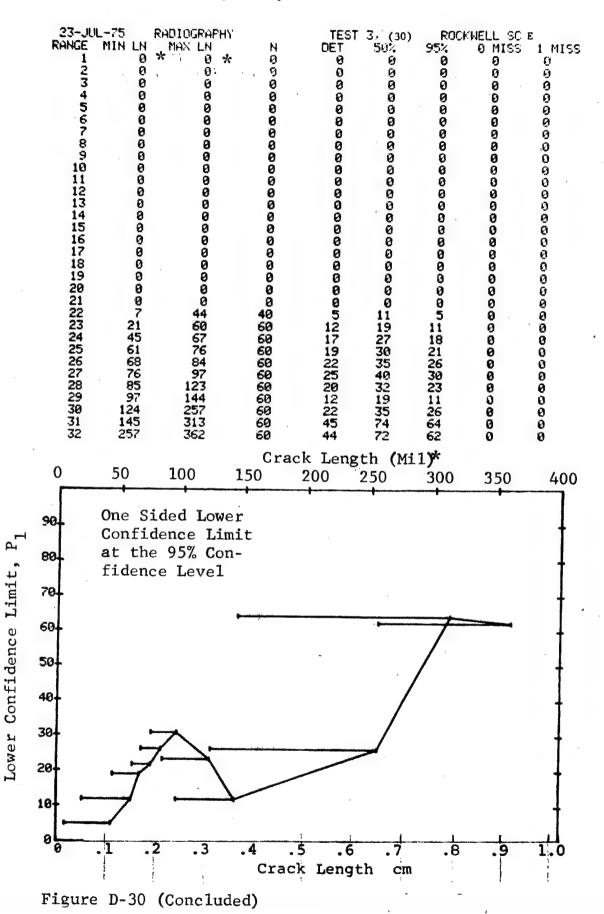


Figure D-30 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation



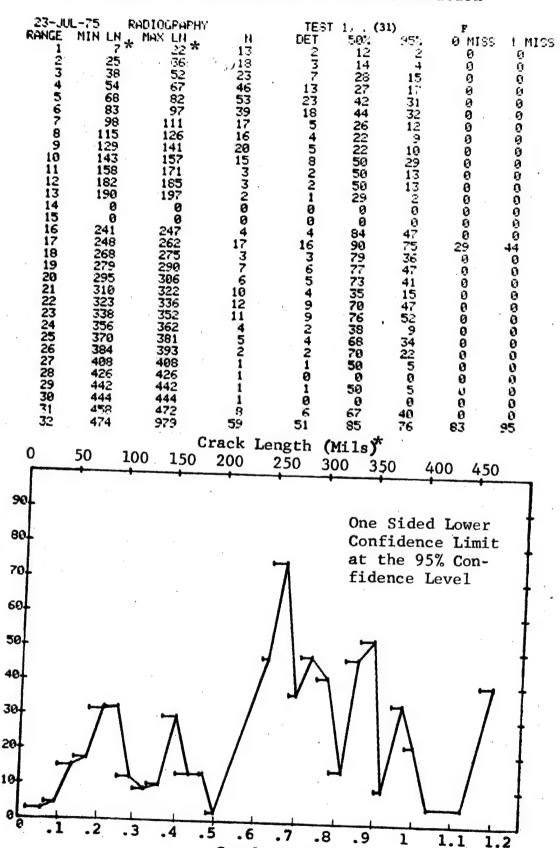


Figure D-31 Probability of Detection for 2219-T87 Al Using X-ray. Etched Fatigue Cracks in Flat Plates Measured by Operator F. Lab. Env.

Lower Confidence Limit, \mathbf{P}_1

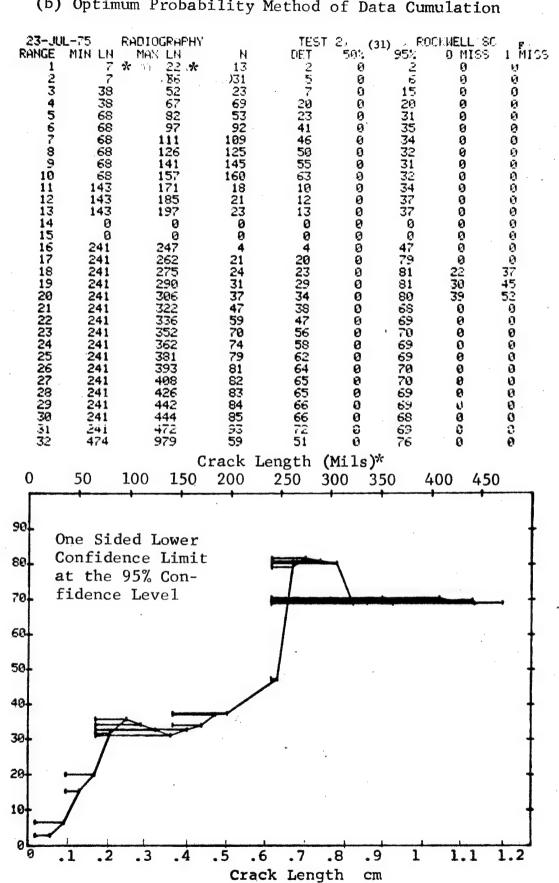


Figure D-31 (Continued)

Lower Confidence Limit,

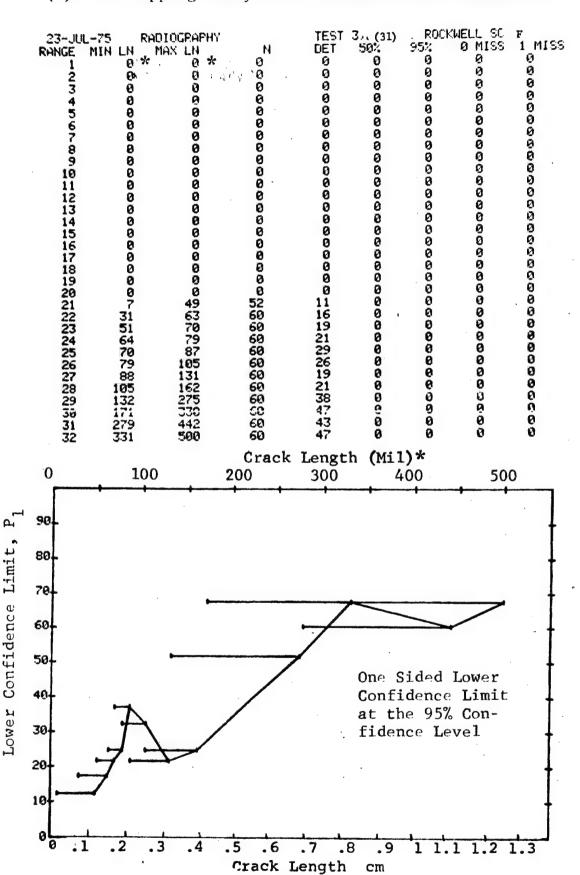


Figure D-31 (Concluded)

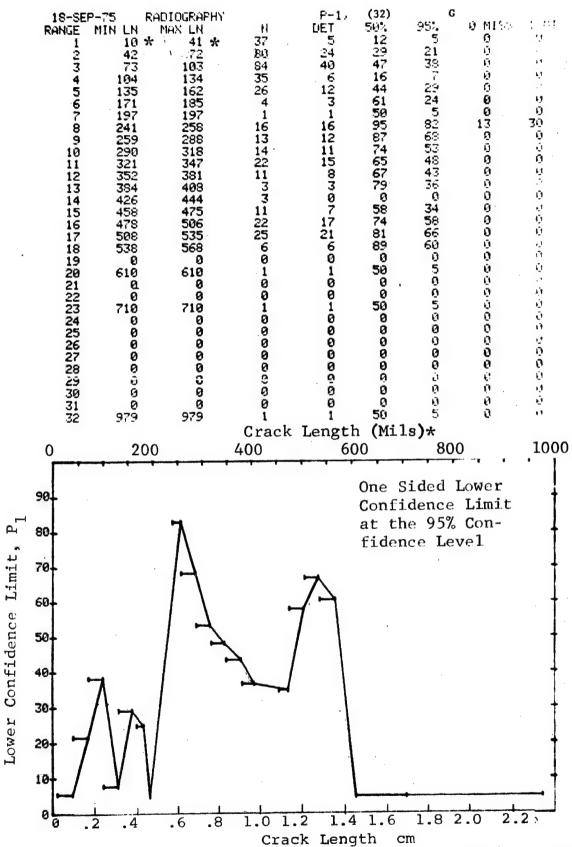


Figure D-32 Probability of Detection for 2219-T87 Al Using X-ray. Etched Fatigue Cracks in Flat Plates Measured by Operator G. Lab. Env.

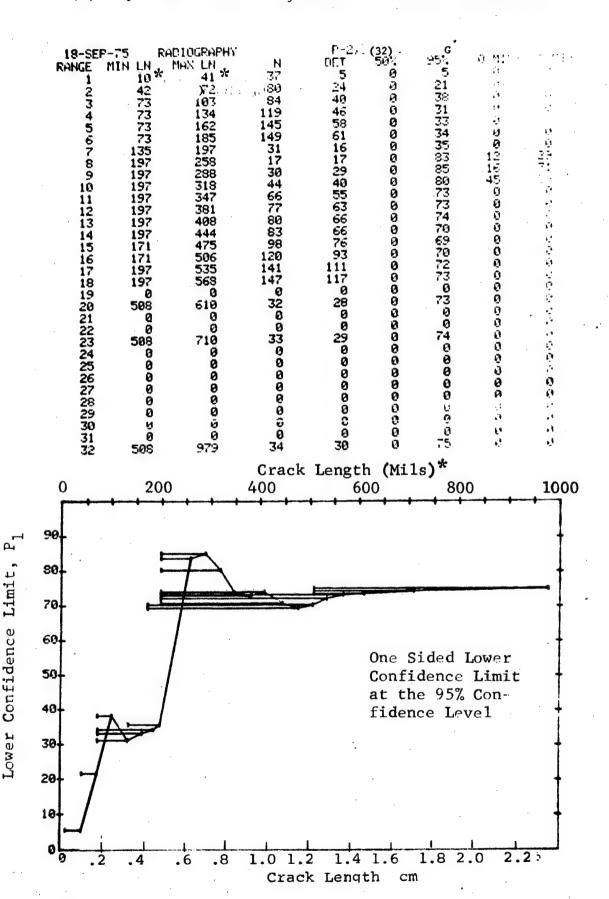
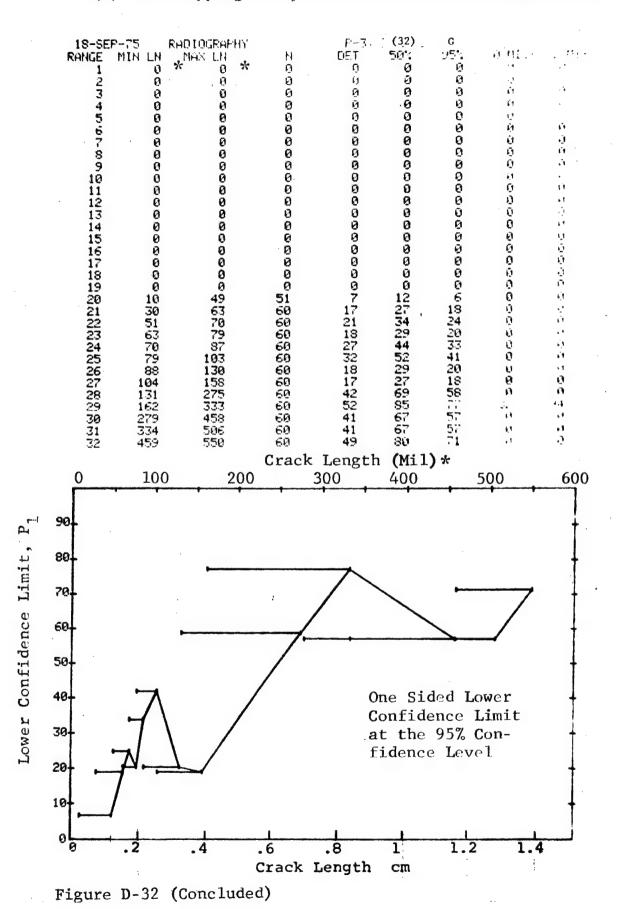


Figure D-32 (Continued)



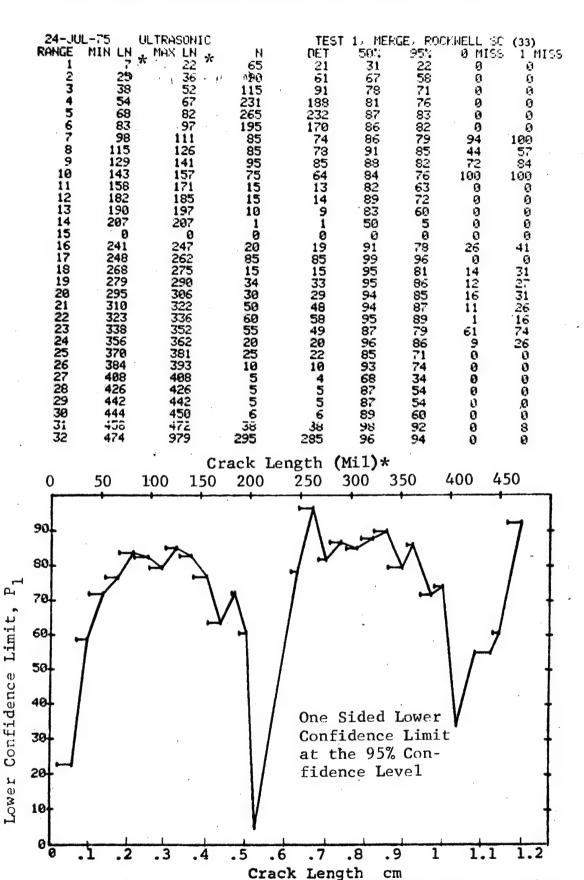


Figure D-33 Probability of Detection for 2219-T87 Al Using Ultrasonics. Etched Fatigue Cracks in Flat Plates Merged for 5 Operators. Lab. Env. D-99

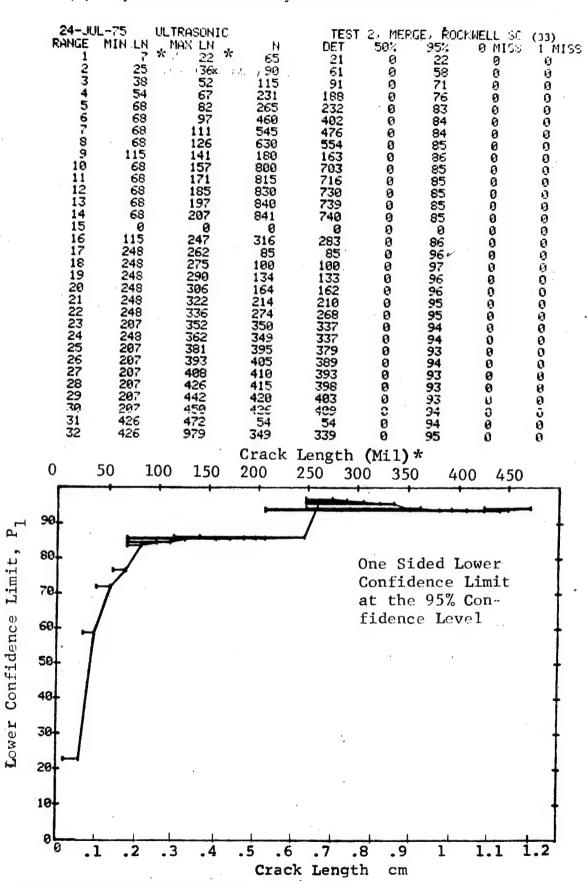
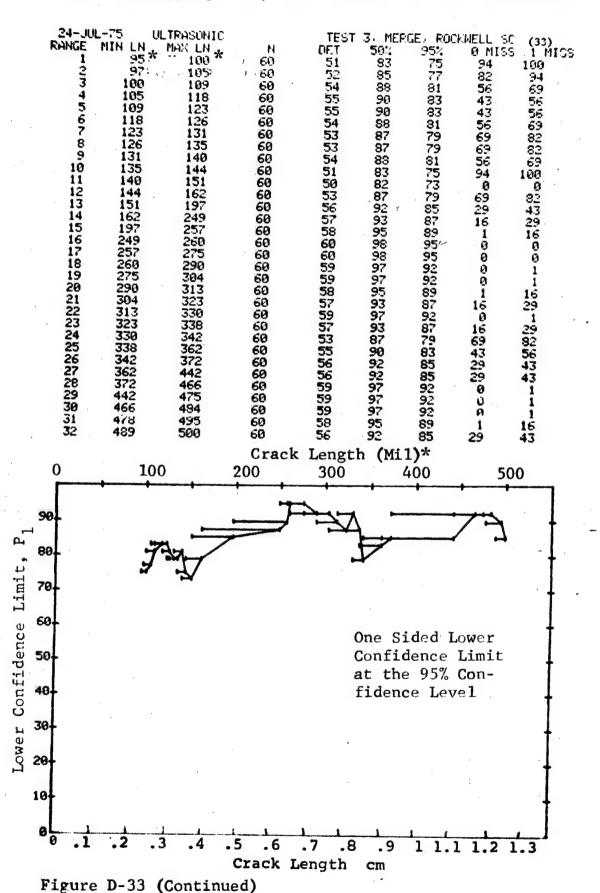


Figure D-33 (Continued



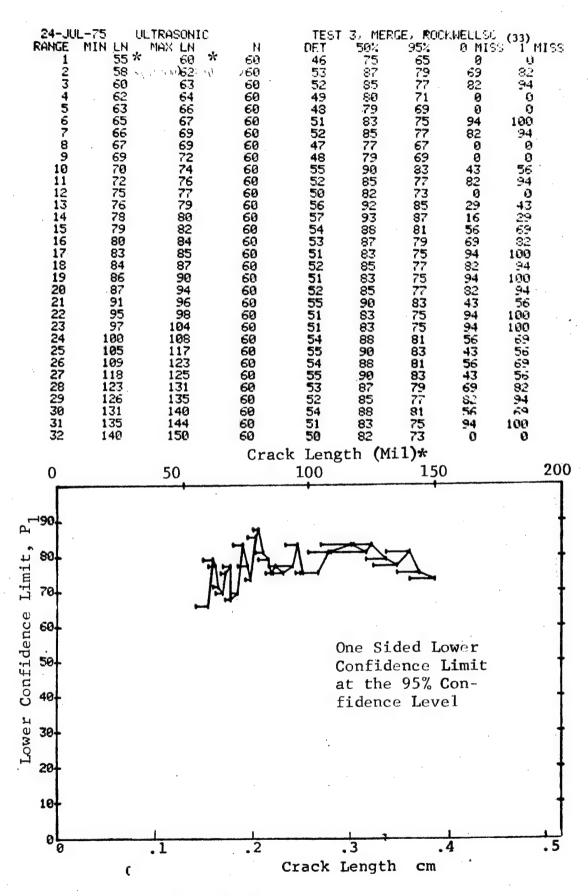


Figure D-33 (Continued)

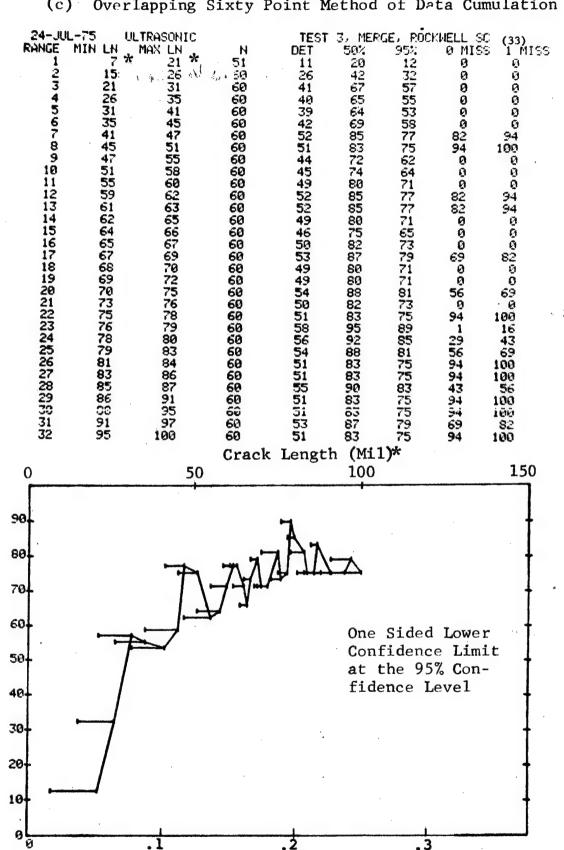


Figure D-33 (Concluded)

Lower Confidence Limit,

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR

Crack Length

cm

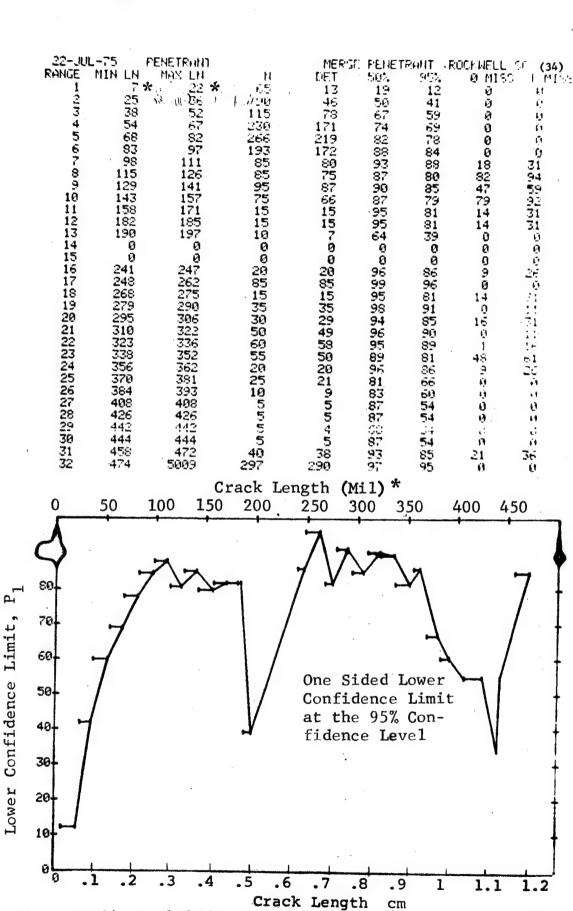


Figure D-34 Probability of Detection for 2219-T87 Al Using Liquid Penetrant. Etched Fatigue Cracks in Flat Plates Merged for 7 Operators. D-104

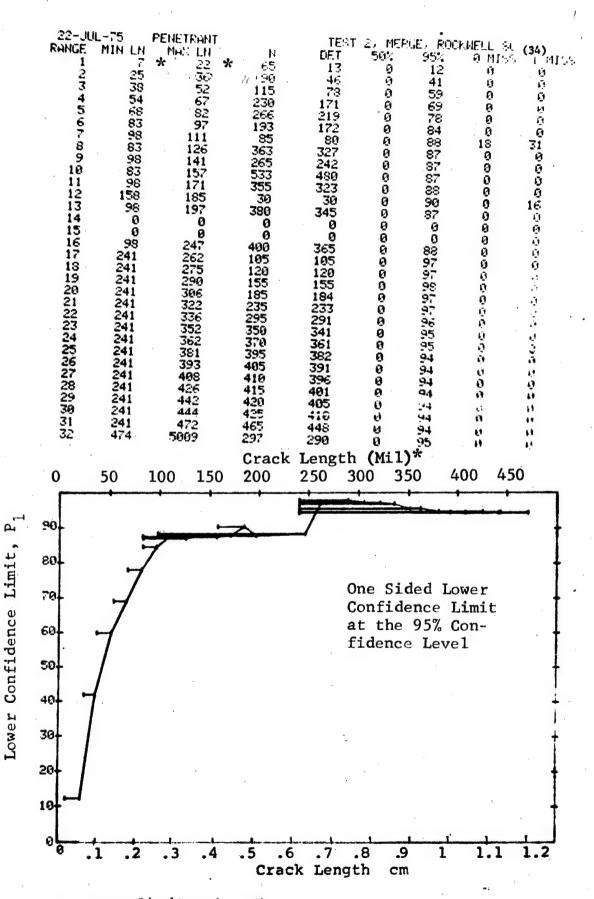


Figure D-34 (Continued)
D-105

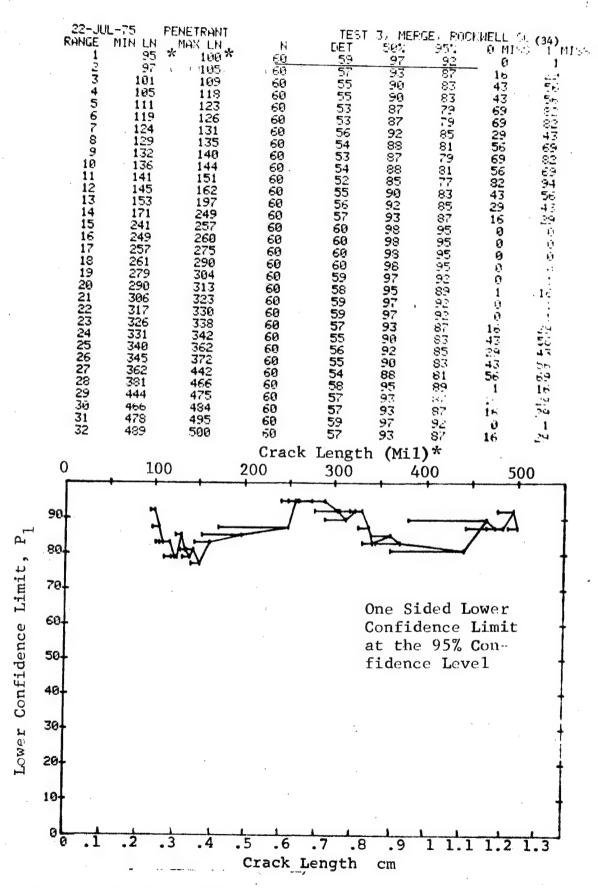
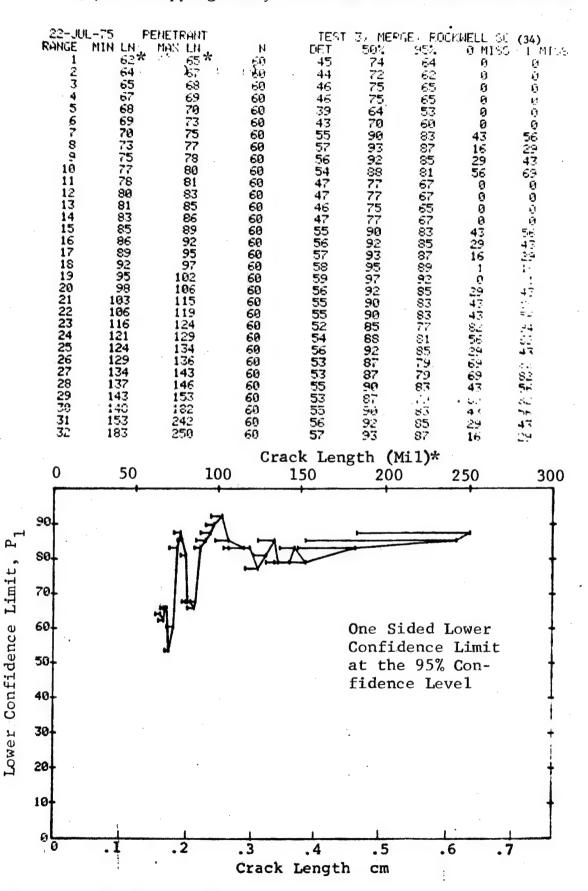


Figure D-34 (Continued)



Figure' D-34 (Concluded)

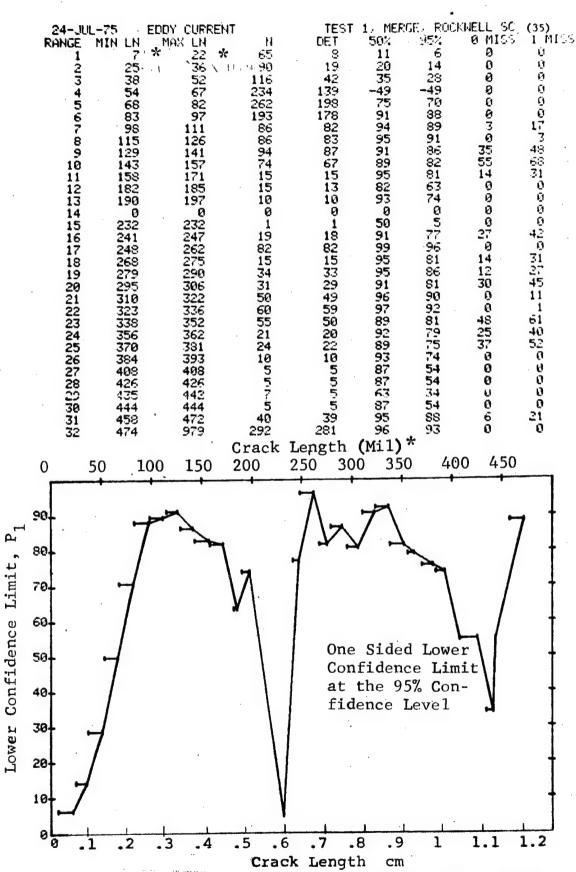


Figure D-35 Probability of Detection for 2219-T87 Al Using
Eddy Current. Etched Fatigue Cracks in Flat
Plates Merged for 5 Operators. D-108 Lab. Env.

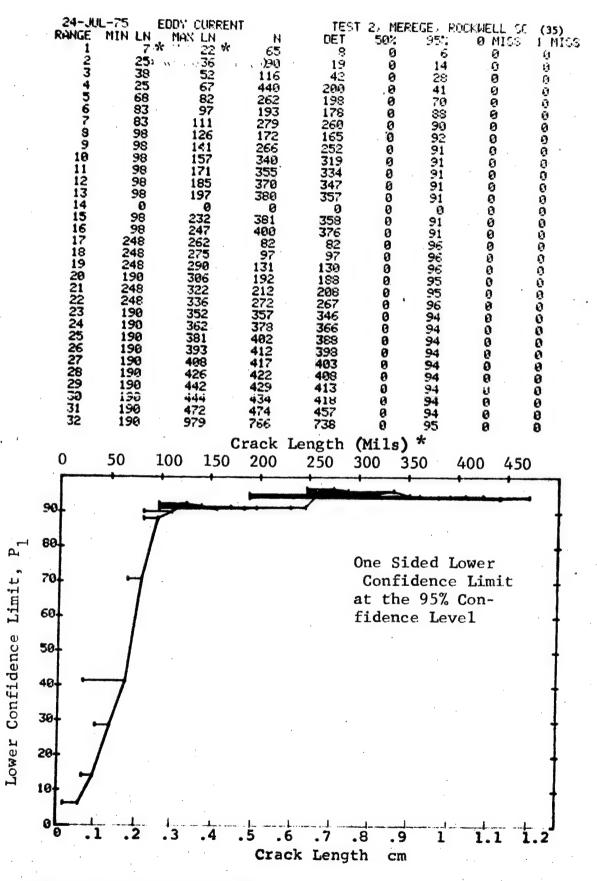


Figure D-35 (Continued)

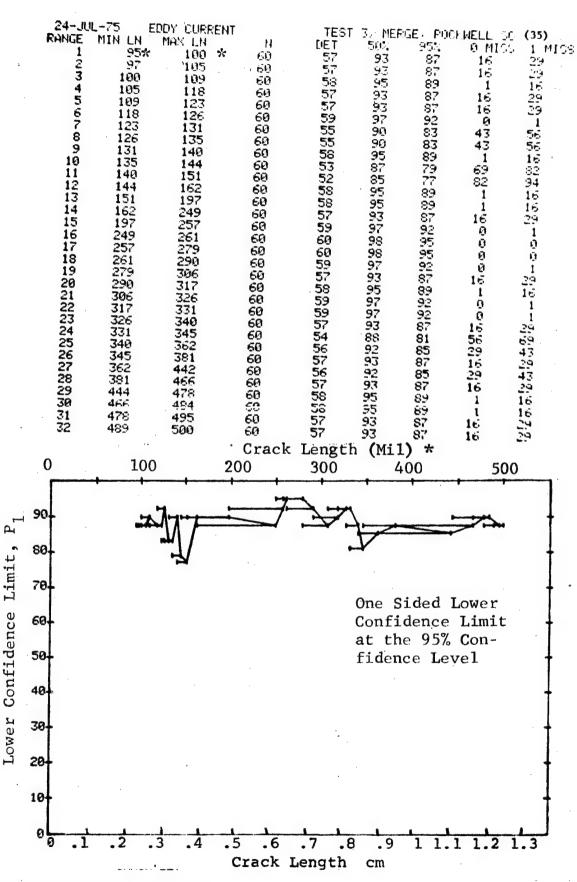


Figure D-35 (Continued)

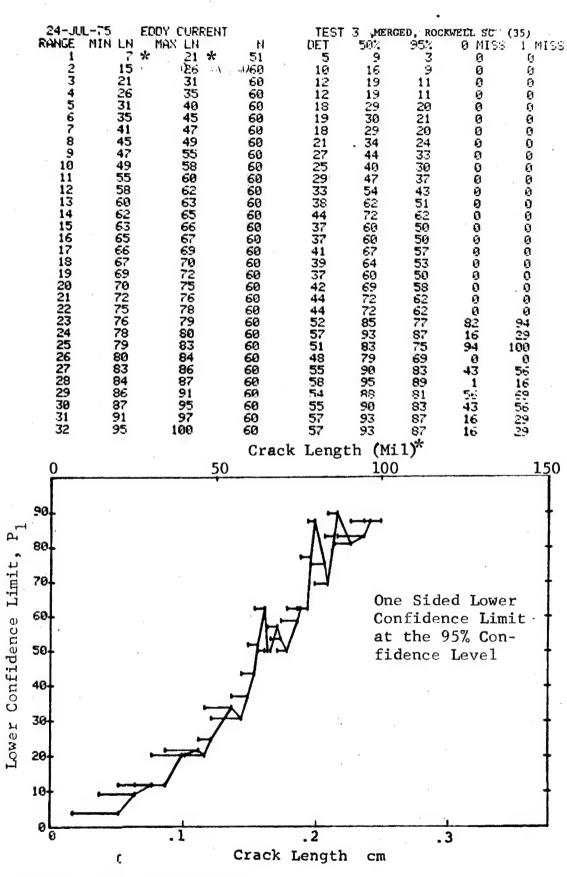


Figure D-35 (Concluded)

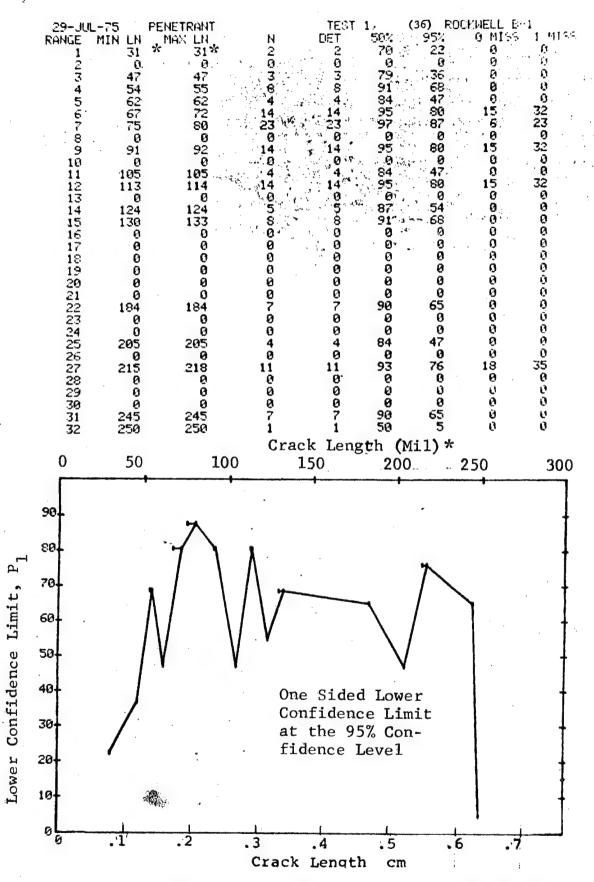


Figure D-36 Probability of Detection for Ti-6A1-4V Using Liquid Penetrant. Fatigue Cracks in Flat Plates. Prod. Env.

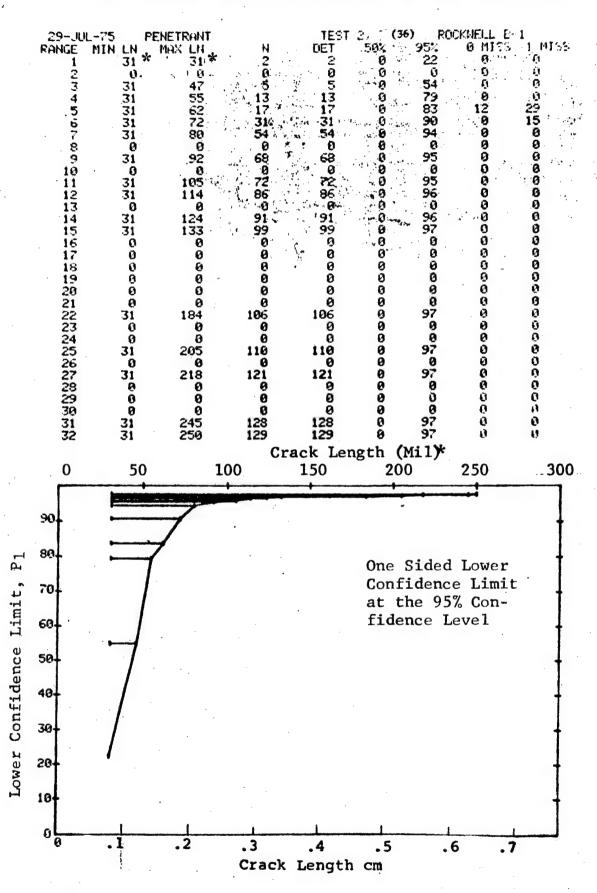


Figure D-36 (Continued)

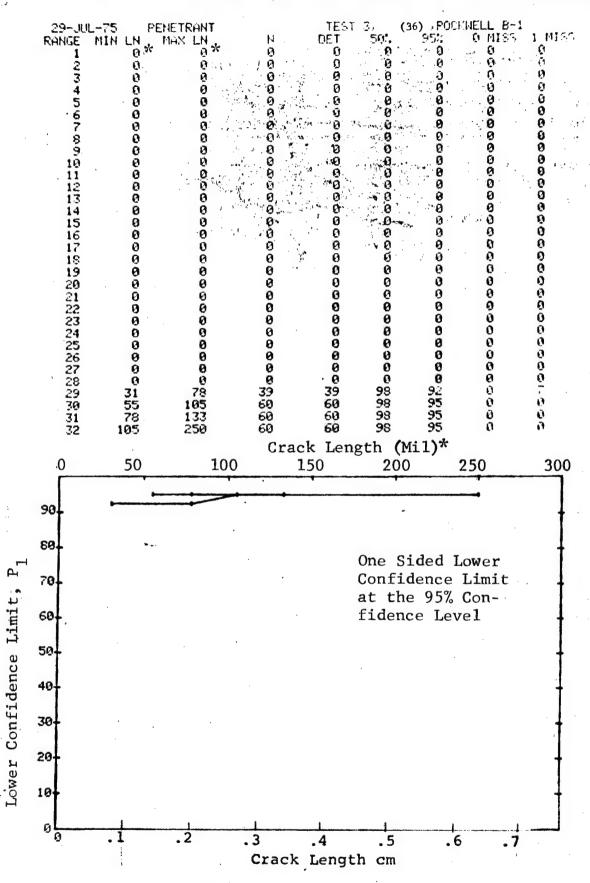


Figure D-36 (Concluded)

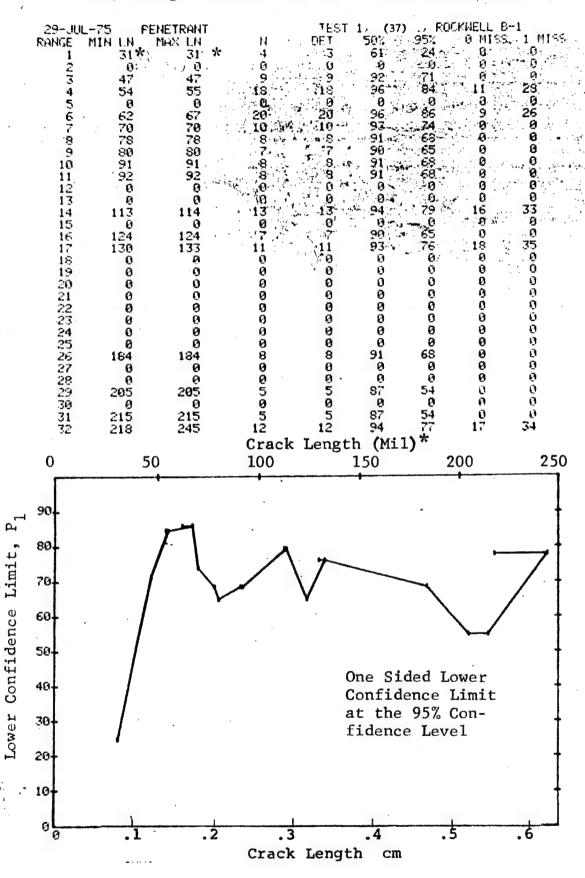


Figure D-37 Probability of Detection for Ti-6A1-4V Using Liquid Penetrant. Fatigue Cracks in Flat Plates.

Prod. Env.

D-115

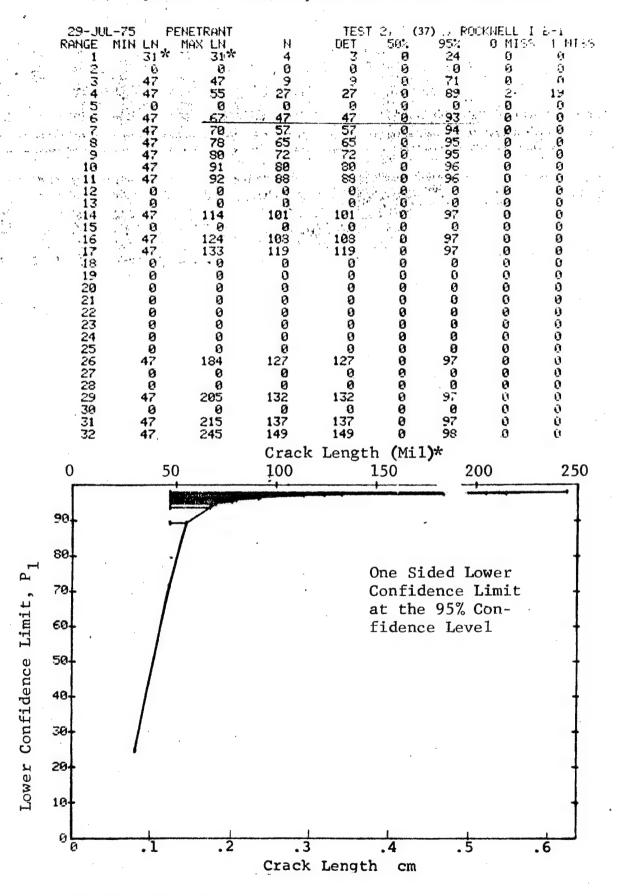


Figure D-37 (Continued)

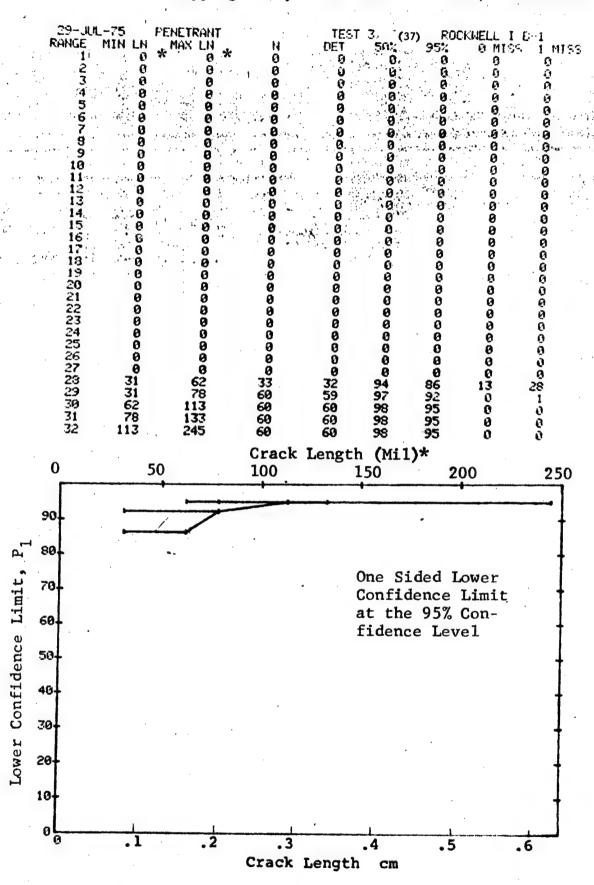
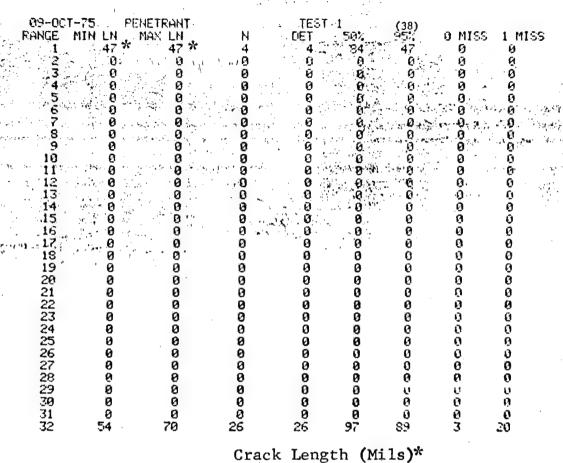


Figure D-37 (Concluded)



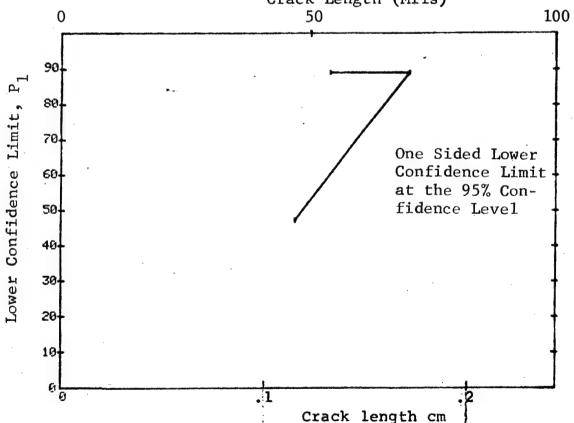
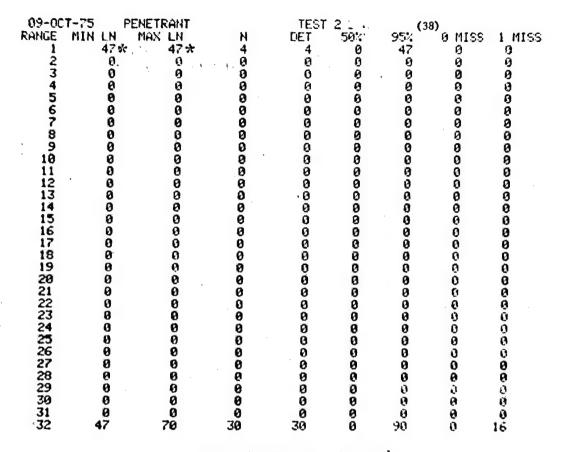


Figure D-38 Probability of Detection for Ti-6A1-4V Using Liquid Penetrant. Fatigue Cracks in Flat Plates. Prod. Env. D-118



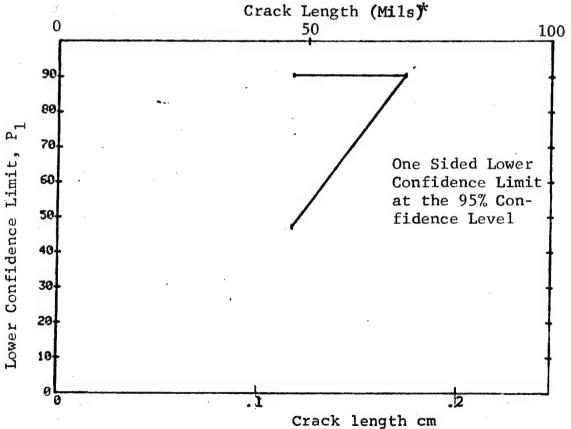
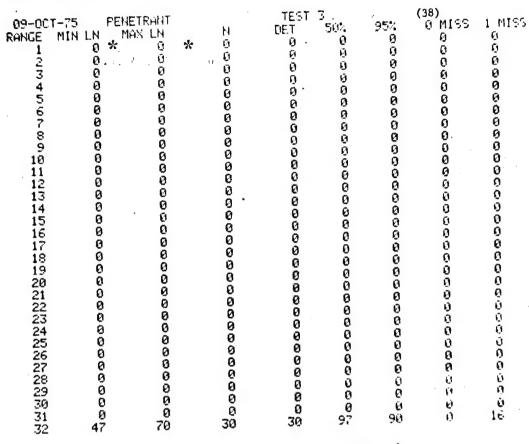


Figure D-38 (Continued)



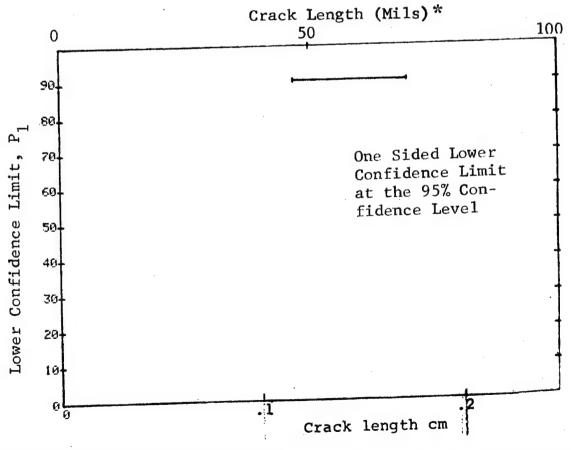


Figure D-38 (Concluded)

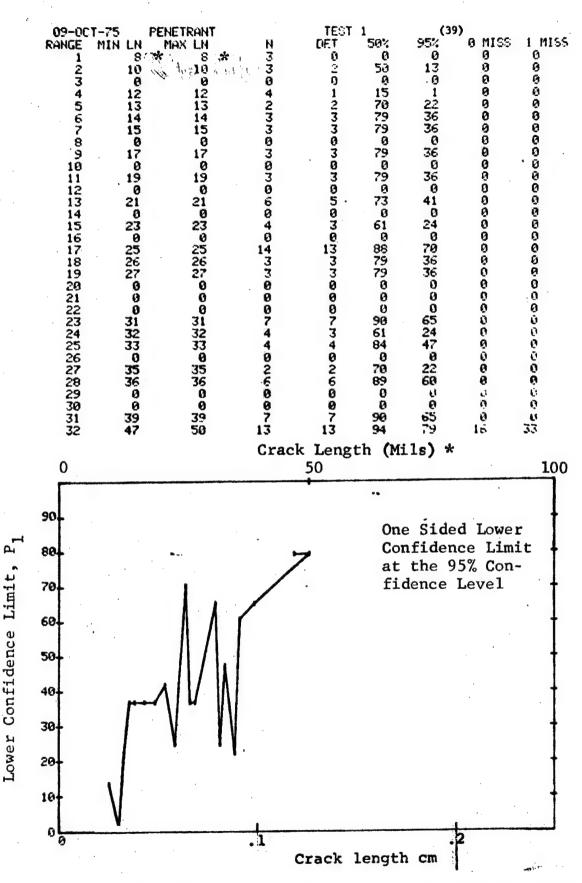


Figure D-39 Probability of Detection for Ti-6Al-4V Using Liquid Penetrant. Fatigue Cracks in Flat Plates. Prod. Env.

D-121

00:00									
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	MIN LH .	MAX LN	н	DET	50%	95%	0 MISS	1 MIS	S
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14	13	0	0	0	Ø	Ø	Ω	õ	
15	13	23	24	22	0	75		32 8 8 8 8	
16	0	9 25 26 27	9	13	9	0	Ø	0	
17	13	25	38	35	0	80 82	38	51 48	
18	13	26	41	38	0	82	35	48	
19	13	. 27	44	41	0	83	32	45	
20	8	9	, 0	ø	9	0	0	0	
21	0	9	0	9	0	8	9	А	
22	0	_0	_0	0	0	0	.0	0	
20 21 22 23 24 25 26 27 28 29	13	0 0 31 32	51	48	Ø	85	25	9 38 48 44	
24	13	32	55	51	0	84	34	48	
20	13	33 0 35	59	55	Ø	85	30	44	
20	.0	_ U	0	0 57	0	0	O	42	
26	13	32	61	57	Ø	85	28	42	
20	13	36	67	63	9	86	22	36	
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31	0	Ø Ø 39	0	9	0	0	ıì	Ü	
32	13 33	50	74	70	9	88	15	23	
36	33	313	32	32	0	91	ŭ	14	

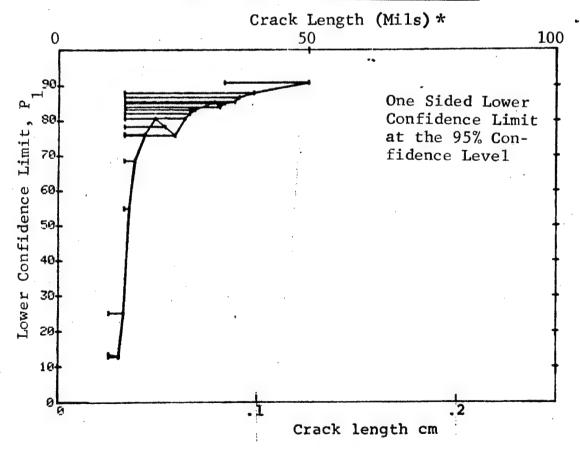


Figure D-39 (Continued)

30 8 25 37 28 74 61 6 6 31 12 33 60 53 87 79 69 82 32 25 50 60 58 95 89 1 16	123456789011234567890112 11234567890112 11234567890112	00000000000000000000000000000000000000	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	888888888888888888888888888888888888888	\$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$	99999999999999999999999999999999999999	79	69	37999999999999999999999999999999999999
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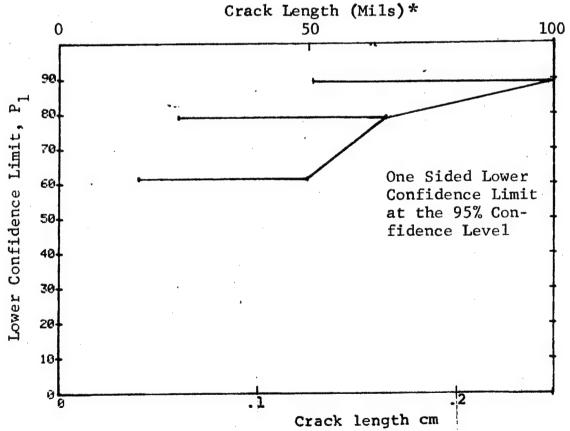


Figure D-39 (Concluded)

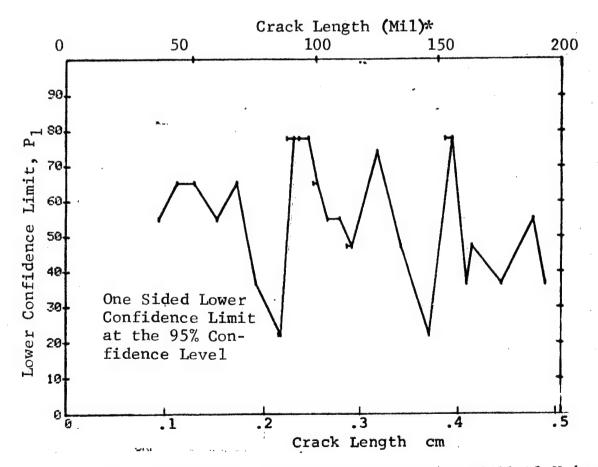


Figure D-40 Probability of Detection for 7075-T6511 Al Using Liquid Penetrant. Fatigue Cracks in Flat Plates.

Prod. Env. D-124

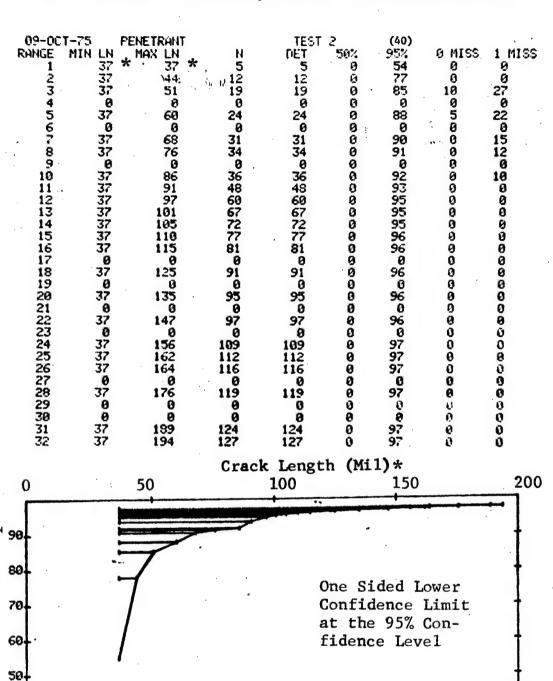


Figure D-40 (Continued)

,1

Confidence Limit

Lower

40

30

20

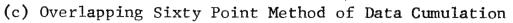
10

0

.2

.3 Crack Length

cm



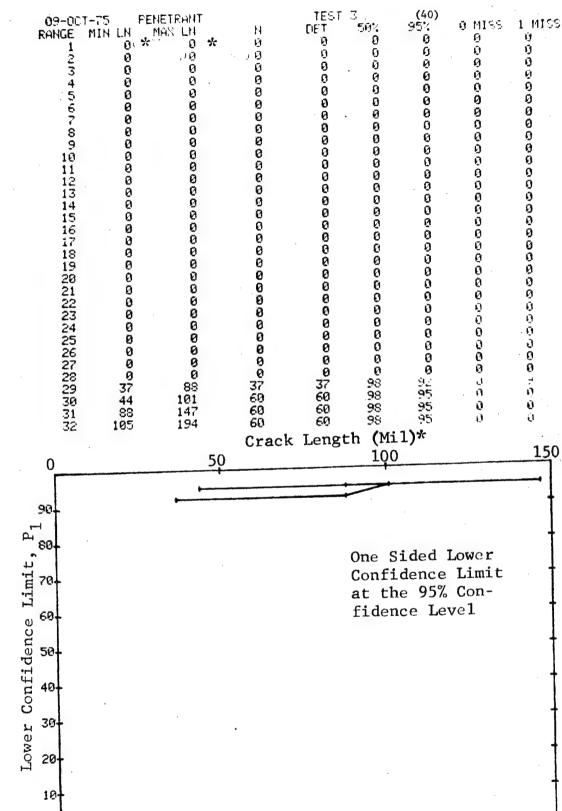


Figure D-40 (Concluded)

Ç

.1

Crack Length

. 3

cm

(a)	Range	Interval	Method	of	Data	Cumulation

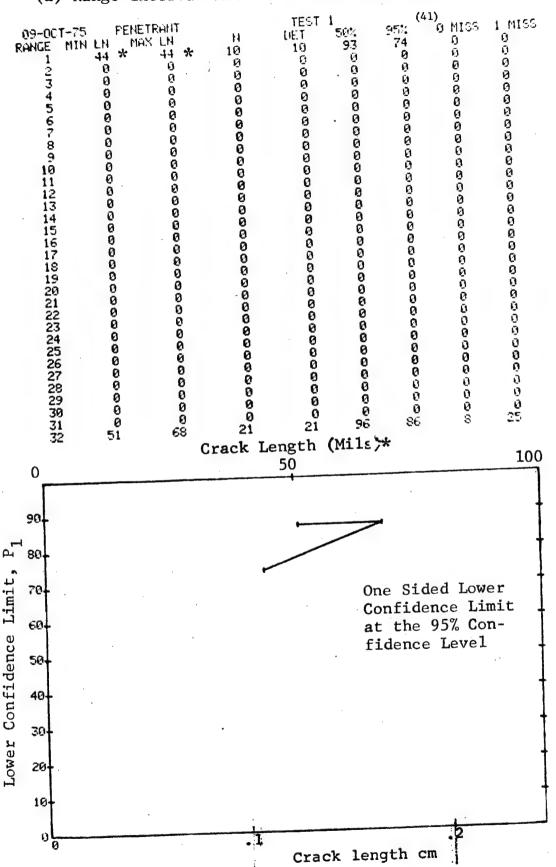


Figure D-41 Probability of Detection for 7075-T6511 Al Using Liquid Penetrant. Fatigue Cracks in Flat Plates.

Prod. Env. D-127

09-00T	-75 PEI	NETRANT		TEST	2	(41)			
RANGE	MIN LN 4476	MAY IN	H ·	DET	50%	95%	0 MISS	1 MISS	
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	0	0	9	Q.	Ø	Ø	Ø	£1	
5	0	0	0	Ø	0	9	9	ឆ្	
4 5 6 7 8 9	0	0	0	0	0	ø	Ø	O	
7	0	0	0	0	0	ø	9	ø	
8	8	0	0	0	0	ß	Ø	0	
9	0	Ø	Ø	0	Ø	8	0	0	
10 11	0	ø	8	0	Ø	Ø	0	<u> 0</u>	
11	0	0	0	0	Ø	ø	0	Ø	
12 13	ø	0 0 0	Ø	ø	0	Ø	Ø	g g	
13	0	0	Ø	ଡ ଡ ଡ	0	0 8	0	<u>ស</u>	
14	8	Ø	0	ø	9	Ø	0	ã	
14 15 16	0	Ø	ø	Ø	Ø	ด	Ø.,	ଡ ଉ ଉ ଉ ତ	
16	9	8	Ø	0	0	0	ହ	ų.	
17	8	Ø	ø.	Ø	0	8	0	U O	
18	0	ଡ	Ø	9	0	9	0	0	
19	Ø	ø	0	9	0	Đ	ପ୍ର	o o	
20	0	9	0	0	Ø	9	0	8	
21	Ø	Ø	0	0	0	0	0	0	
21 22 23	0	Ø	0	0	0	0	0	en G	
23	0	Ø	0	0	0	0	0	ଷ ବ ବ ବ ବ	
24	0	Ø	Ø	0	Ø	0	0	e e	
25	0	ø	0	0	0	0	0	ig O	
26	Ø	0	0	0	0	0	Ö	63	
27	Ø	9	0	0	0	0	0	0	
28	ē	0	0	0	0	0		ล์ อ	
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39	0	9 9	9	0	e e	0	0		
31	.0	Ø	9	.0	Ø	0	6	9	
32	44	68	31	31	9	90	θ	15	

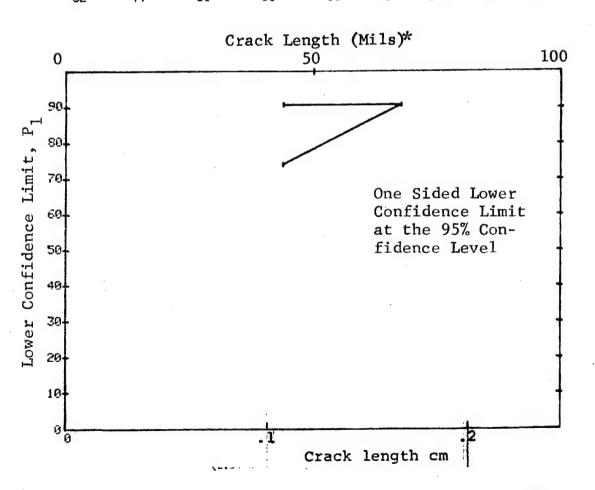
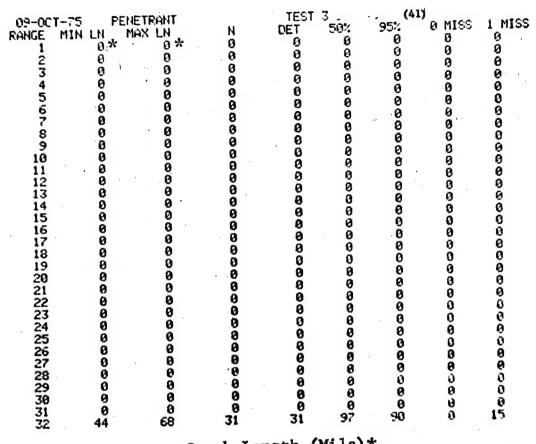


Figure D-41 (Continued)

REPRODUCIBILITY OF THE



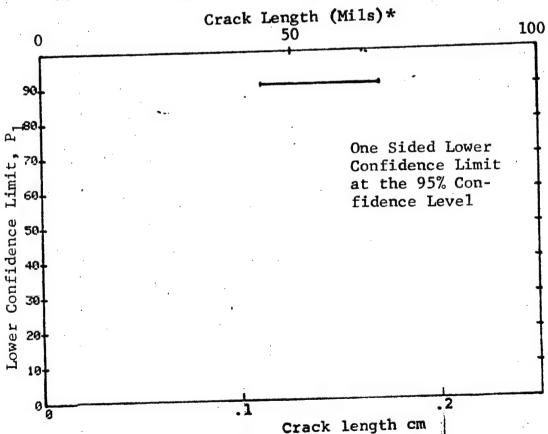


Figure D-41 (Concluded)

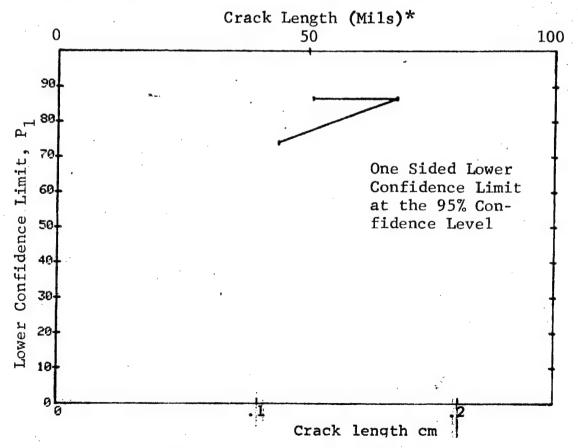


Figure D-42 Probability of Detection for 7075-T6511 Al Using Liquid Penetrant. Fatigue Cracks in Flat Plates.

Prod. Env. D-130

(b) Optimum Probability Method of Data Cumulation

09-0CT-75	TEST 2 50% 10 00 00 00 00 00 00 00 00 00 00 00 00	(42) 9 10 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0 0 0 0
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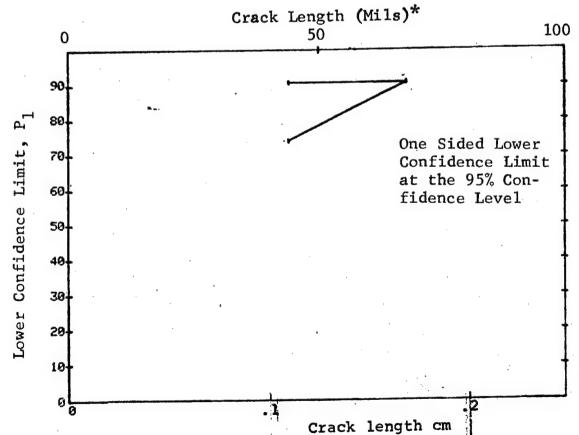


Figure D-42 (Continued)

(0)	Overlapping	Sixty	Point	Method	of	Data	Cumulation
()	•	CACTIONATIVE	DIALLY	LOTTIC	riceriou	OI	Data	Odinaracion

09-001-75 PEHETRAHT RANGE MIN LN MAX LH H	TEST 3	95% 0 MISS	1 MISS
RANGE MIN LN MAX LH N	DET 50%	9 9	9
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8 0 0 0 9 0 0 0	. 0	9	ø
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	0 0 0 0	9 9 9 9	0 0
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17 0 0 0	0 0	0 0	
18 0 0 0 - 19 0 0 0 20 0 0	0 0 0 0	9 9 9 9	0 0
20 0 0 0	0 0	0 0	ឲ្
21 0 0 0	0 0	0 0 0 0	9 9
22 0 0 0 23 0 0 0	0 0 0 0	9 9	ő
24 0 0 0	0 0		0
24 0 0 0 25 0 0 0	0 0	0 0	0
26 0 0 0 27 0 0 0	0 0	6 6	ŏ
27	0 0	9 9	0
29 0 0 0	0 0 0 0	0 0 0 0	0
21	0 0	8 8	15
31 0 0 0 32 44 68 31	31 97	90 0	15

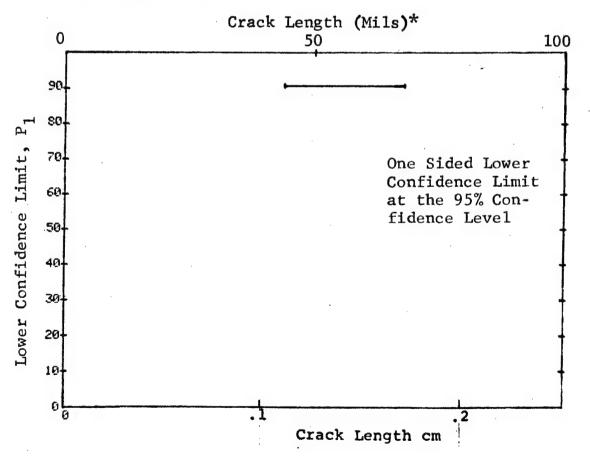
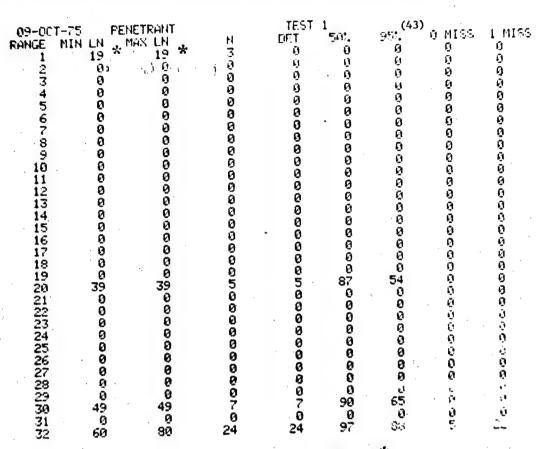


Figure D-42 (Concluded)



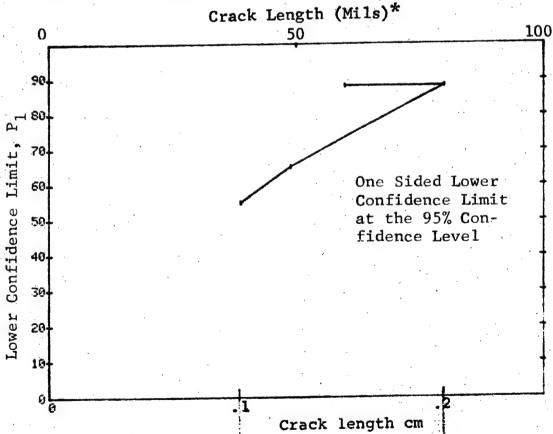


Figure D-43 Probability of Detection for PH13-8 Mo. St. Using Liquid Penetrant. Fatigue Cracks in Flat Plates. Prod. Env. D-133

(b) Optimum Probability Method of Data Cumulation

RANGE MIN LN MA 1 19 * 2 3 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	19 * 300000000000000000000000000000000000	TEST 50% 99999999999999999999999999999999999	© 0 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5	0 MISS 1 MISS 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
30 39 31 0 32 39				

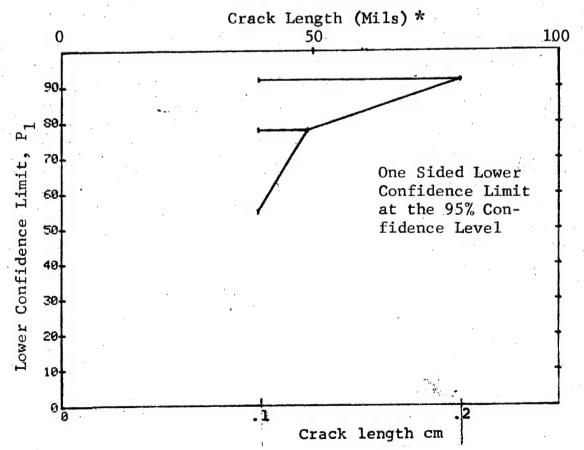


Figure D-43 (Continued)

09-0CT-75	TET 00000000000000000000000000000000000	% 95% 95% 95% 95% 95% 95% 95% 95% 95% 95	(43) © MG B G G G G G G G G G G G G G G G G G	1 MISS 00000000000000000000000000000000000
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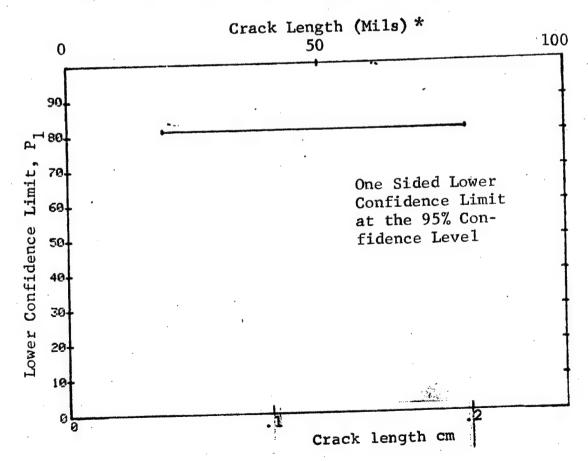


Figure D-43 (Concluded)

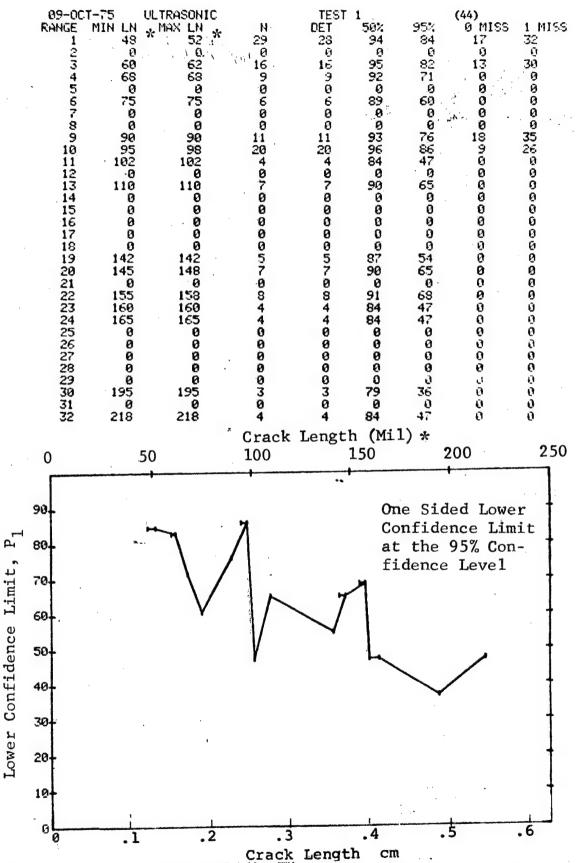
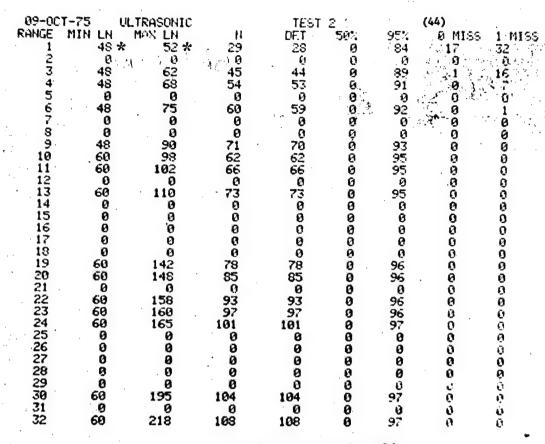


Figure D-44 Probability of Detection for Ti-6Al-4V Using Ultrasonic Shear Wave. Fatigue Cracks in Welded Flat Corroded Plates. Prod. Env.

졟



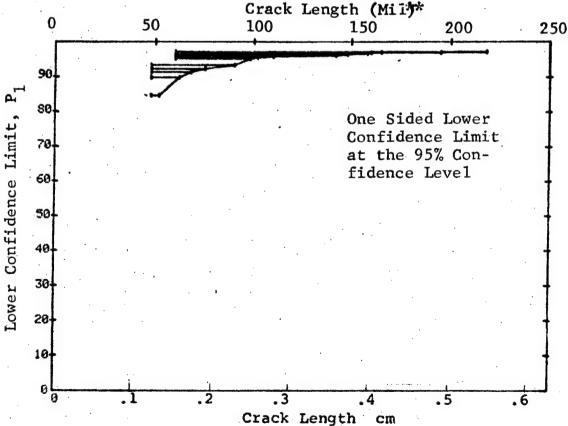


Figure D-44 (Continued)

(c)	Overlapping	Sixty	Point	Method	of	Data	Cumulation
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RANGE MIN LN * 000000000000000000000000000000000000	TRASONIC ** 1A	^{ମ ନ୍} ତ୍ରତ୍ତର୍ଭର ବର୍ଷ ବର୍ଷ ବର୍ଷ ବର୍ଷ ବର୍ଷ ବର୍ଷ ବର୍ଷ ବର୍	TEST 00000000000000000000000000000000000	ා ලා ඉහැරි	න ය අත ය ගත ය ගත ය ය ය ය ය ය ය ය ය ය ය ය ය ය	େ ଅପ୍ରତ୍ରତ୍ତ୍ରତ୍ତ୍ରତ୍ତ୍ରତ୍ତ୍ରତ୍ତ୍ରତ୍ତ୍ରତ୍ତ	1 ଓ ପ୍ରତ୍ର ପ୍ରତ୍ର ପ୍ରତ୍ର ପ୍ରତ୍ର ପ୍ରତ୍ର ପ୍ରତ୍ର କଳ ଓ ସହର ପ୍ରତ୍ର କଳ ଓ ସହର ପ୍ରତ୍ର ପ୍ରତ୍ର ପ୍ରତ୍ର ପ୍ରତ୍ର ଓ ସହର ପ୍ରତ୍ର ପ୍ରତ୍ର ଓ ସହର ପ୍ରତ୍ର ପ୍ରତ୍ର ଓ ସହର ପ୍ରତ୍ମ ଓ ସହର ପ୍ରତ୍ର ଓ ସହର ପର ପ୍ରତ୍ର ଓ ସହର ପର ପର ପ୍ରତ୍ର ଓ ସହର ପ୍ରତ୍ର ଓ ସହର ପ୍ରତ୍ର ଓ ସହର ପ୍ରତ୍ର ଓ ସହର ପର
30 50 31 68 32 95	95 142 218	69 69 69	60 60 60	98 98 98	95 95 95	-	

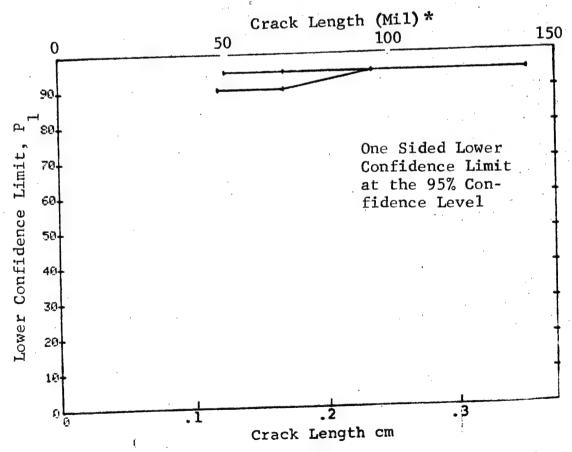


Figure D-44 (Concluded)

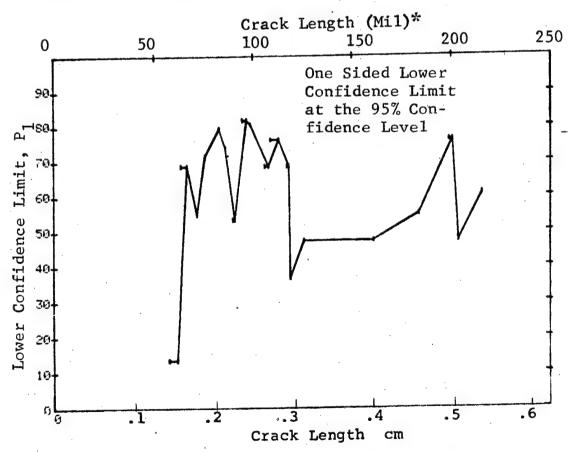


Figure D-45 Probability of Detection for 4330V Steel Using Ultrasonic Shear Wave. Simulated Weld Flaws Prod. Env. D-139

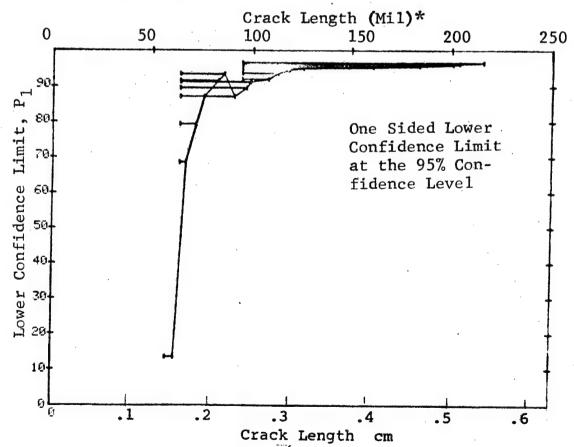
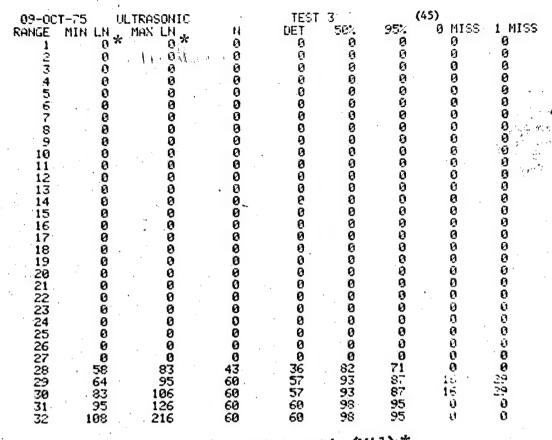


Figure D-45 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation



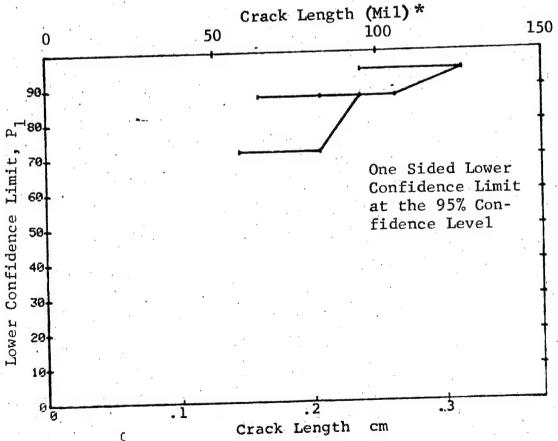


Figure D-45 Concluded)

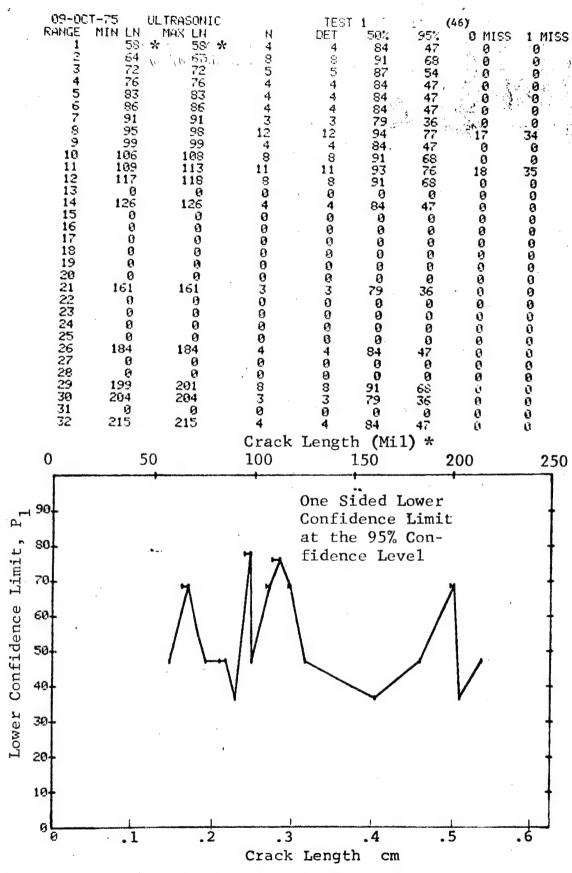


Figure D-46 Probability of Detection for PH17-4 Steel Using Ultrasonic Shear Wave. Simulated Flaws in Wrought Steel. Prod. Env.

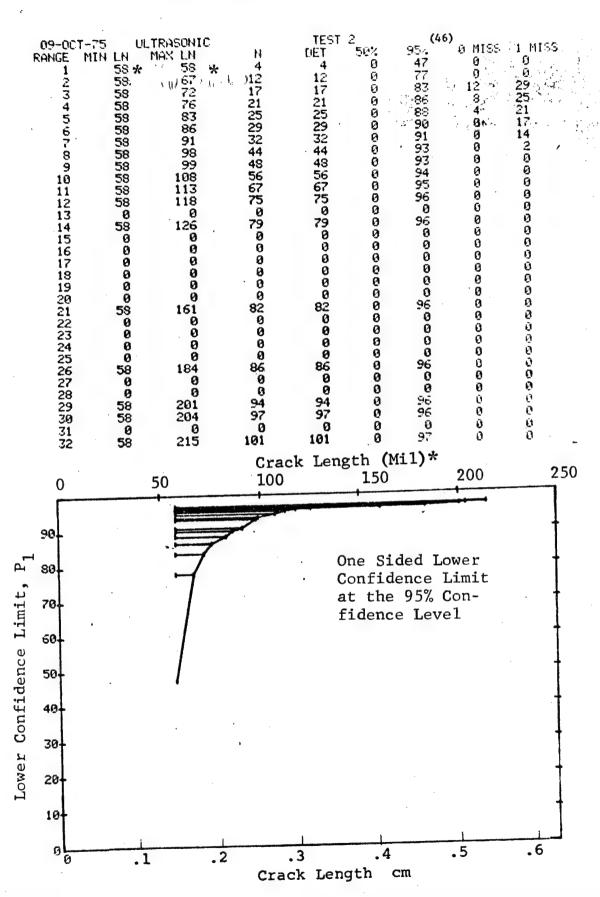


Figure D-46 (Continued)

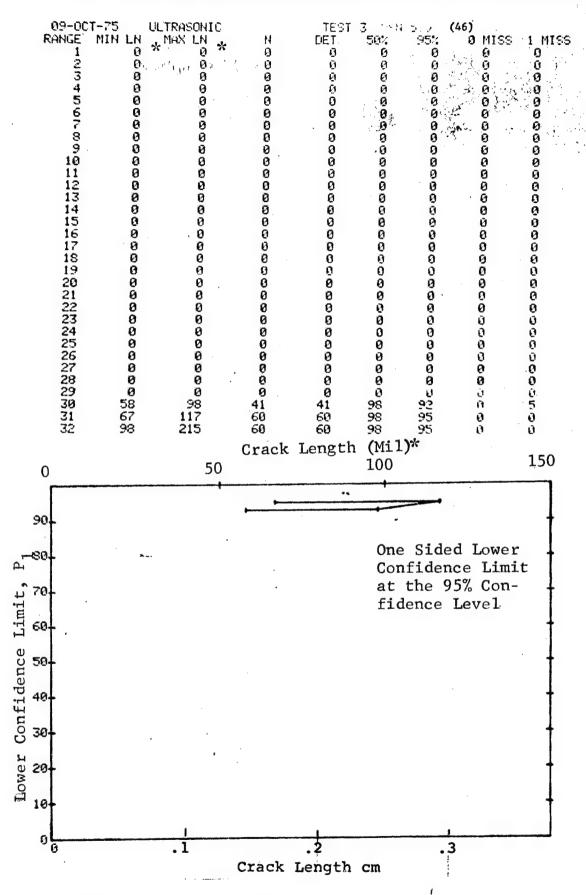


Figure D-46 (Concluded)

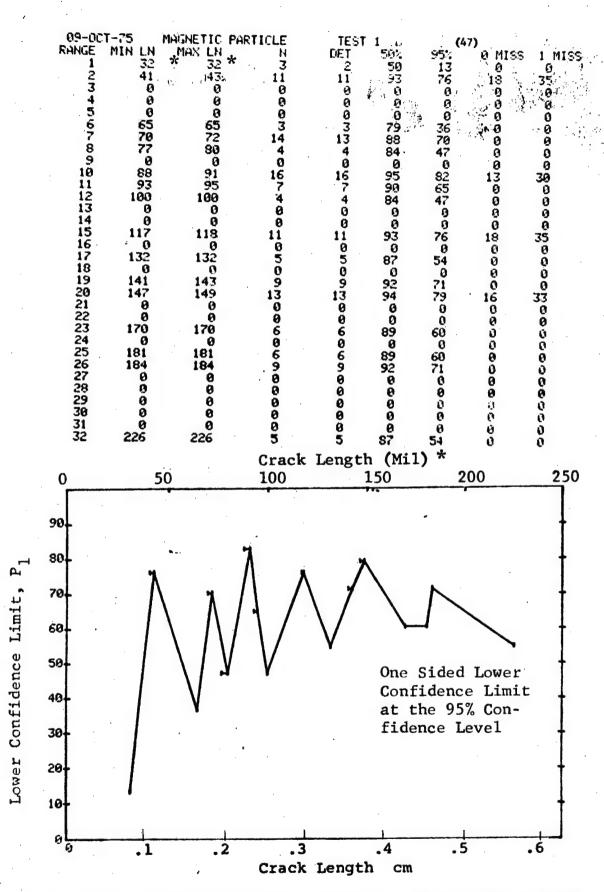


Figure D-47 Probability of Detection for PH17-4 Steel Using Ultrasonic Shear Wave. Fatigue Cracks in Flat Plates. Prod. Env. D-145

DEPRODUCIBILITY OF THE

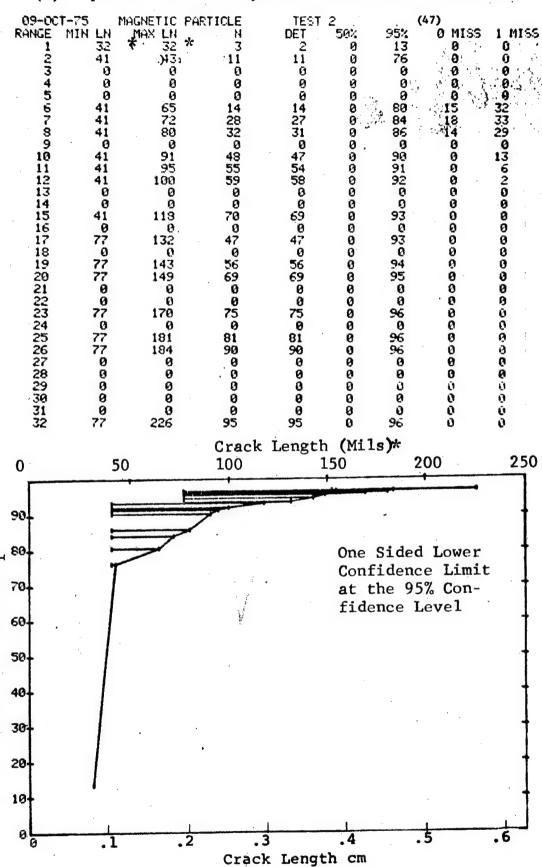


Figure D-47 (Continued)

Lower Confidence Limit,

09-00 RANGE 123456789 10112314 1561789 10112314 1561789 2012234 226278 22728 230331 32	T-75 N ** N *	SETIC PA MAX LN * , 00000000000000000000000000000000000	RT 1CLE N 3 3 5 5 5 5 5 6 6 6 6 6 6 6 6 6 6 6 6 6	TET 9 0 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9	% 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	$\frac{1}{2} \cdot C \cdot D \cdot C \cdot D \cdot D \cdot D \cdot D \cdot D \cdot D \cdot D$	MI	1 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
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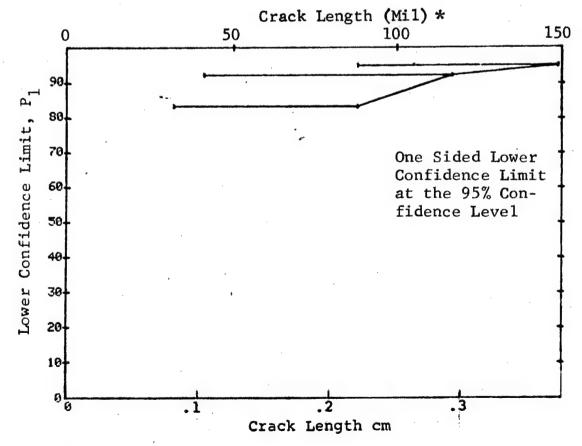


Figure D-47 (Concluded)

10-00T RANGE		HETRHUT MAG IN	- ti	TEST OF T	1 (MERG	E ALL ALI	MUNUM) ((48) MISS
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3	70	100	5.37 204		ବର ବର	89	Ö	Ö,
4	101	131	281	260 203	92 87	83	Ö	õ
5	132	162	231	45	94	97 97	14	29
6	164	190	47	7	71	45	.0	<u>.</u>
15 '61- 00 o	194	197	9 70	69	97	93	ĕ	Õ
8	241	253		119	99	97	้ ดี	õ
	256	283	119 112	109	22 96	93	õ	ĕ
10	287	317 347	175	165	93 93	90	õ	ĕ
11	318	247	70	63	89	82	59	72
12	352 381	372 408	28	27	94	.34	13	33
13		426	7	-1	90	€5	ė	ő
14	426	420	70	67	94	89 89	6	19
15	442	472 503	154	149	96	93	õ	.
16	474	534	182	180	98 98	96	õ	ĕ
17	504	559	49	47	94	87 87	12	27
18	5 35	335	77	77	90	65	• 6	៊ីម៉
19	568	568	<u>(</u>	ŕ	90 90	65	õ	ő
20	610	610	(é	9	0	õ	ő,
21	Ø	9	ø		0	õ	õ	ő
22 23	710	Ø	9 7	Ø 7	90	65	õ	ยั
23	710	710		Ó	90	8	õ	õ
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26	0	0	0	0	ø	õ	ĕ	õ
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31	9	9	0	õ	Ö	Ü	i u	وا
32	979	979	7	2	લાફે	65	E.	<u>و</u> با
32	215	212		'	#15.1		•••	<u>~</u>

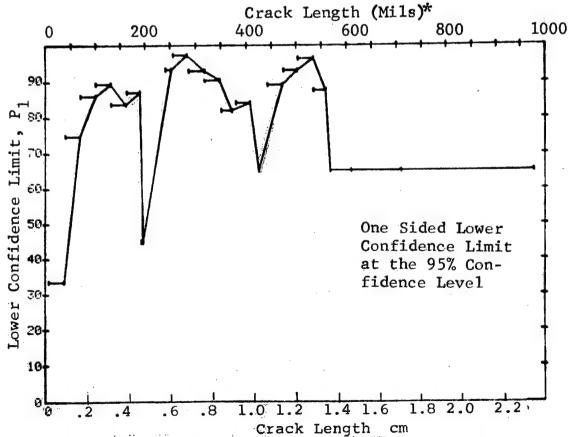
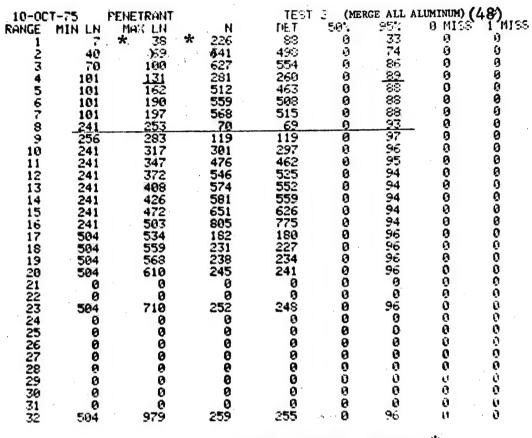


Figure D-48 Probability of Detection for Aluminum Using Liquid Penetrant Fatigue Cracks in Flat Plates.

Prod. & Lab. Env.

D-148



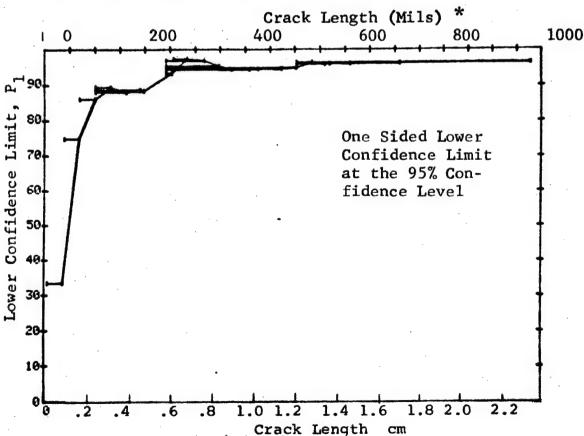


Figure D-48 (Continued)

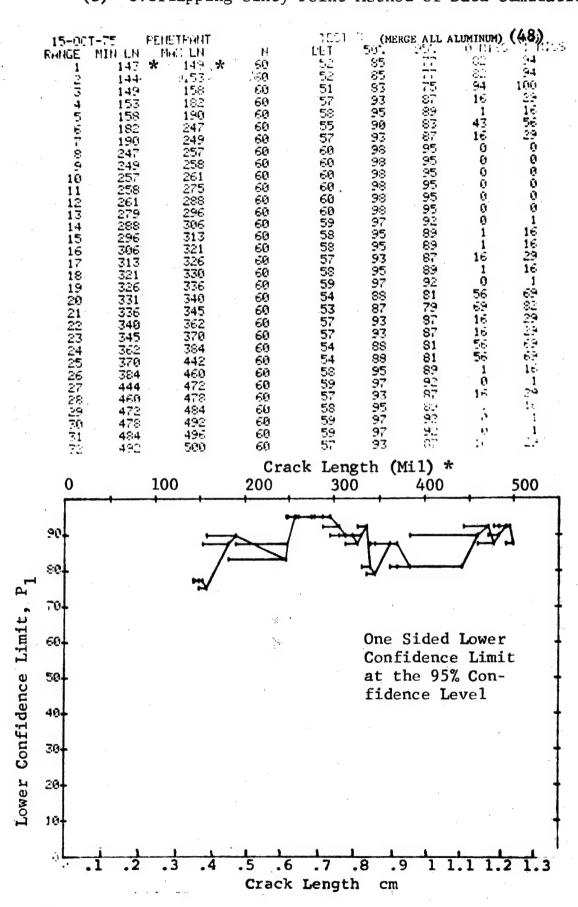


Figure D-48 (Continued)

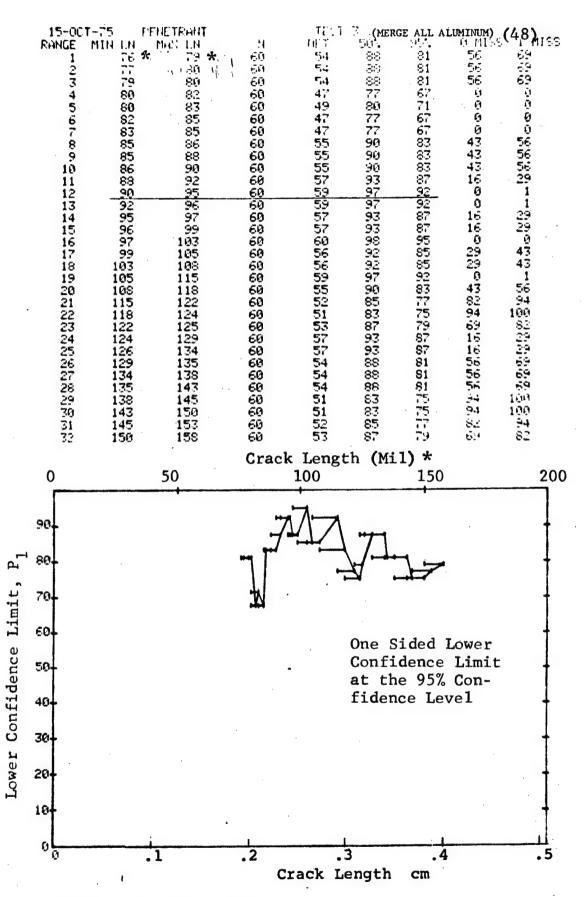


Figure D-48 (Continued)

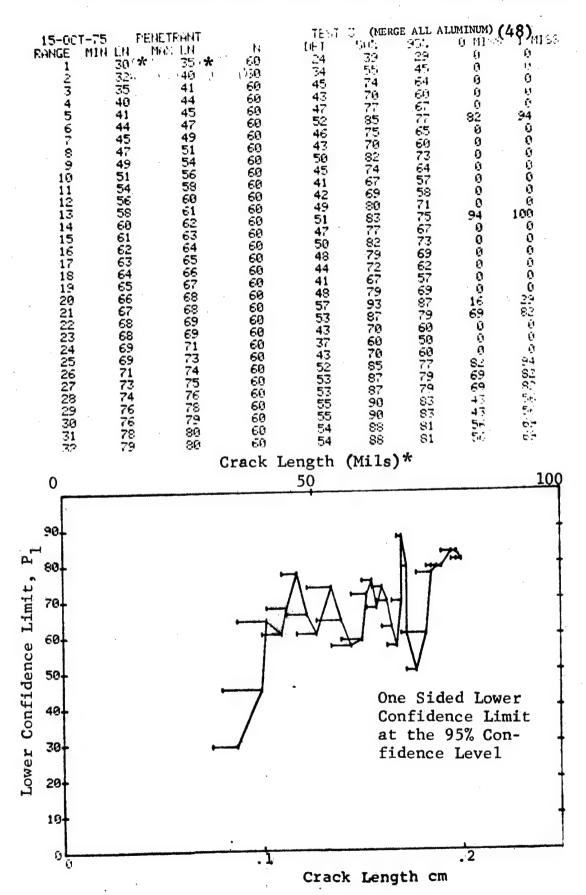
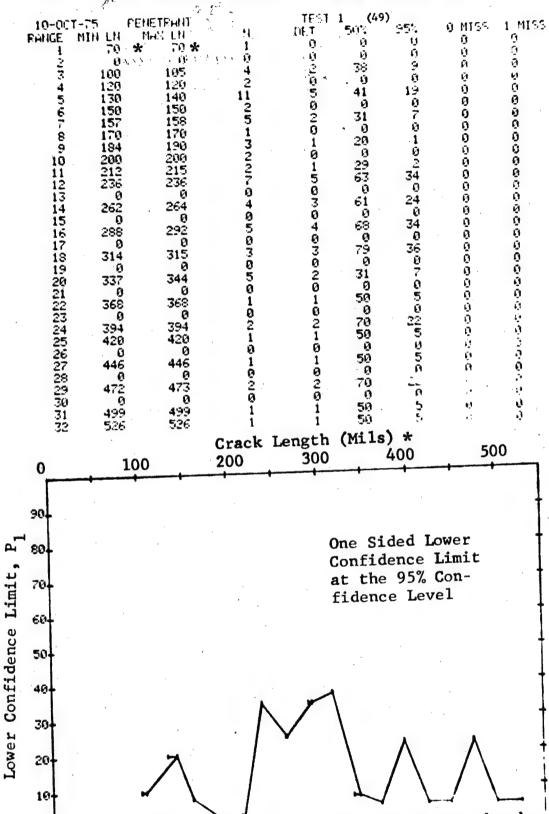


Figure D-48 (Concluded)



Crack Length Probability of Detection for 4330V Steel Using Figure D-49 Fatigue Cracks in Cylindrical Liquid Penetrant. Shell Specimens. Lab. Env.

6

cm

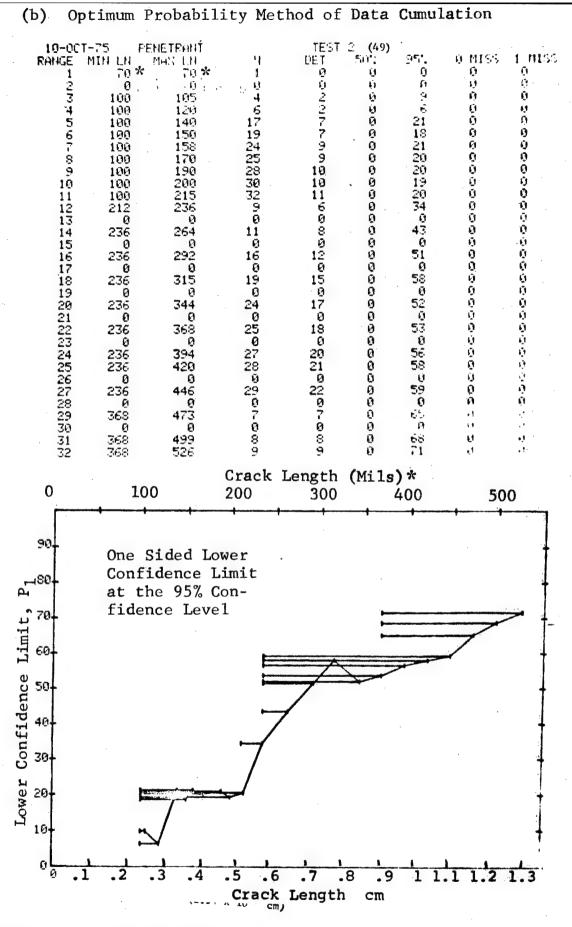


Figure D-49 (Continued)

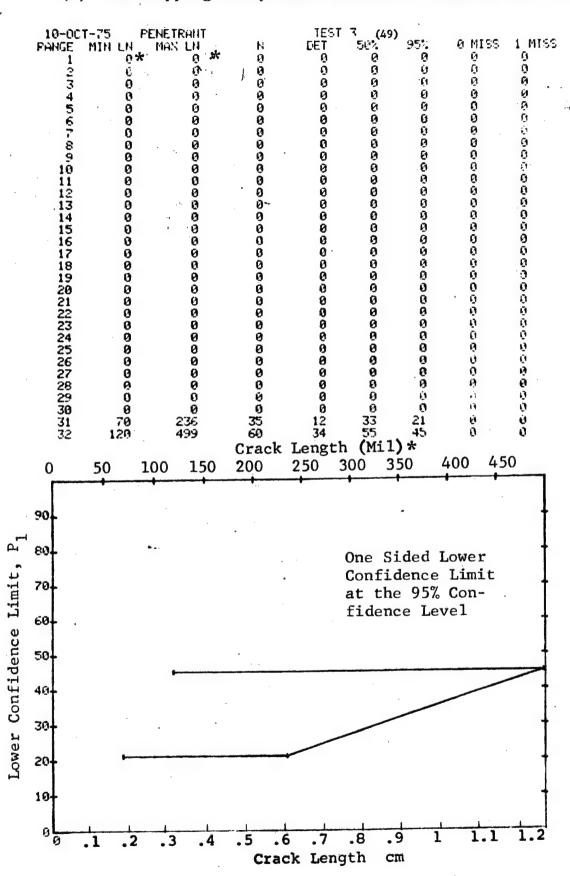


Figure D-49 (Concluded)

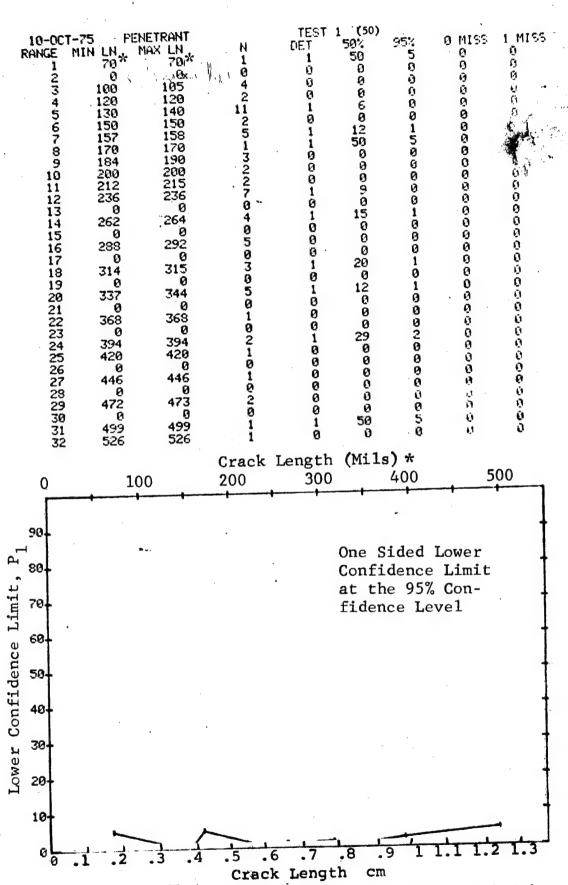


Figure D-50 Probability of Detection for 4330V Steel Using Magniflux ZL-2. Fatigue Cracks in Cylindrical Shell Specimens. Prod. Env.

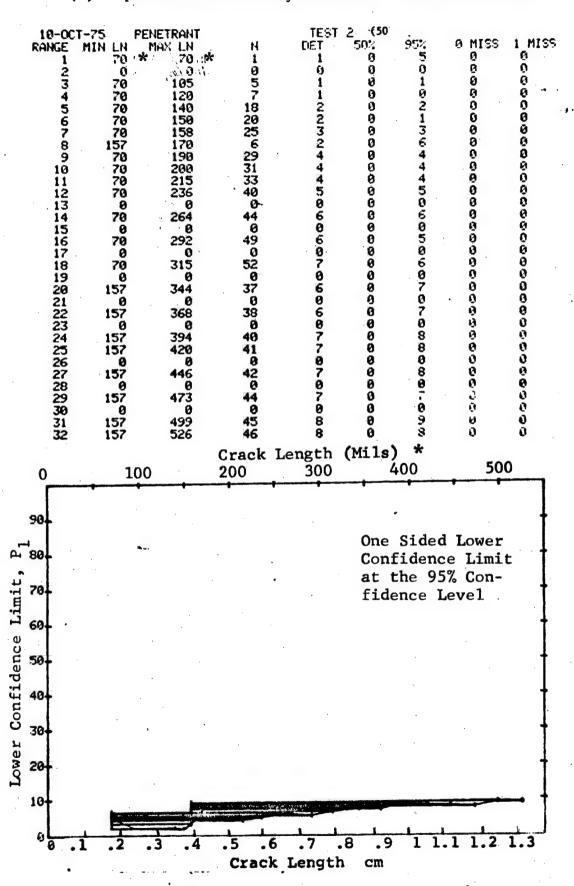


Figure D-50 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

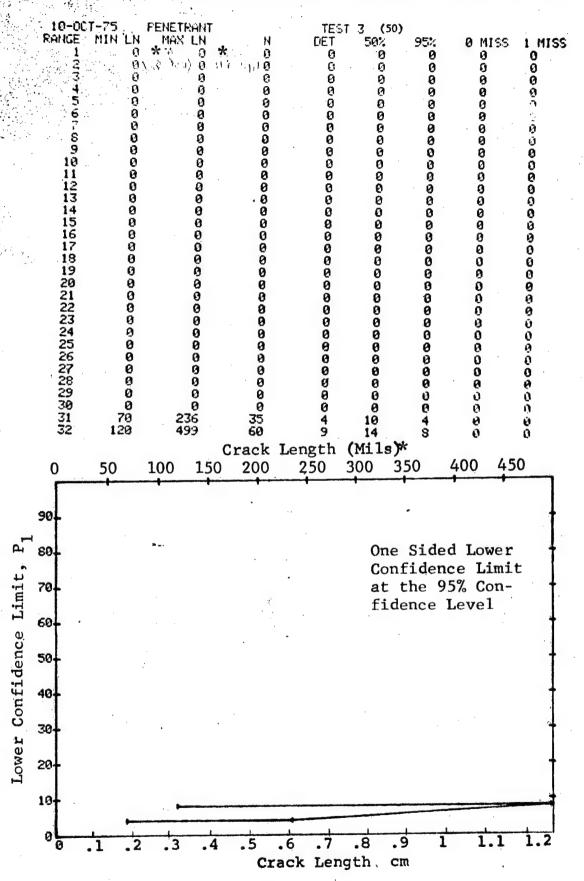


Figure D-50 (Concluded)

F .	10-00T-77 NGE 011 123 45 67 8 9 10 112 134 156 17 18 19 20 122 223 225 229 331 32	H LM MA 52차	92 193 193 193 193 193 193 193 193 193 193	N45555502632232232203333312022010100021 Lei	TEST DET 113402322013222010100021 h	1 50°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 120°:000 12	(51) (51) (51) (51) (51) (51) (51) (51) (51) (51) (52) (52) (53) (53) (53) (53) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54) (54)	6 MG G G G G G G G G G G G G G G G G G G	1 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
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Using Magniflux ZL-2. Fatigue Cracks in Cylindrical Shell Specimens. Lab. Env.

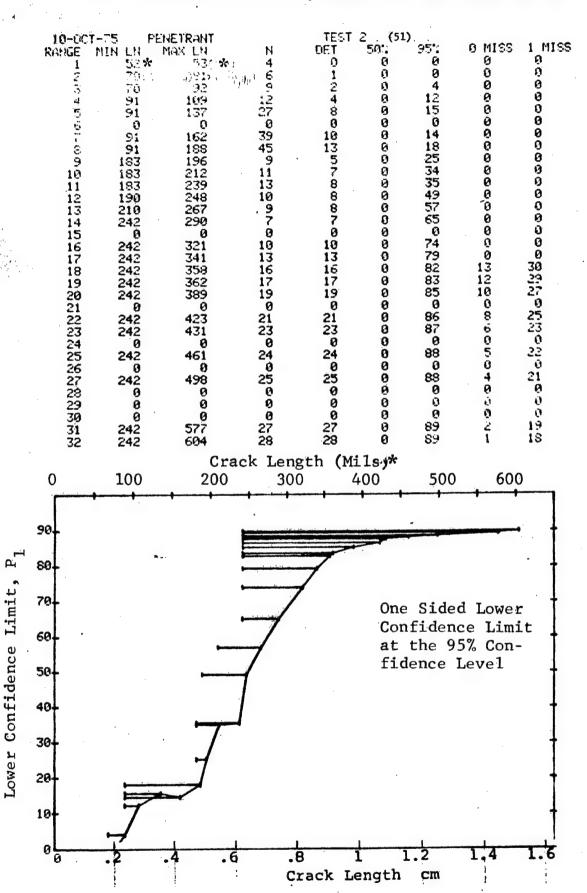


Figure D-51 (Continued)

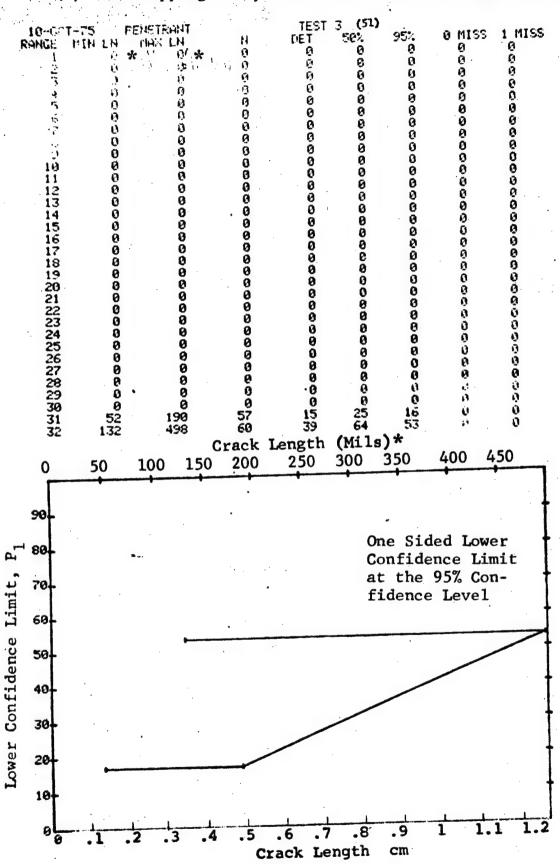


Figure D-51 (Concluded)

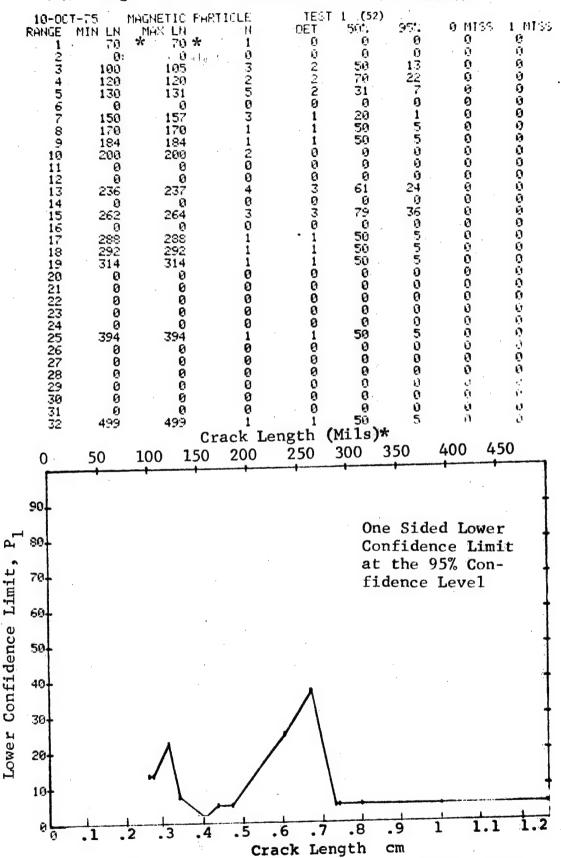
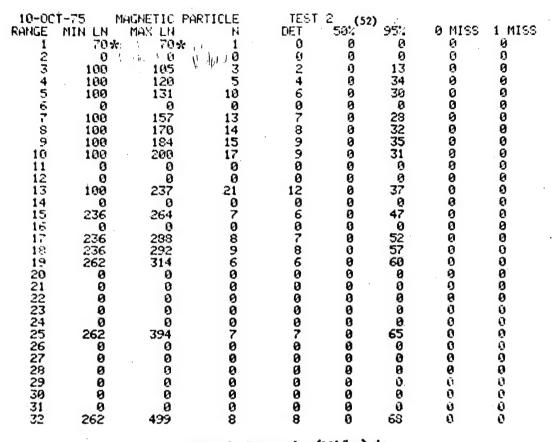


Figure D-52 Probability of Detection for 4330V Steel
Using Magnetical Particle. Fatigue Cracks
in Cylindrical Shell Specimens. Lab. Env.



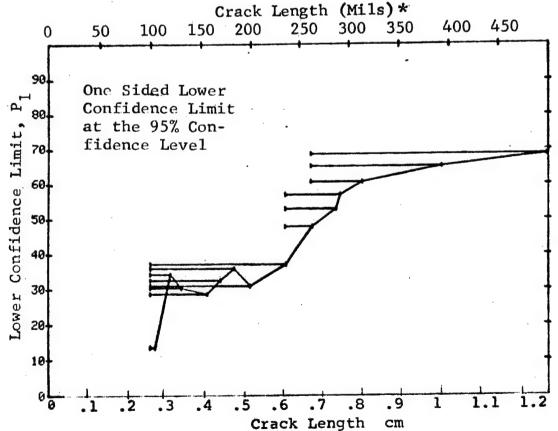


Figure D-52 (Continued)

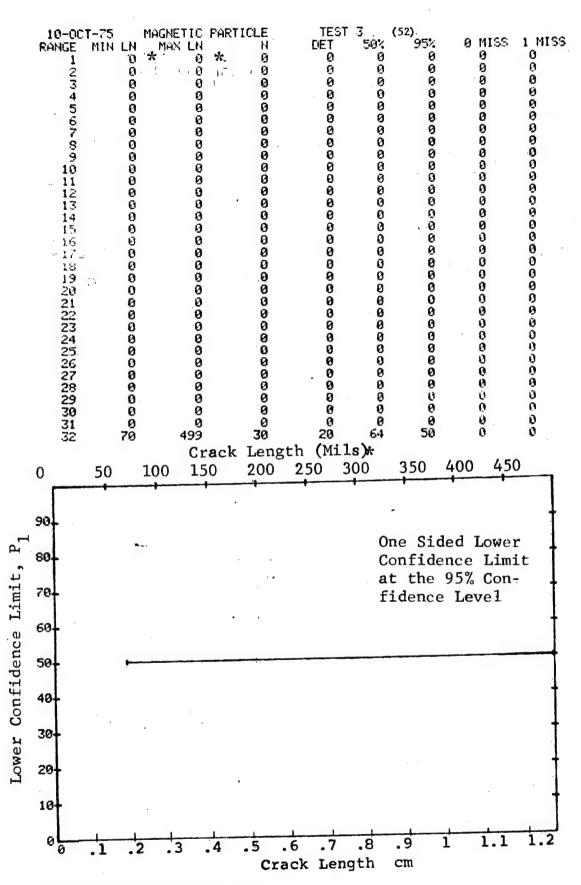


Figure D-52 (Concluded)

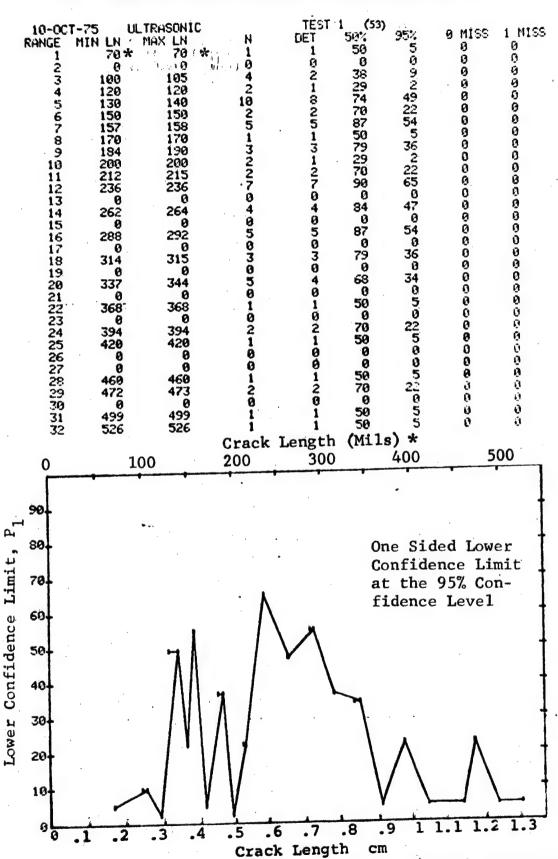


Figure D-53 Probability of Detection for 4330V Steel Using Ultrasonic Shear Wave. Fatigue Cracks in Cylindrical Shell Specimens. Lab. Env. D-165

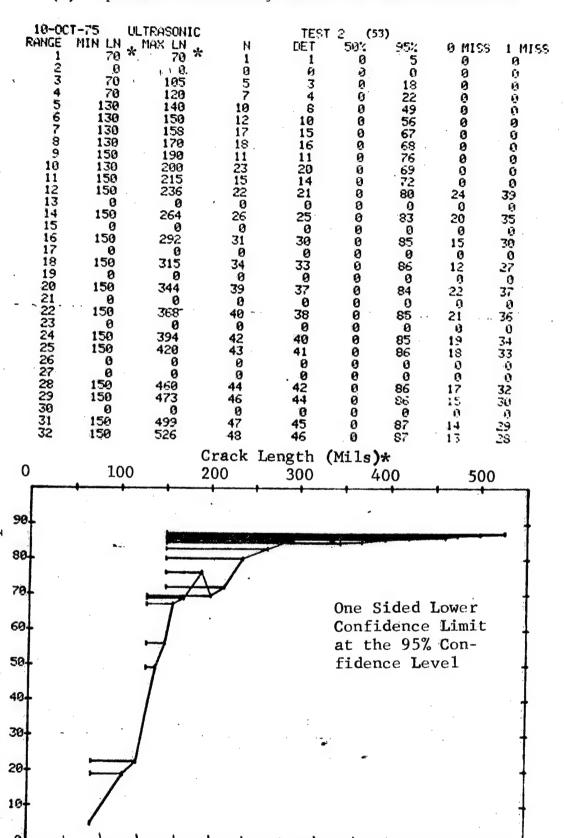


Figure D-53 (Continuea)

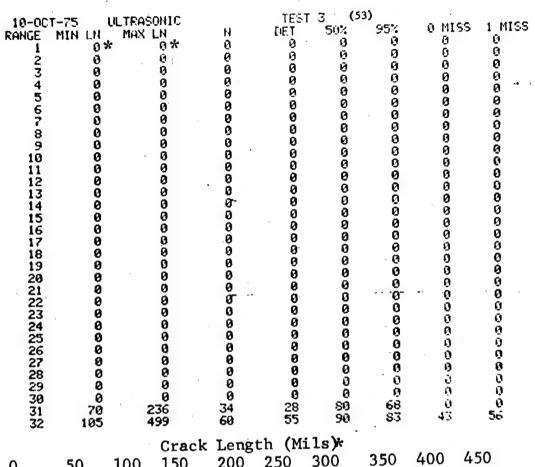
Lower Confidence Limit, P,

Crack Length cm

.8

. 5

.6



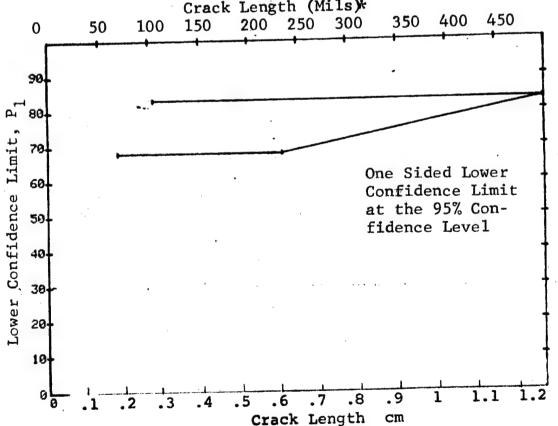


Figure D-53 (Concluded)

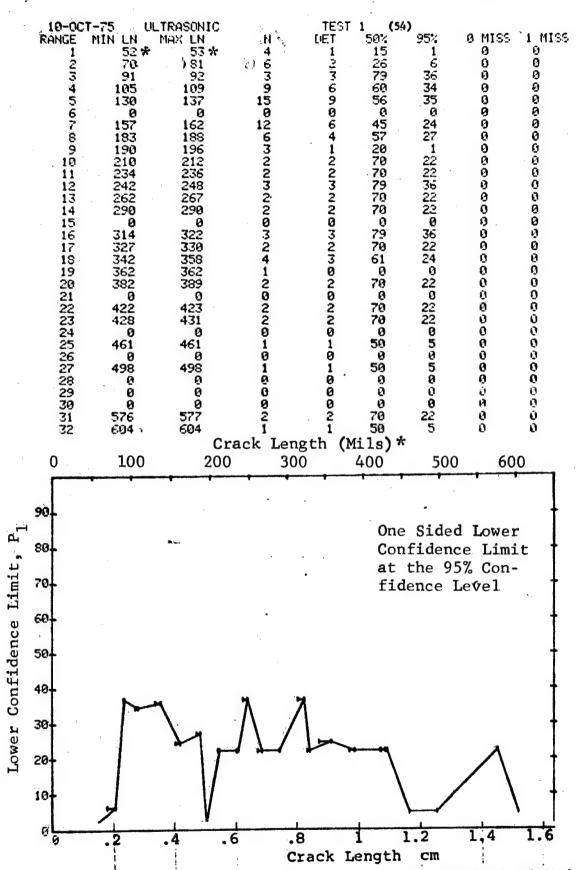


Figure D-54 Probability of Detection for 7075-T6511 Al Using Ultrasonic Shear Wave. Fatigue Cracks in Cylindrical Shell Specimens. Lab. Env..

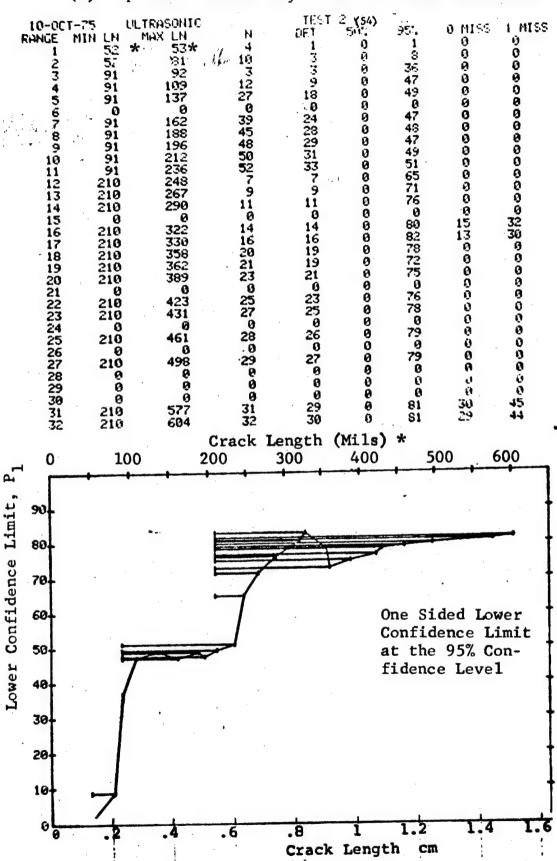


Figure D-54 (Continued)

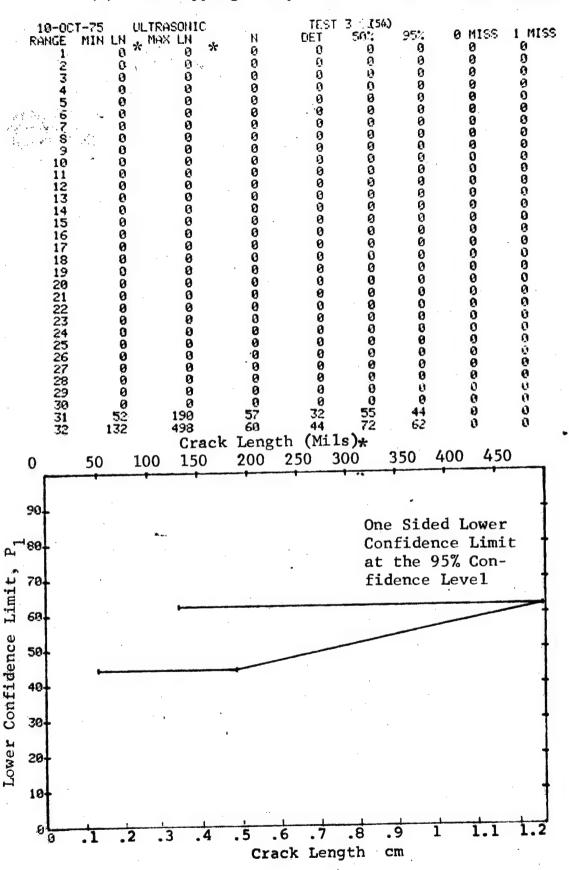


Figure D-54 (Concluded)

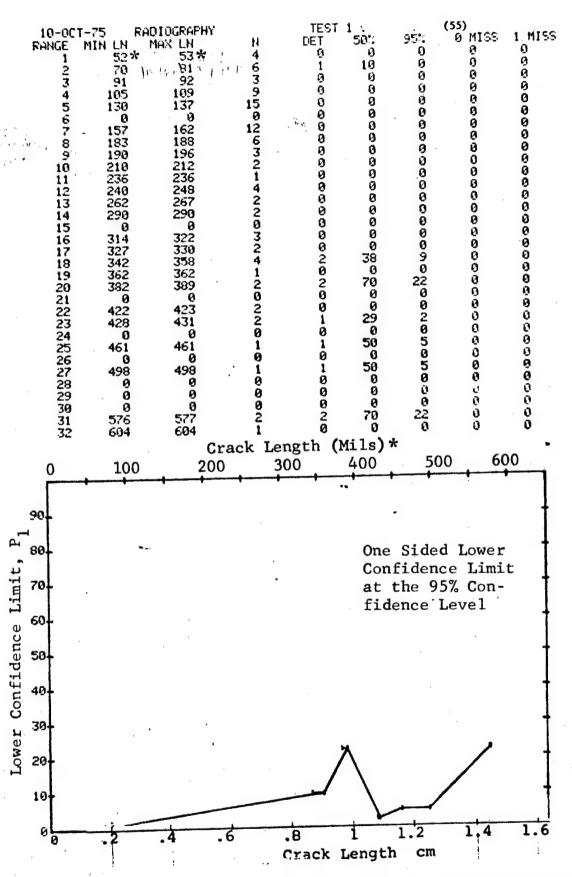


Figure D-55 Probability of Detection for 7075-T6511 Al Using X-ray. Fatigue Cracks in Cylindrical Shell. Lab. Env.

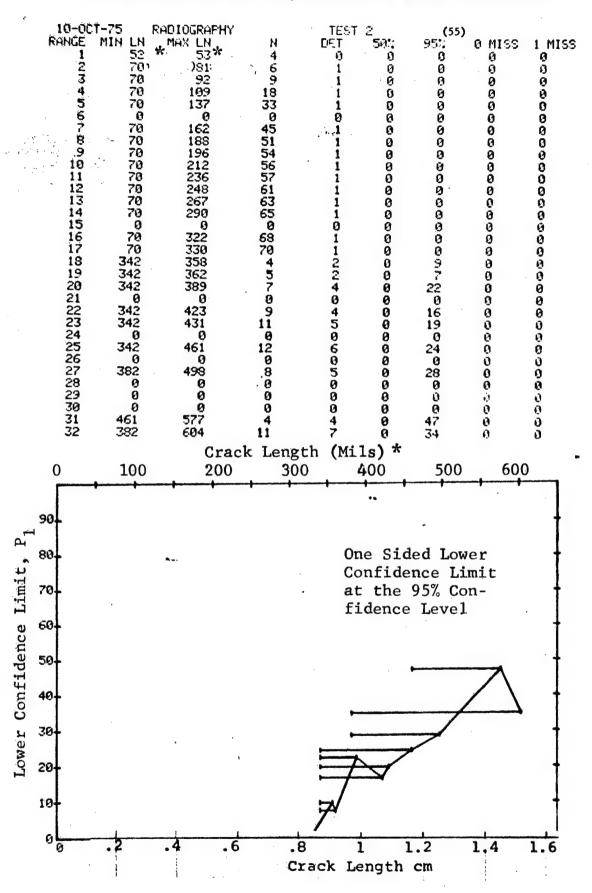


Figure D-55 (Continued)

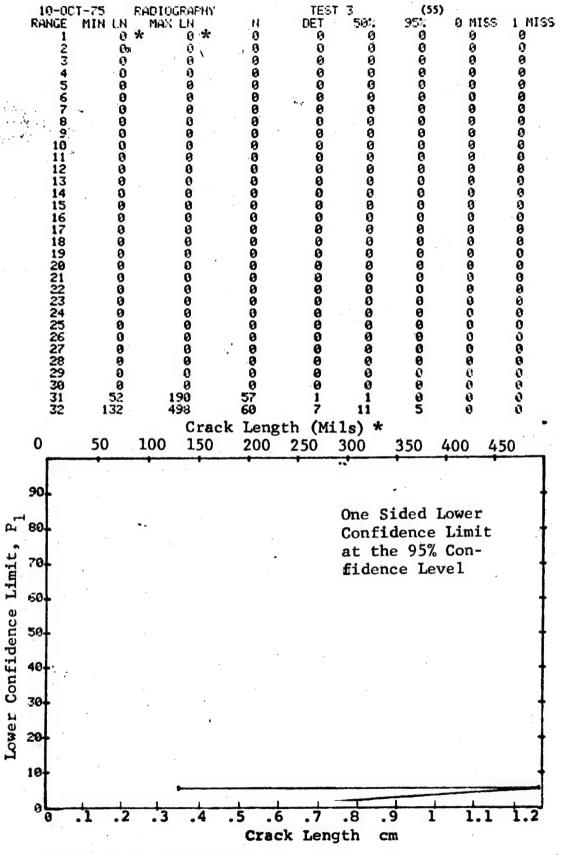


Figure D-55 (Concluded)

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR

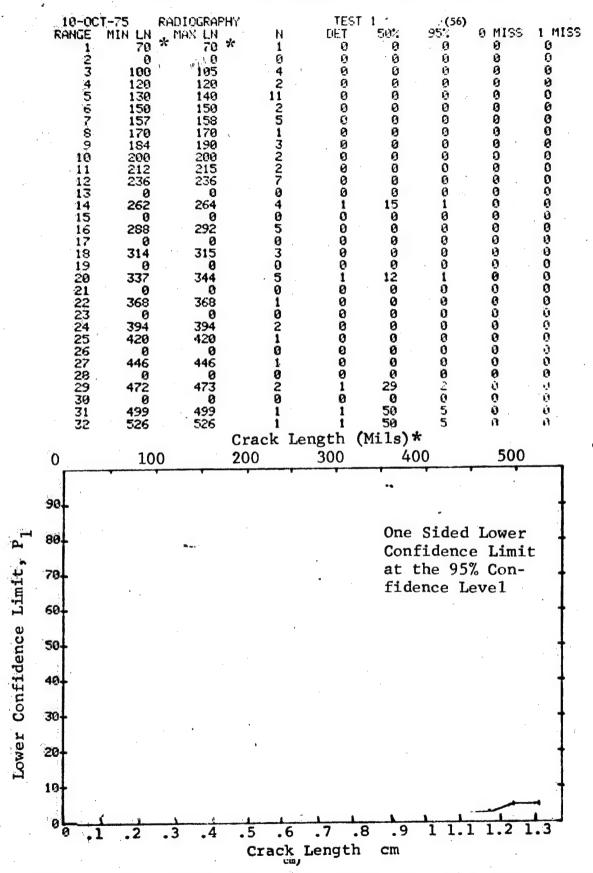


Figure D-56 Probability of Detection for 4330V Steel Using X-ray. Fatigue Cracks in Cylindrical Shell. Lab. Env.

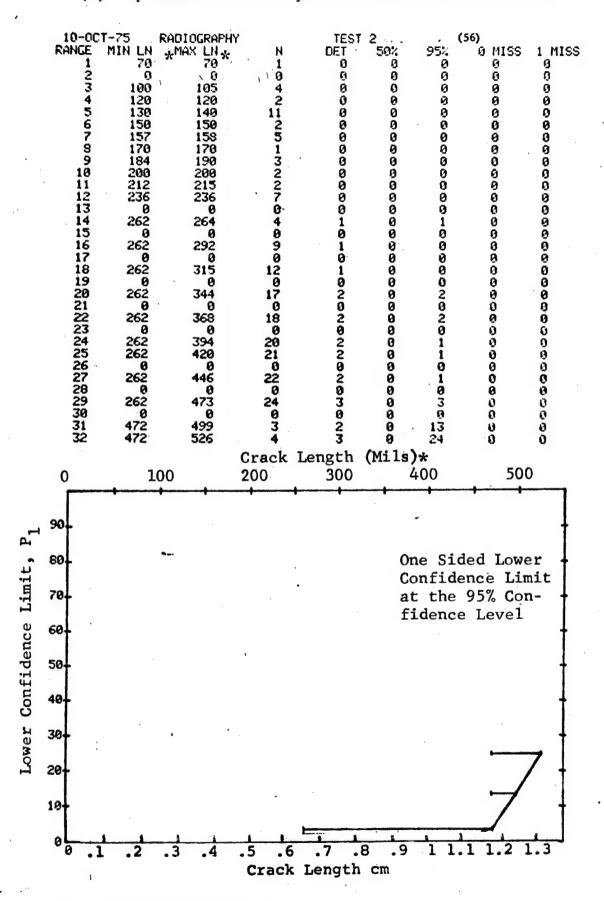


Figure D-56 (Continued)

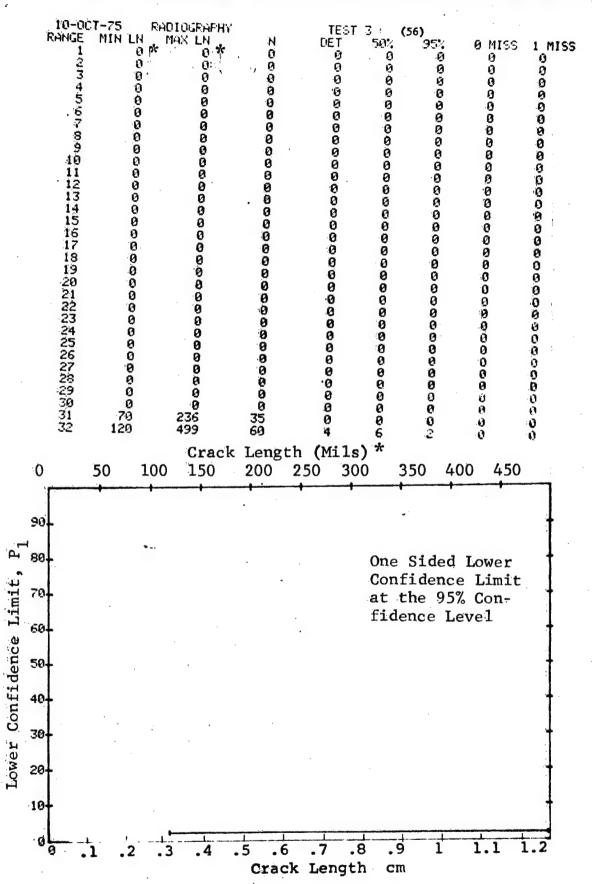


Figure D-56 (Concluded)

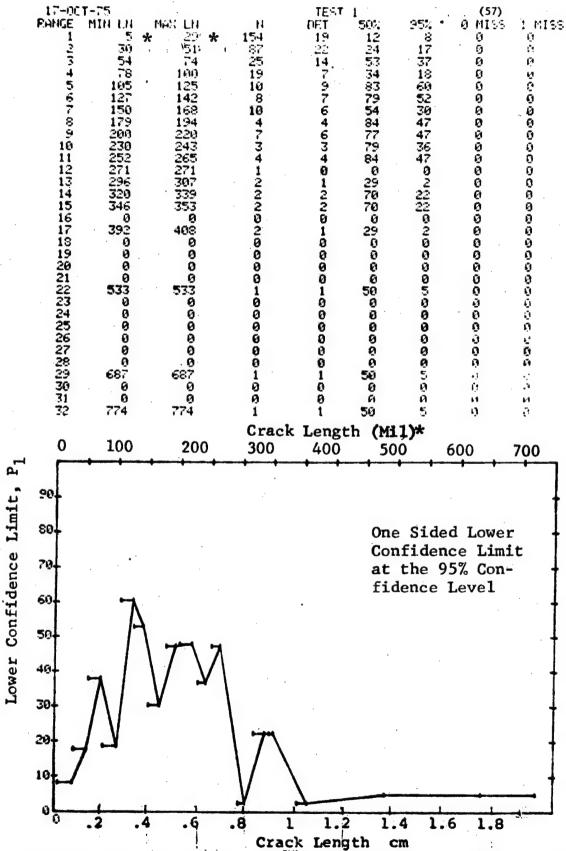


Figure D-57 Probability of Detection for 7178-T651 Al Using Eddy Current. Fatigue Cracks in Fastener Holes Measured by Team 2. Field Env.

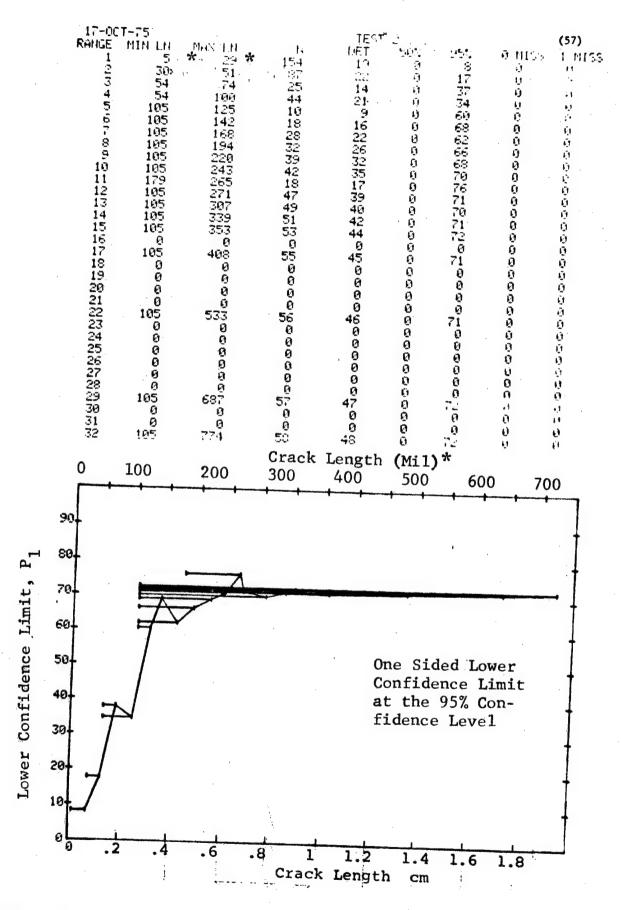


Figure D-57 (Continued)

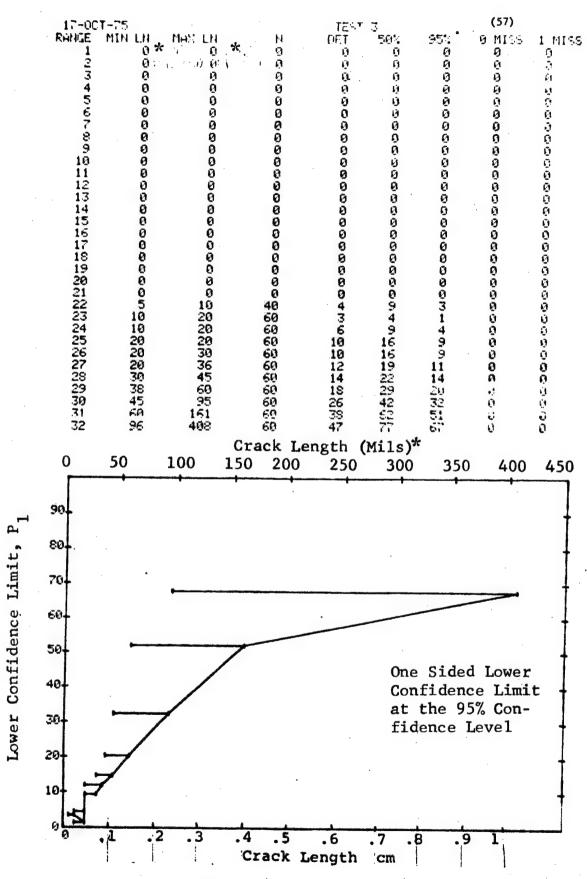


Figure D-57 (Concluded)

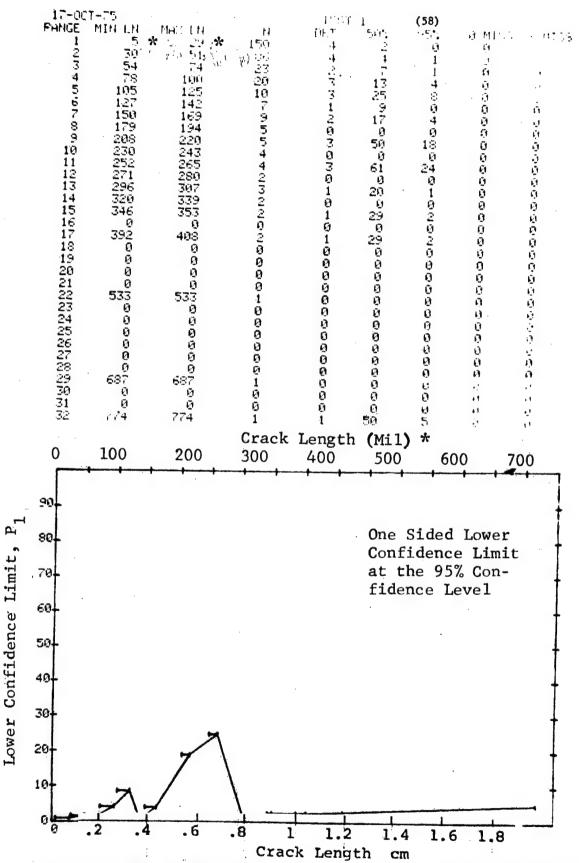


Figure D-58 Probability of Detection for 7178-T651 Al Using Eddy Current. Fatigue Cracks in Fastener Holes Measured by Team 4. Field Env.

D - 180

(b) Optimum Probability Method of Data Cumulation

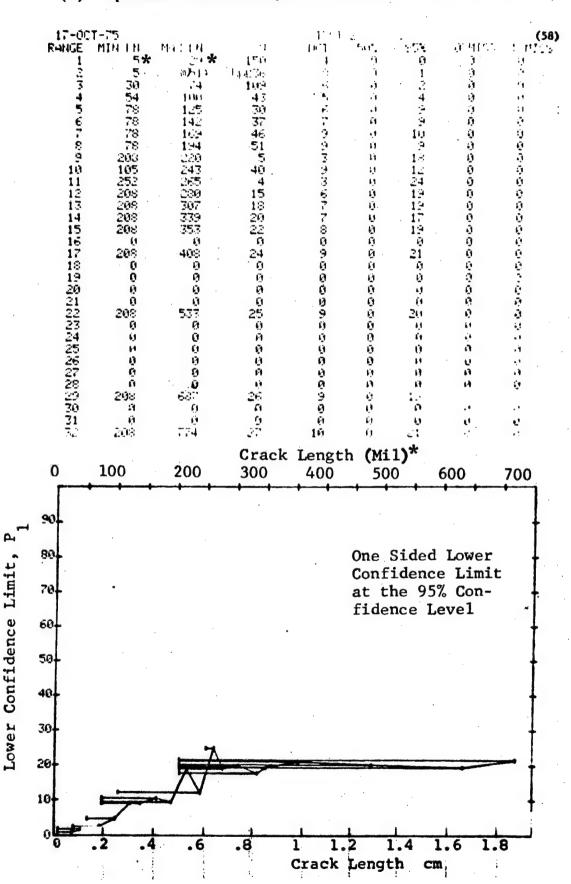
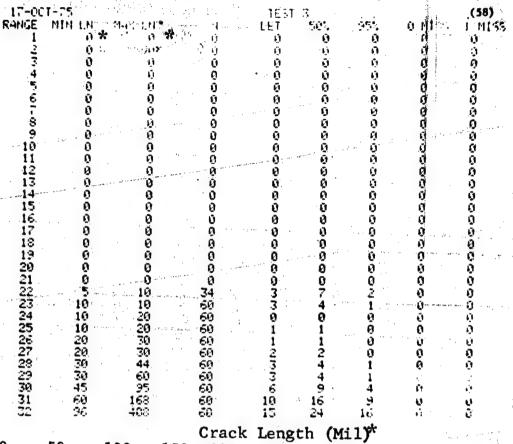


Figure D-58 (Continued)



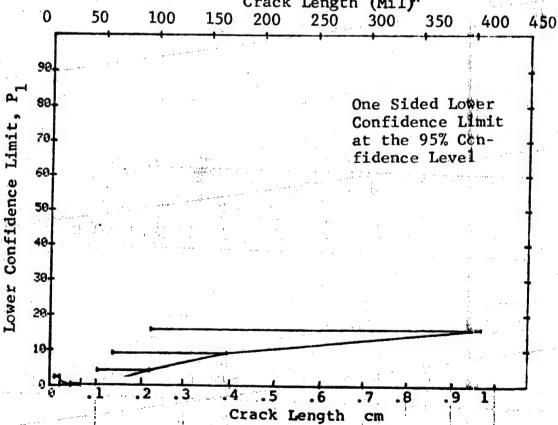


Figure D-58 (Concluded)

(a) Range Interval Method of Data Cumulation

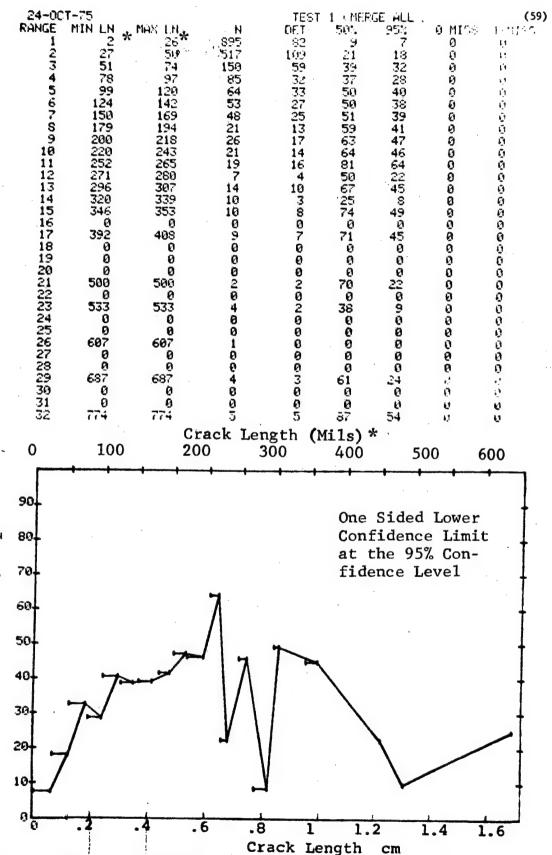


Figure D-59 Probability of Detection for 7178-T651 Al Using Eddy Current. Fatigue Cracks in Fastener Holes Measured by 5 Merged Teams. Field Env.

D-183

Lower Confidence Limit,

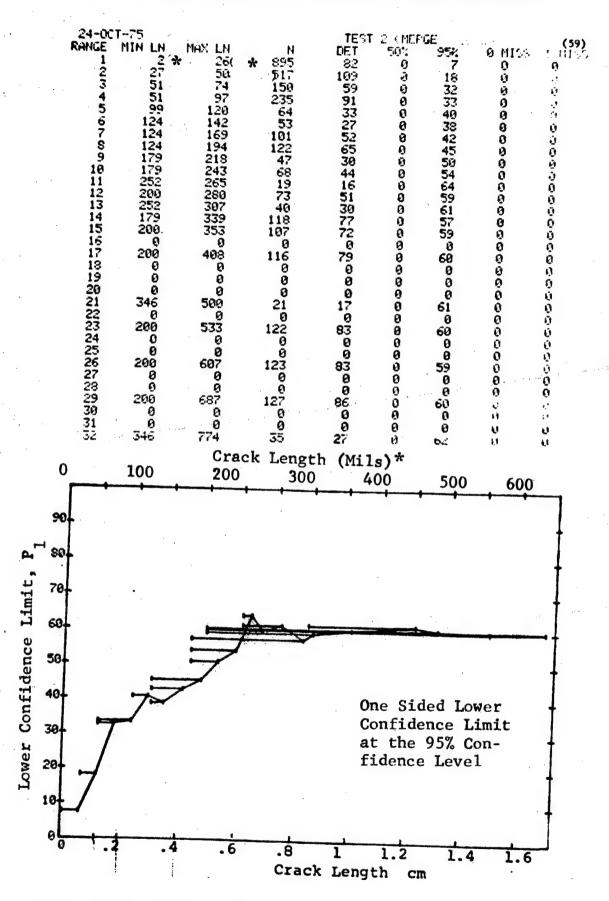


Figure D-59 (Continued)

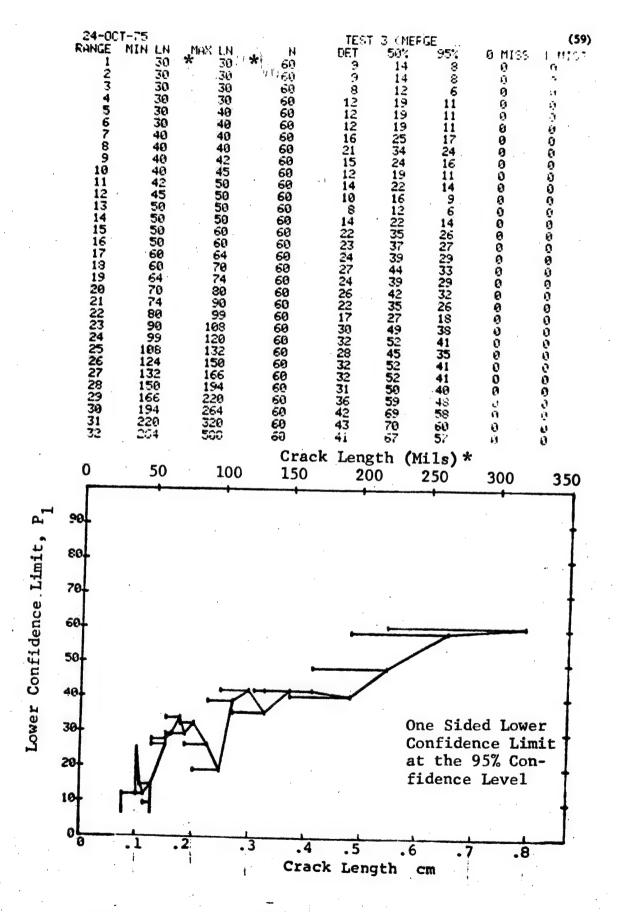


Figure D-59 (Concluded)

(a) Range Interval Method of Data Cumulation

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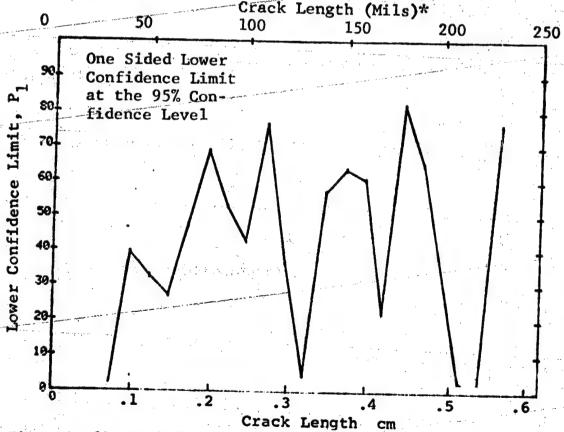


Figure D-60 Probability of Detection for 2024-T6 Al Using Liquid Penetrant. Compressed Notch Flaws in Tandem T Sections. Prod. Env.

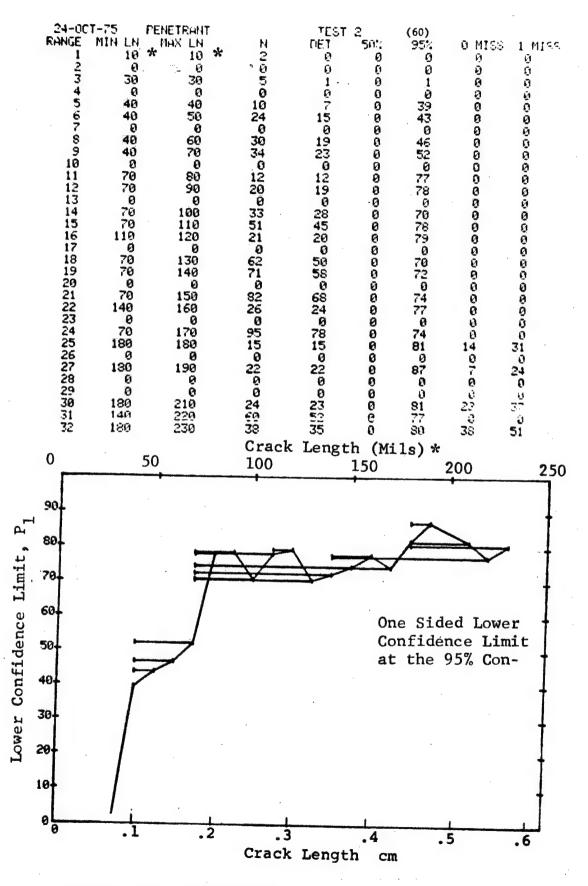


Figure D-60 (Continued)

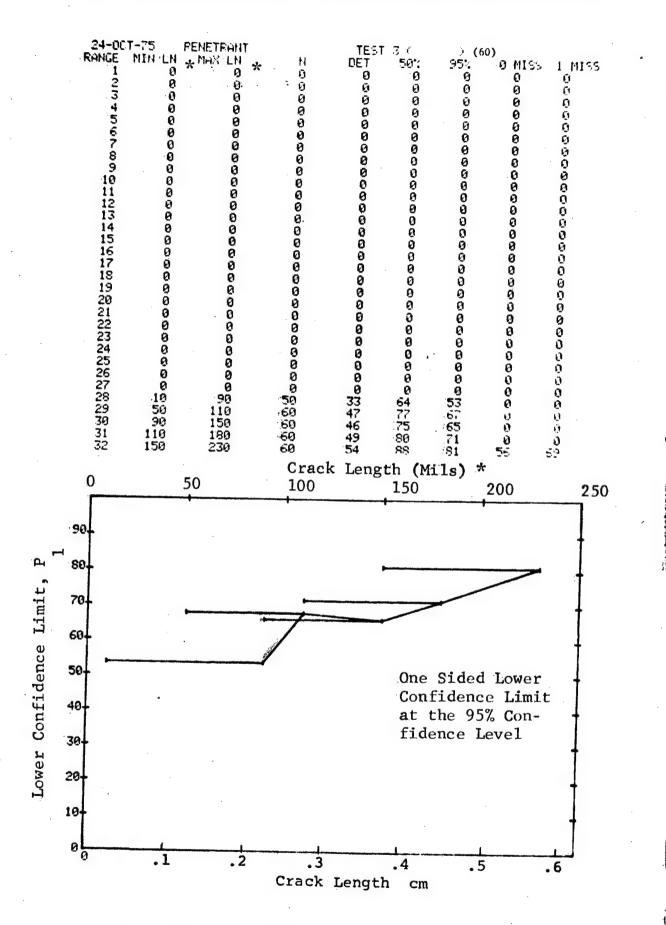


Figure D-60 (Concluded)

(a) Range Interval Method of Data Cumulation

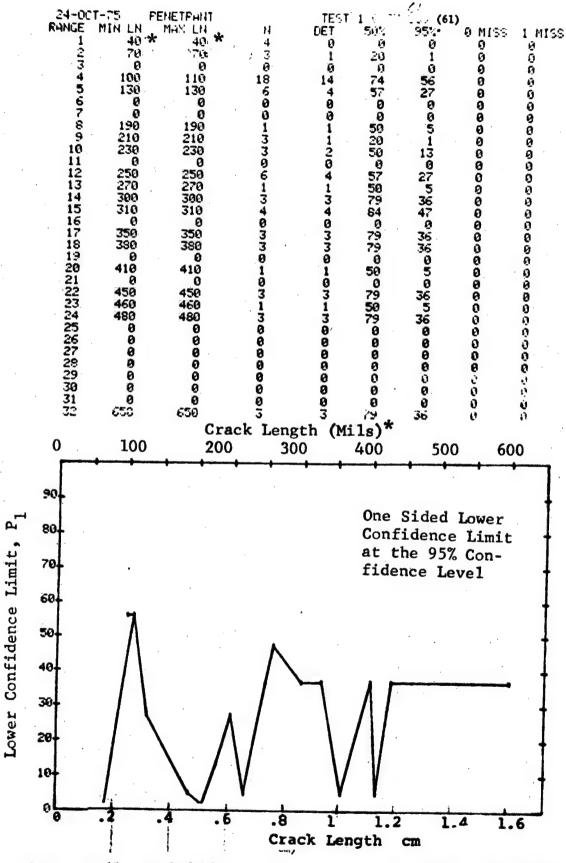


Figure D-61 Probability of Detection for 4340M Steel Using Liquid Penetrant. Compressed Notch Flaws in Solid Threaded Cylinder. Prod. Env.

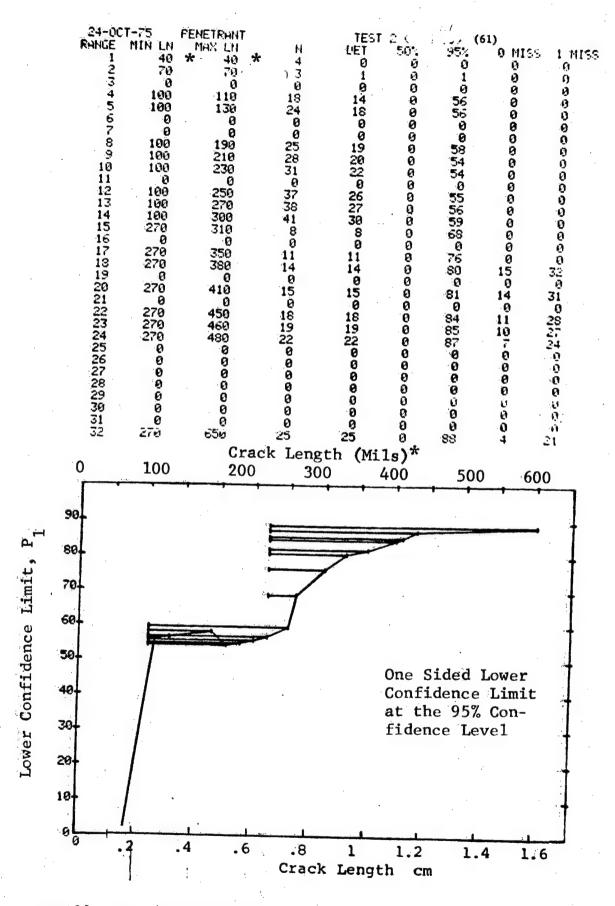


Figure D-61 (Continued)

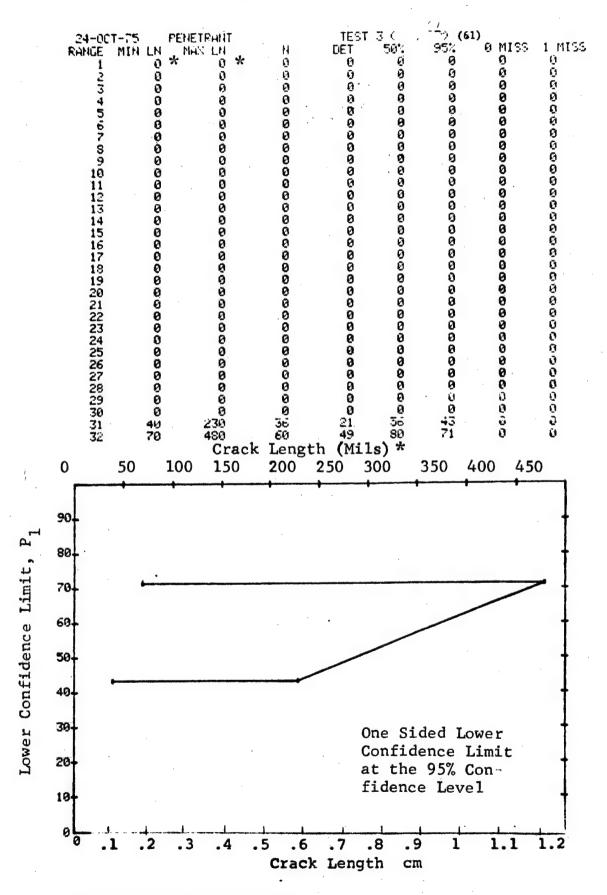


Figure D-61 (Concluded)

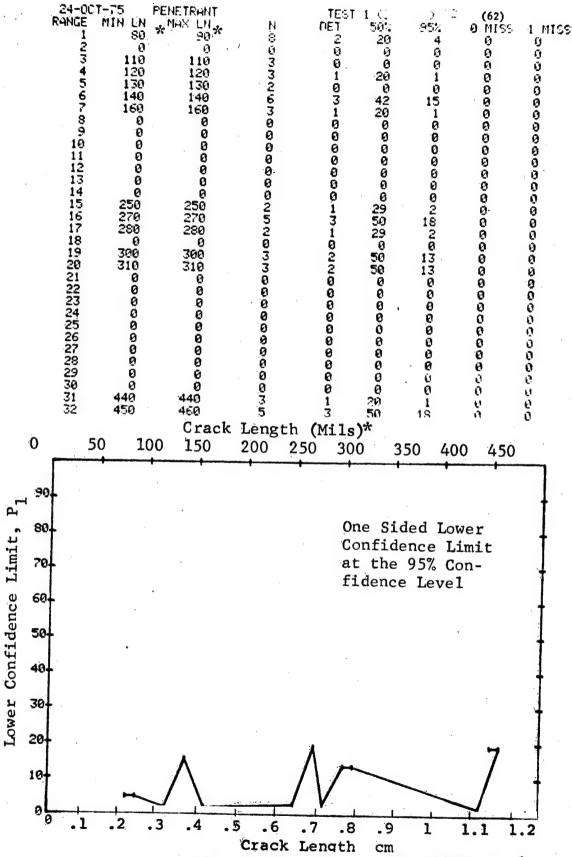


Figure D-62 Probability of Detection for 4340M Steel
Using Liquid Penetrant. Compressed Notch
Flaws in Hollow Cylinder. Prod. Env.

D-192

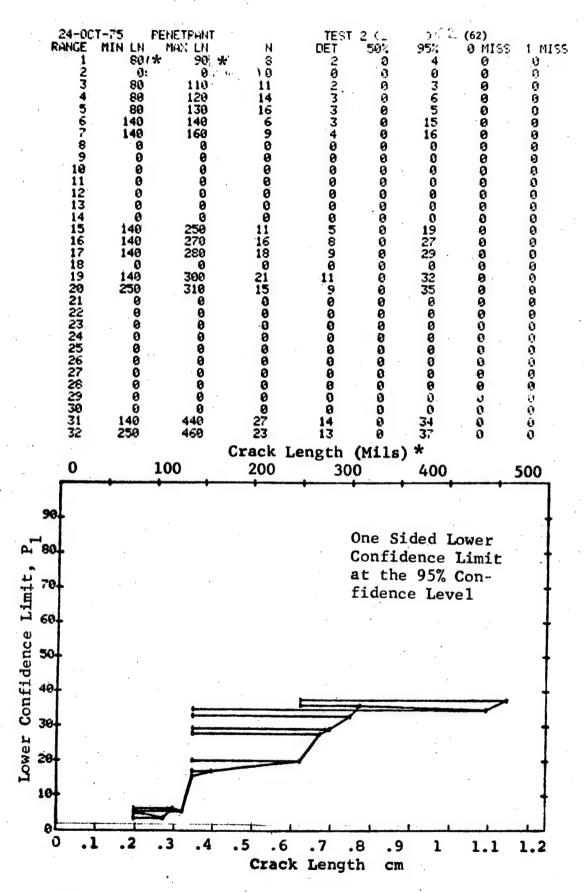


Figure D-62 (Continued)

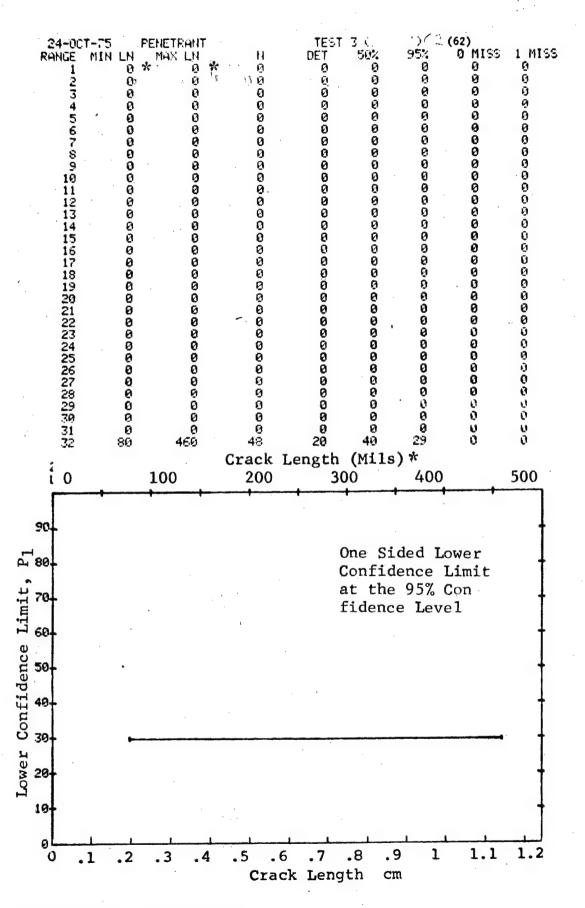
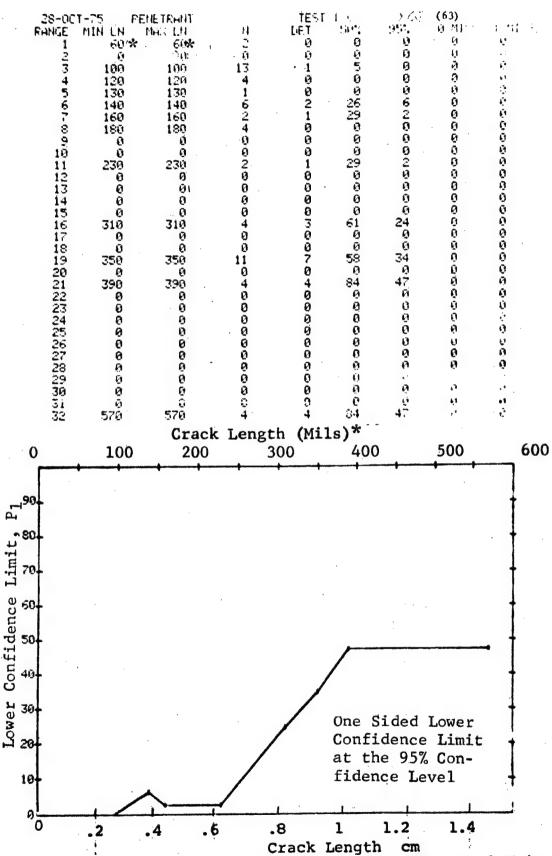


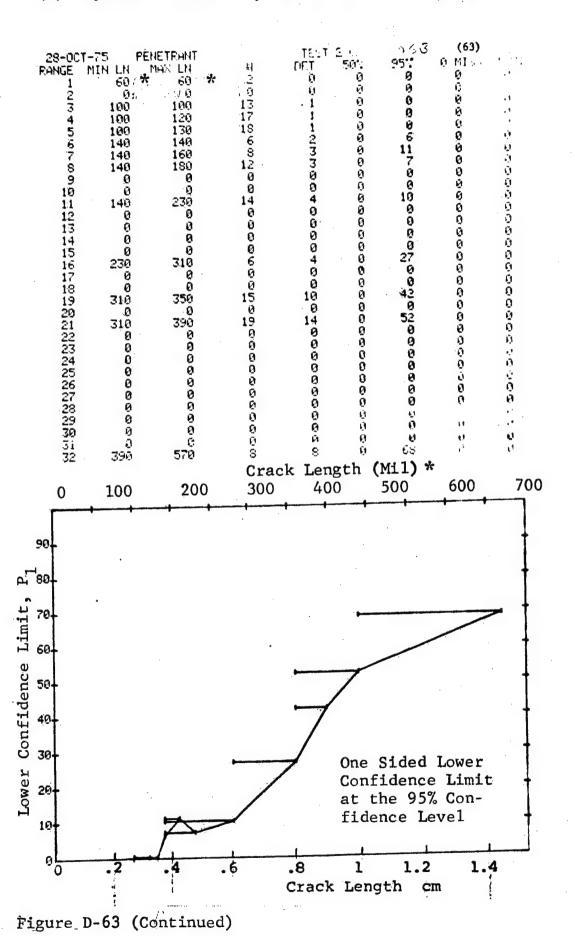
Figure D-62 (Concluded)

(a) Range Interval Method of Data Cumulation



Probability of Detection for 4340M Steel Using Figure D-63 Liquid Penetrant. Compressed Notch Flaws in Filleted Hollow Cylinder. Prod. Env. D-195

Confidence



(c) Overlapping Sixty Point Method of Data Cumulation

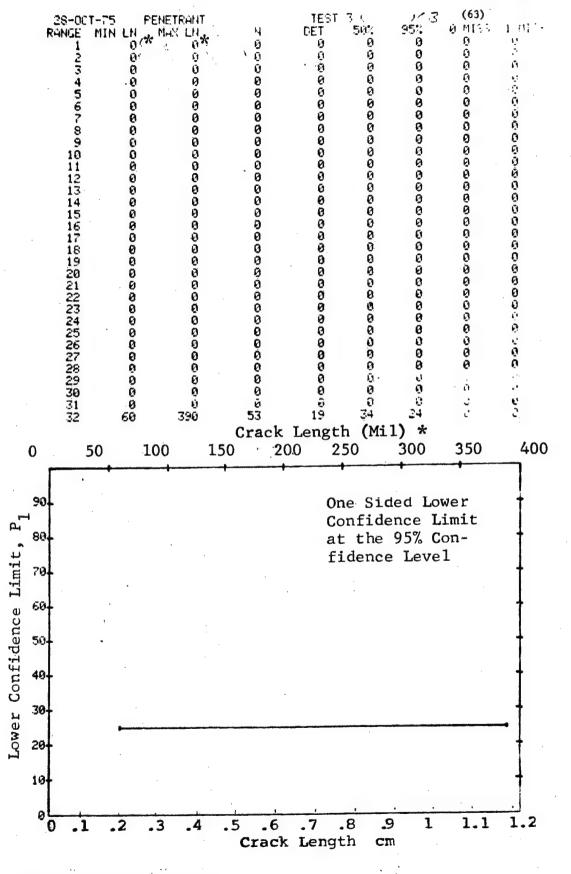


Figure D-63 (Concluded)

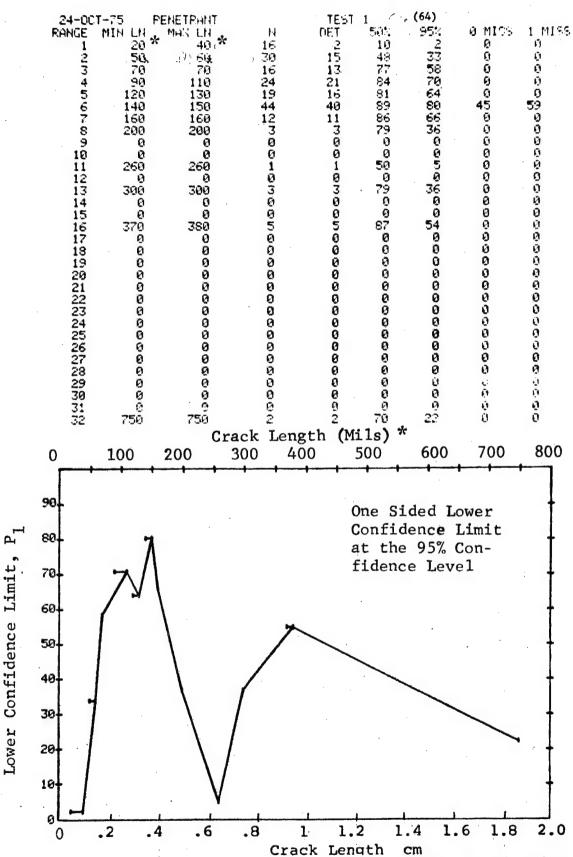


Figure D-64 Probability of Detection for 4340M Steel Using
Liquid Penetrant. Compressed Notch Flaws in
Solid Cylinder. Prod. Env.
D-198

REPRODUCIBILITY OF THE

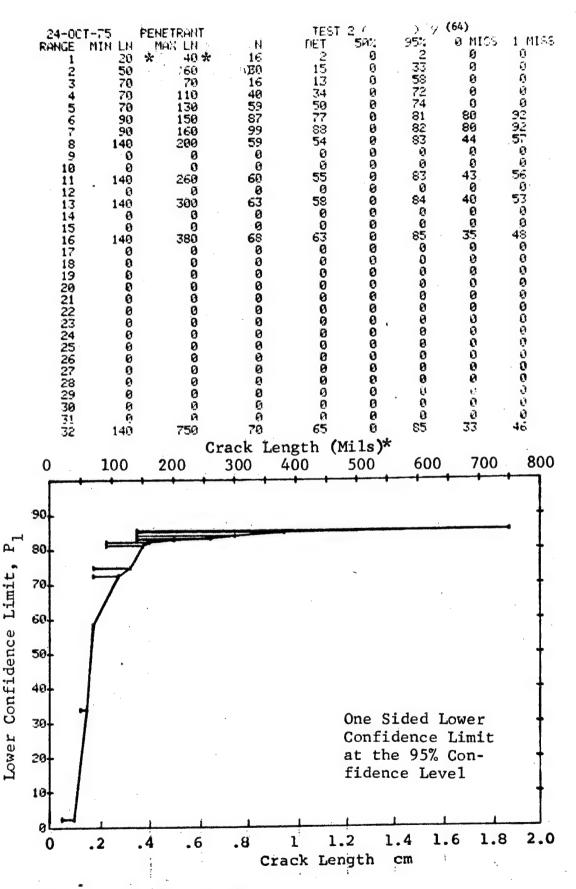


Figure D-64 (Continued)

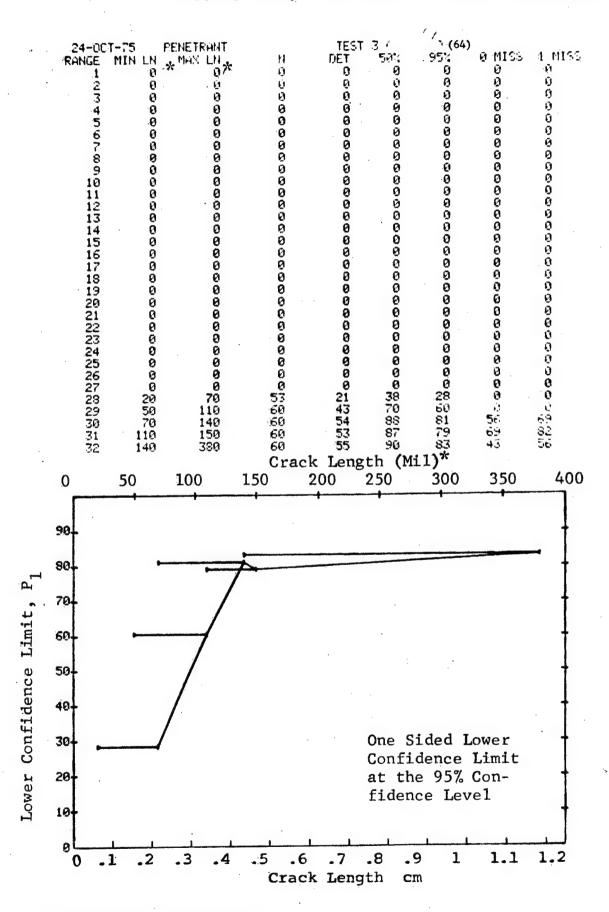


Figure D-64 (Concluded)

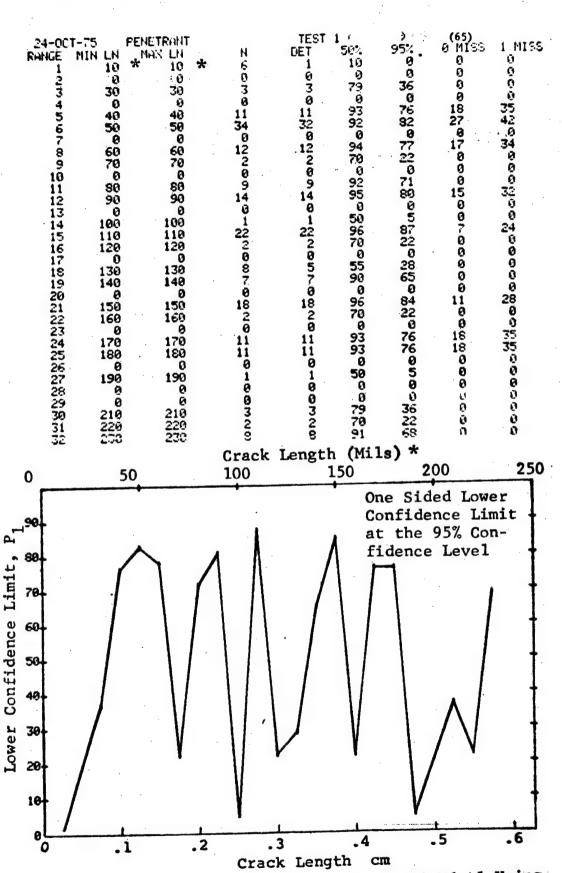


Figure D-65 Probability of Detection for 2024-T6 Al Using Liquid Penetrant. Compressed Notch Flaws in Tandem T Specimen. Lab. Env.

D-201

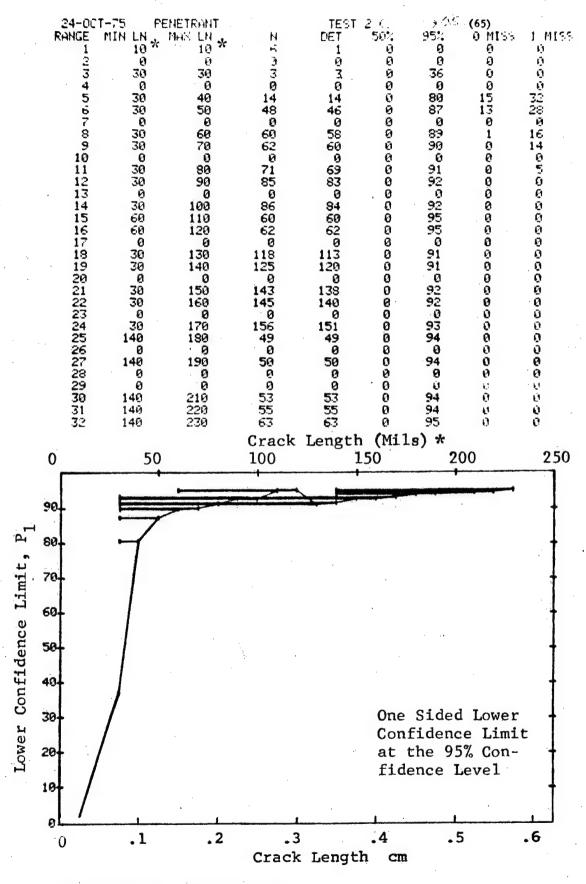


Figure D-65 (Continued)

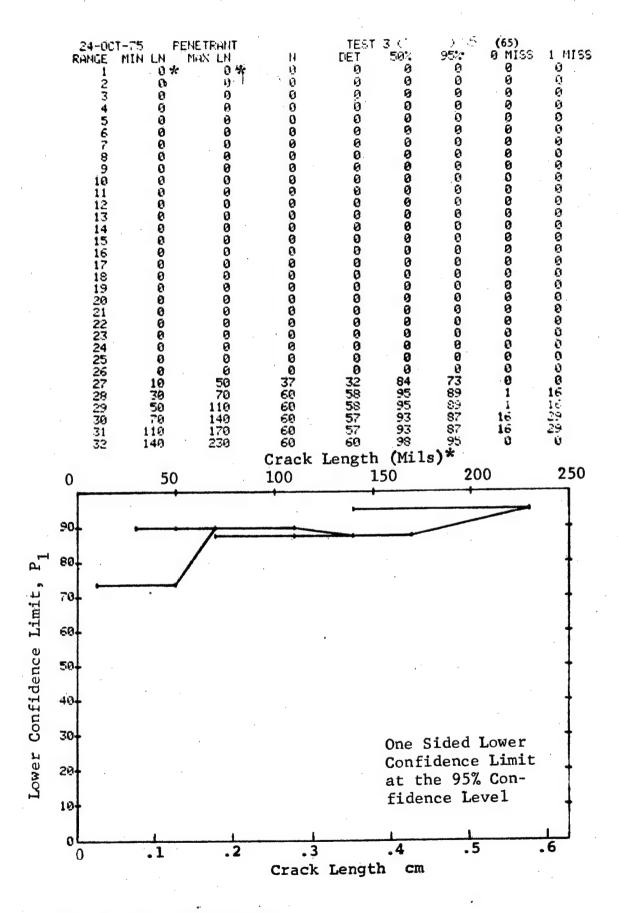


Figure D-65 (Concluded)

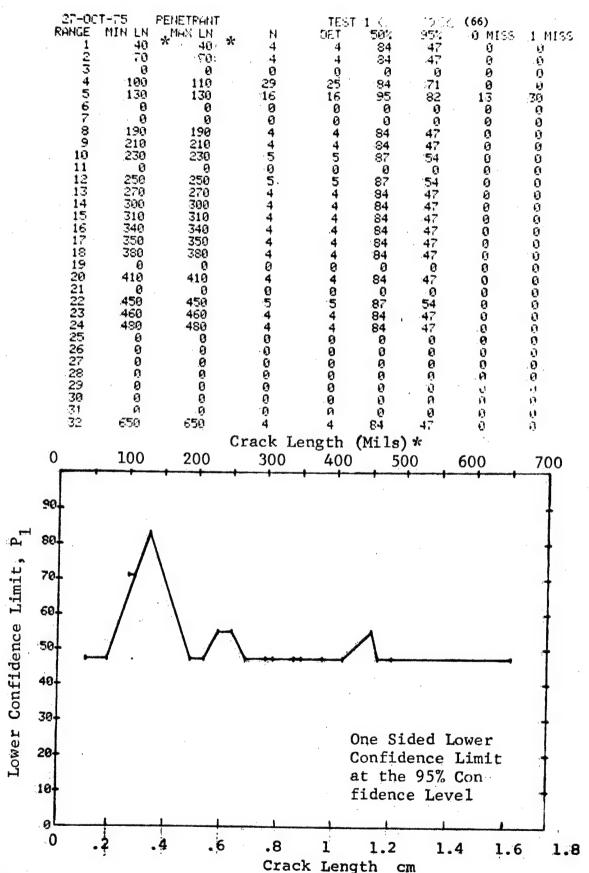


Figure D-66 Probability of Detection for 4340M Steel Using Liquid Penetrant. Compressed Notch Flaws in Solid Cylinder.

D-204

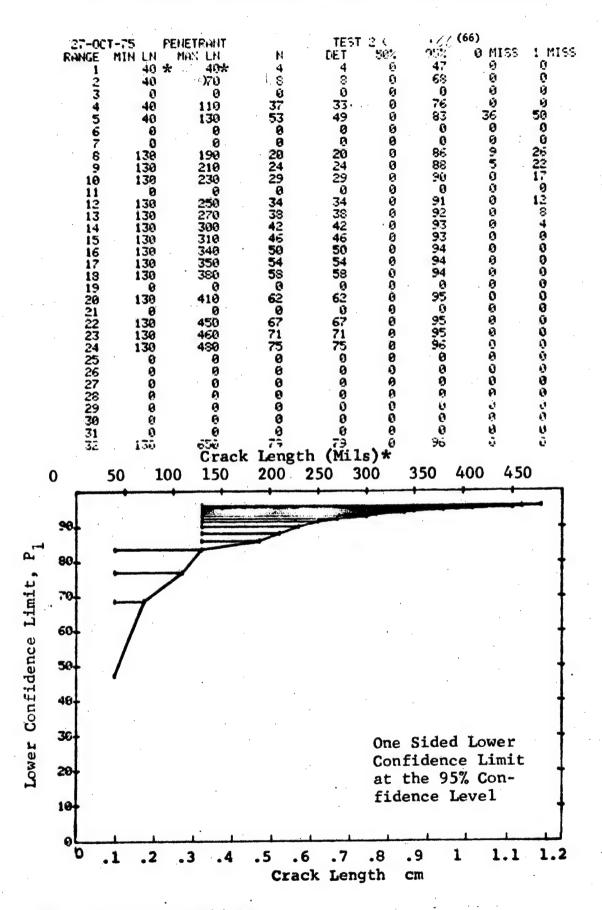


Figure D-66 (Continued)

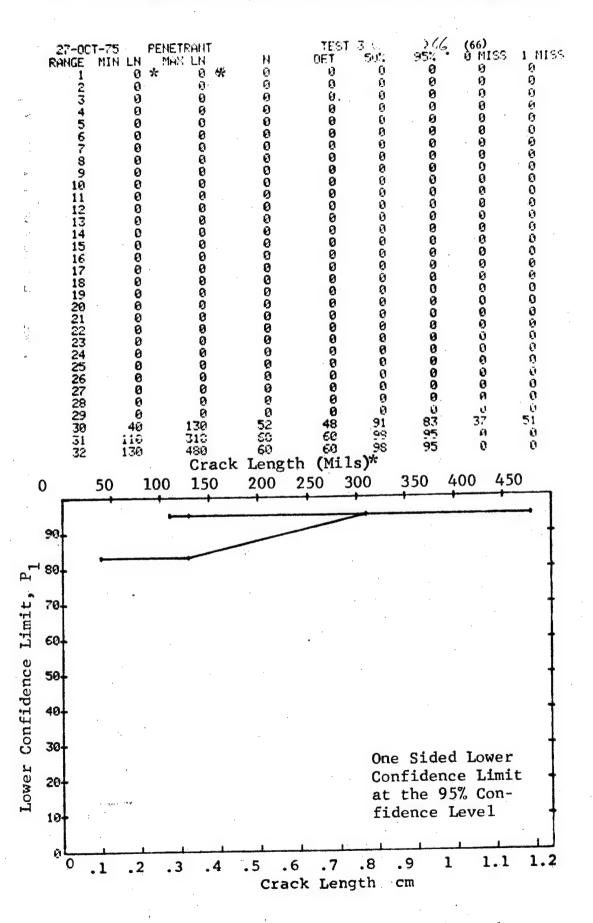


Figure D-66 (Concluded)

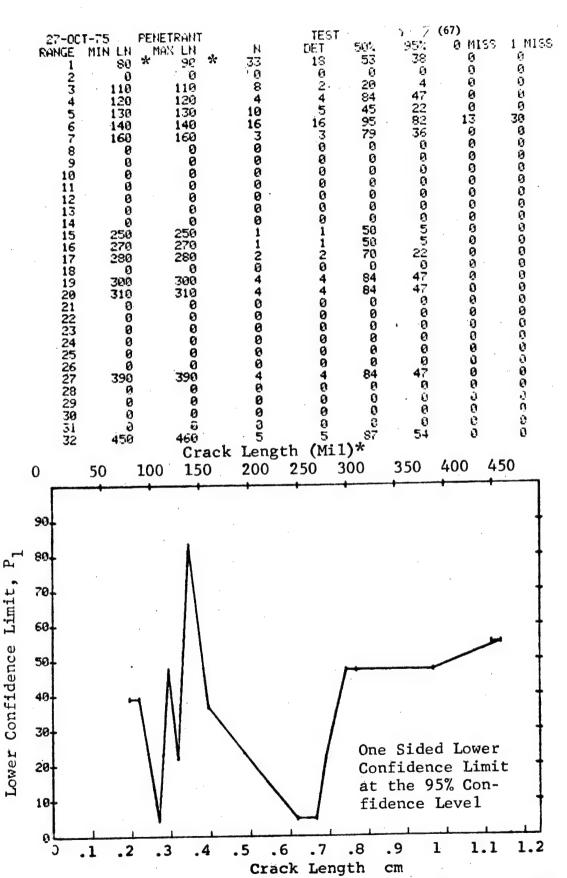


Figure D-67 Probability of Detection for 4340M Steel Using Liquid Penetrant. Compressed Notch Flaws in Hollow Cylinder. Lab. Env.

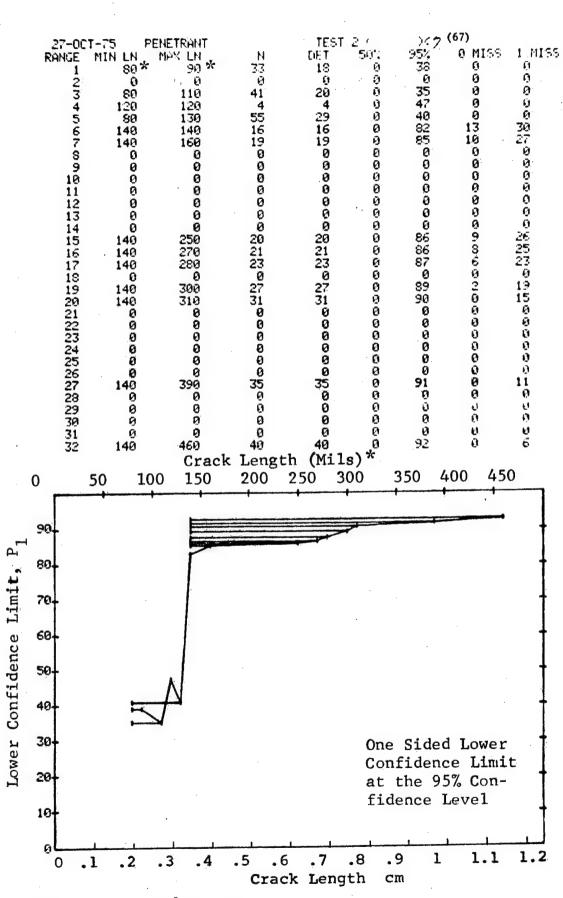


Figure D-67 (Continued)

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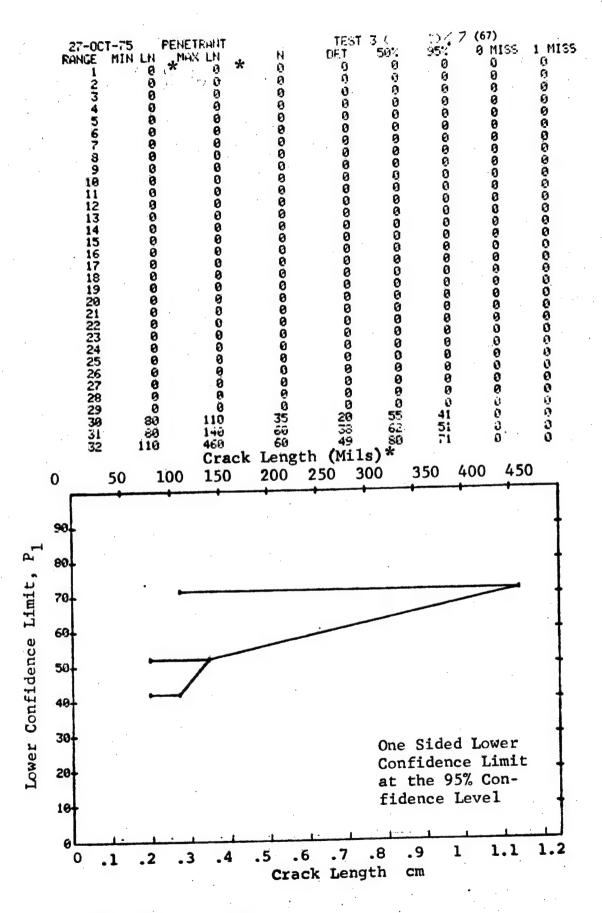


Figure D-67 (Concluded)

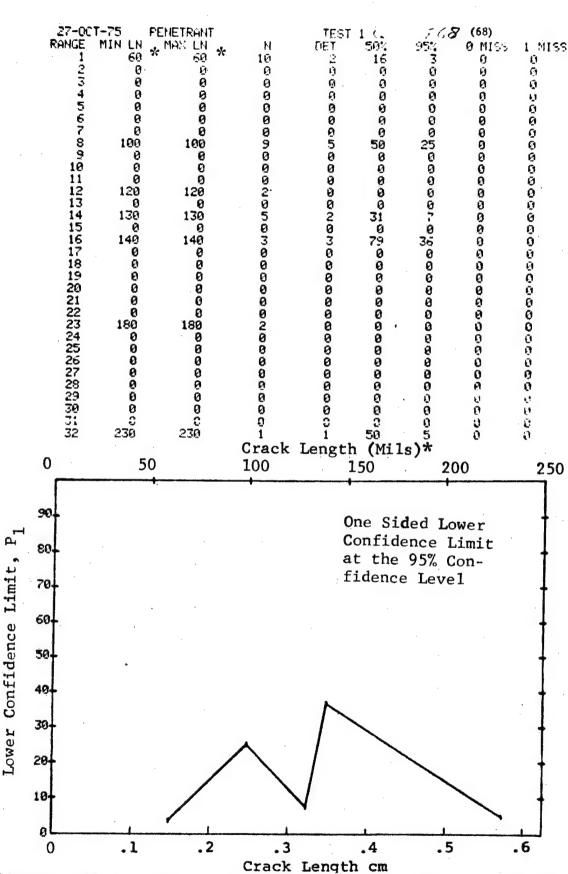


Figure D-68 Probability of Detection for 4340M Steel Using Liquid Penetrant. Compressed Notch Flaws in Filleted Hollow Cylinder. Lab. Env. D-210

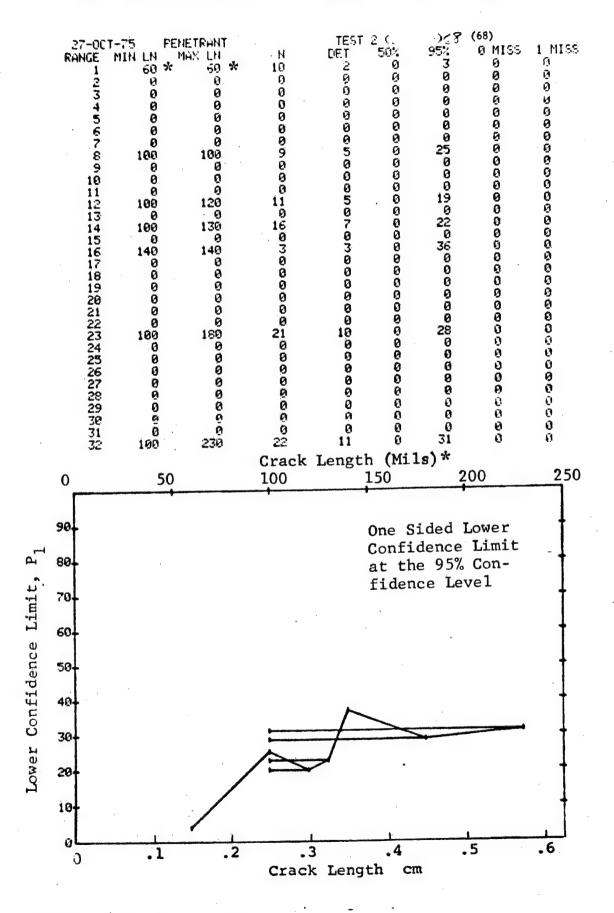


Figure D-68 (Continued)

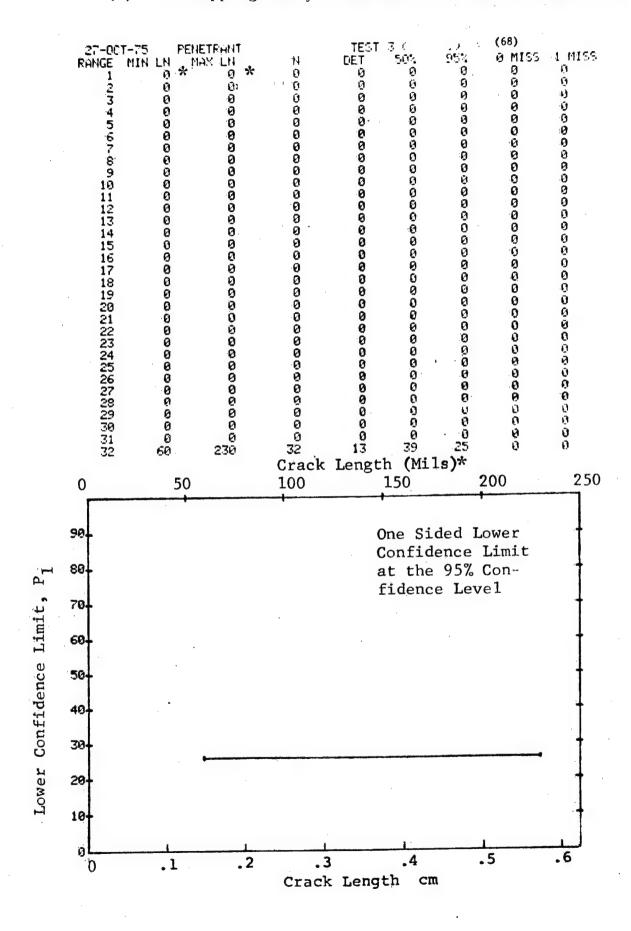
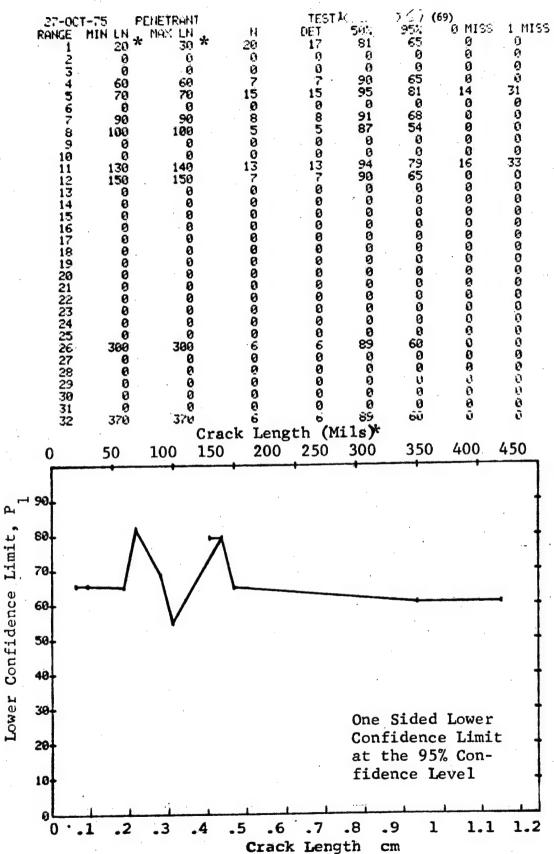


Figure D-68 (Concluded)



Probability of Detection for 4340M Steel Using Figure D-69 Liquid Penetrant. Compressed Notch Flaws in Lab. Env. Filleted Solid Cylinder. D-213

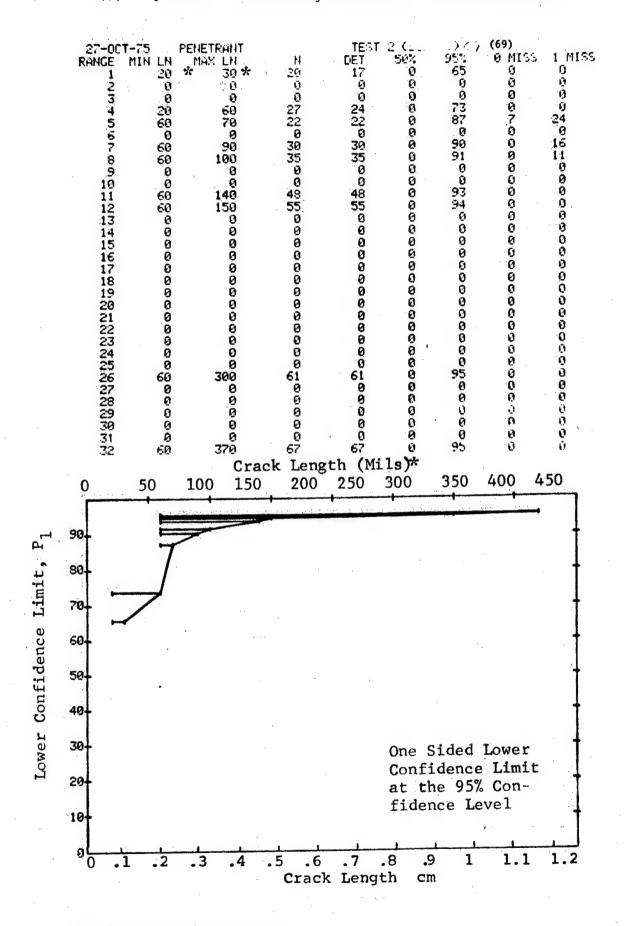


Figure D-69 (Continued)

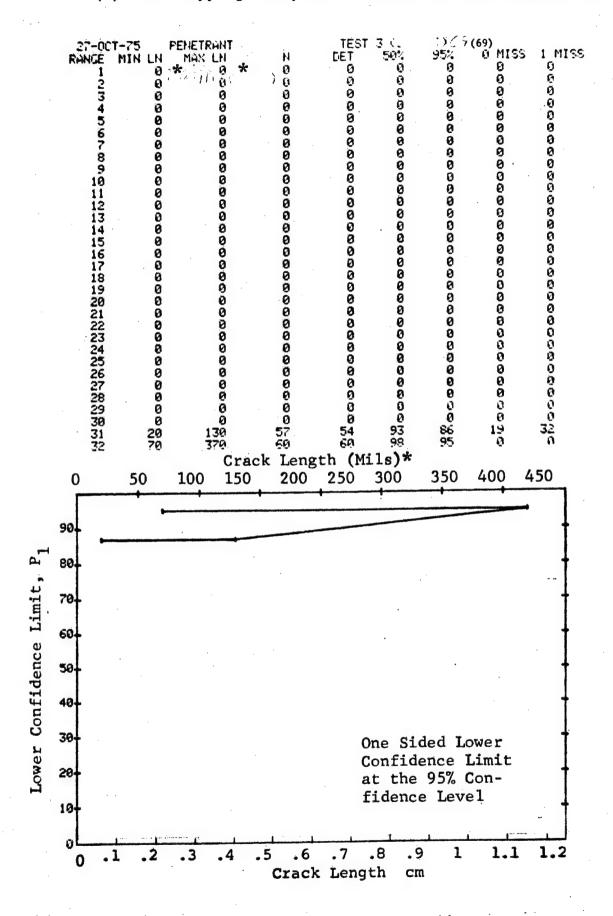


Figure D-69 (Concluded)

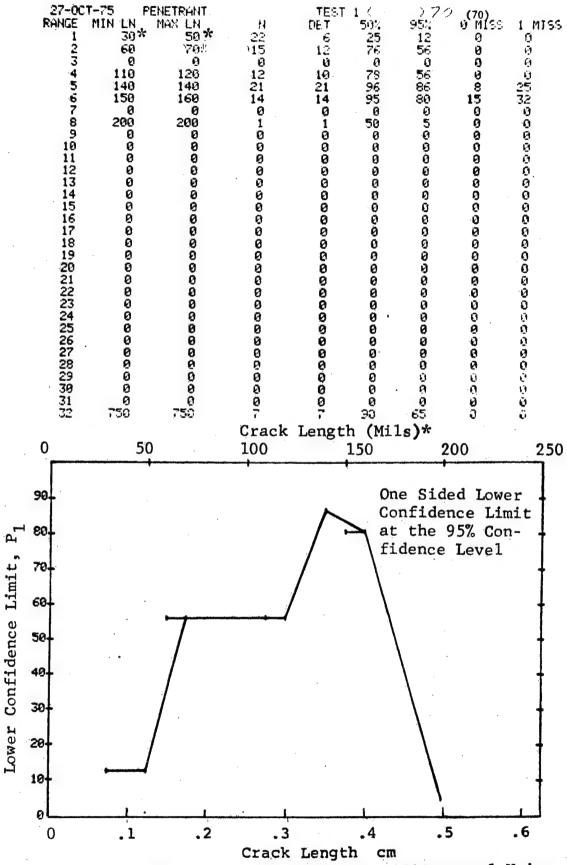


Figure D-70 Probability of Detection for 4340M Steel Using Liquid Penetrant. Compressed Notch Flaws in Filleted Solid Cylinder. Lab. Env.

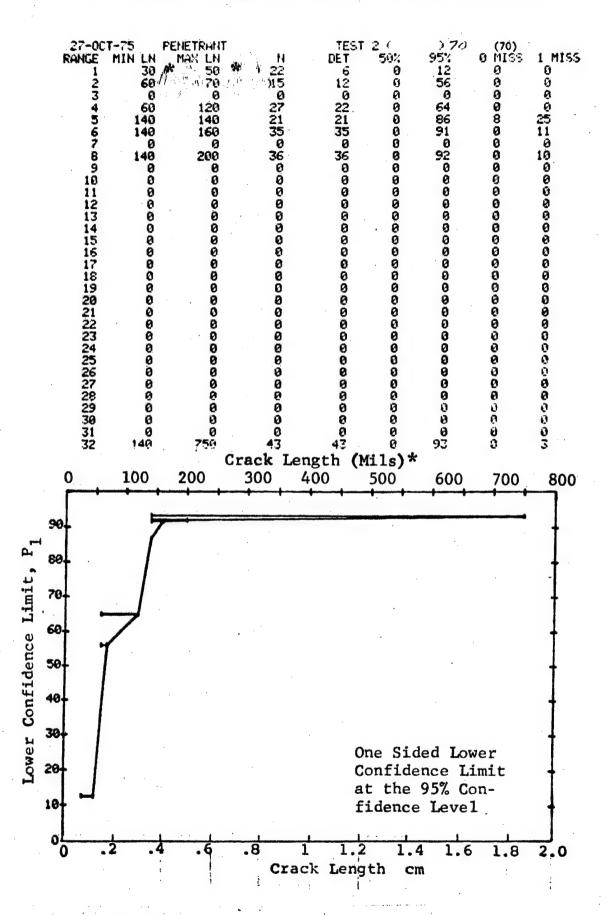


Figure D-70 (Continued)

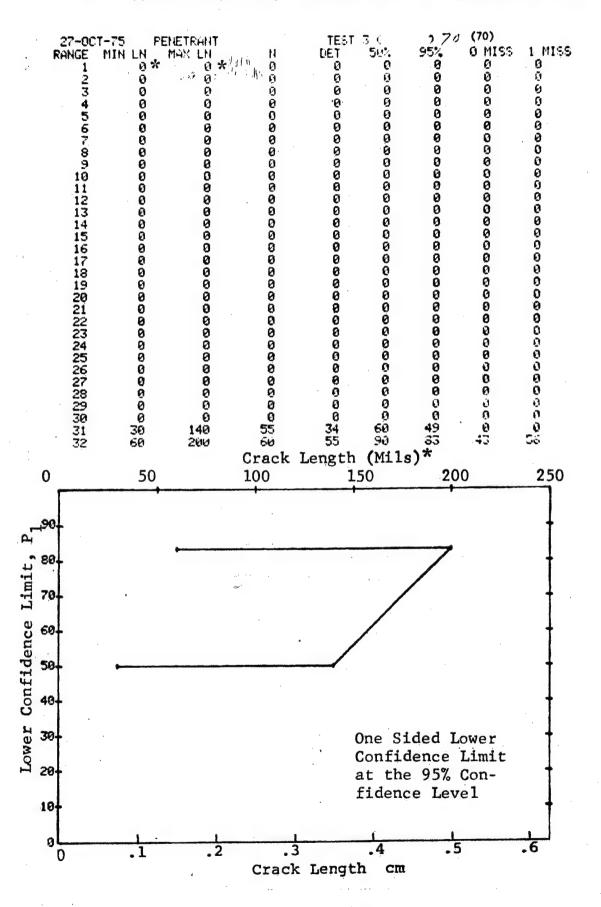


Figure D-70 (Concluded)

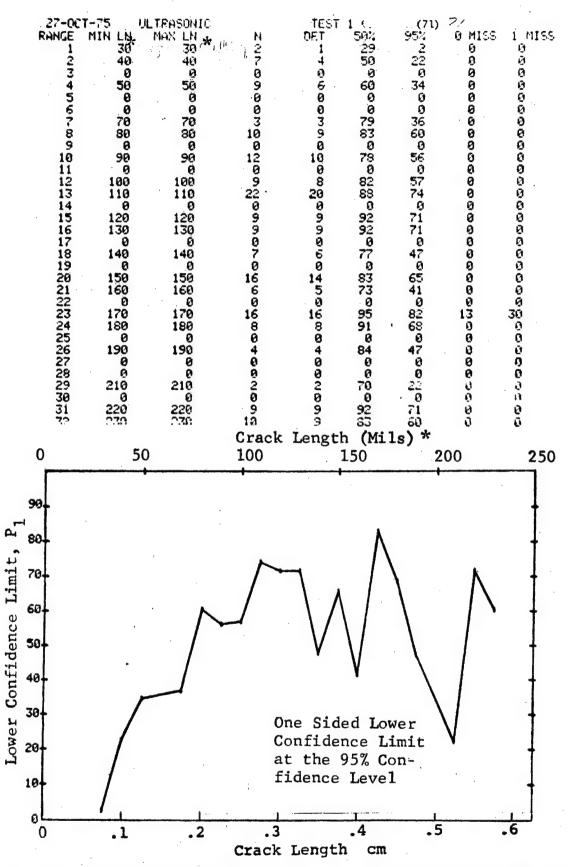


Figure D-71 Probability of Detection for 2024-T6 Al for Ultrasonic Shear and Surface Waves. Compressed Notch Flaws in Tandem T. Prod. Env.

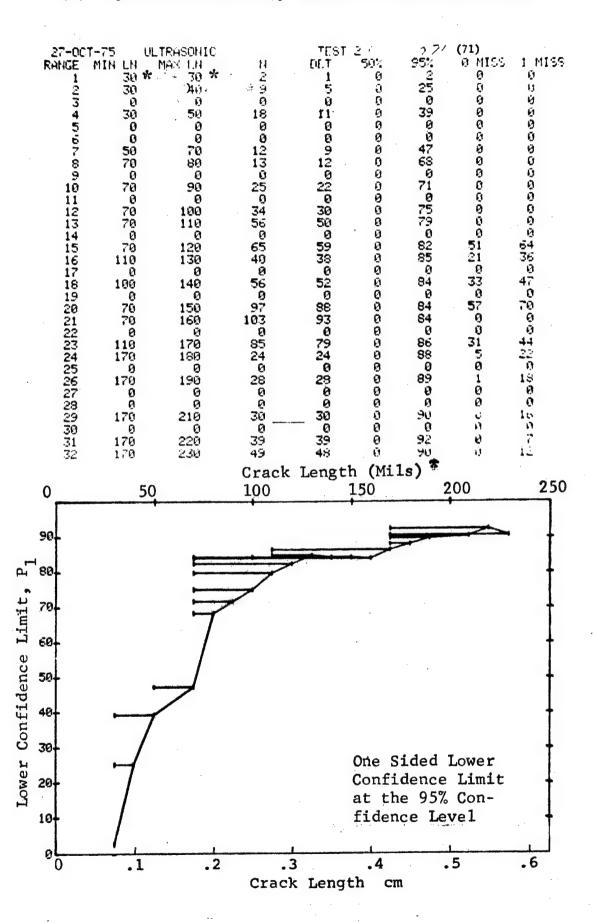


Figure D-71 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

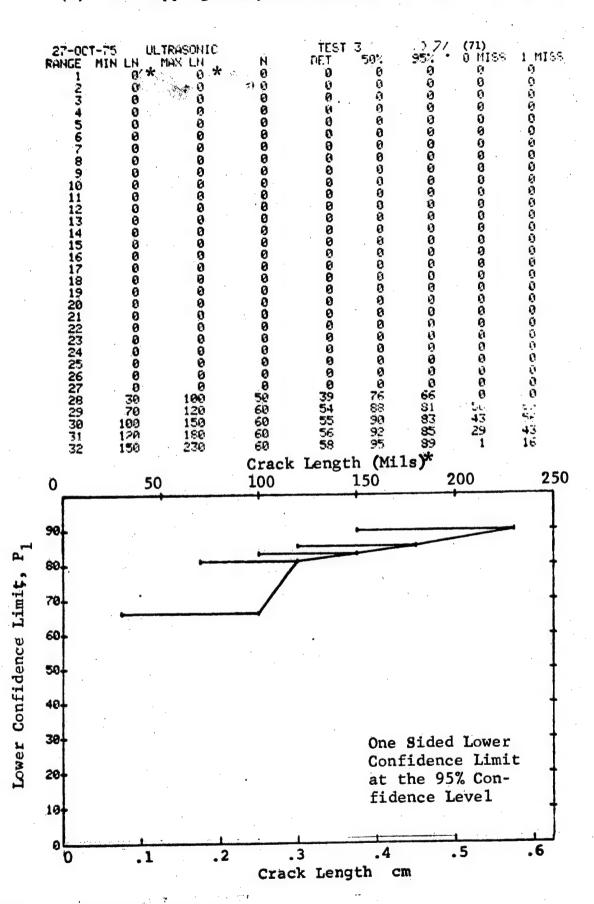


Figure D-71 (Concluded)

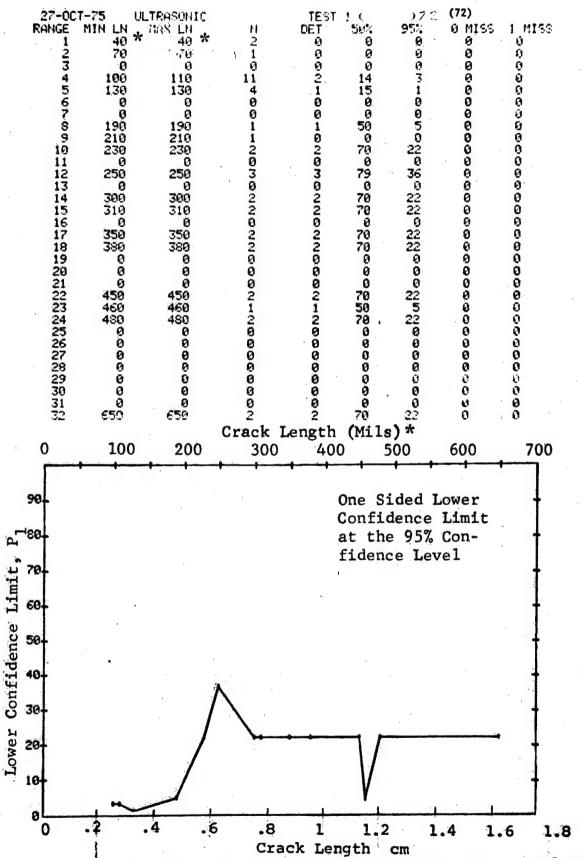


Figure D-72 Probability of Detection for 4340M Steel Using Ultrasonic Shear and Surface Waves. Compressed Notch Flaws in Solid Threaded Cylinder. Prod. Env.

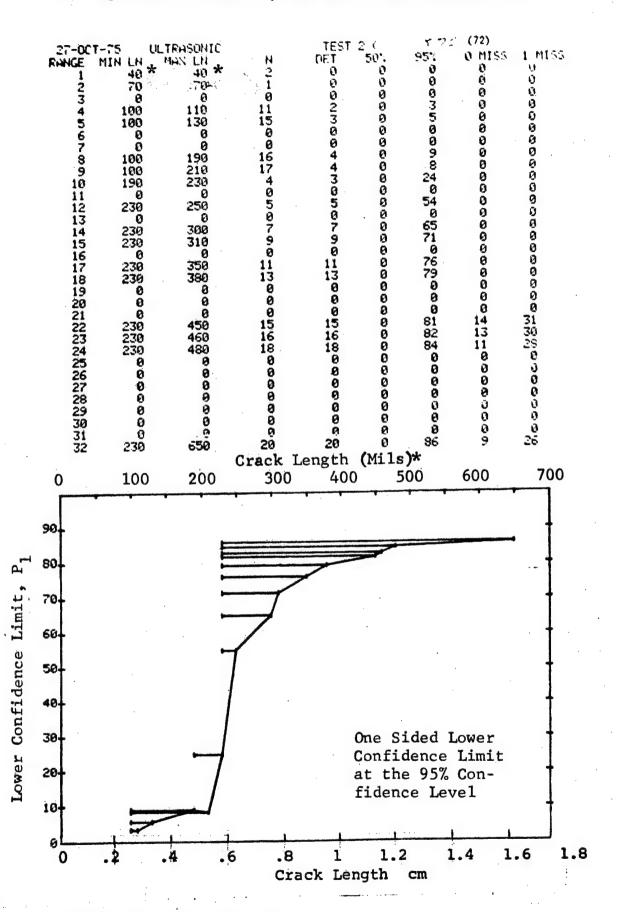


Figure D-72 (Continued)

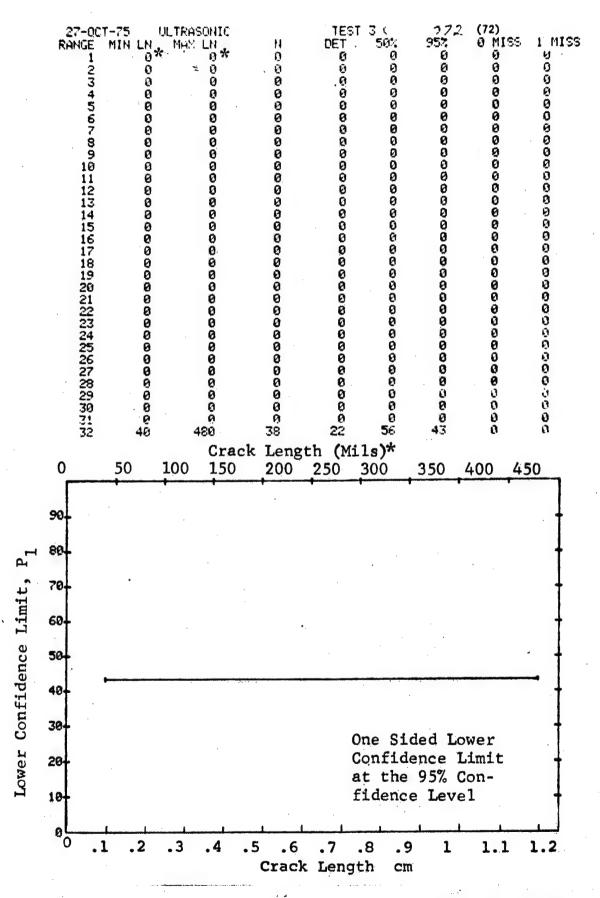
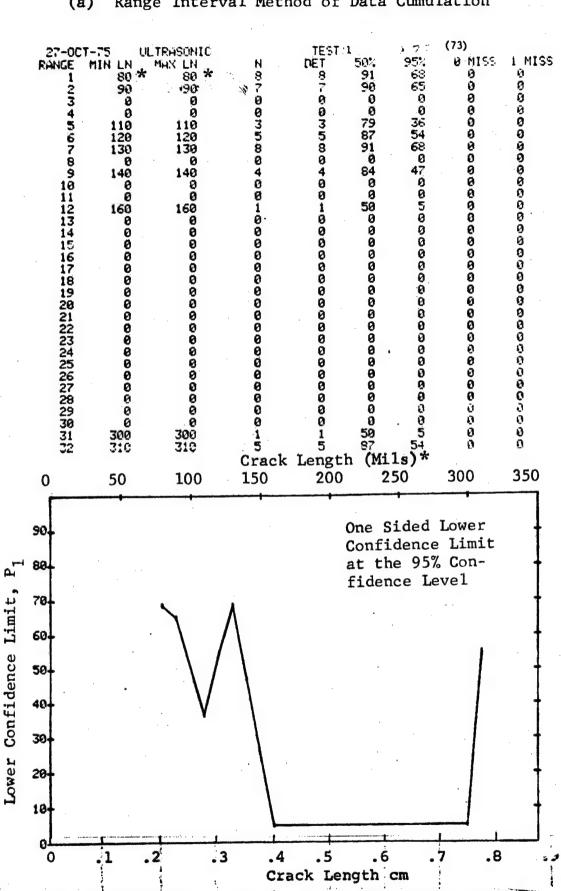


Figure D-72 (Concluded)



Probability of Detection for 4340M Steel Using Figure D-73 Ultrasonic Shear and Surface Waves. Compressed Notch Flaws in Hollow Cylinder. Prod. Env.

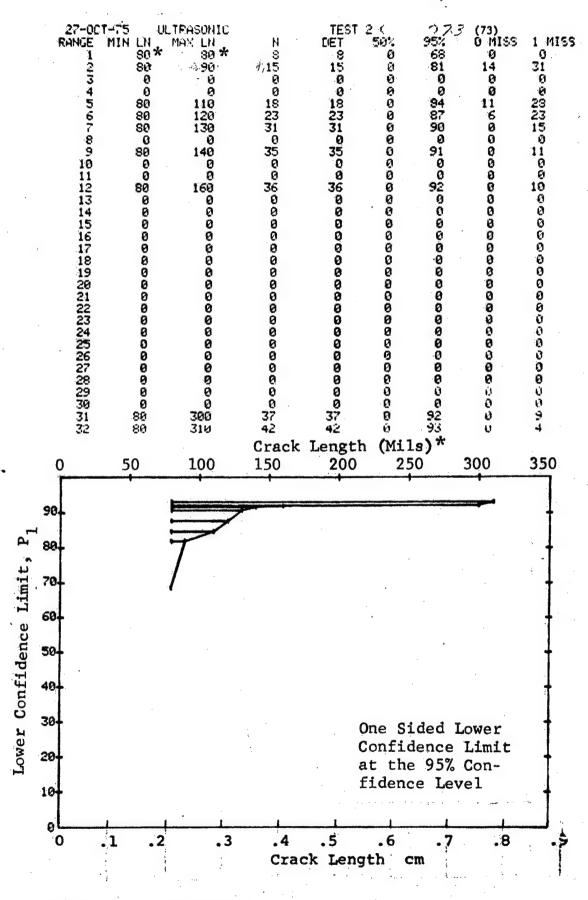


Figure D-73 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

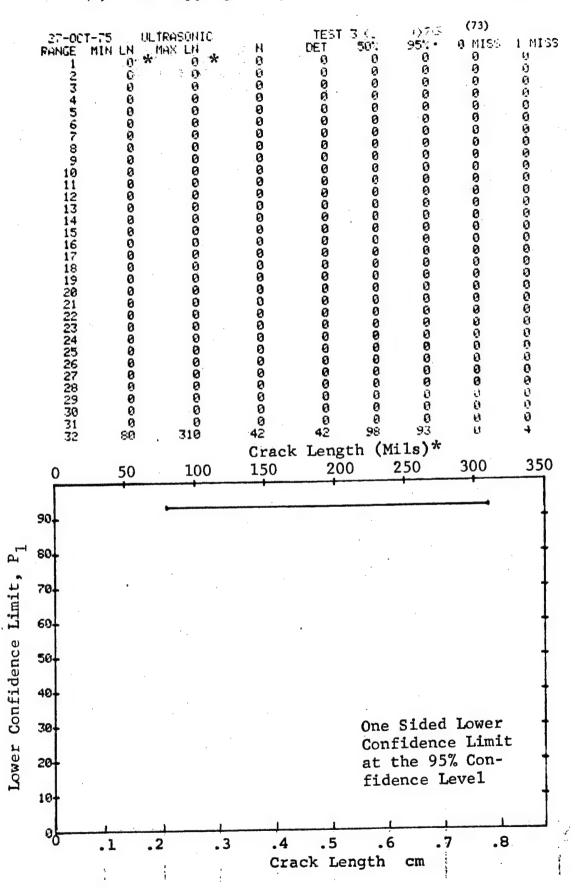
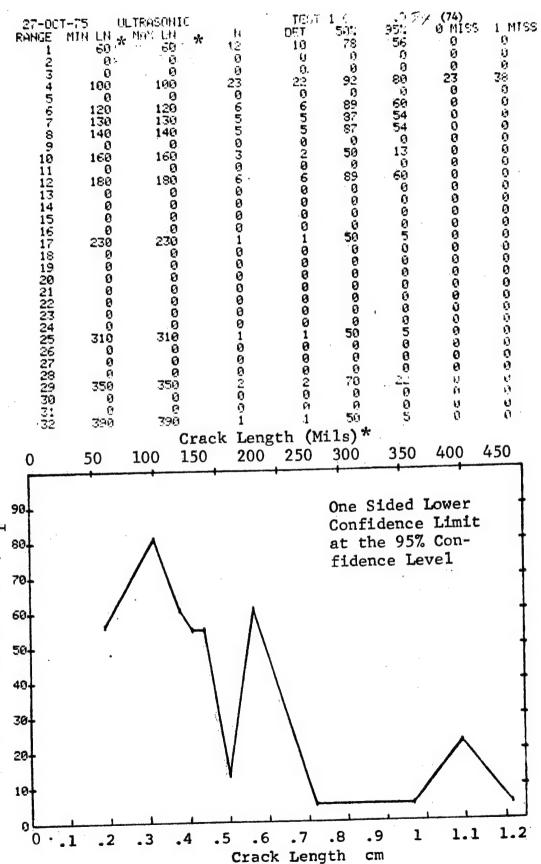


Figure D-73 (Concluded)



Lower Confidence Limit, P,

Figure D-74 Probability of Detection for 4340M Steel Using
Ultrasonic Shear and Surface Waves. Compressed Notch
Flaws in Filleted Hollow Cylinder.

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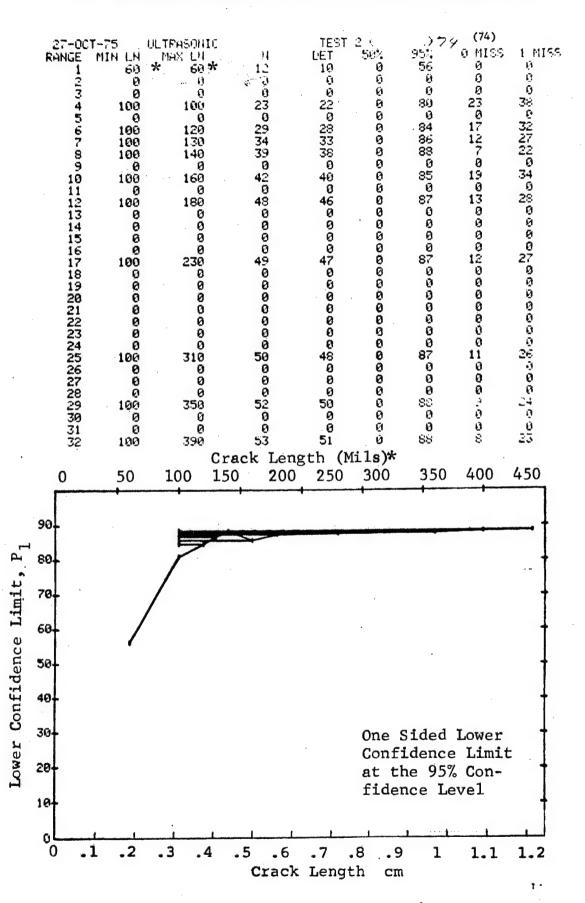


Figure D-74 (Continued)

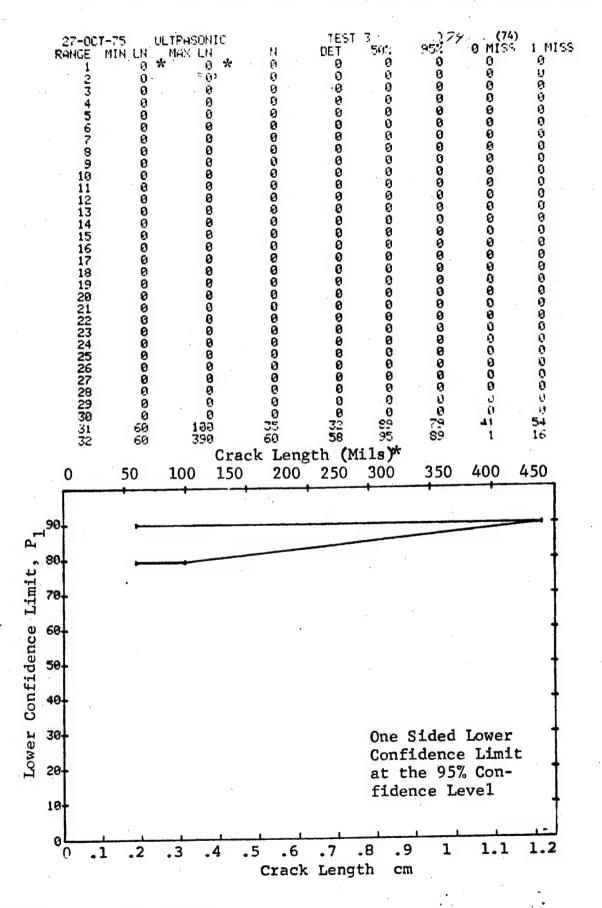


Figure D-74 (Concluded)

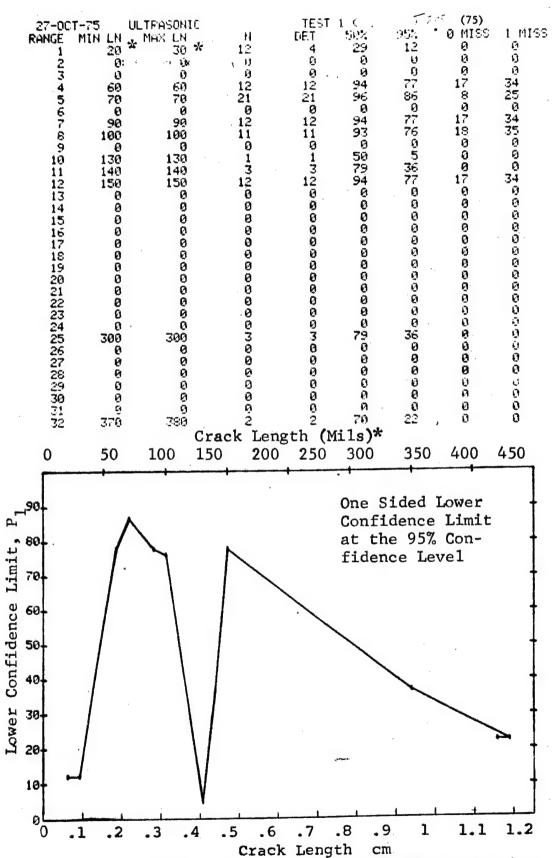


Figure D-75 Probability of Detection for 4340M Steel Using Ultrasonic Shear and Surface Waves. Compressed Notch Flaws in Filleted Solid Cylinder. Prod Env.

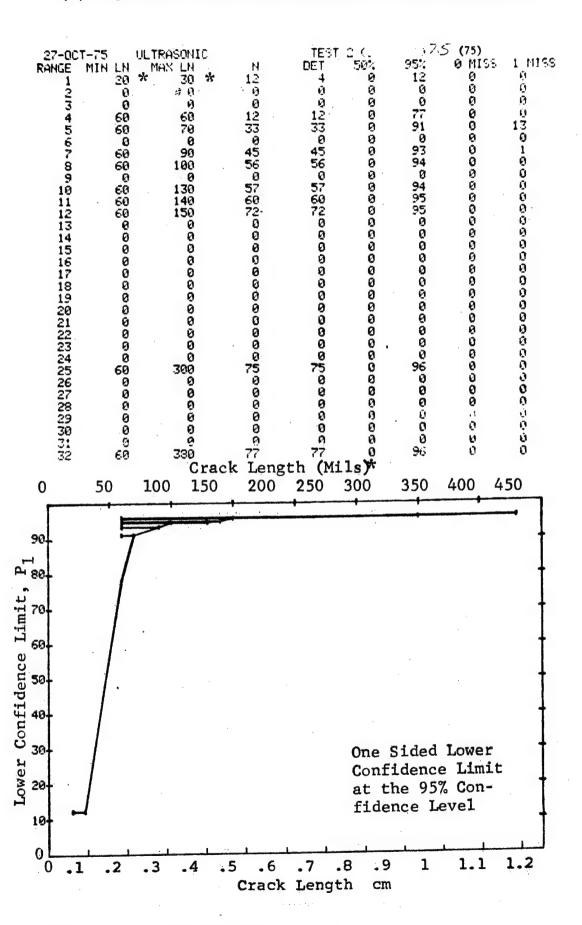


Figure D-75 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

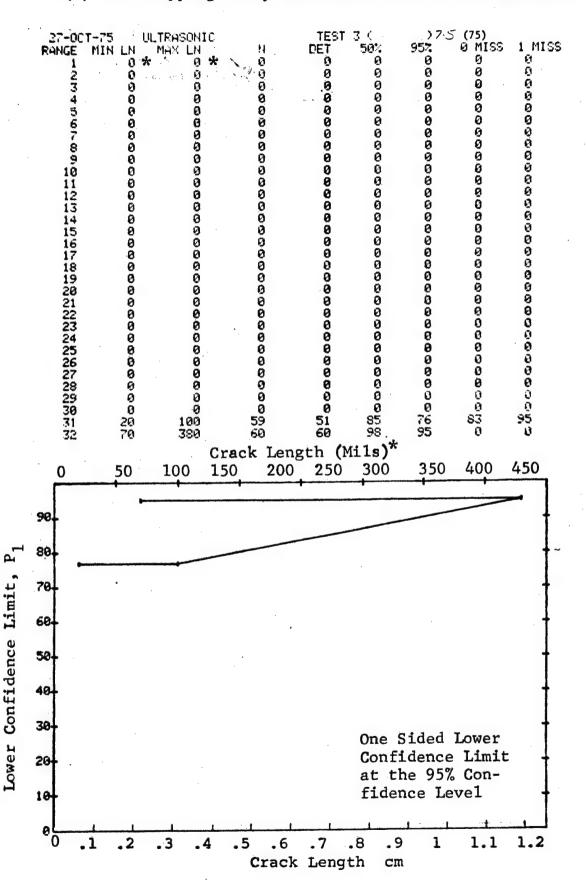


Figure D-75 (Concluded)

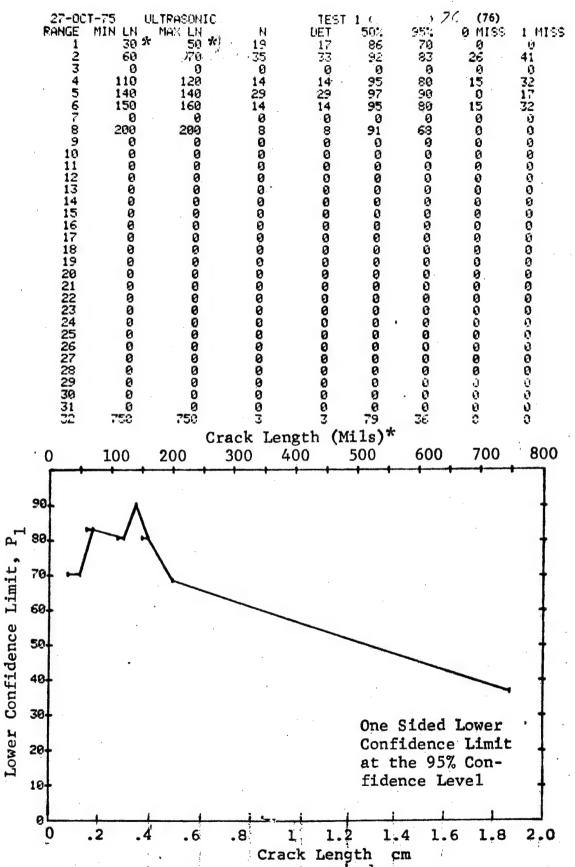


Figure D-76 Probability of Detection for 4340M Steel Using Ultrasonic Shear and Surface Waves. Compressed Notch Flaws in Filleted Solid Cylinder. Prod. Env.

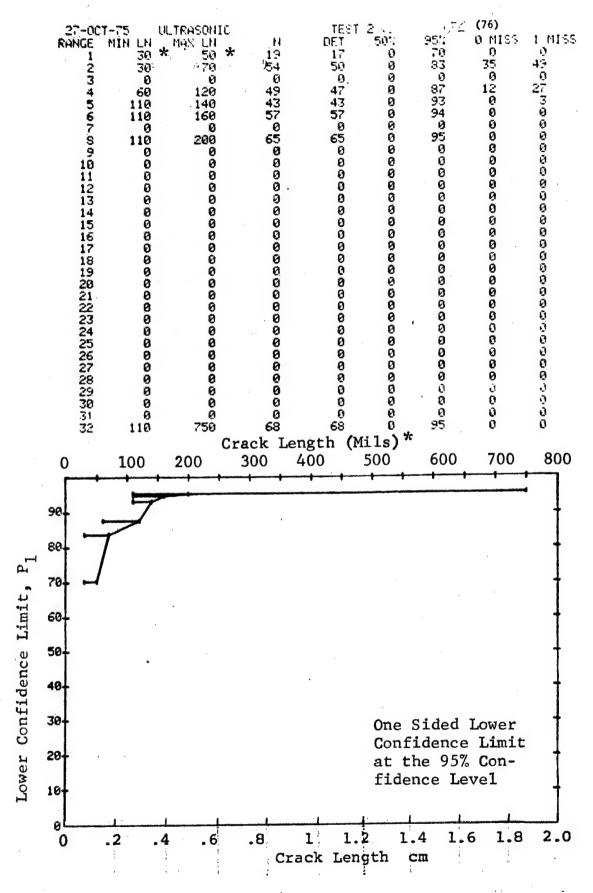


Figure D-76 (Continued)

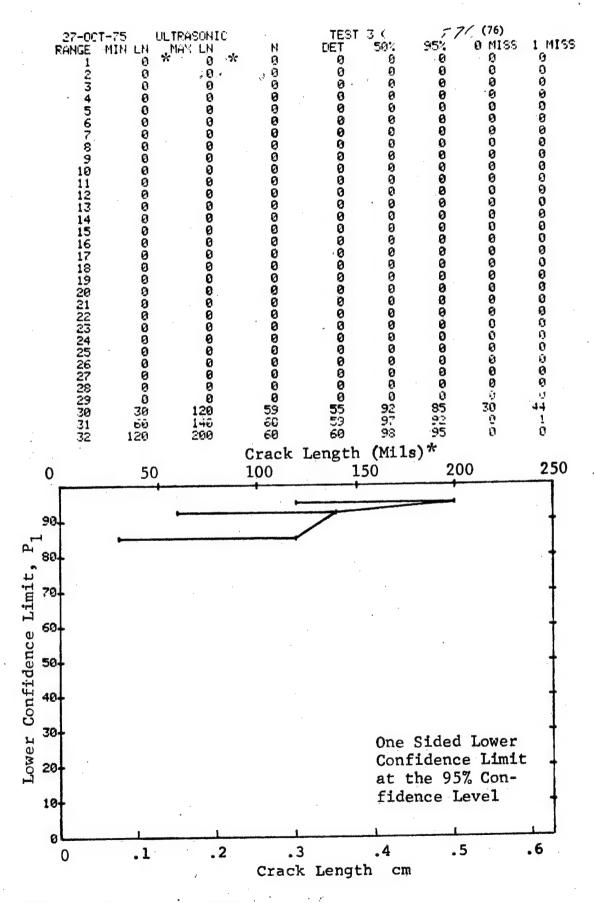


Figure D-76 (Concluded)

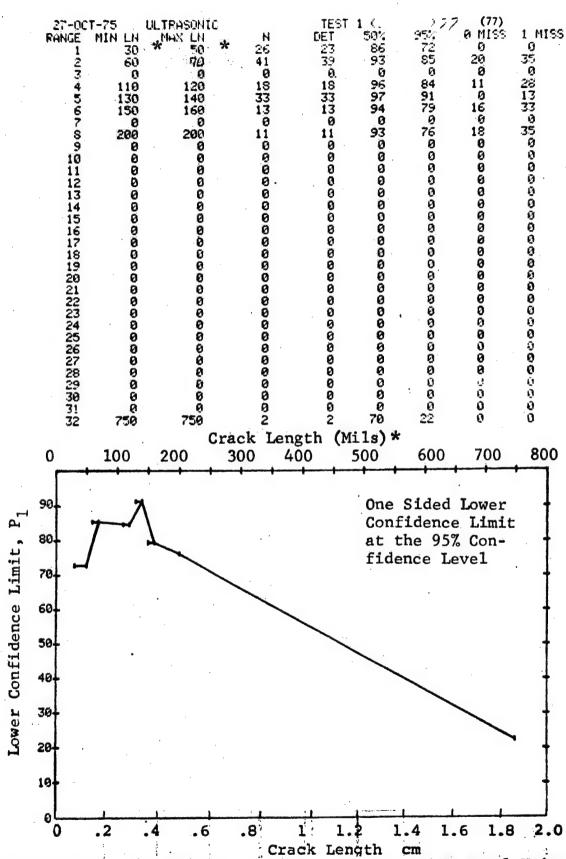


Figure D-77 Probability of Detection for 4340M Steel Using Ultrasonic Shear and Surface Waves, Compressed Notch Flaws in Filleted Solid Cylinder. Lab. Env.

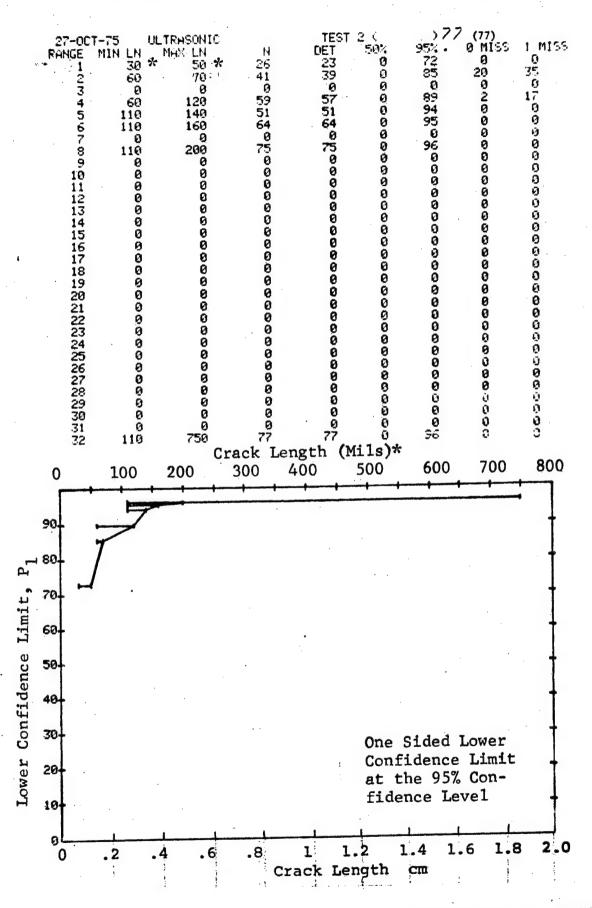


Figure D-77(Continued)

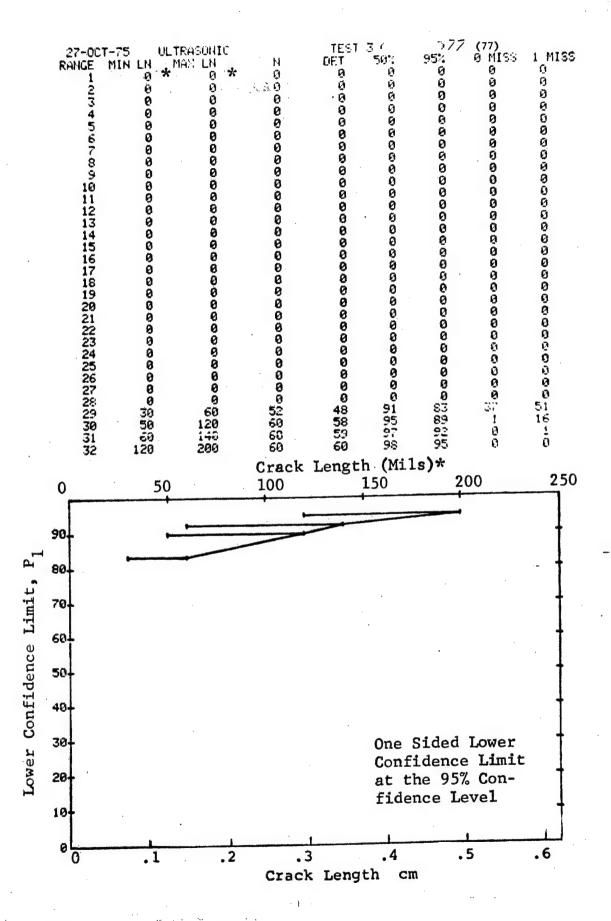


Figure D-77 (Concluded)

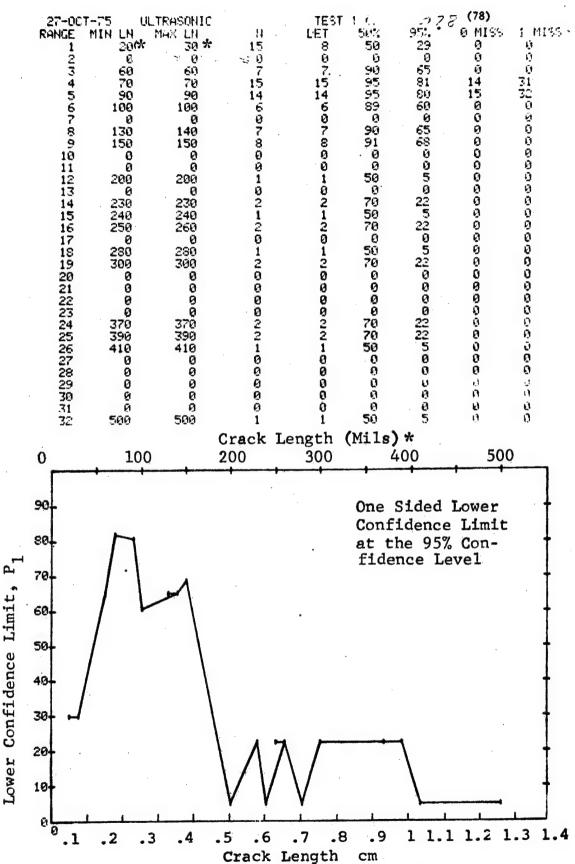


Figure D-78 Probability of Detection for 4340M Steel Using
Ultrasonic Shear and Surface Waves. Compressed
Notch Flaws in Filleted Solid Cylinder. Lab. Env.
D-240

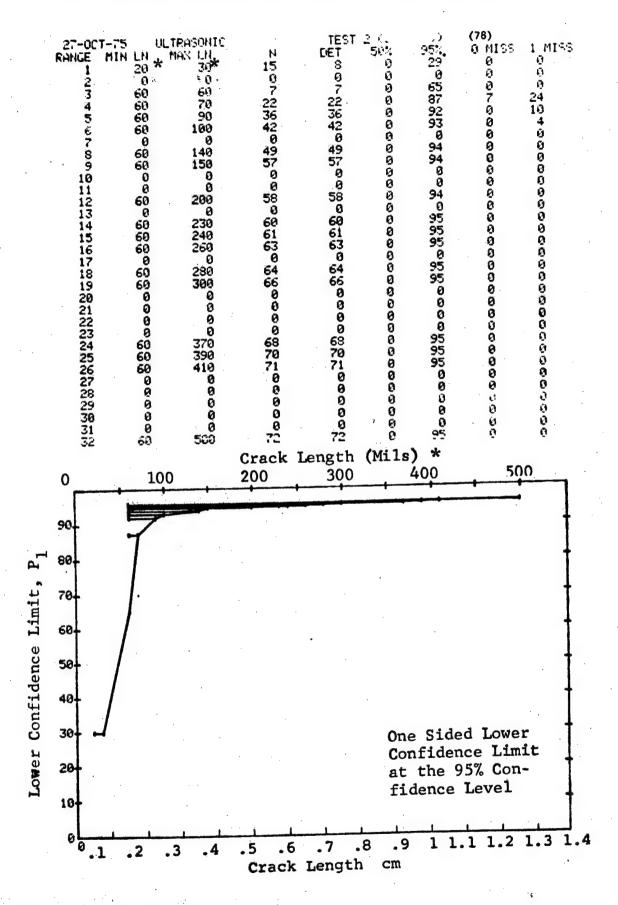


Figure D-78 (Continued)

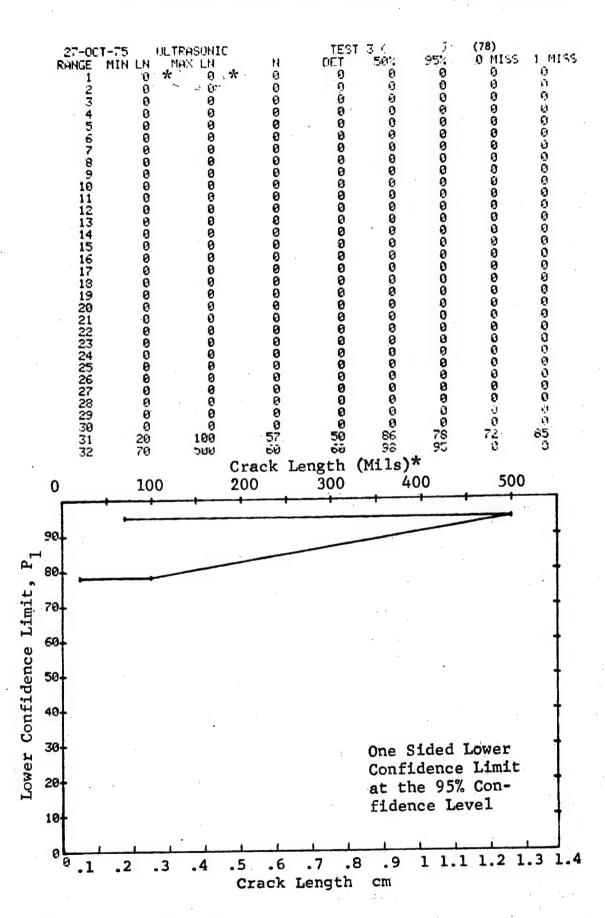


Figure D-78 (Concluded)

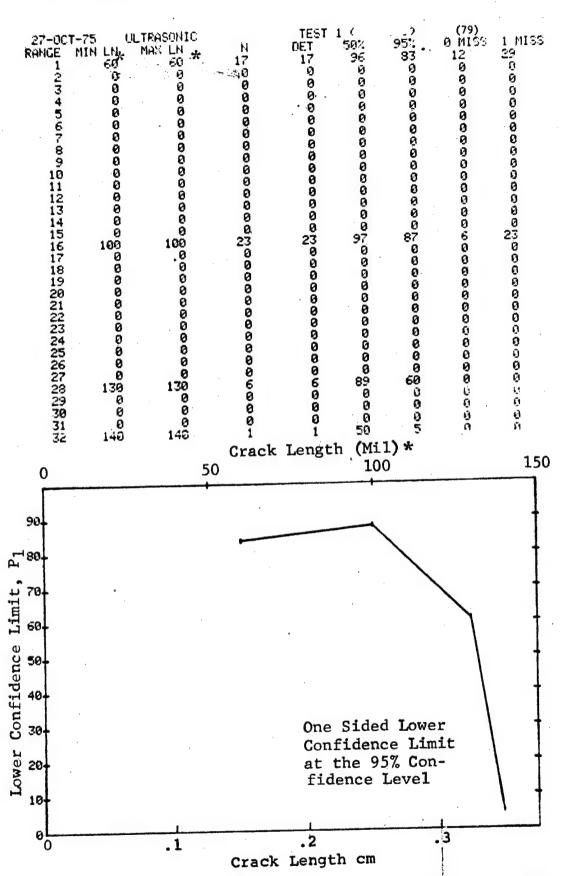


Figure D-79 Probability of Detection for 4340M Steel Using Ultrasonic Shear and Surface Waves. Compressed Notch Flaws in Hollow Cylinder. Lab. Env. D-243

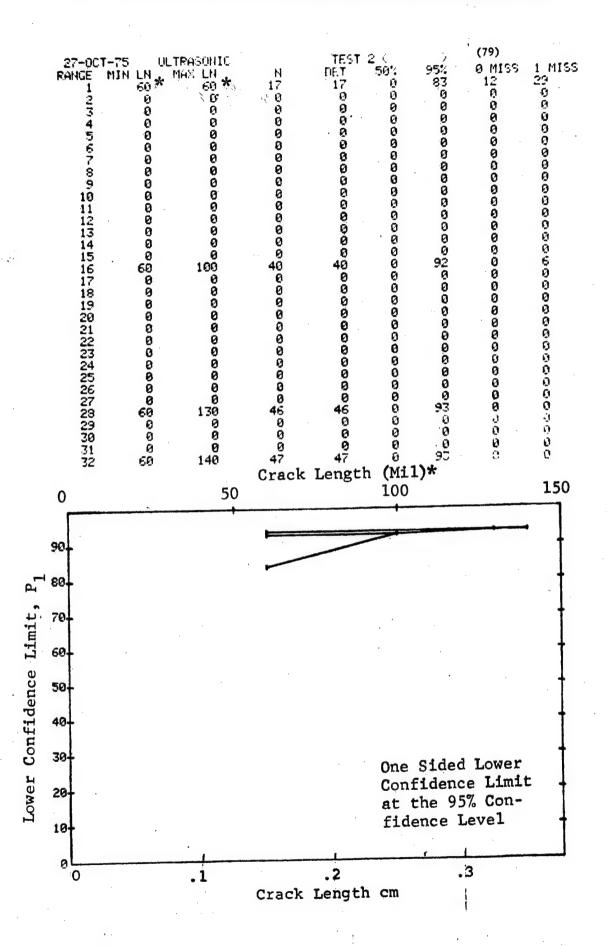


Figure D-79 (Continued)

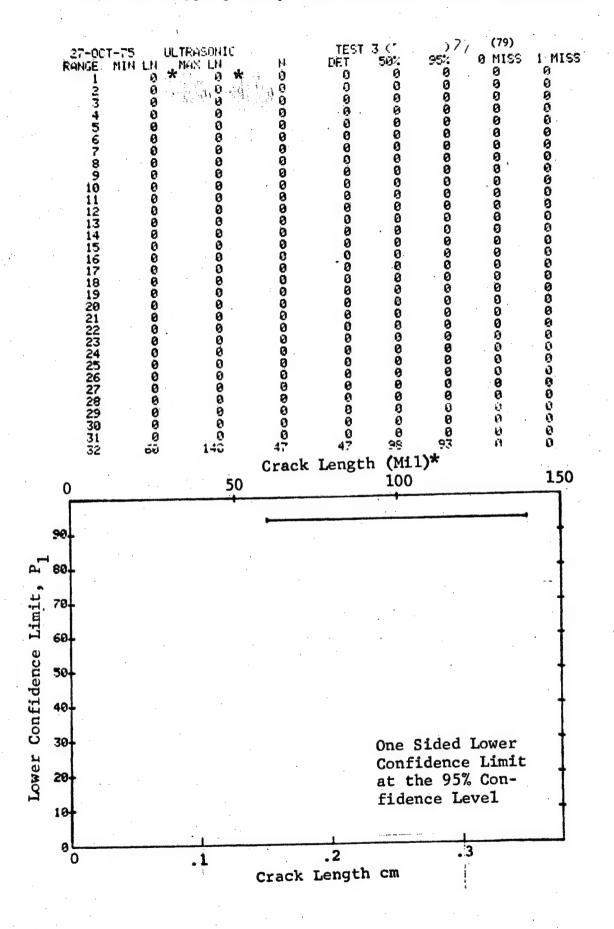


Figure D-79 (Concluded)

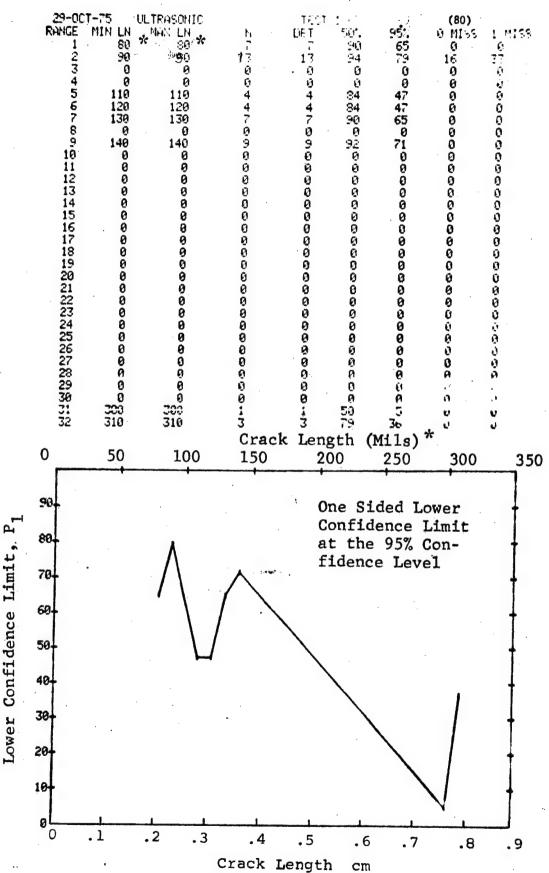


Figure D-80 Probability of Detection for 4340M Steel Using Ultrasonic Shear and Surface Waves. Compressed Notch Flaws in Filleted Hollow Cylinder. Lab. Env.

(b) Optimum Probability Method of Data Cumulation

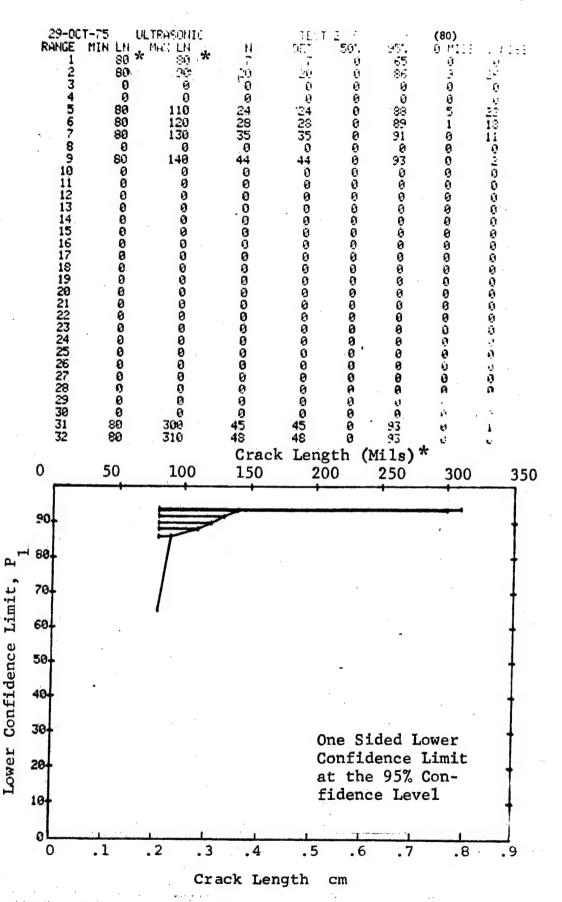


Figure D-80 (Continued)

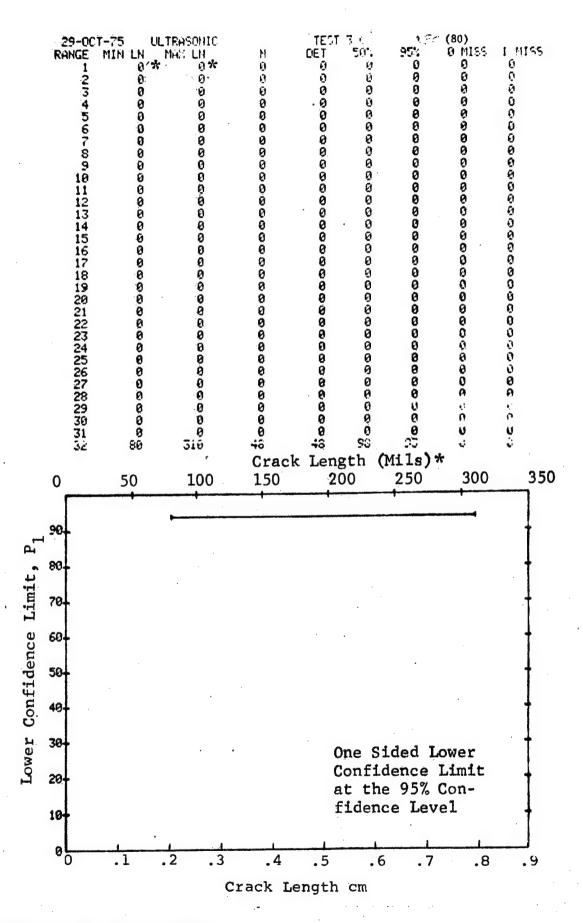


Figure D-80 (Concluded)

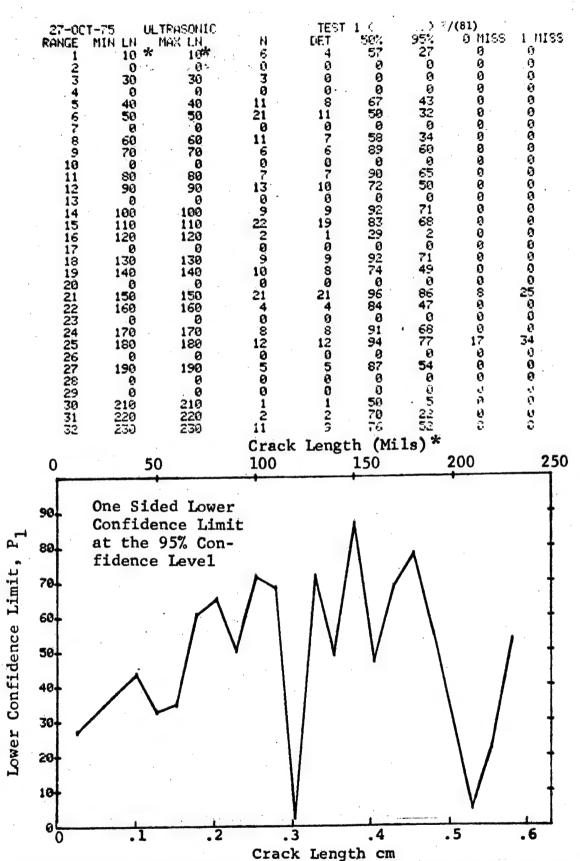


Figure D-81 Probability of Detection for 2024-T6 Al Using Ultrasonic Shear and Surface Waves. Compressed Notch Flaws in Tandem T Specimen. Lab. Env.

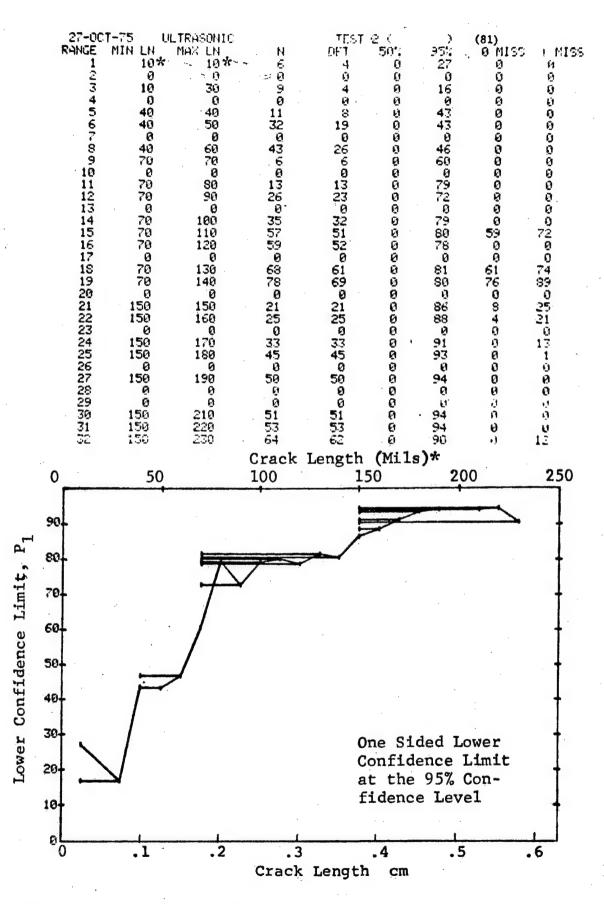


Figure D-81 (Continued)

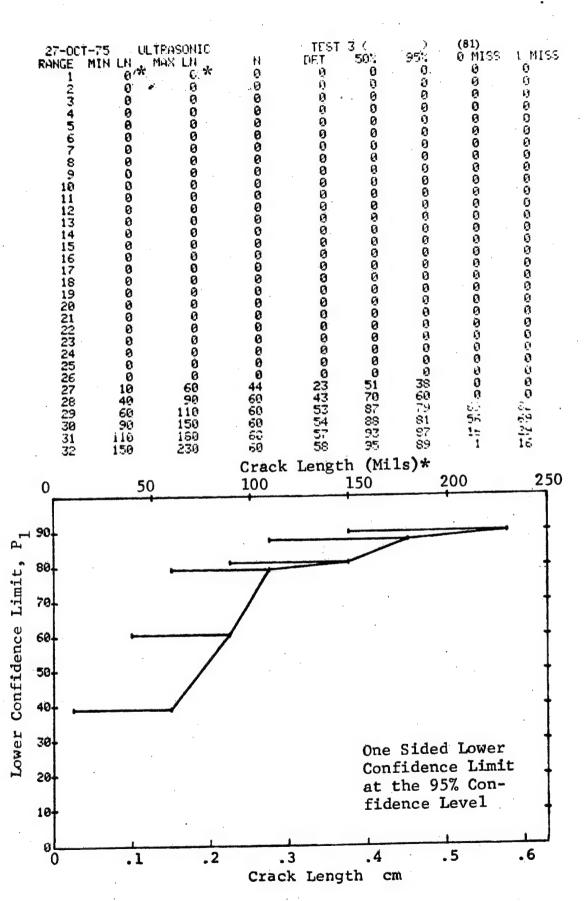
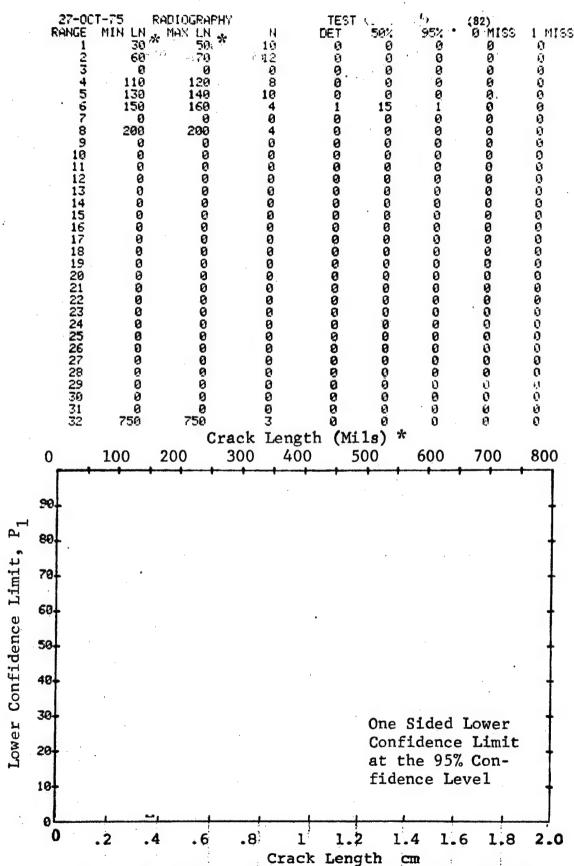
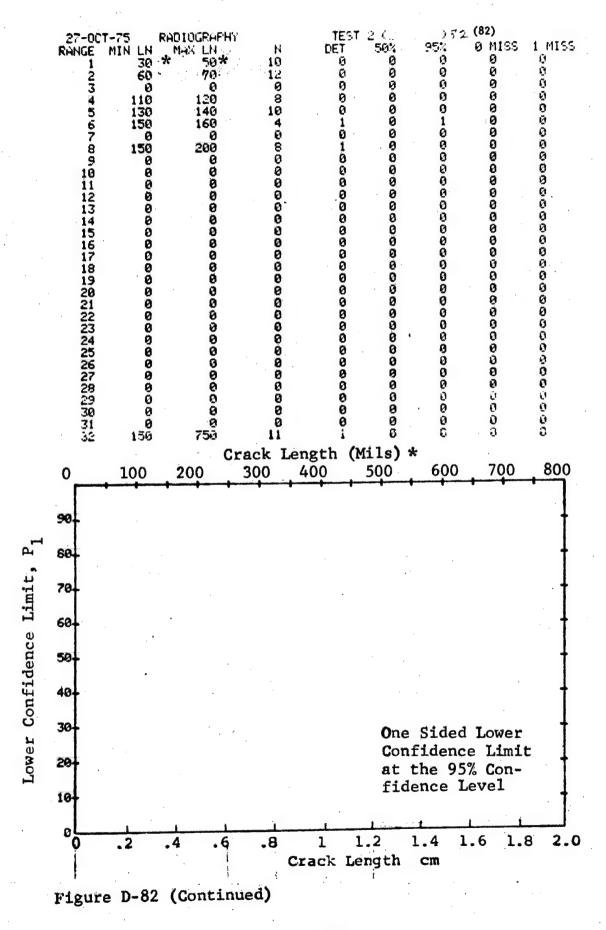


Figure D-81 (Concluded)



Probability of Detection for 4340M Steel Using Figure D-82 Compressed Notch Flaws in Solid Cylinder. X-ray. REPRODUCIBILITY OF THE Prod. Env. ORIGINAL PAGE IS POOR

n-252



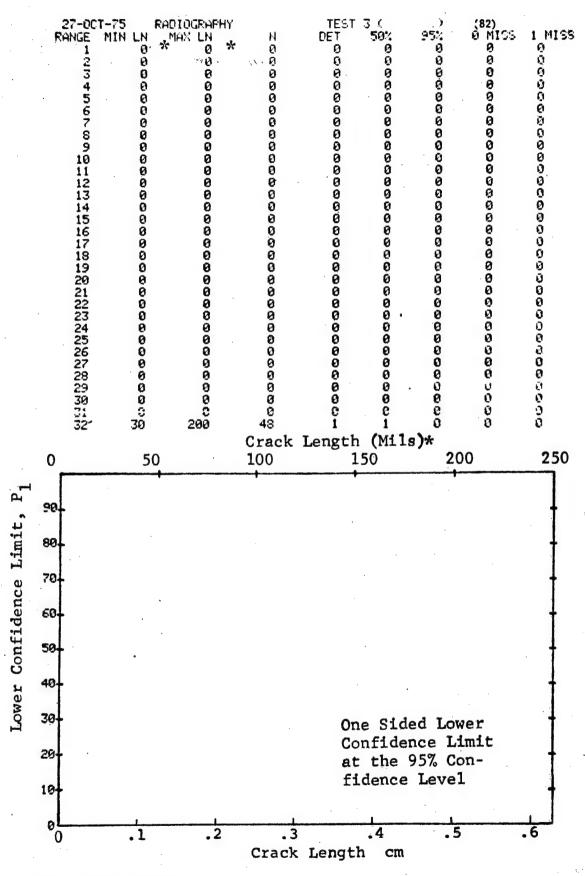


Figure D-82 (Concluded)

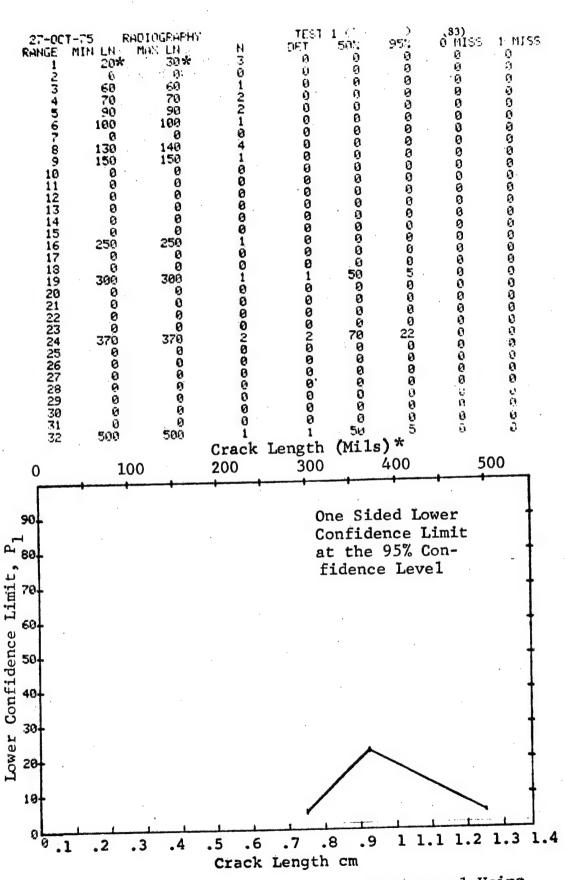


Figure D-83 Probability of Detection for 4340M Steel Using X-ray. Compressed Notch Flaws in Filleted Solid Cylinder. D-255 Prod. Env.

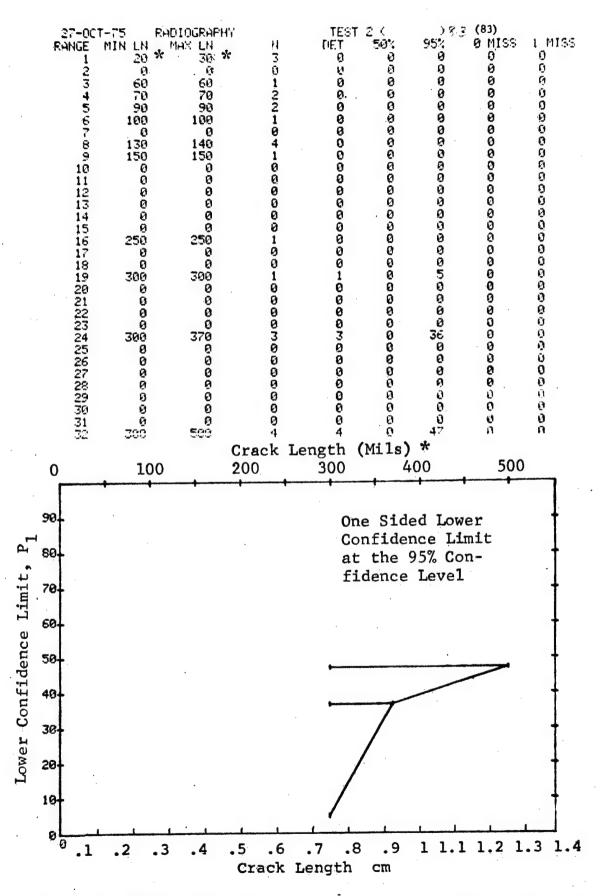


Figure D-83 (Continued)

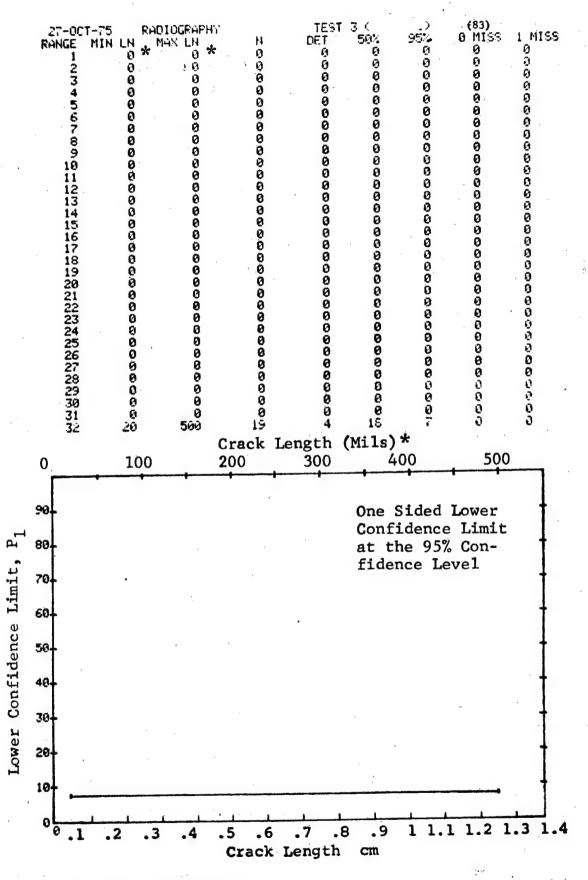


Figure D-83 (Concluded)

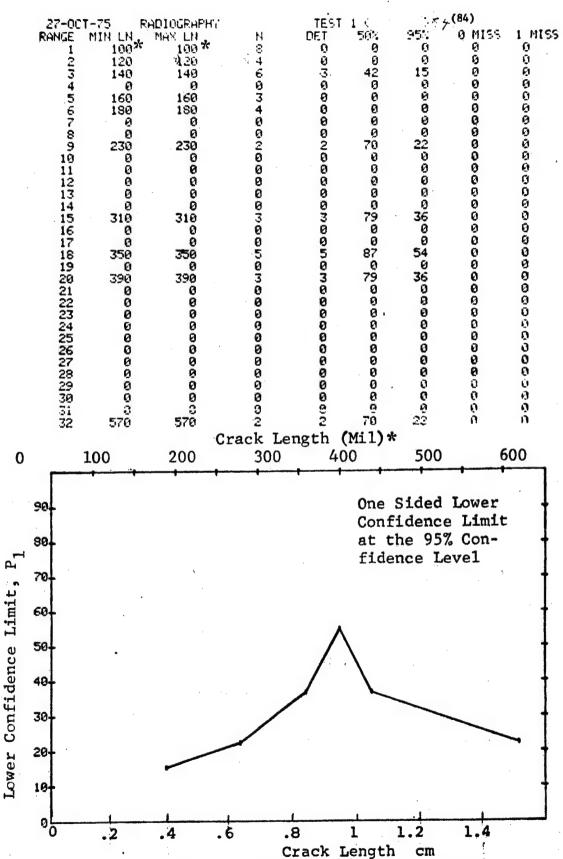


Figure D-84 Probability off Detection for 4340M Steel Using X-ray. Compressed Notch Flaws in Hollow Filleted Cylinder. Prod. Env. D-258

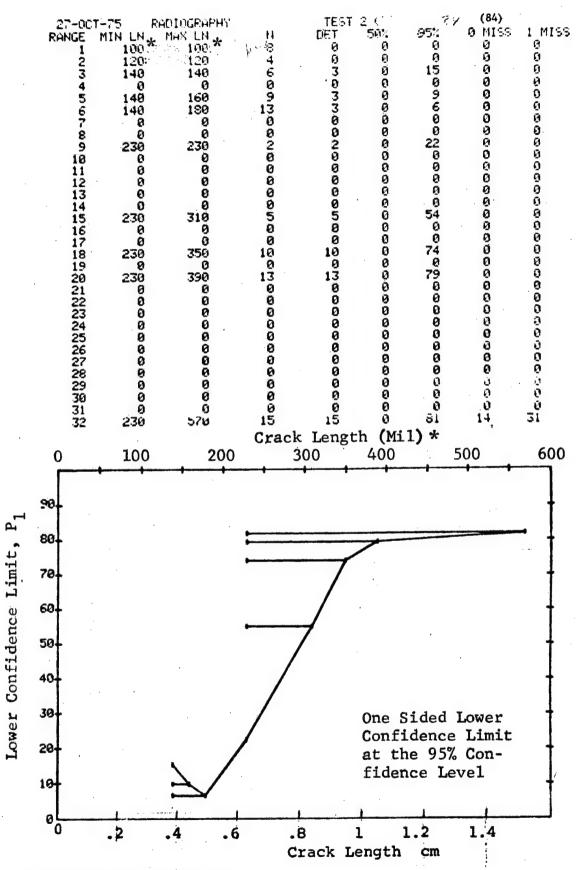


Figure D-84 (Continued)

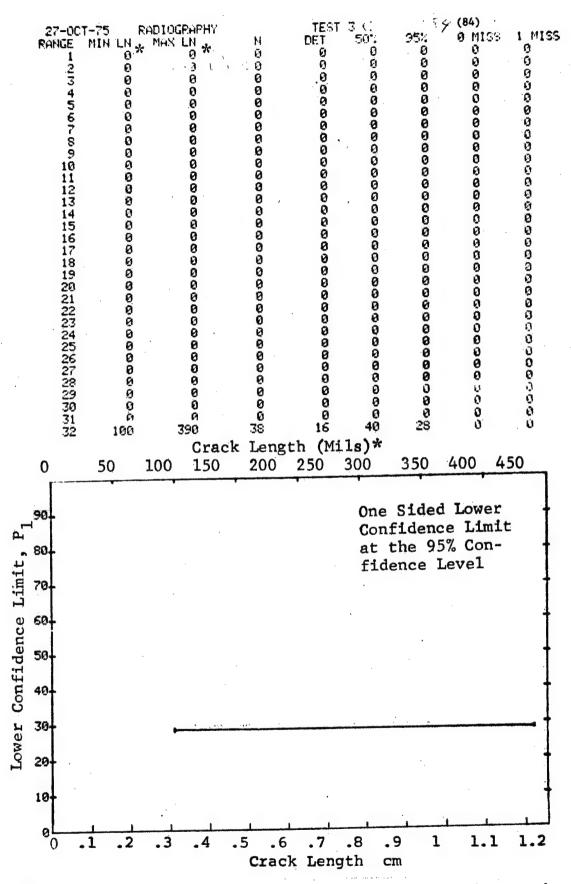


Figure D-84 (Concluded)

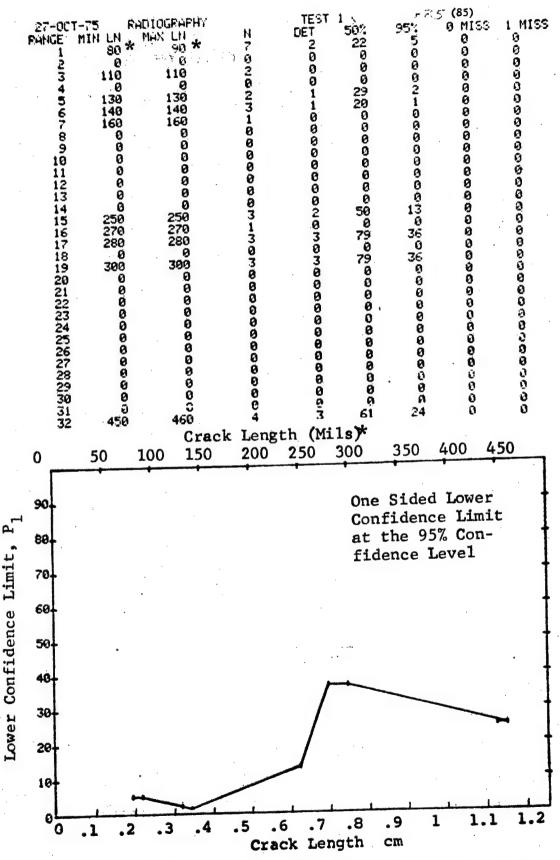


Figure D-85 Probability of Detection for 4340M Steel Using X-ray. Compressed Notch Flaws in Hollow Filleted Cylinder. Prod. Env.

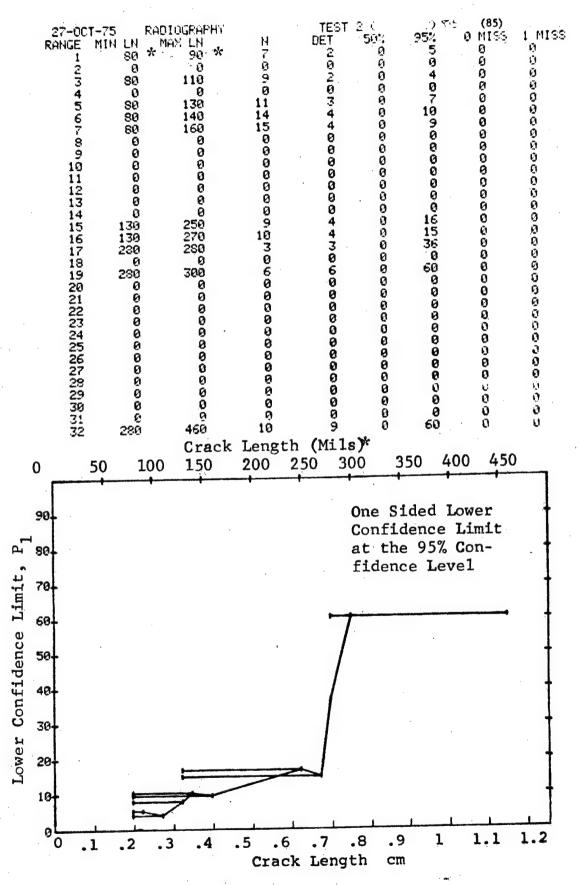


Figure D-85 (Continued)

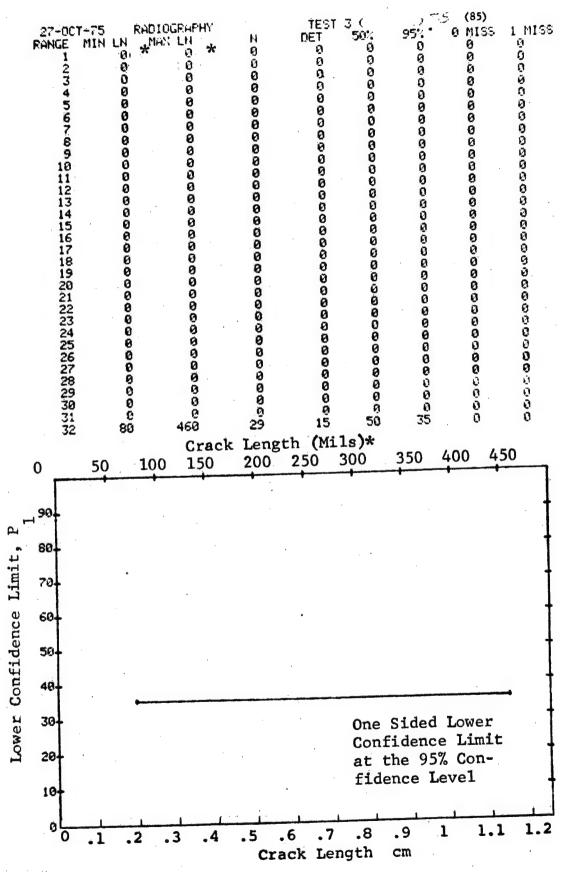


Figure D-85 (Concluded)

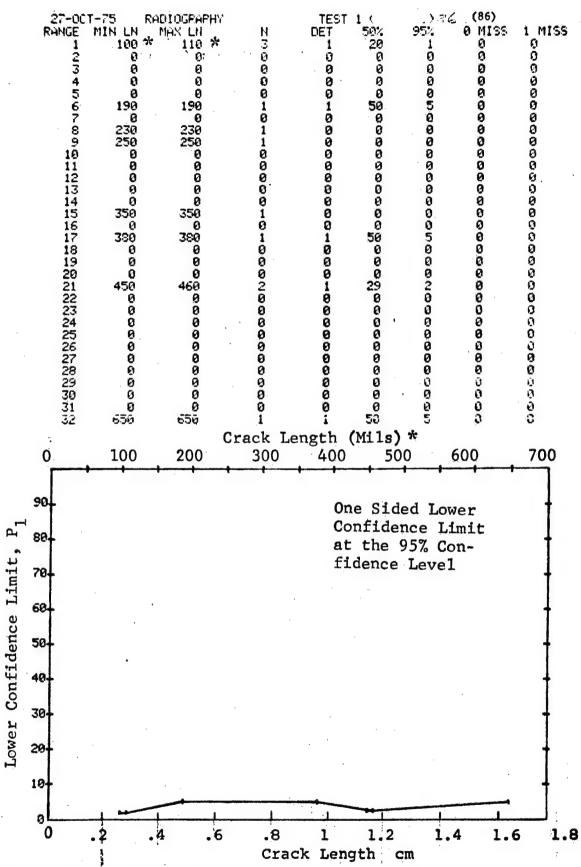
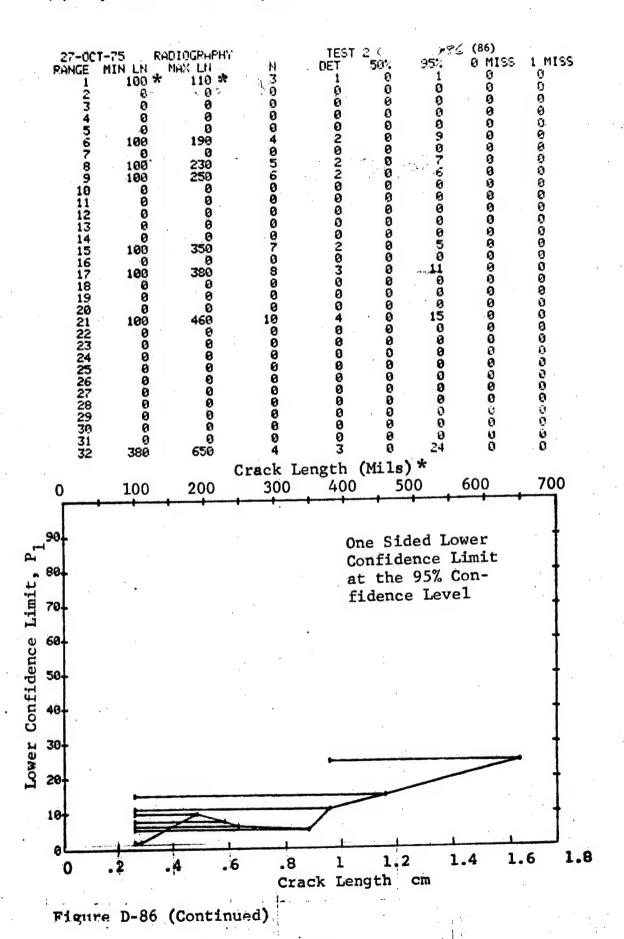


Figure D-86 Probability of Detection for 4340M Steel Using X-ray.

Compressed Notch Flaws in Solid Threaded Cylinder.

Prod. Env.

(b) Optimum Probability Method of Data Cumulation



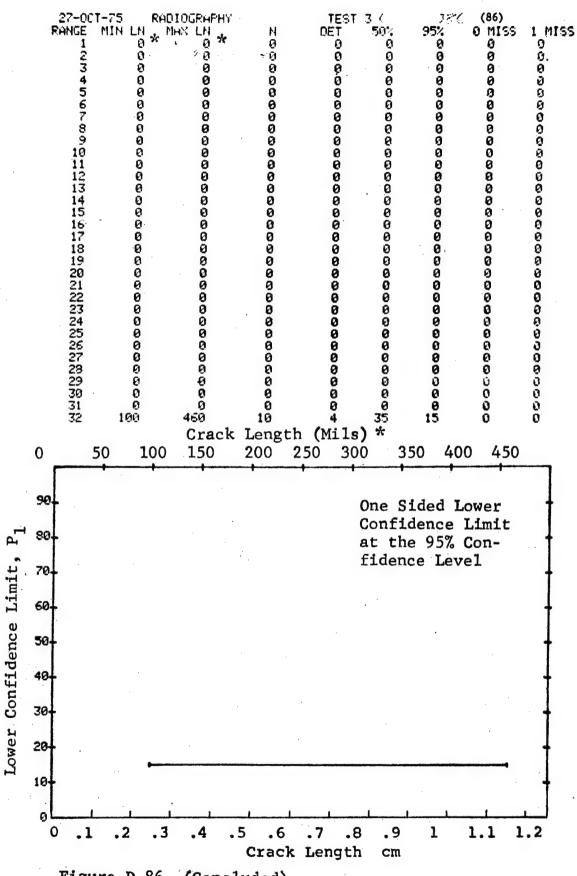


Figure D-86 (Concluded)

Figure D-87 Probability of Detection for 2024-T6 Al Using X-ray. Compressed Notch Flaws in Tandem-T Specimen. Prod. Env.

D-267

Lower Confidence Limit, P

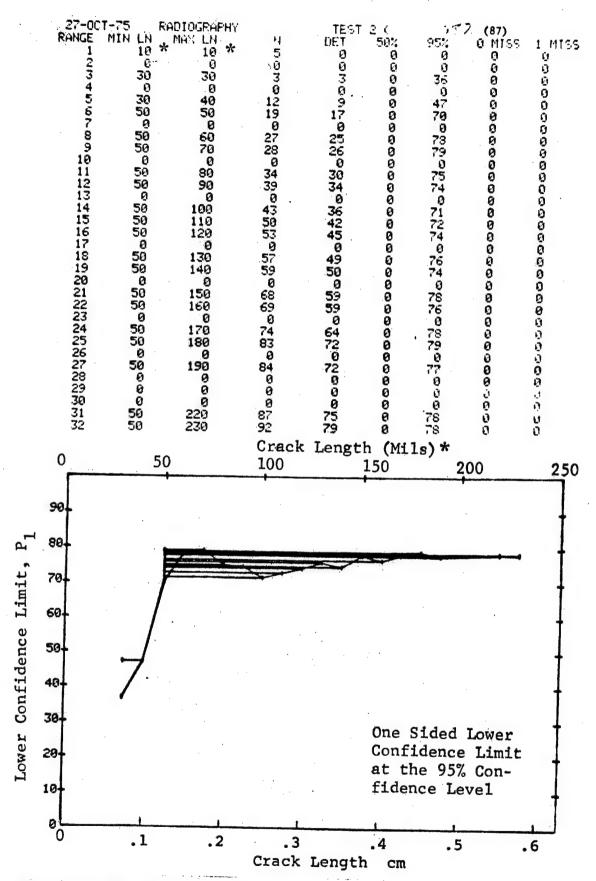


Figure D-87 (Continued)

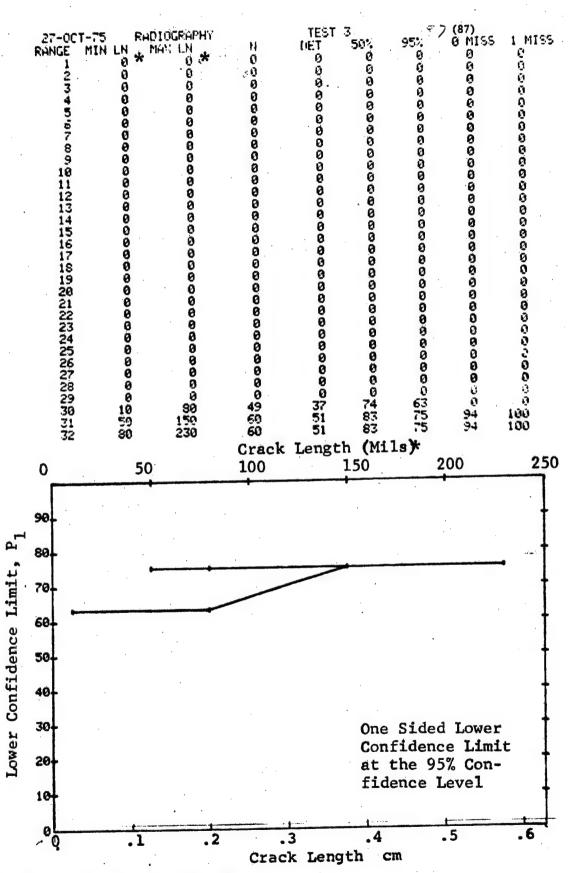


Figure D-87 (Concluded)

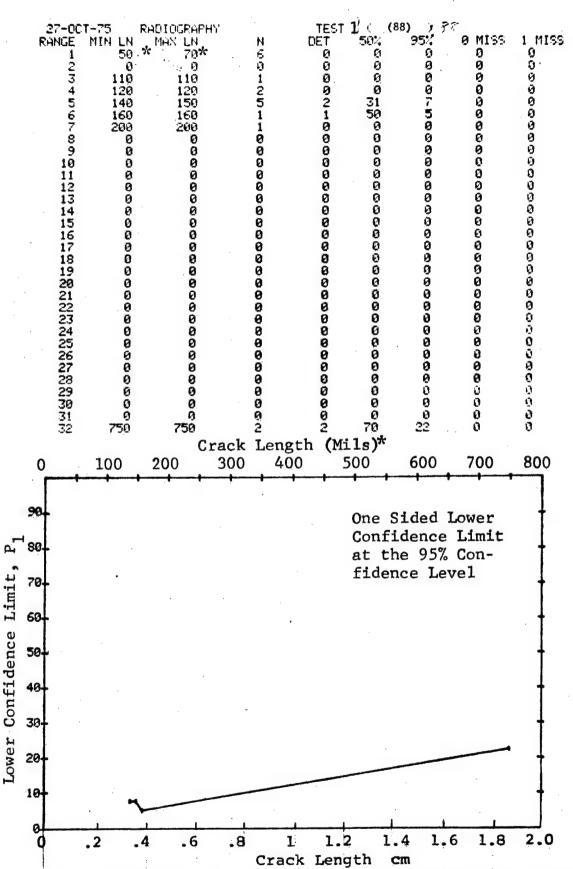
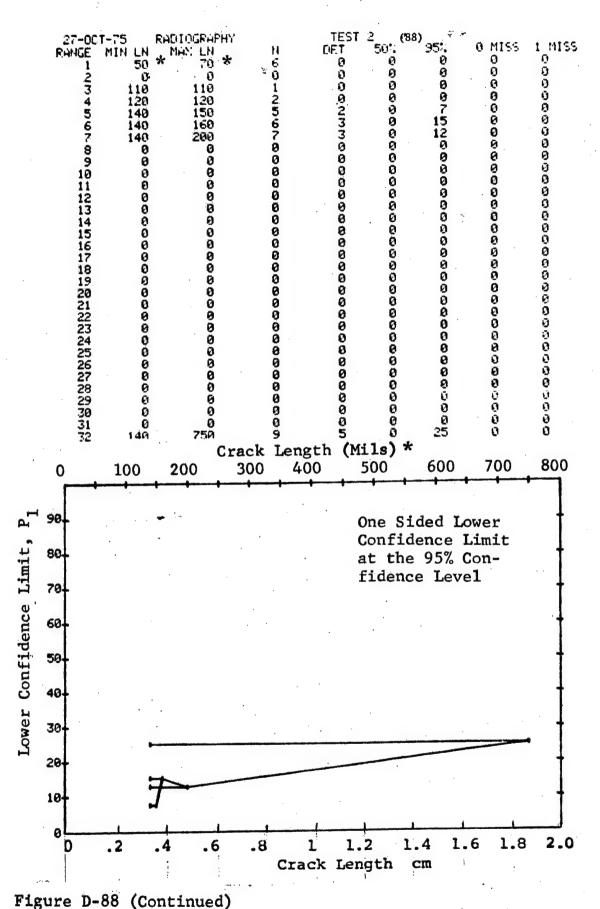


Figure D-88 Probability of Detection for 4340M Steel Using X-ray. Compressed Notch Flaws in Solid Cylinder. Lab. Env.

D-270 REPRODUCIBILITY OF THE



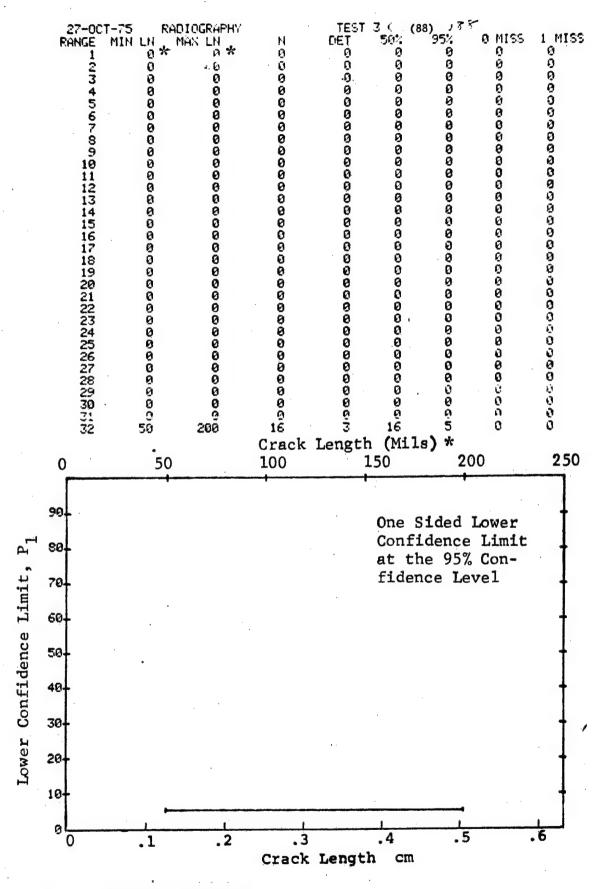


Figure D-88 (Concluded)

(a) Range Interval Method of Data Cumulation

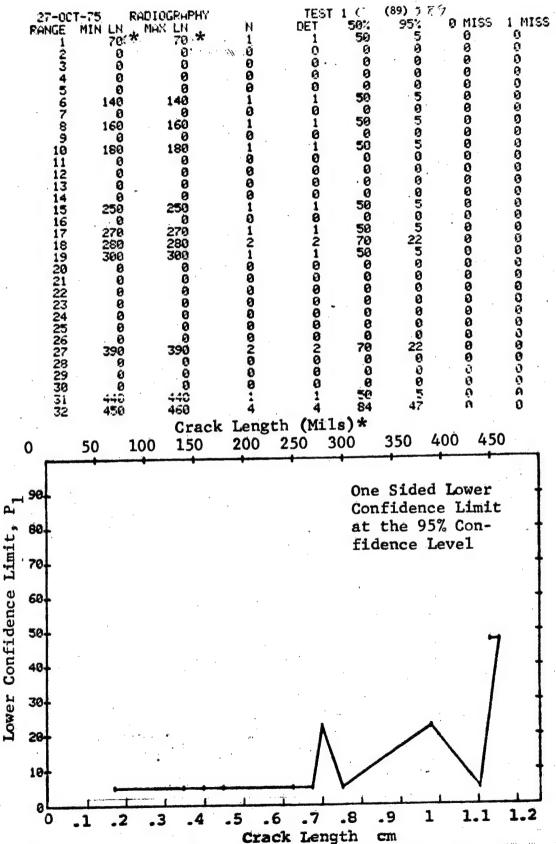


Figure D-89 Probability of Detection for 4340M Steel Using X-ray.

Compressed Notch Flaws in Hollow Filleted Cylinder.

Lab. Env. D-273

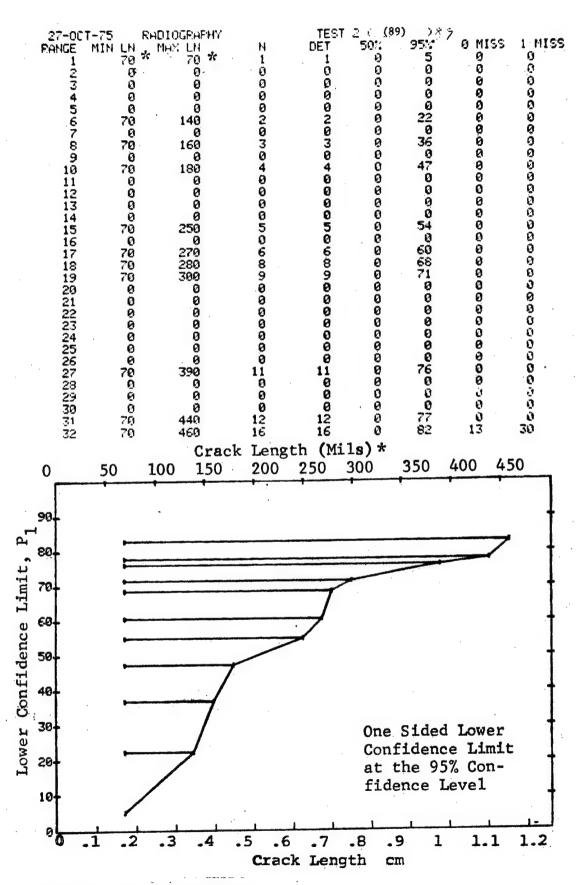


Figure D-89 (Continued)

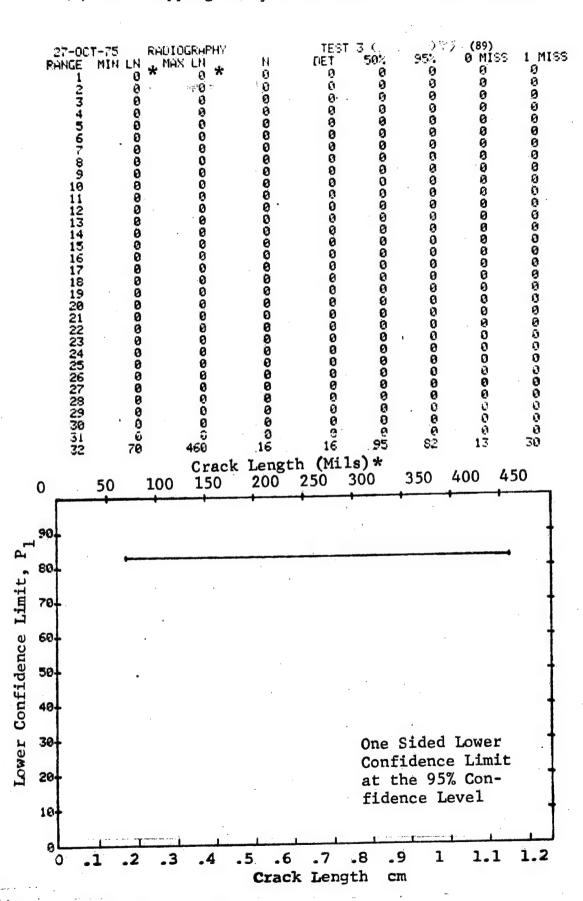


Figure D-89 (Concluded)

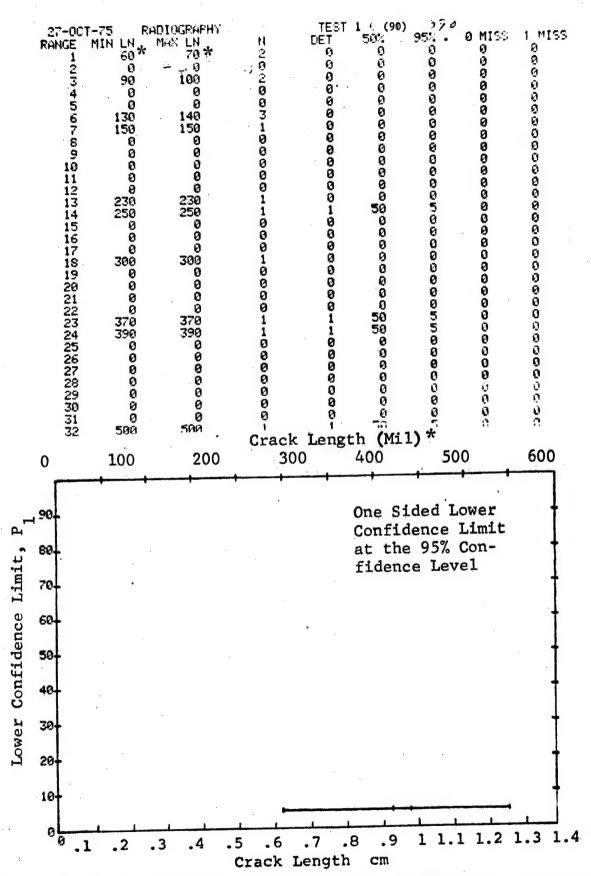


Figure D-90 Probability of Detection for 4340M Steel Using X-ray.

Compressed Notch Flaws in Solid Filleted Cylinder.

Lab. Env.

D-276

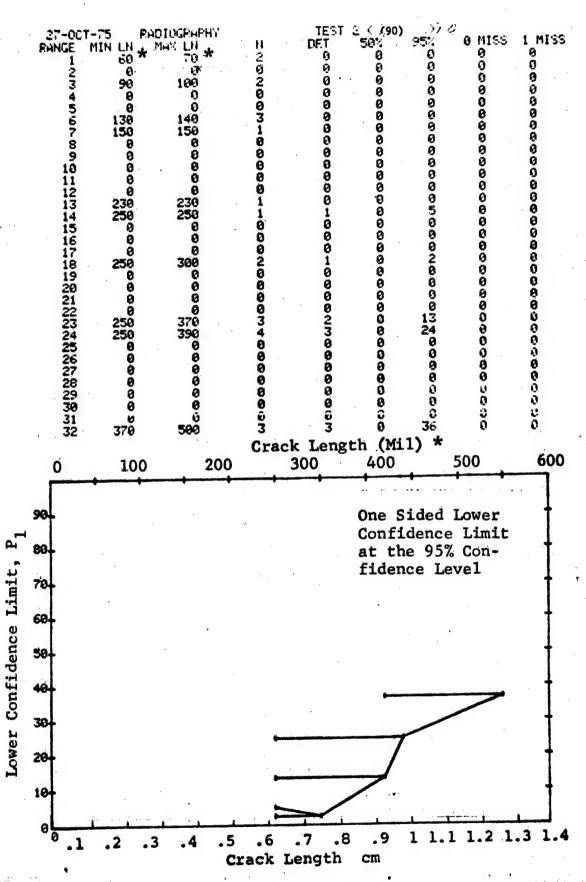


Figure D-90 (Continued)

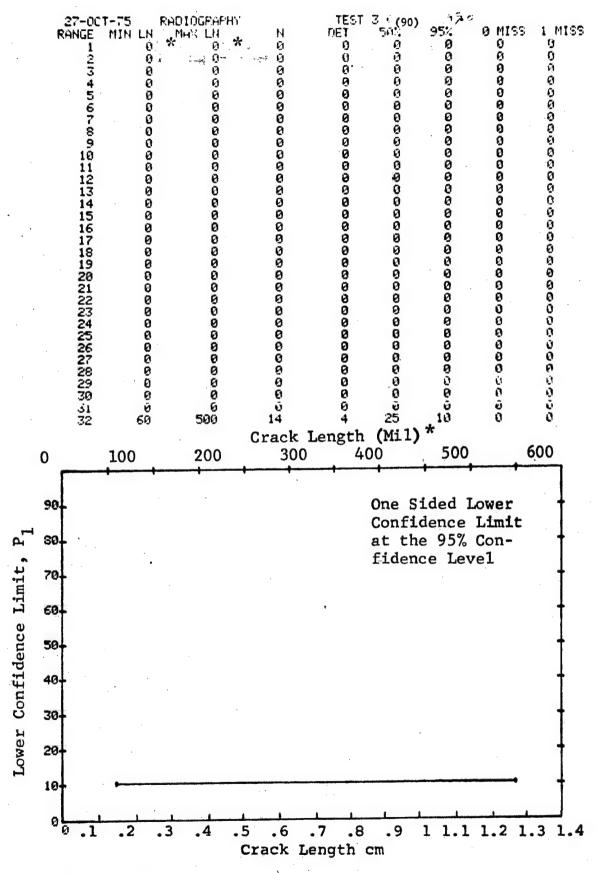


Figure D-90 (Concluded)

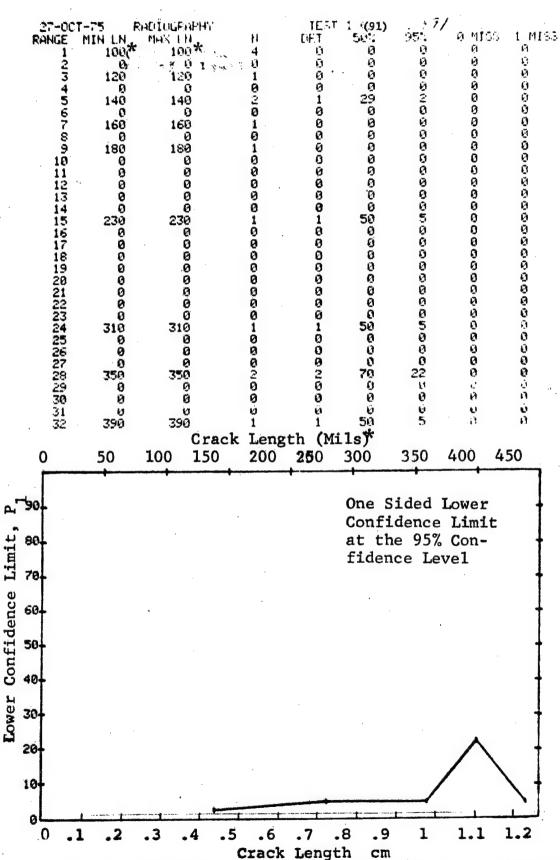


Figure D-91 Probability of Detection for 4340M Steel Using X-ray.

Compressed Notch Flaws in Hollow Filleted Cylinder.
Lab. Thv.

D-279

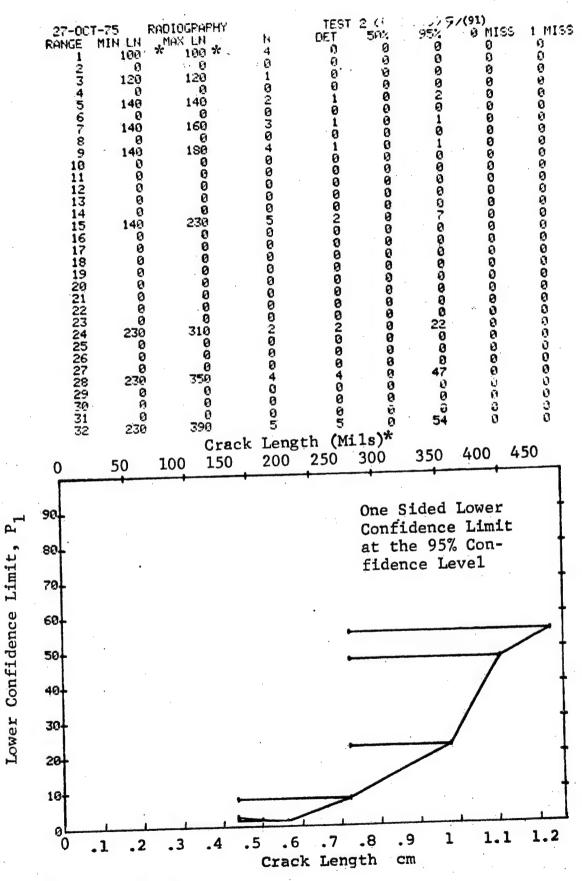


Figure D-91 (Continued)

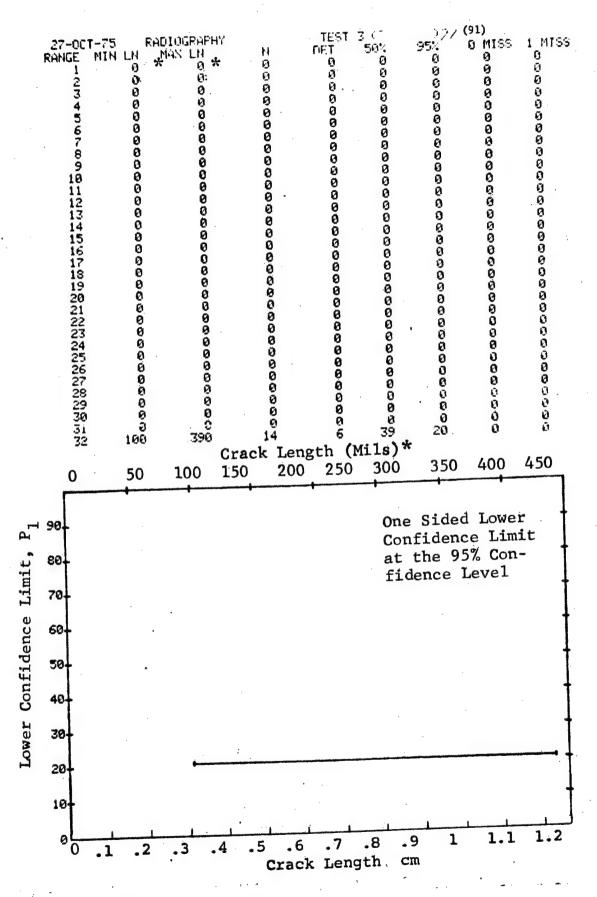


Figure D-91 (Concluded)

(a) Range Interval Method of Data Cumulation

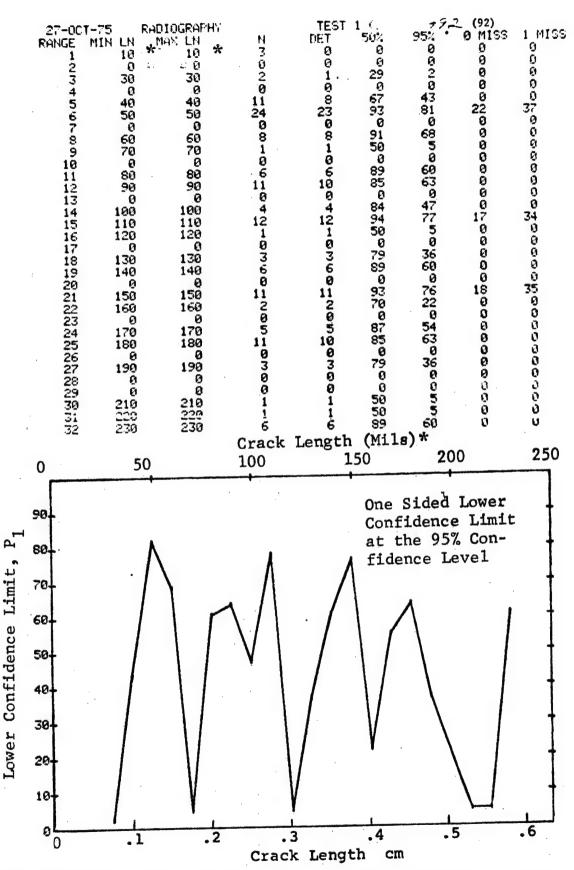


Figure D-92 Probability of Detection for 2024-T6 Al Using X-ray.
Compressed Notch Flaws in Tandem T Specimen. Lab. Env.

(b) Optimum Probability Method of Data Cumulation

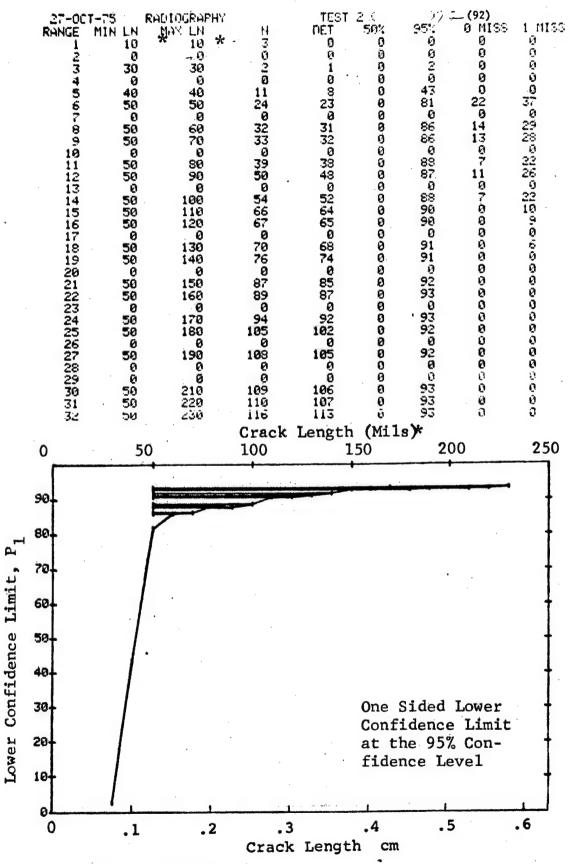


Figure D-92 (Continued)

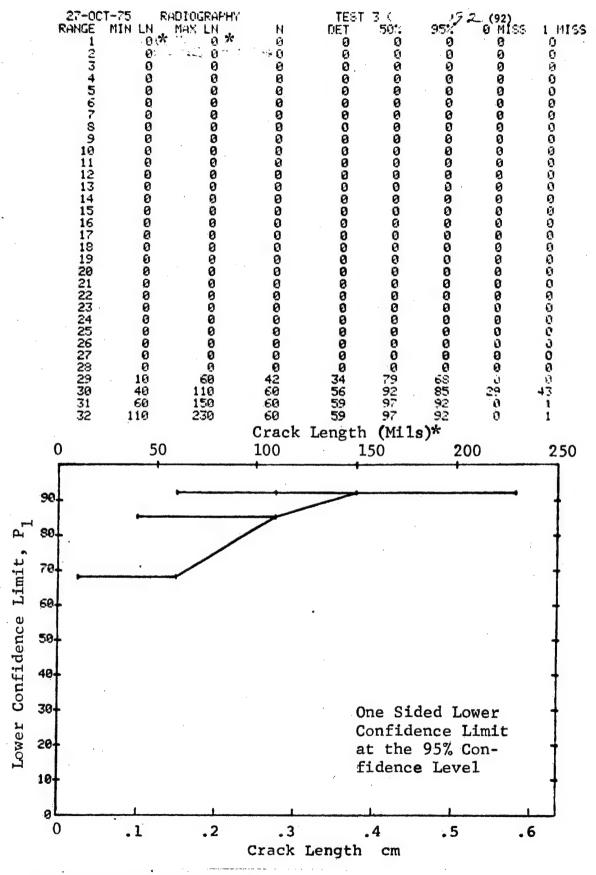
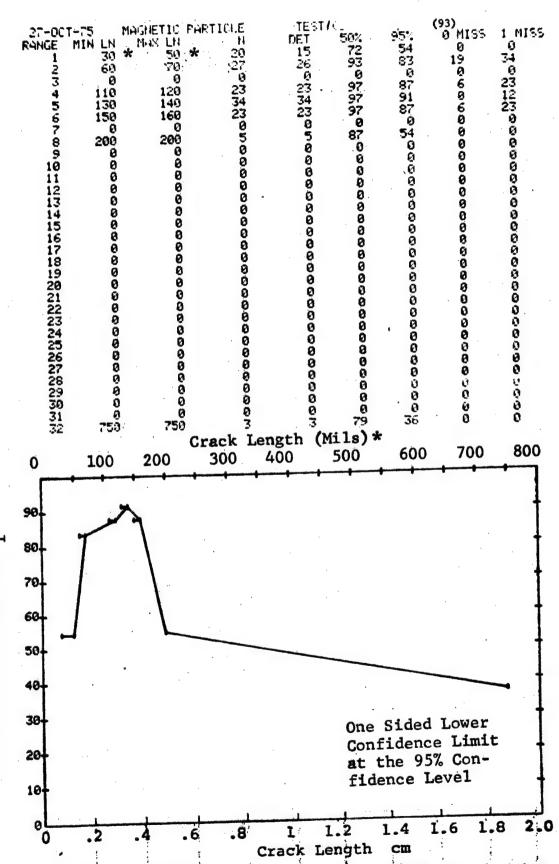


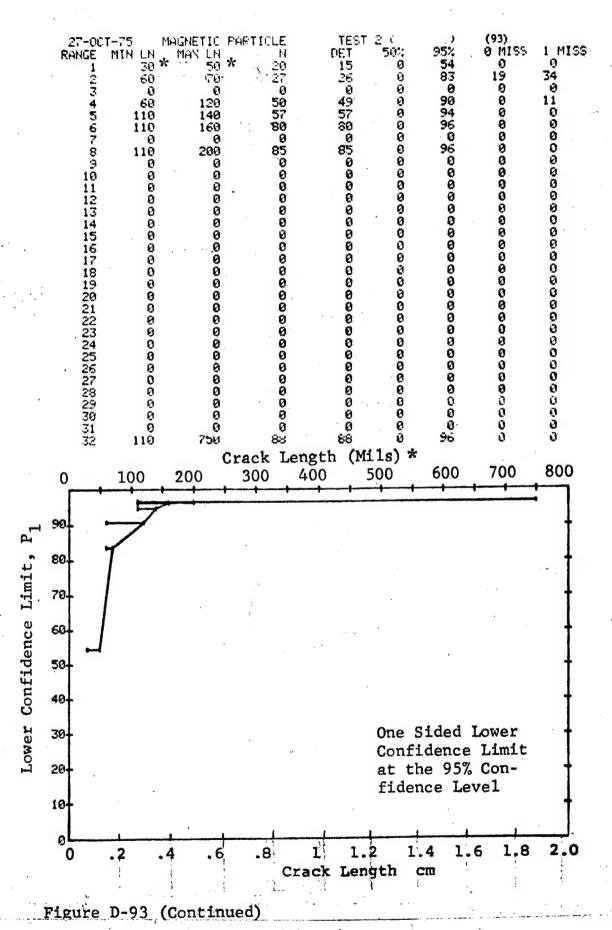
Figure D-92 (Concluded)

(a) Range Interval Method of Data Cumulation



Lower Confidence Limit,

Figure D-93 Probability of Detection for 4340M Steel Using Magnetic Particles. Compressed Notch Flaws in Solid Cylinder. Prod. Env.



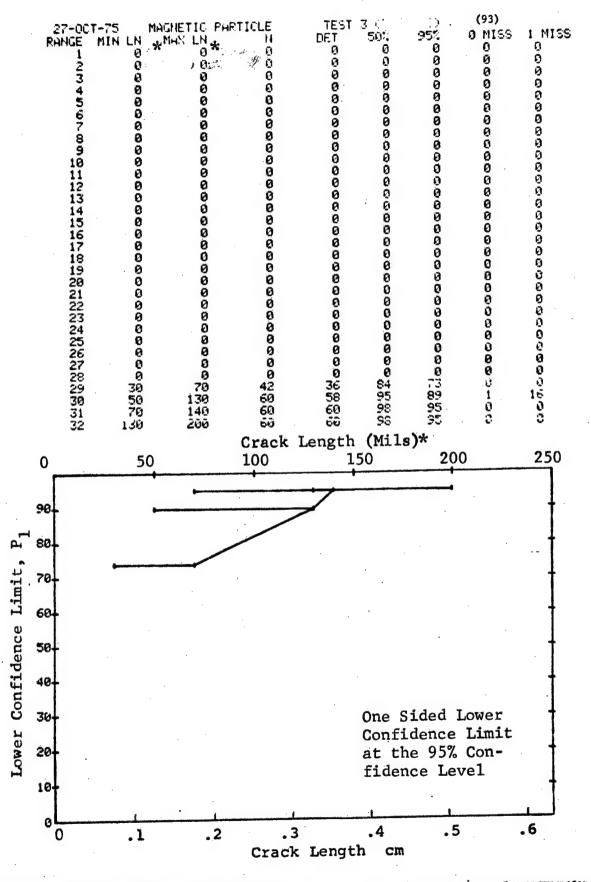


Figure D-93 (Concluded)

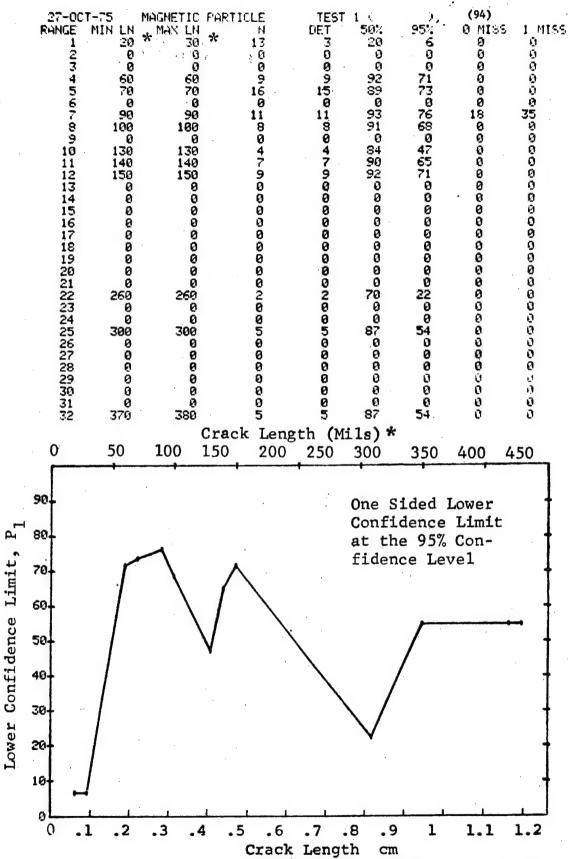


Figure D-94 Probability of Detection for 4340M Steel Using Magnetic Particles. Compressed Notch Flaws in Hollow Filleted Cylinder. Prod. Env.

(b) Optimum Probability Method of Data Cumulation

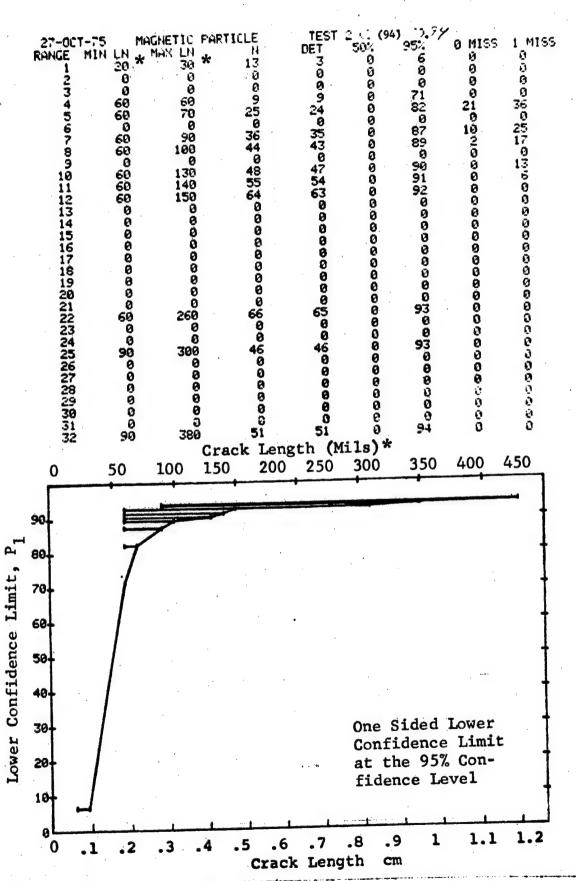


Figure D-94 (Continued)

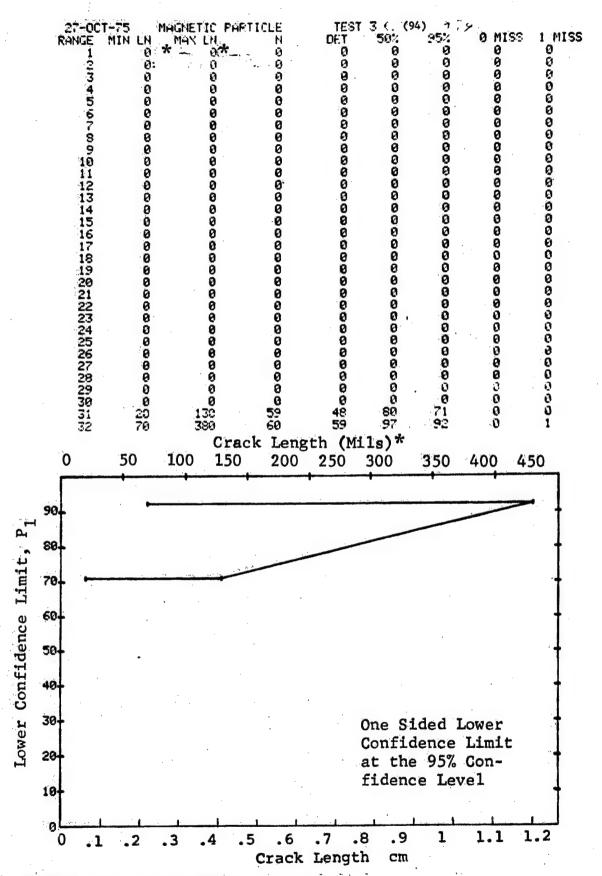
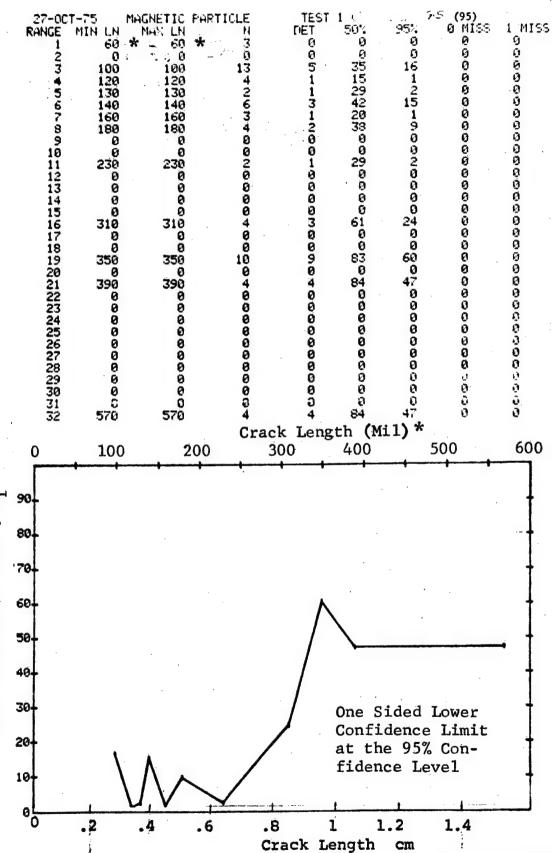
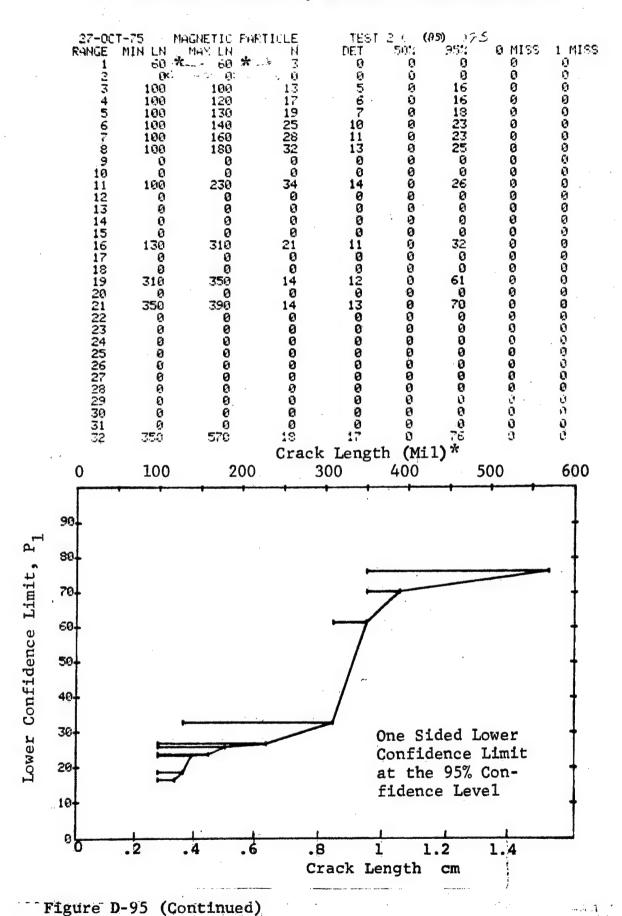


Figure D-94 (Concluded)



Lower Confidence Limit,

Figure D-95 Probability of Detection for 4340M Steel Using Magnetic Particles. Compressed Notch Flaws in Hollow Filleted Cylinder. Prod. Env. D-291



(c) Overlapping Sixty Point Method of Data Cumulation

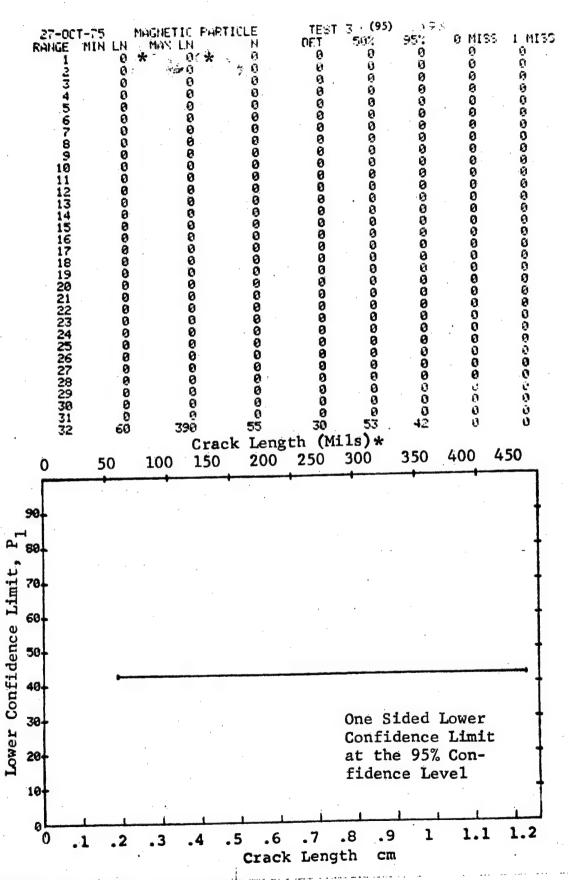


Figure D-95 (Concluded)

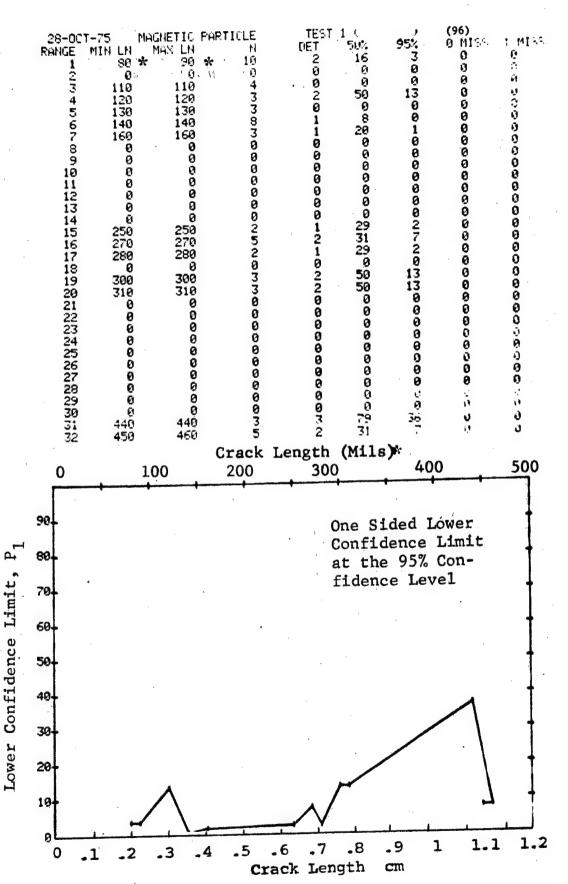


Figure D-96 Probability of Detection for 4340M Steel Using Magnetic Particles. Compressed Notch Flaws in Hollow Filleted Cylinder Prod. Env. D-294

(b) Optimum Probability Method of Data Cumulation

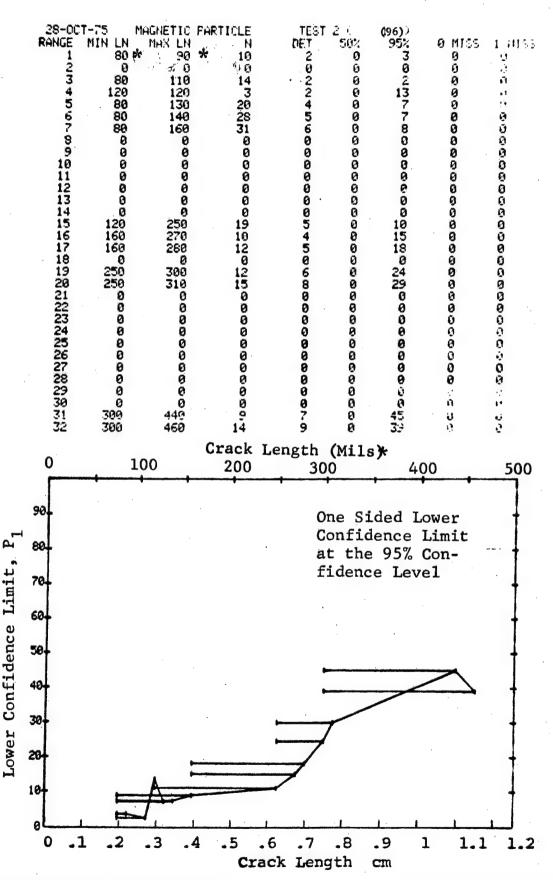


Figure D-96 (Continued)

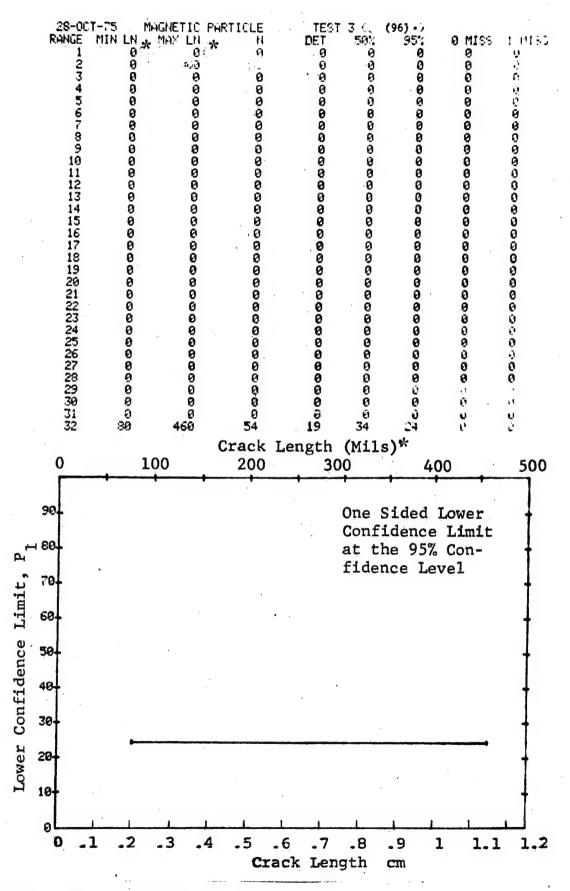
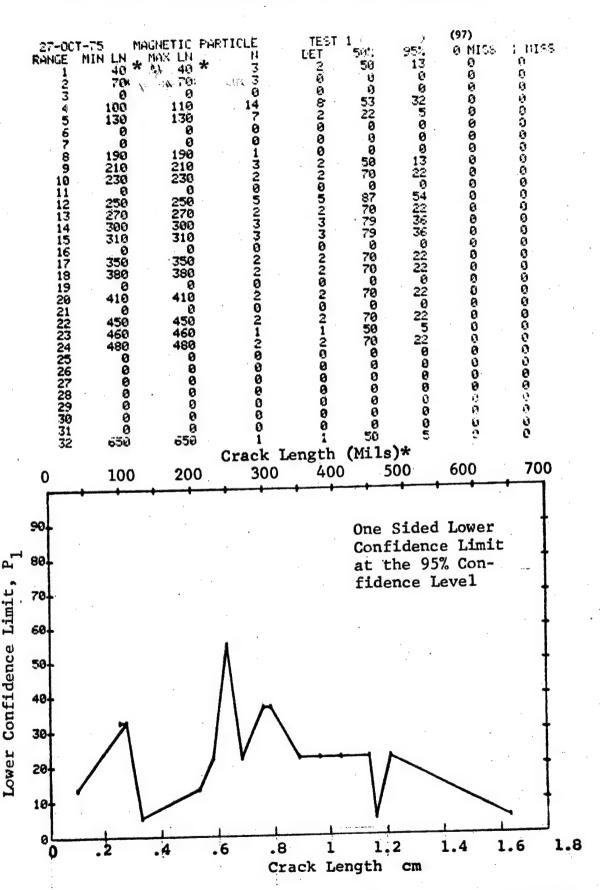


Figure D-96 (Concluded)



Probability of Detection for 4340M Steel Using Magnetic Figure D-97 Compressed Notch Flaws in Solid Threaded Cylinder. Particles. Prod. Env. REPRODUCIBILITY OF THE

D-297 AND TOUR PAGE IS POOT

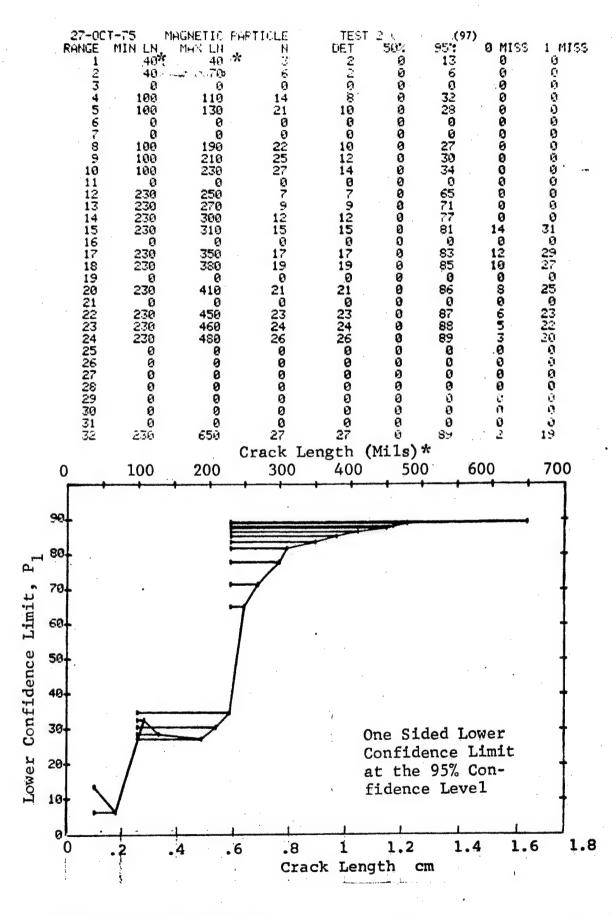


Figure D-97 (Continued)

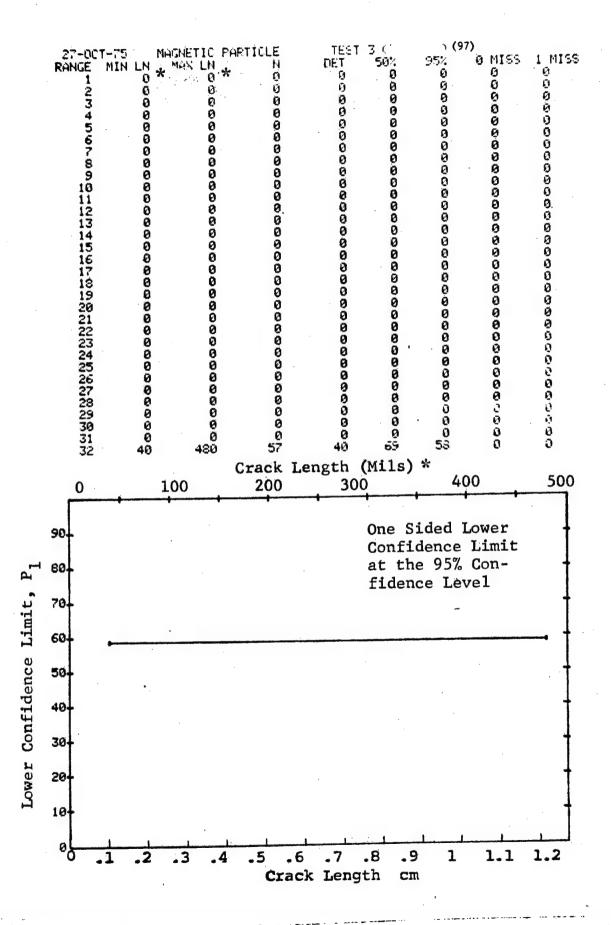


Figure D-97(Concluded)

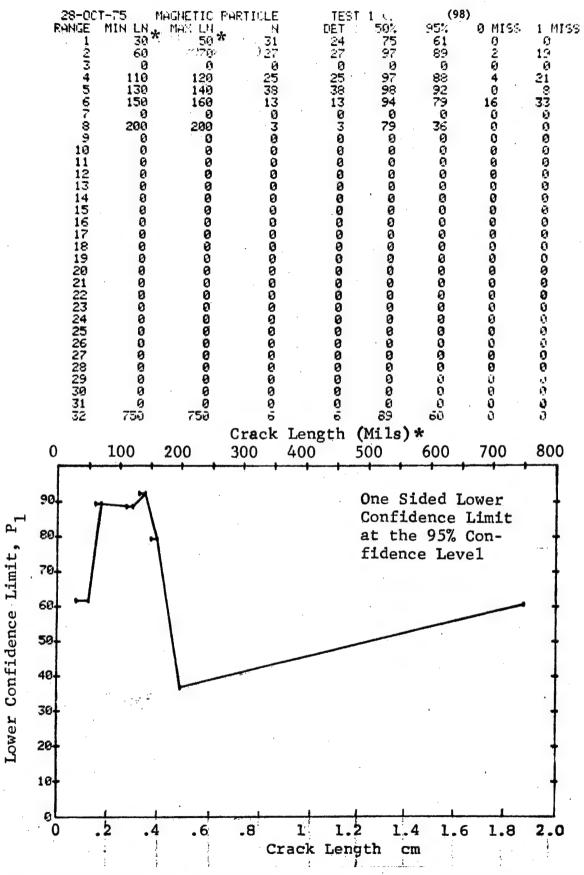


Figure D-98 Probability of Detection for 4340M Steel Using Magnetic, Particles. Compressed Notch Flaws in Solid Cylinder. Lab. Env.

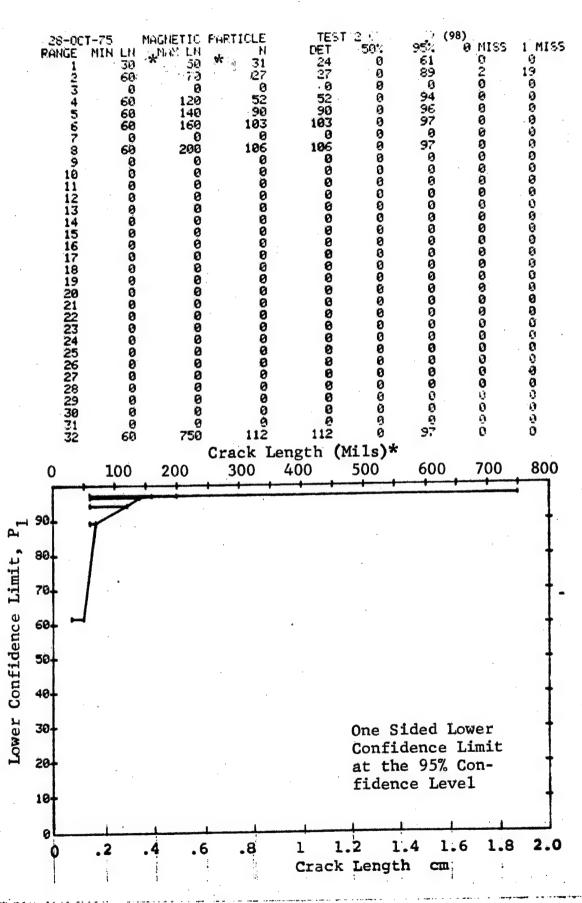


Figure D-98 (Continued)

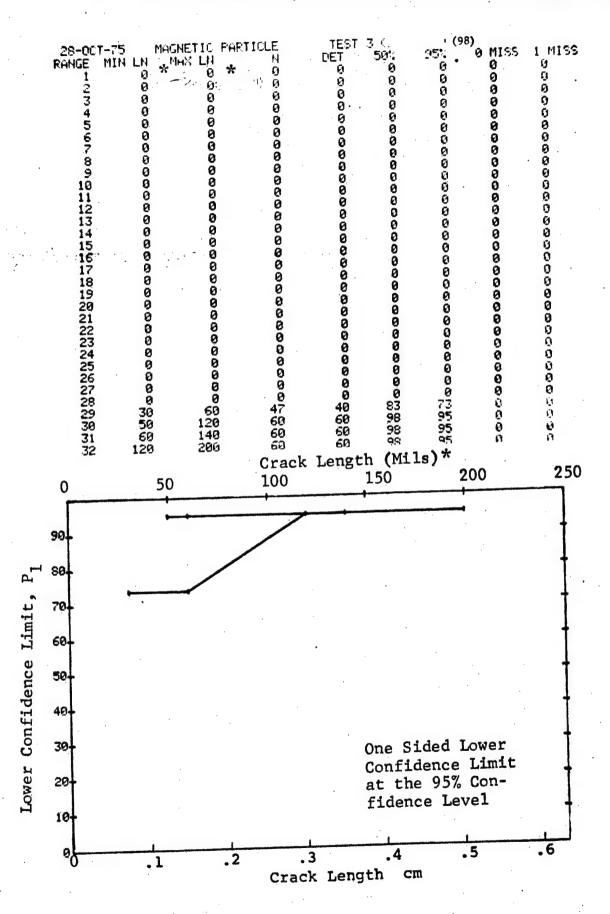


Figure D-98 (Concluded)

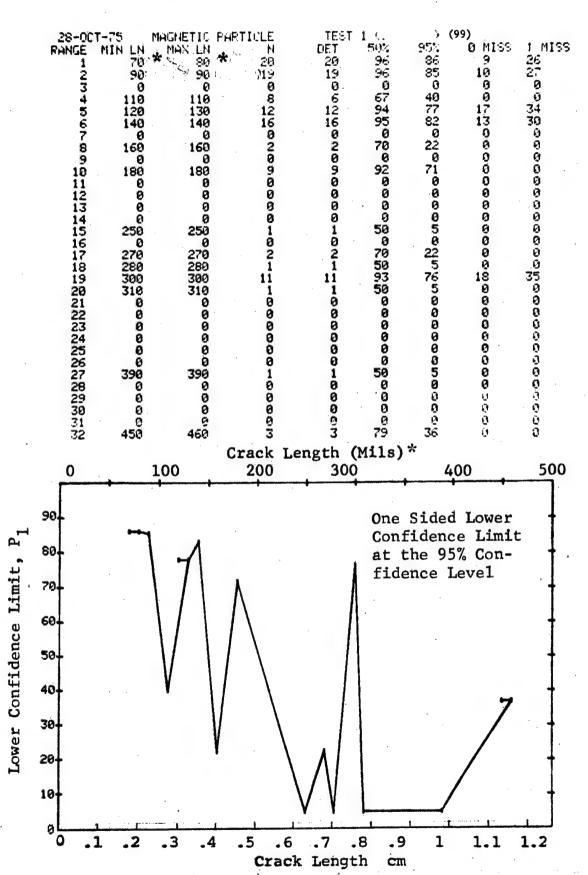


Figure D-99 Probability of Detection for 4340M Steel Using Magnetic Particles. Compressed Notch Flaws in Hollow Filleted Cylinder. Lab. Env.

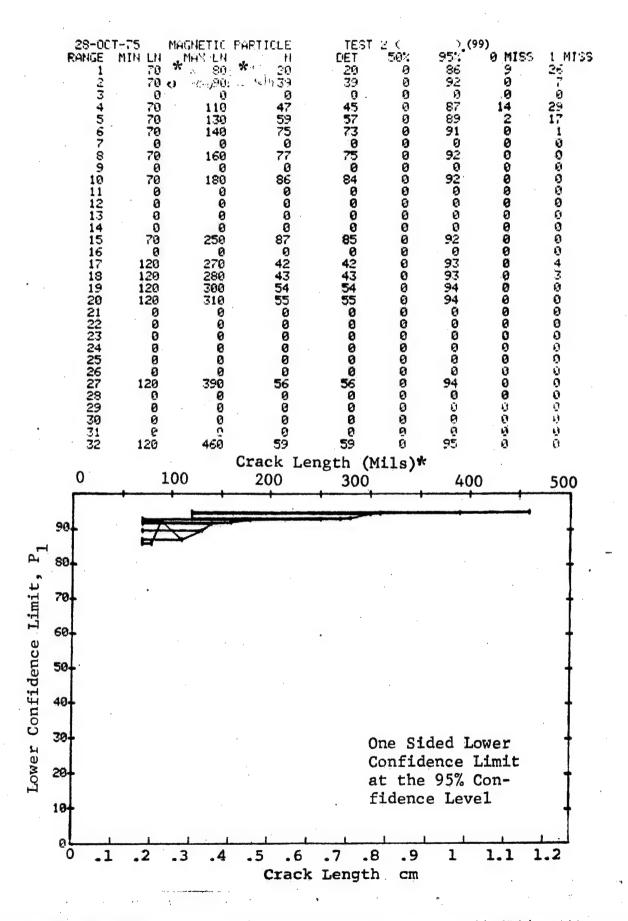


Figure D-99 (Continued)

REPRODUCIBILITY OF TH

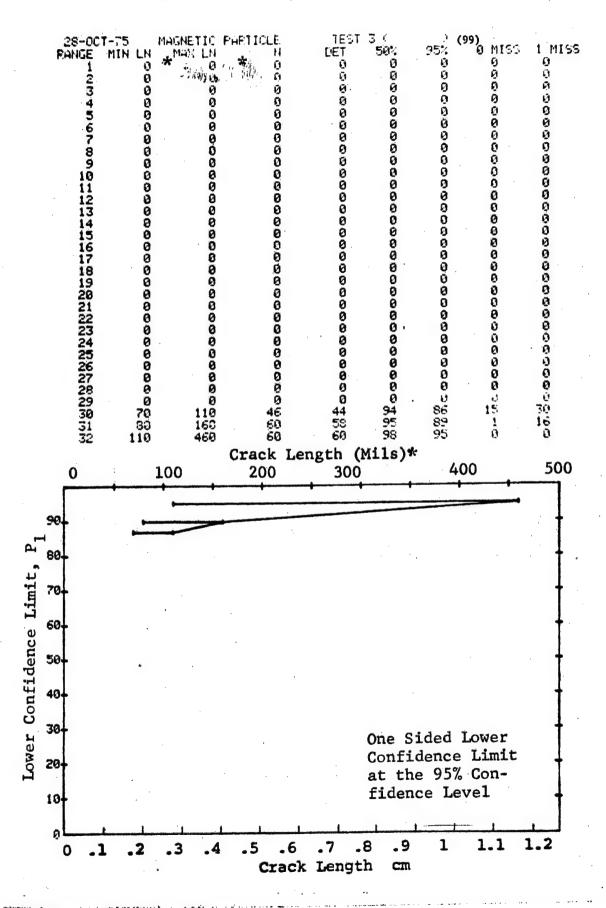


Figure D-99 (Concluded)

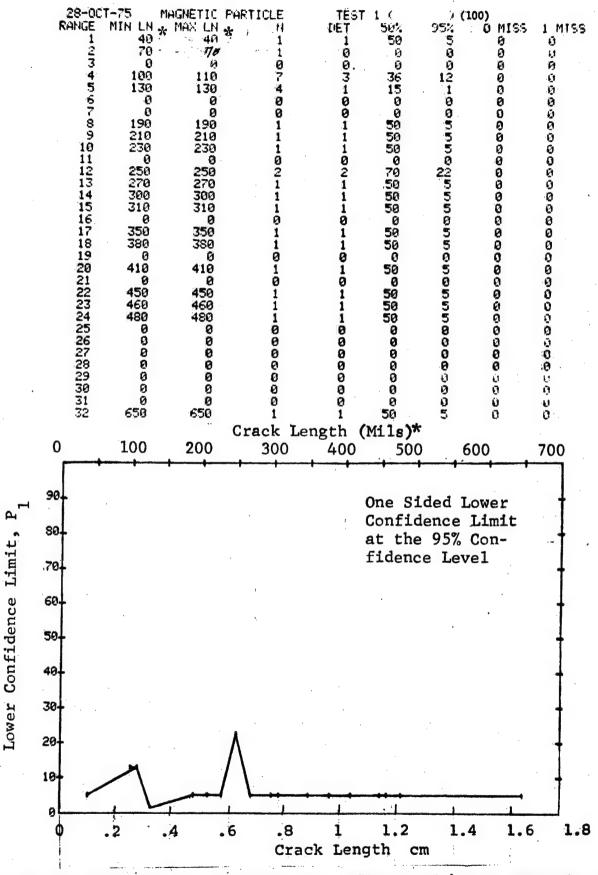


Figure D-100 Probability of Detection for 4340M Steel Using Magnetic Particles.

Compressed Notch Flaws in Solid Threaded Cylinder. Lab. Env.

(b) Optimum Probability Method of Data Cumulation

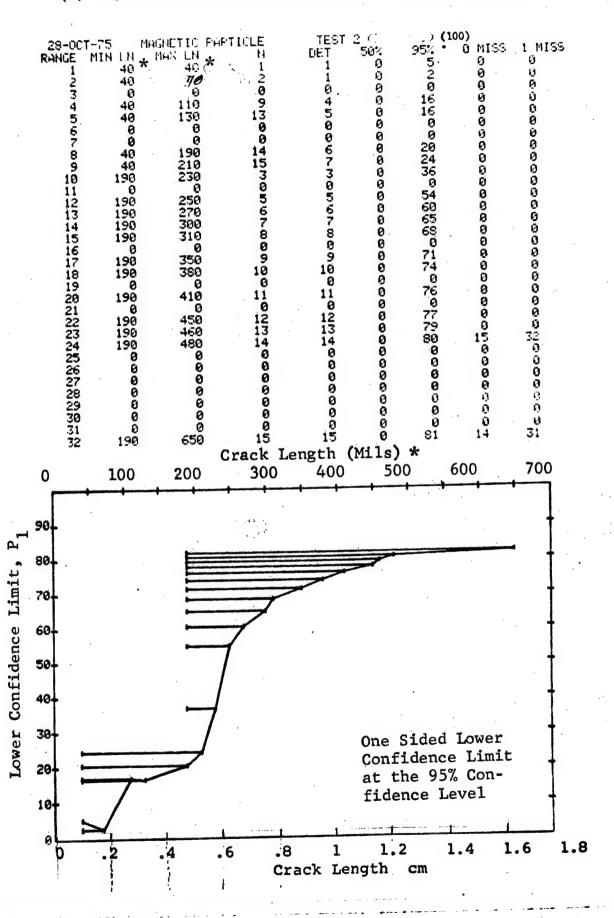


Figure D-100 (Continued)

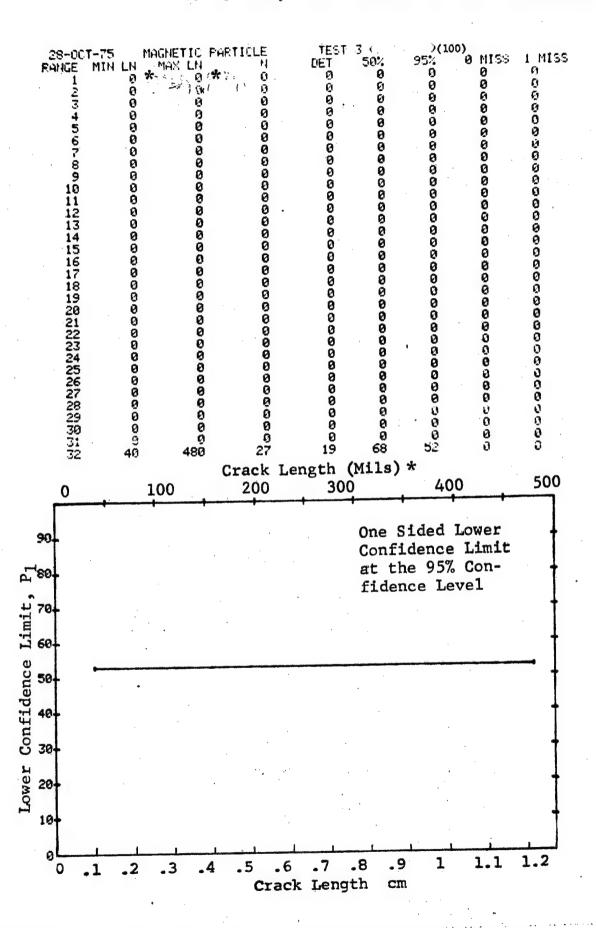


Figure D-100 (Concluded)

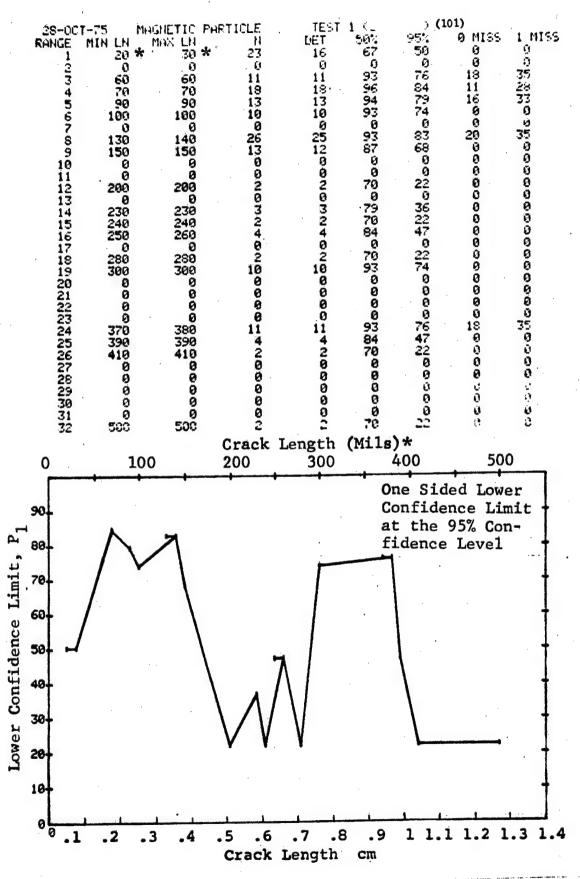


Figure D-101 Probability of Detection for 4340M Steel Using Magnetic Particles.

Compressed Notch Flaws in Solid Filleted Cylinder. Lab. Env.

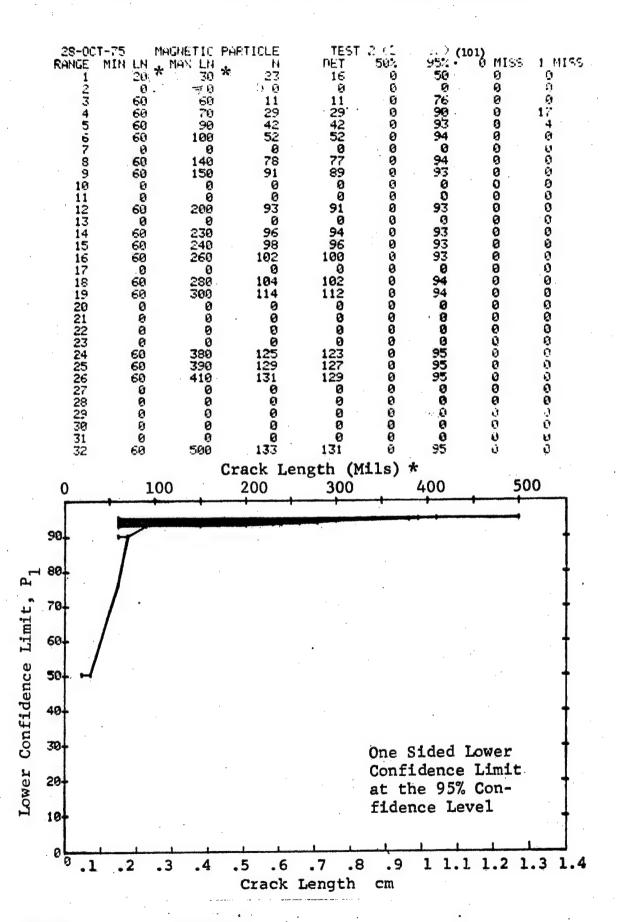


Figure D-101 (Continued)

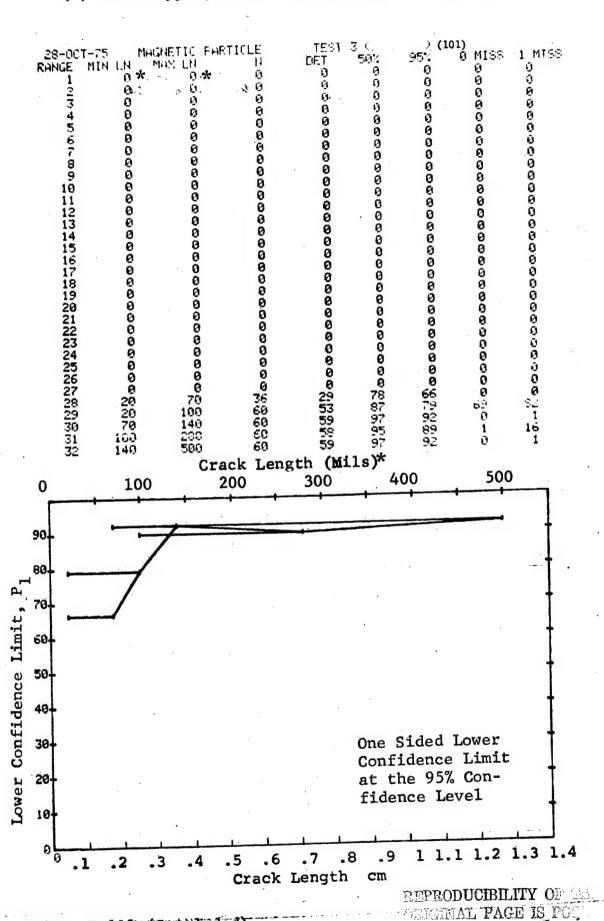


Figure D-101 (Concluded)

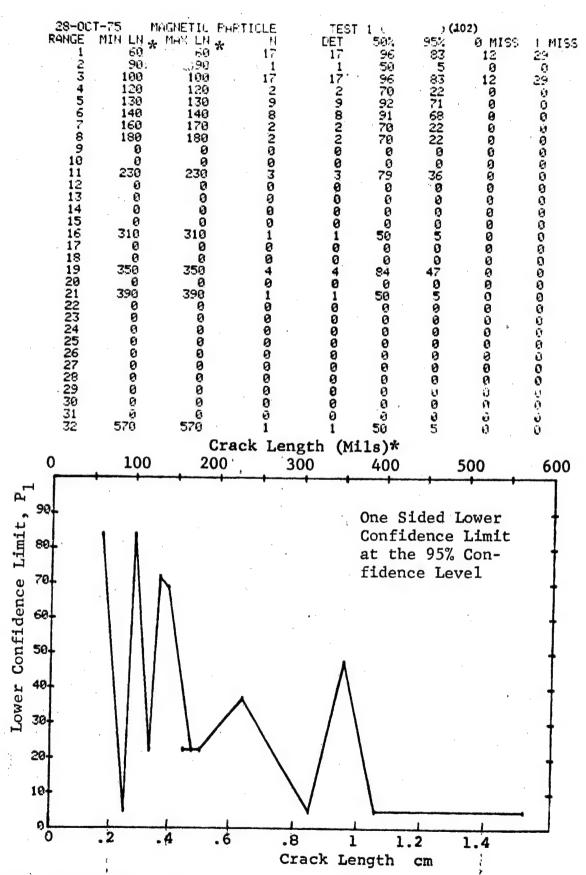


Figure D-102 Probability of Detection for 4340M Steel Using Magnetic Particles. Compressed Notch Flaws in Hollow Cylinder. Lab. Env.

(b) Optimum Probability Method of Data Cumulation

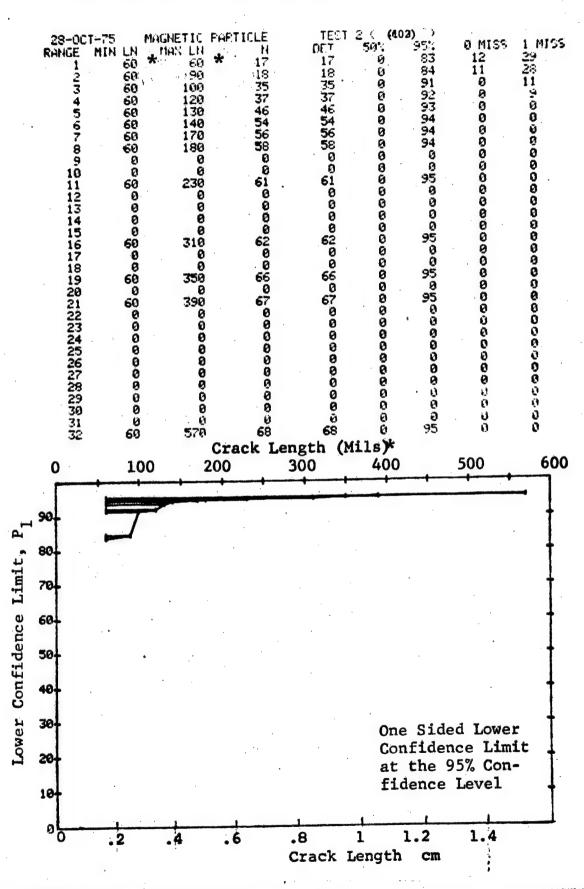


Figure D-102 (Continued)

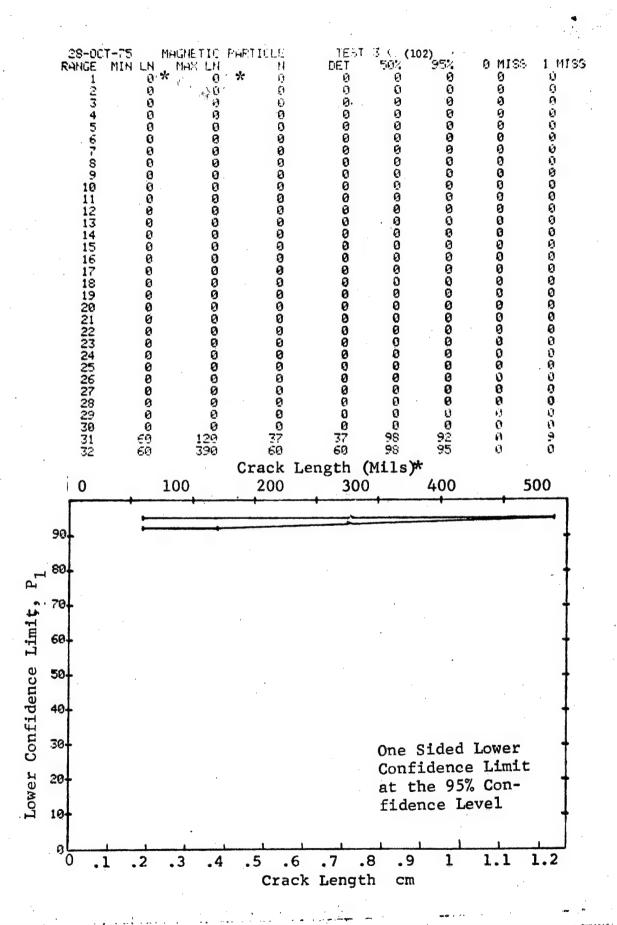


Figure D-102 (Concluded)

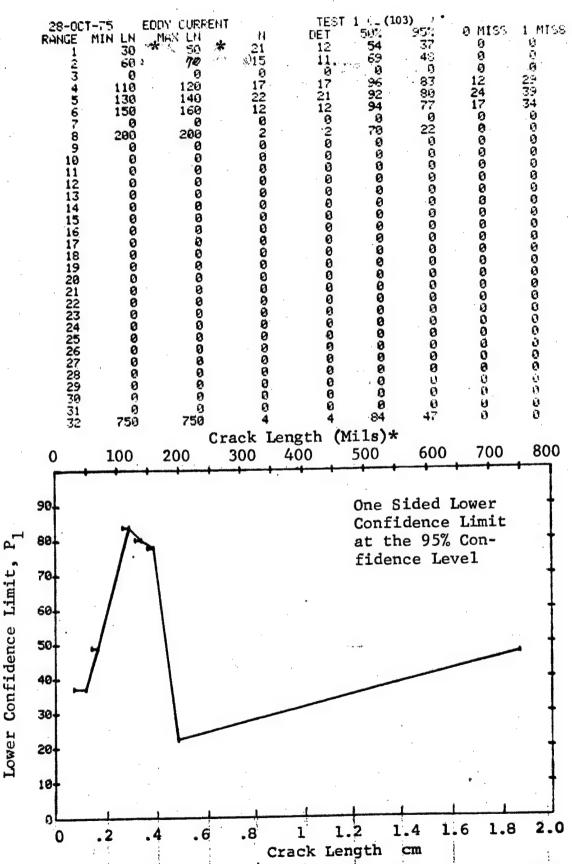
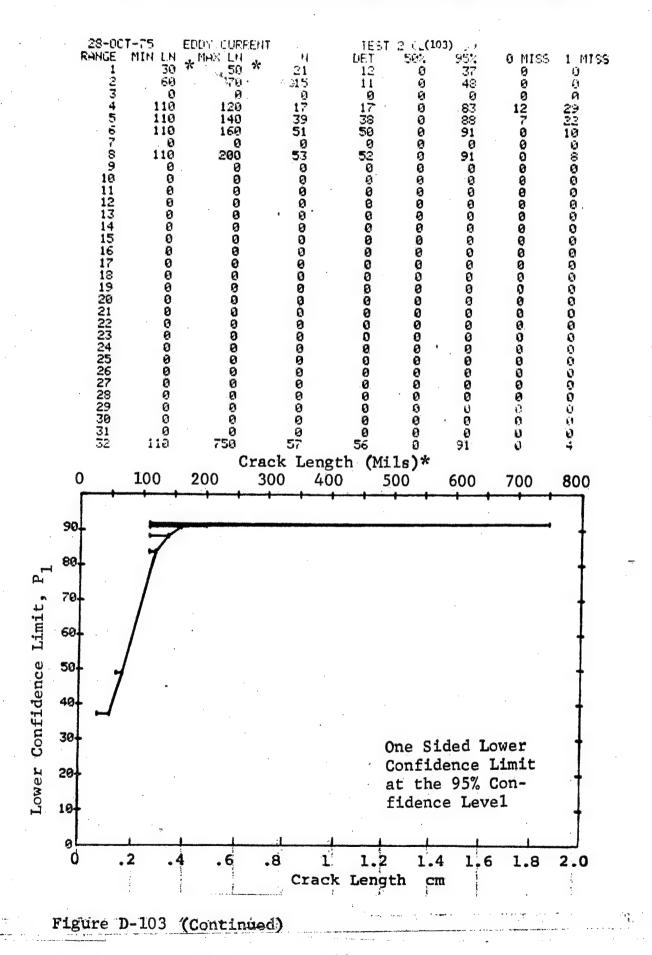


Figure D-103 Probability of Detection for 4340M Steel Using Eddy Current.
Compressed Notch Flaws in Solid Cylinder. Prod. Env.



(c) Overlapping Sixty Point Method of Data Cumulation

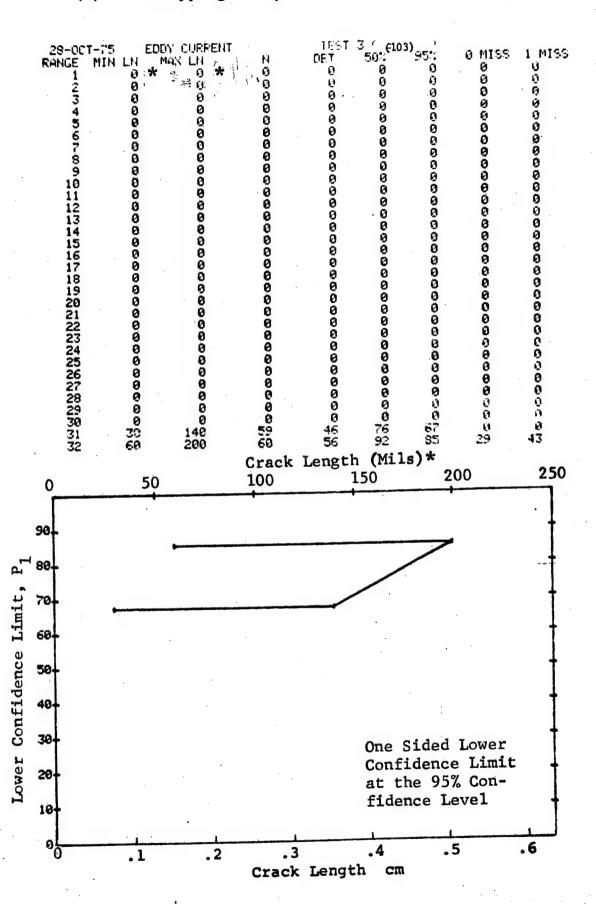


Figure D-103 (Concluded)

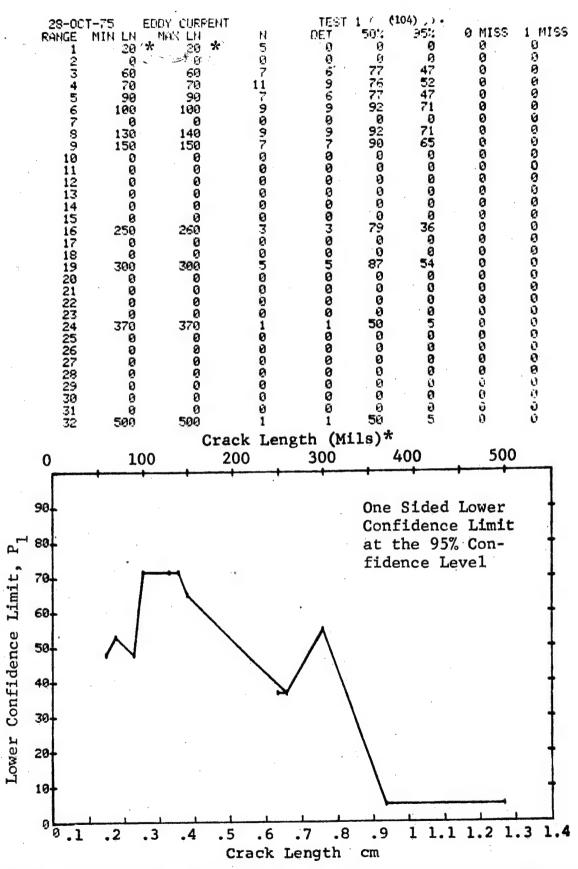


Figure D-104 Probability of Detection for 4340M Steel Using Eddy Current. Compressed Notch Flaws in Solid Filleted Cylinder. Prod. Env.

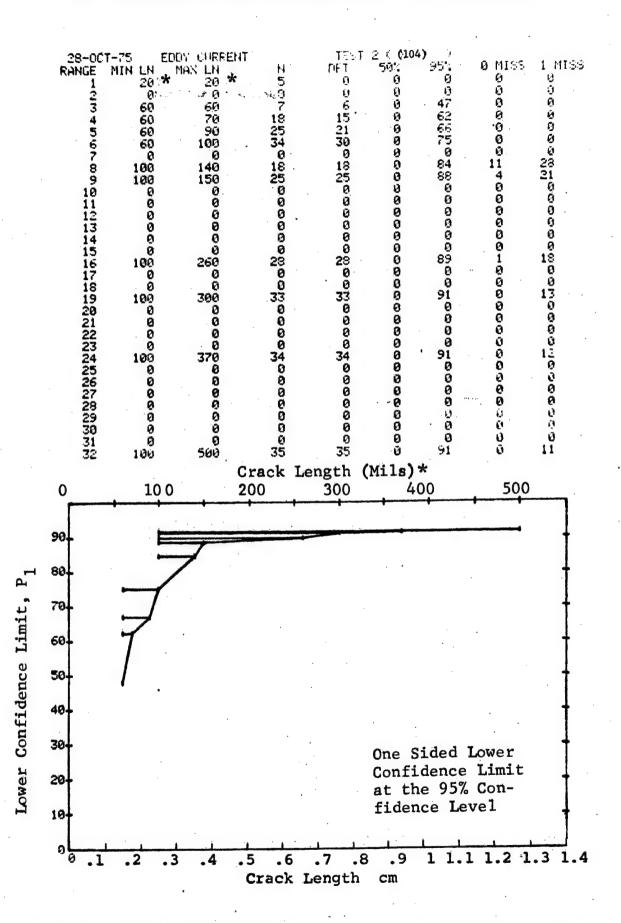


Figure D-104 (Continued)

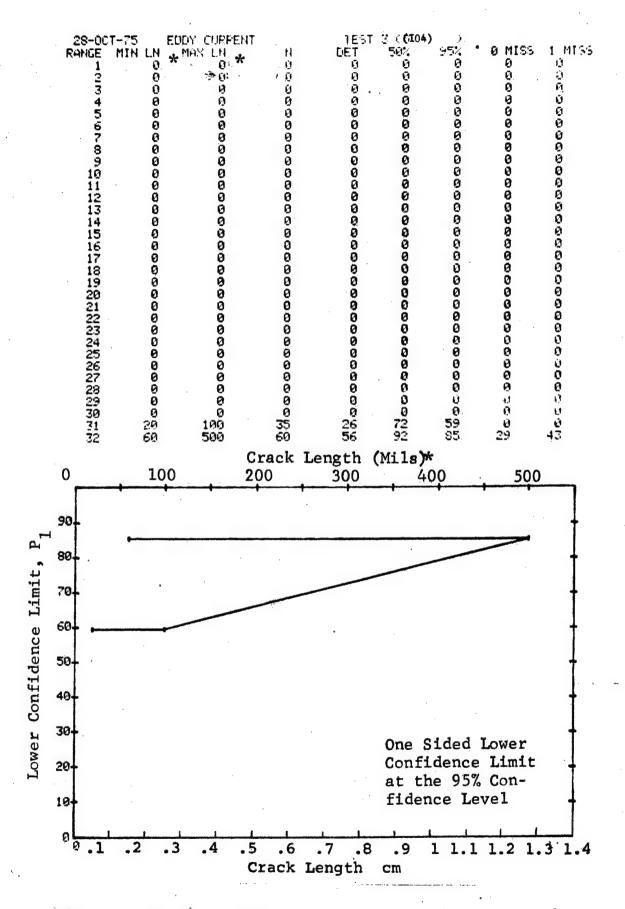


Figure D-104 (Concluded)

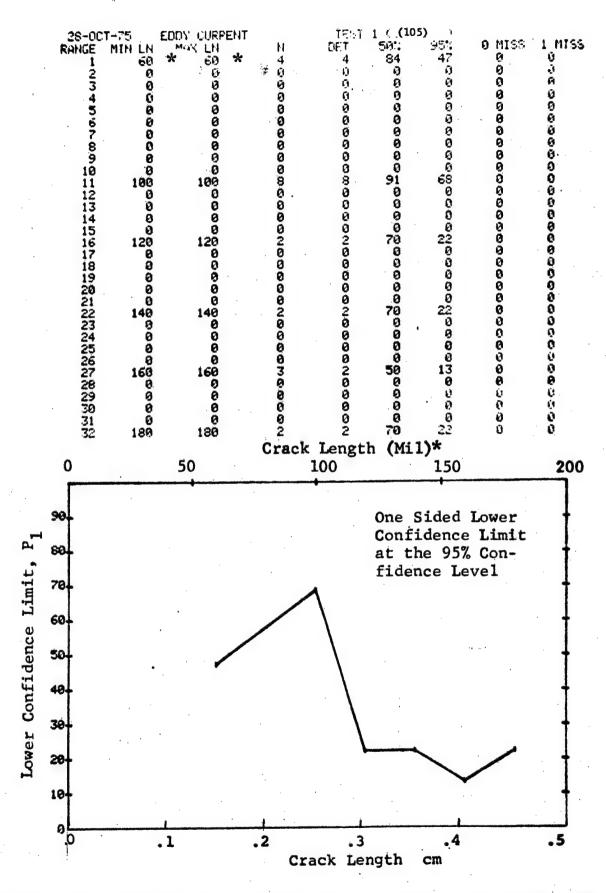
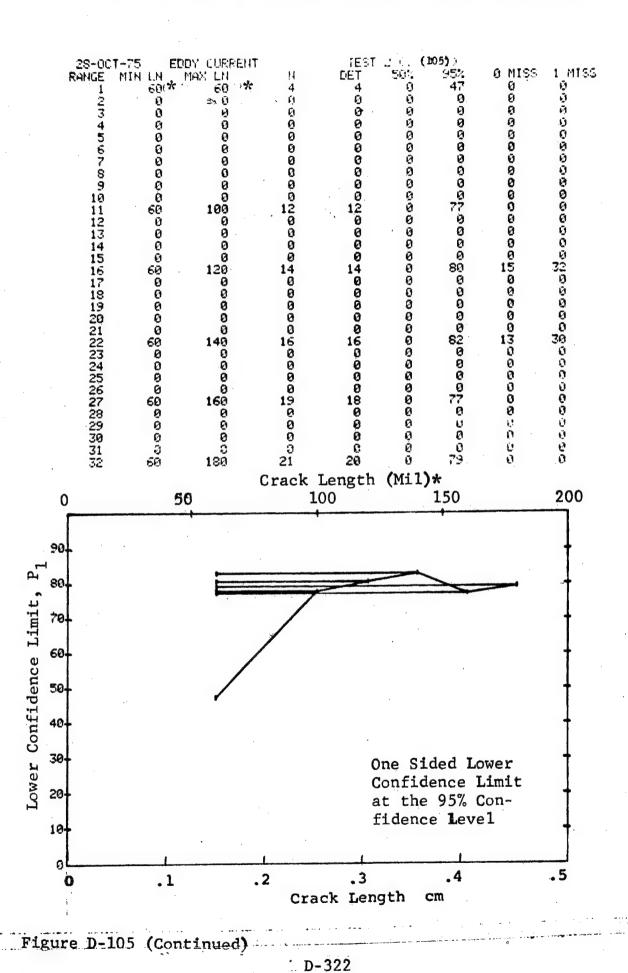
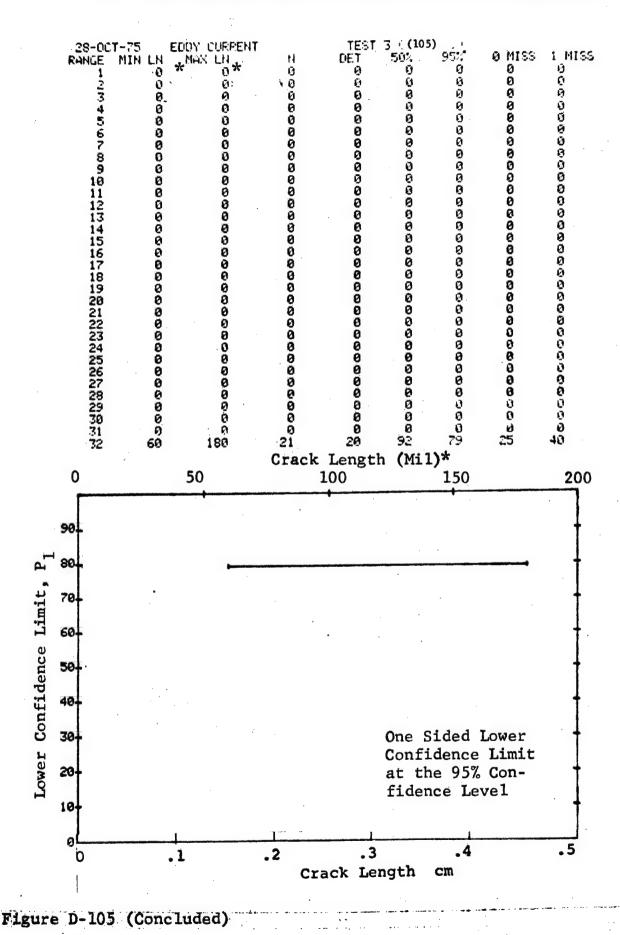


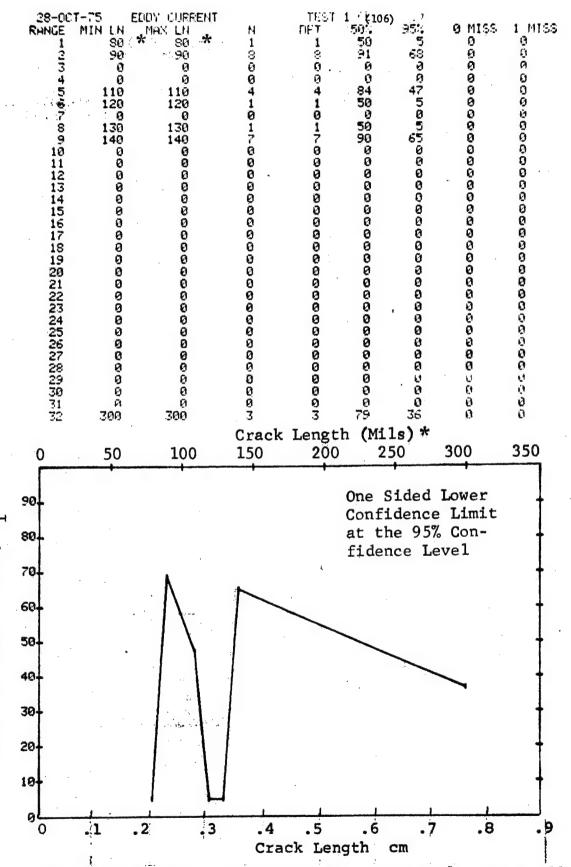
Figure D-105 Probability of Detection for 4340M Steel Using Eddy Current.

Compressed Notch Flaws in Hollow Cylinder. Prod. Env.



(c) Overlapping Sixty Point Method of Data Cumulation





Lower Confidence Limit,

Figure D-106 Probability of Detection for 4340M Steel Using Eddy Current Compressed Notch Flaws in Hollow Cylinder. Prod. Env.

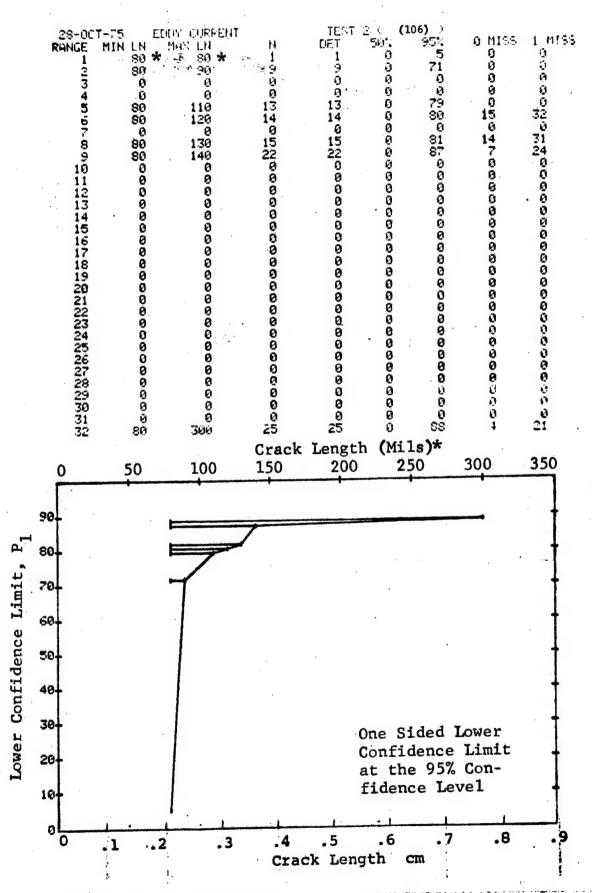


Figure D-106 (Continued)

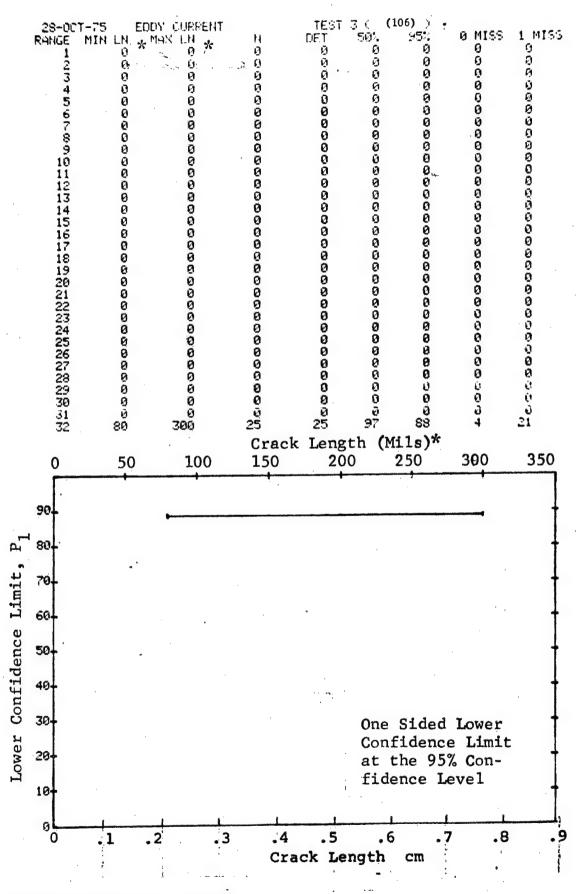


Figure D-106 (Concluded)

(a) Range Interval Method of Data Cumulation

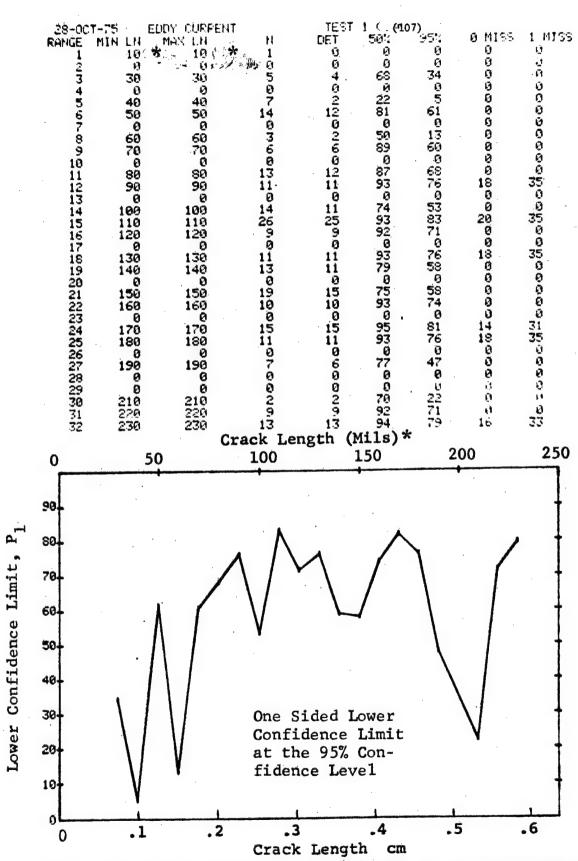


Figure D-107 Probability of Detection for 2024-T6 Al Using Eddy Current.

Compressed Notch Flaws in Tandem T Specimens. Production

Environment.

~D-327

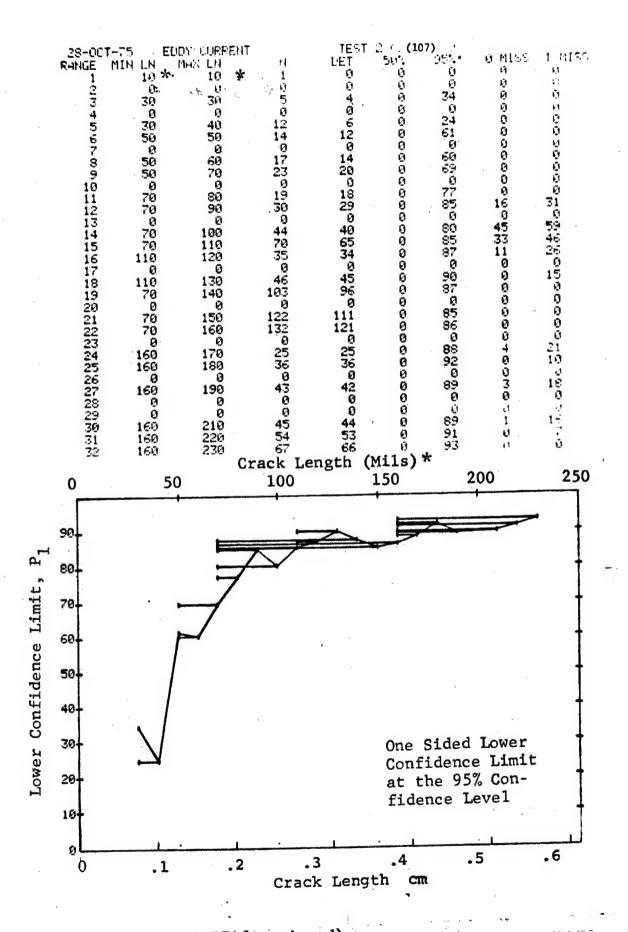


Figure D-107 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

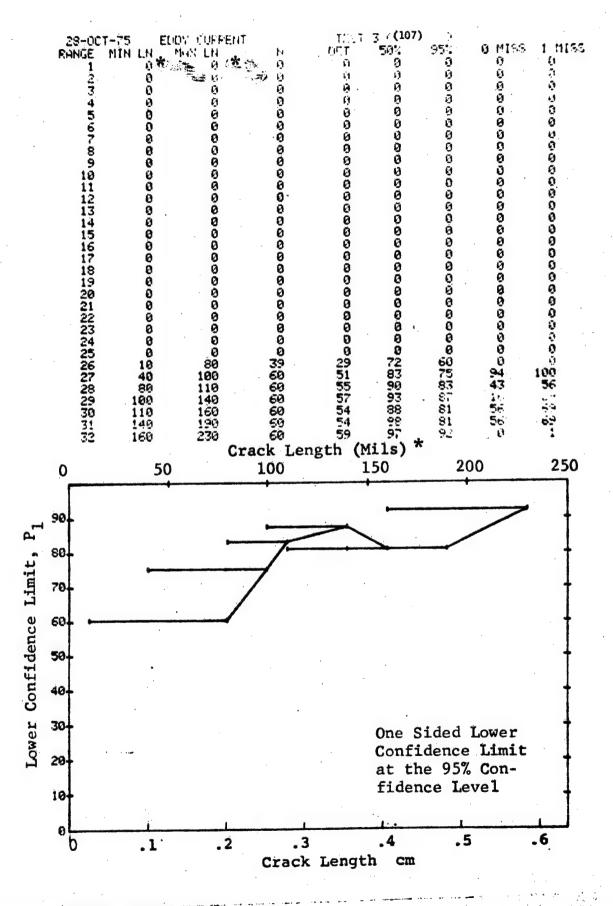


Figure D-107 (Concluded)

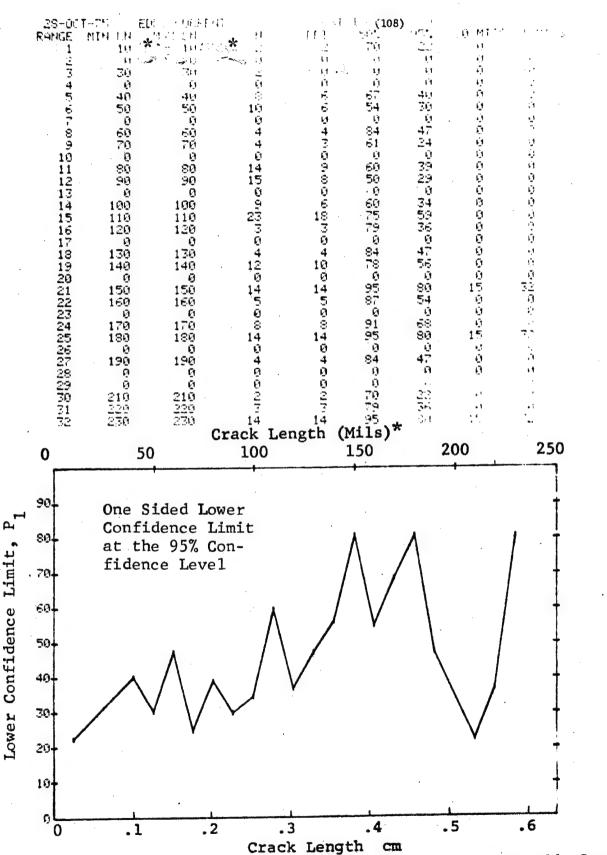


Figure D-108 Probability of Detection for 2024-T6 Al Using Eddy Current.

Compressed Notch Flaws in Tandem T Specimen. Laboratory

Environment.

D - 330

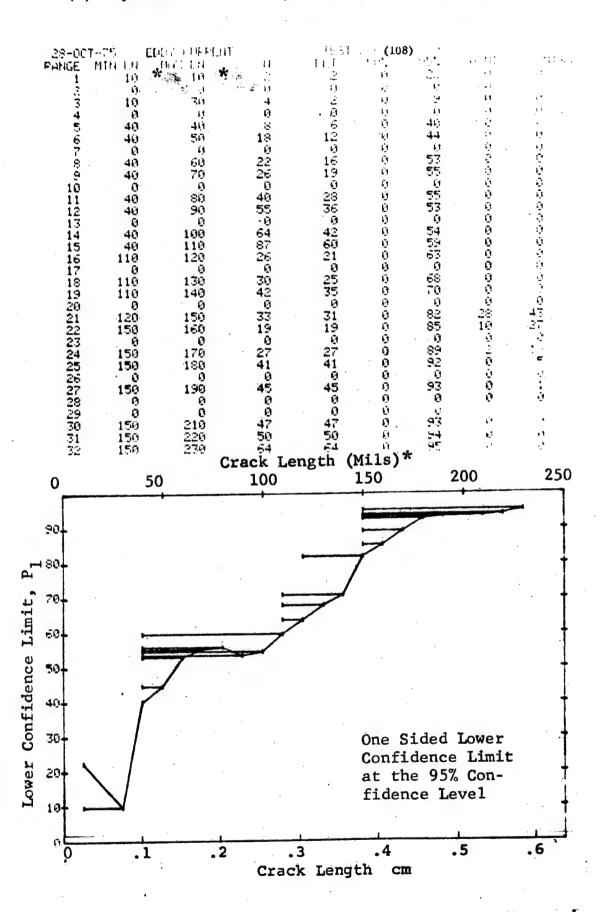


Figure D-108 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

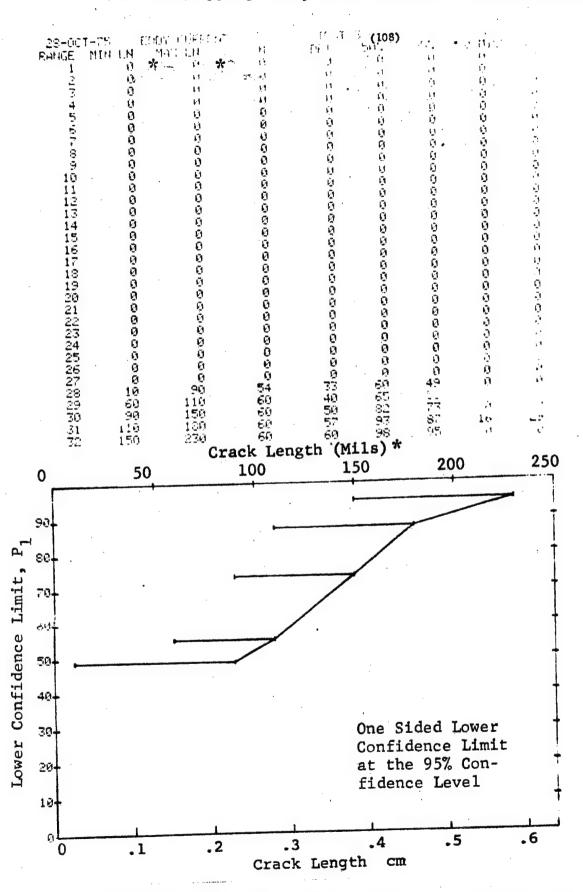


Figure D-108 (Concluded)

(a) Range Interval Method of Data Cumulation

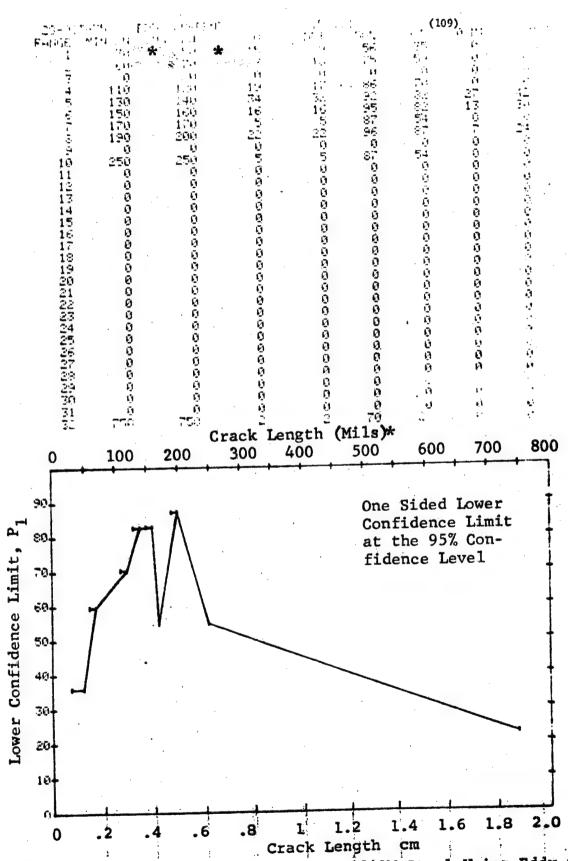


Figure D-109 Probability of Detection for 4340M Steel Using Eddy Current.

Compressed Notch Flaws in Solid Cylinder. Laboratory Environment.

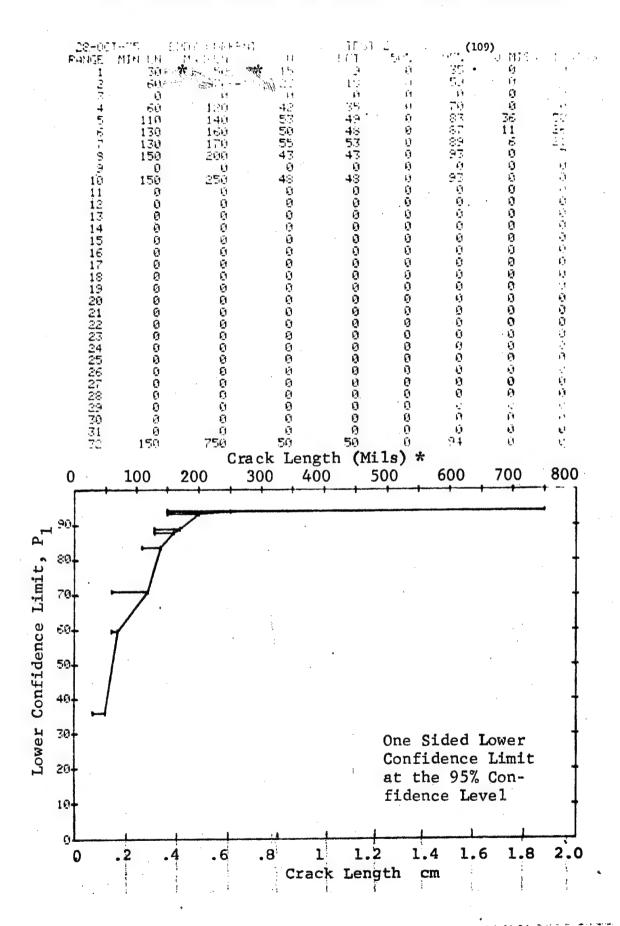


Figure D-109 (Continued)

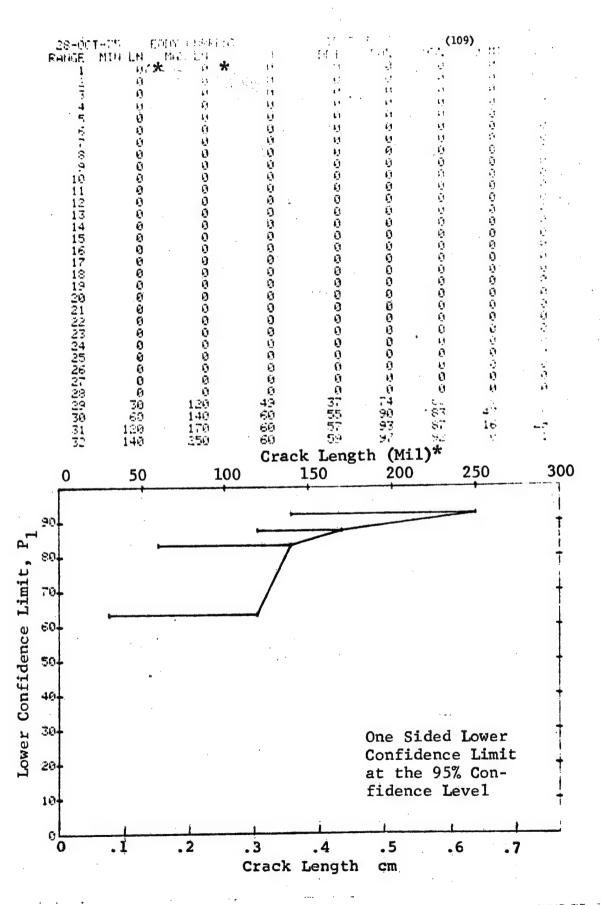


Figure D-109 (Concluded)

(a) Range Interval Method of Data Cumulation

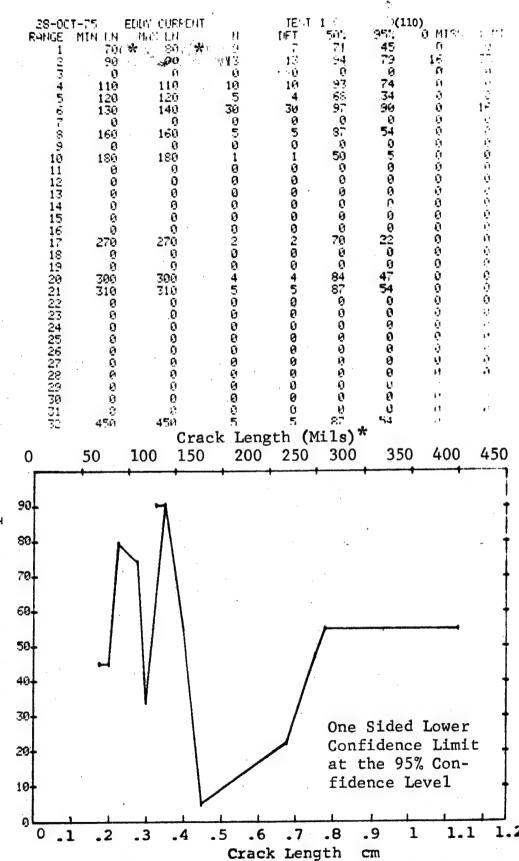


Figure D-110 Probability of Detection for 4340M Steel Using Eddy Current.

Compressed Notch Flaws in Hollow Filleted Cylinder. Laboratory
Environment.

D-336

Lower Confidence Limit,

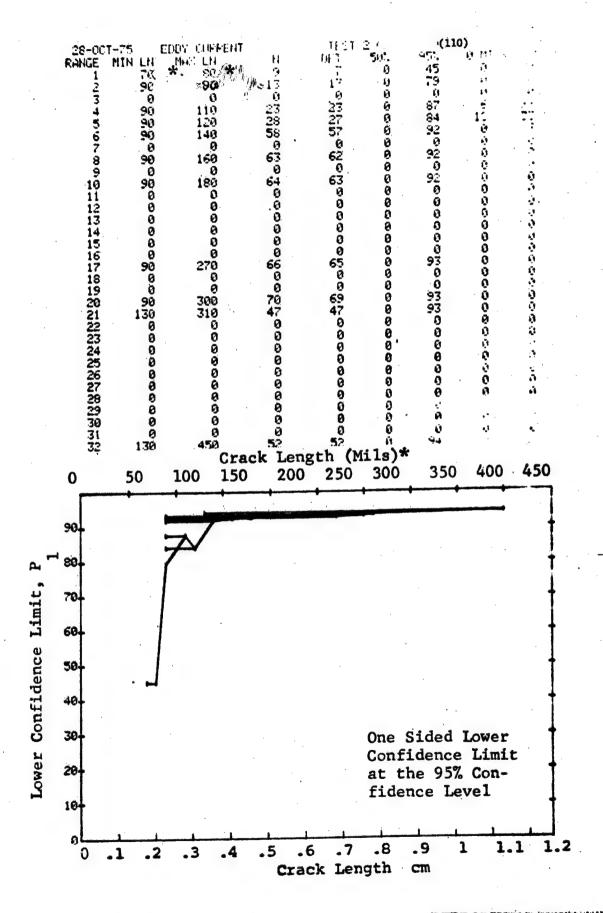


Figure D-110 (Continued)

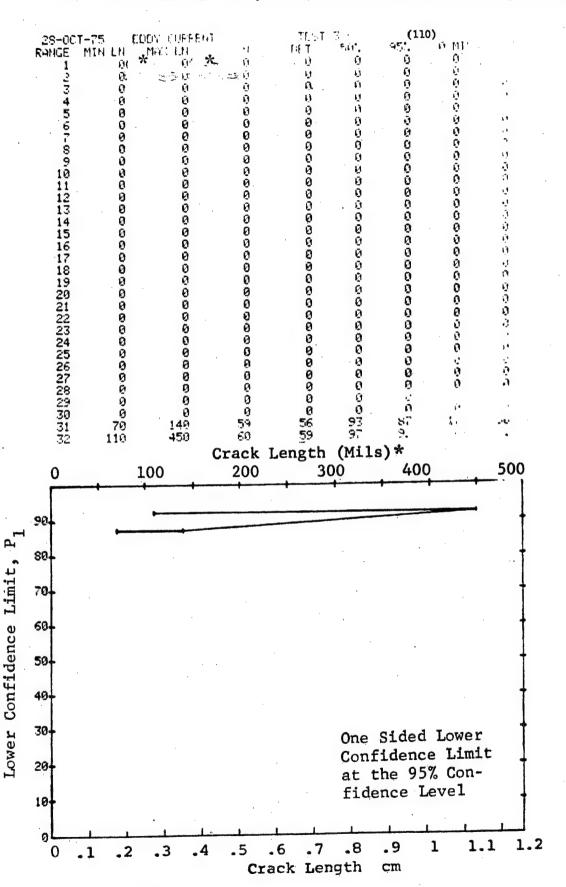


Figure D-110 (Concluded)

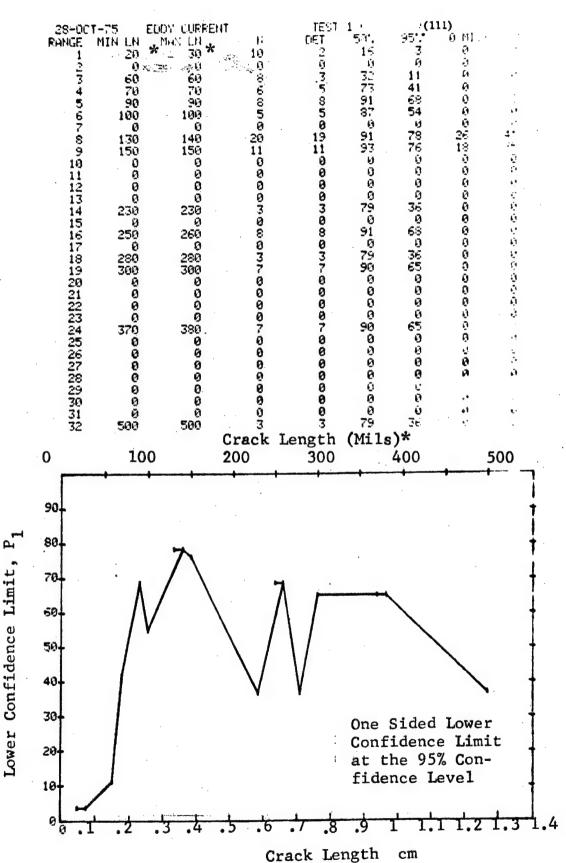


Figure D-111 Probability of Detection for 4340M Steel Using Eddy Current.

Compressed Notch Flaws in Solid Filleted Cylinder. Laboratory
Environment

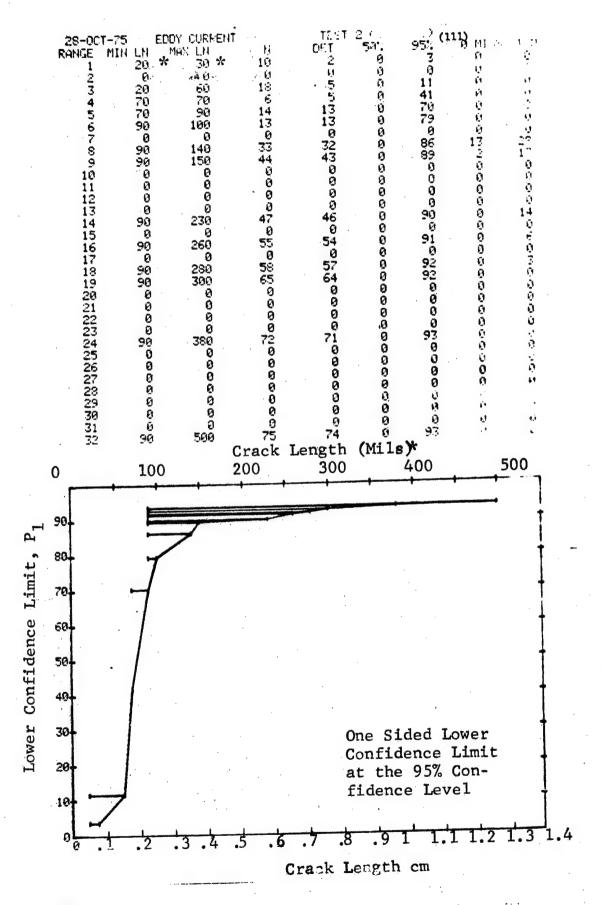


Figure D-111 (Continued)

(c) Overlapping Sixty Point Method of Data Cumulation

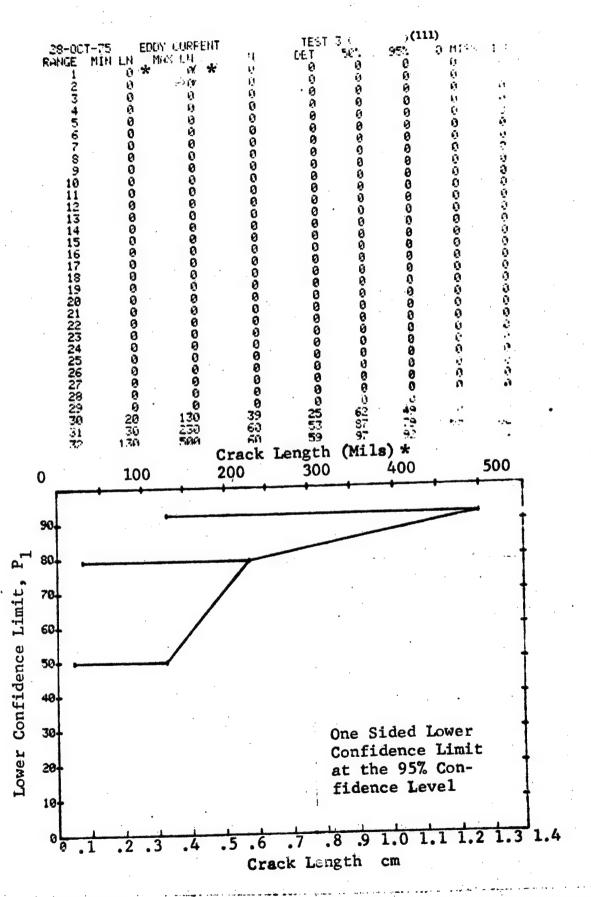


Figure D-111 (Concluded)

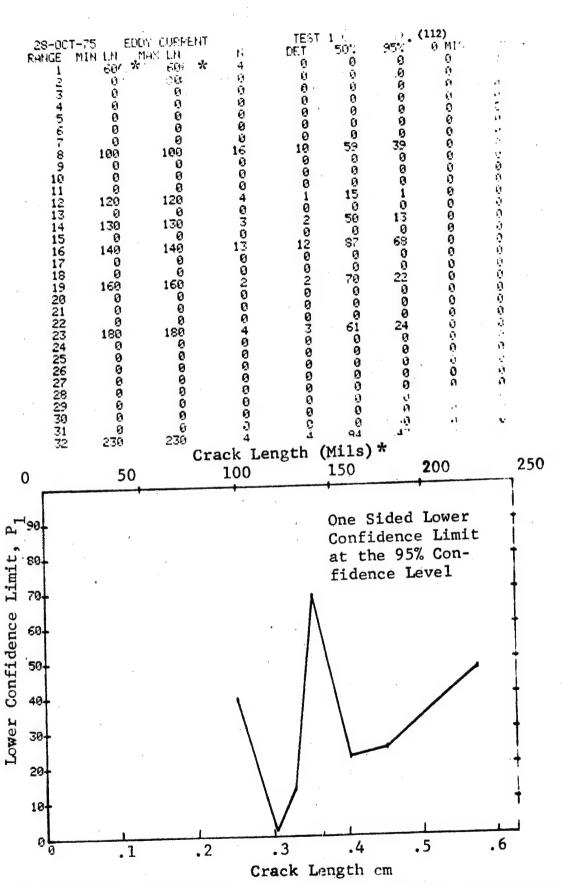


Figure D-112 Probability of Detection for 4340M Steel Using Eddy Current.

Compressed Notch Flaws in Hollow Cylinder. Laboratory Environment.

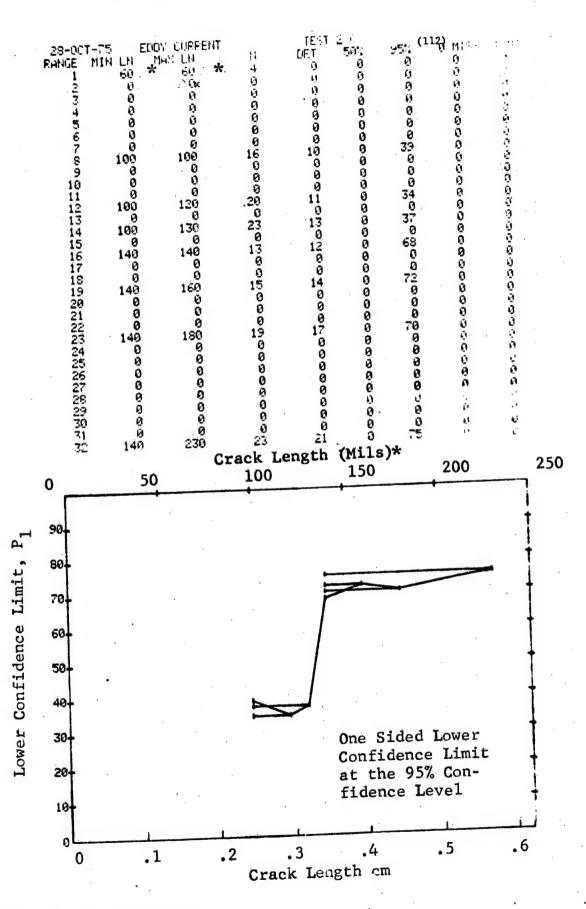


Figure D-112 (Continued)

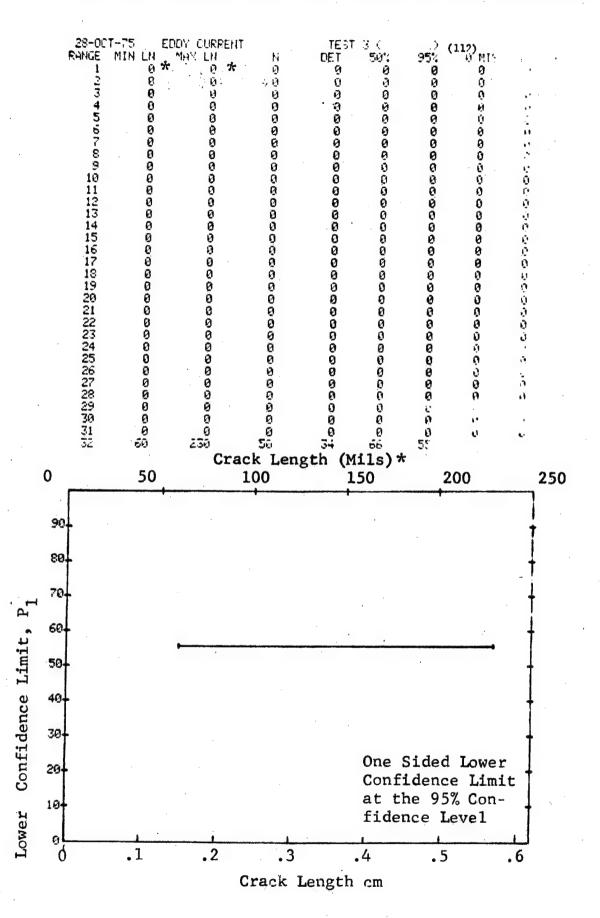


Figure D-112 (Concluded)

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